



## TBM 930

# PILOT'S INFORMATION MANUAL

**P/N DMJPIPYEE0 - Edition 0 - Revision 5**

**▲ CAUTION ▲**

This information manual is a non-official copy of the pilot's operating handbook and may be used for general information purposes only. It is not kept current and therefore cannot be used as a substitute for airworthiness authorities approved manual which is the only one intended for operation of the airplane.

The list of effective pages in this manual corresponds to that of the basic Pilot's Operating Handbook.



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**PIM - DO NOT USE FOR FLIGHT OPERATIONS**

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## Section 1

### General

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## 1.1 - General

This POH contains 9 sections and includes the material required by FAR Part 23 to be furnished to the pilot for operation of the TBM airplane. It also contains supplemental data supplied by the manufacturer, in accordance with GAMA standard.

Section 1 provides basic data and information of general interest. It also contains definitions or explanations of abbreviations and terminology commonly used.

Whenever this POH refers to the GARMIN integrated Flight Deck Pilot's Guide, it states the one described in section 2.1.

Whenever this POH refers to the MD302 Pilot's Guide, it states the one described in section 2.1.

The general information for complex optional systems are given in section 9, Supplements of the POH.

The installed ADS-B OUT system has been shown to meet the equipment requirements of 14 CFR 91.227.

The installed transponder system is able to respond to interrogations in Modes A, C and S and is fully compliant with the requirements of CS ACNS.D.ELS/EHS (Mode S Elementary/Enhanced Surveillance).

The installed ADS-B OUT system is fully compliant with the requirements of CS ACNS.D.ADSB (1090 MHz Extended Squitter ADS-B OUT).

## Part 135 operations

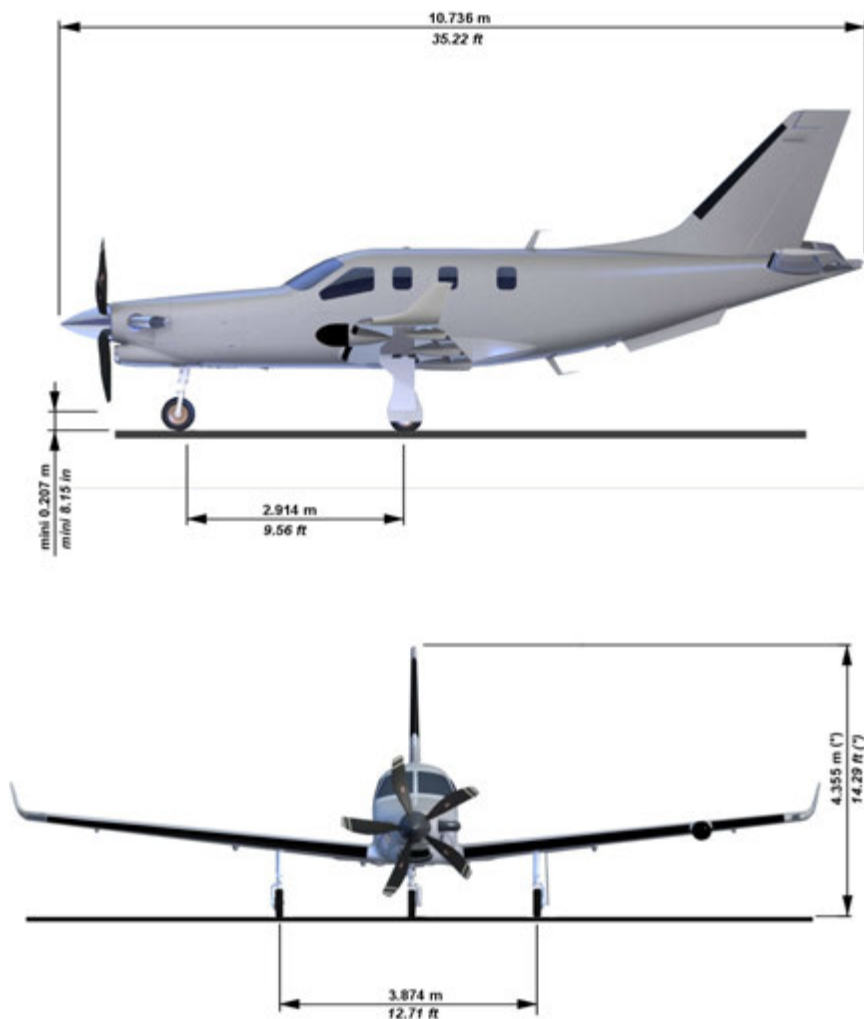
For 14 CFR 135 operations, TBM airplane alternative source of electric power is able to supply 150 percent of the electrical loads of all required instruments and equipment for safe emergency operation of the aircraft for at least 1 hour.

Electrical load shedding procedure provided in section 3 of this POH must be followed in order to meet the requirements of that paragraph under 14 CFR 135.163(f)(2).

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## 1.2 - Three view drawing



\* Airplane on level field with fully extended FWD shock-absorber

Figure 1.2.1 (1/2) - Three view drawing

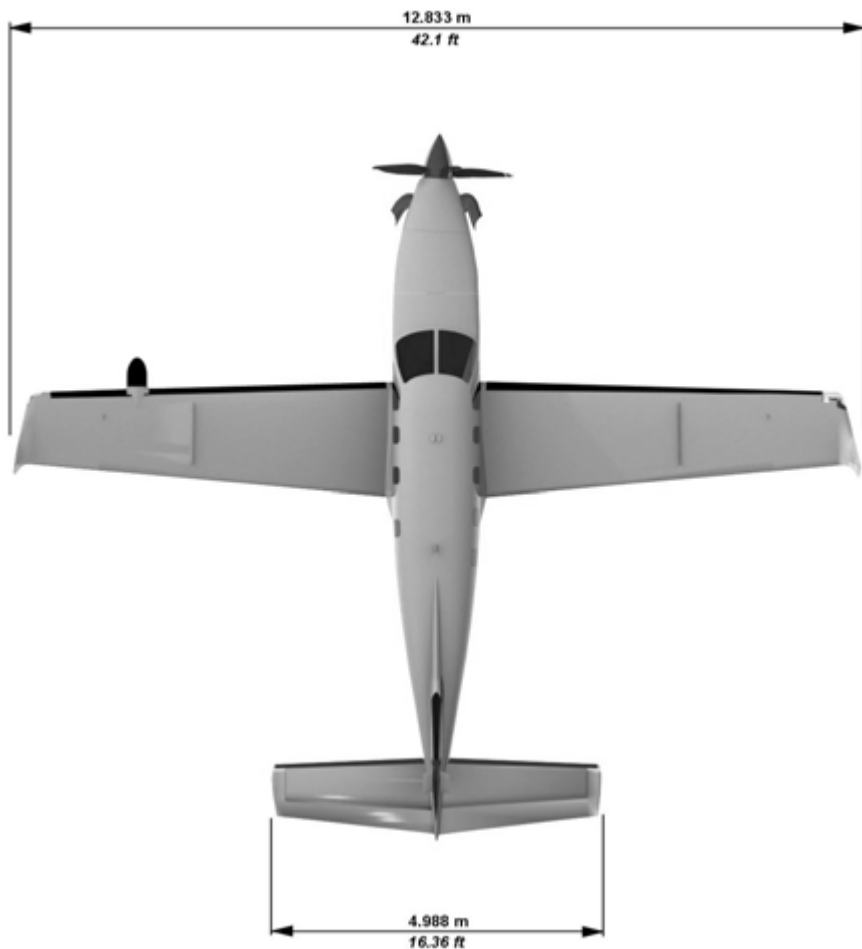


Figure 1.2.1 (2/2) - Three view drawing

### **1.3 - Descriptive data**

#### **Engine**

Number of engines : 1

Engine manufacturer : PRATT & WHITNEY CANADA

Engine model number : PT6A - 66D

Engine type : Free turbine, reverse flow and 2 turbine sections

Compressor type : 4 axial stages, 1 centrifugal stage

Combustion chamber type : annular

Turbine type : 1 gas generator turbine stage, 2 power turbines stages

Horsepower rating and propeller speed : 850 SHP at 2000 RPM

#### **Propeller**

Number of propellers : 1

Propeller manufacturer : HARTZELL

Propeller model number : HC-E5N-3C / NC8834K

Number of blades : 5

Propeller diameter :

Minimum : 90 in (2.286 m)

Maximum : 91 in (2.311 m)

Propeller type : Adjustable constant speed, with feathering and hydraulic control reverse

Propeller blade setting at station 30 in :

Low pitch : 19.5°

Feathering : 85°

Maximum reverse : - 9°

Propeller governor : 8210.007 WOODWARD

## Fuel

Total capacity : 301 USG (1140 litres)

Total capacity each tank : 150.5 USG (570 litres)

Total usable : 292 USG (1106 litres)

### ▲ CAUTION ▲

The fuel used must contain an anti-ice additive, in accordance with specification MIL-I-27686 or MIL-I-85470. Additive concentrations (EGME or DIEGME) shall be comprised between a minimum of 0.06 % and a maximum of 0.15 % by volume. Refer to section 8 Handling, servicing and maintenance for additional information.



### ▲ CAUTION ▲

The use of aviation gasoline (AVGAS) must be restricted to emergency purposes only. AVGAS shall not be used for more than 150 cumulative hours during any period between engine overhaul periods.



### ● NOTE ●

Use of AVGAS to be recorded in engine module logbook.



| US specification<br>(US)                                  | French specification<br>(FR) | English specification<br>(UK) | NATO code                                |
|---|------------------------------|-------------------------------|--|
| ASTM-D1655 JET A<br>ASTM-D1655 JET A1<br>ASTM-D1655 JET B | AIR 3405C Grade F35          | DERD 2494 Issue 9             | F35 without<br>additive                  |
| MIL-DTL-5624<br>Grade JP-4                                | AIR 3407B                    | DERD 2454 Issue 4<br>Amdt 1   | F40 with<br>additive                     |
| MIL-DTL-5624<br>Grade JP-5                                | AIR 3404C Grade F44          | DERD 2452 Issue 2<br>Amdt 1   | F44 with<br>additive when<br>utilization |
| MIL-DTL-83133<br>Grade JP-8                               | AIR 3405C Grade F34          | DERD 2453 Issue 4<br>Amdt 1   | F34 with<br>additive S748                |
|   | AIR 3404C Grade F43          | DERD 2498 Issue 7             | F43 without<br>additive                  |

Figure 1.3.1 - Recommended fuel types  
Reference : Service Bulletin P & W C. No. 14004

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**Engine oil**

System total capacity : 12.7 Quarts (12 litres) (oil cooler included)

Usable capacity : 6 Quarts (5.7 litres)

Maximum oil consumption in 10 hour period : 0.14 qt/hr (0.13 l/hr)  
[0.3 lb/hr (0.136 cc/hr)]

## Specification

| Nominal viscosity | Specification  | NATO code                  |
|-------------------|----------------|----------------------------|
| 5cSt              | MIL-PRF-23699G | O-156 (STD)<br>O-154 (HTS) |

Figure 1.3.2 - Recommended engine oil types

Reference : Service Bulletin P & W C. No. 14001 at the latest revision

**Maximum certificated weights**

Ramp : 7430 lbs (3370 kg)

Takeoff : 7394 lbs (3354 kg)

Landing : 7024 lbs (3186 kg)

Baggage weight

- refer to section 2, paragraph 2.5 for weight and C.G. limits
- refer to section 6 for cargo loading instructions

**Standard airplane weights**

Standard empty weight : 4583 lbs (2079 kg)

Maximum useful load : 2811 lbs (1275 kg)

## **Cabin and entry dimensions**

Maximum cabin width : 3 ft 11.64 in (1.21 m)

Maximum cabin length : 13 ft 3.45 in (4.05 m)

Maximum cabin height : 4 ft (1.22 m)

Number of cabin entries : 1 (standard) + 1 pilot door (if installed)

Entry width (standard) : 3 ft 6.52 in (1.08 m)

Entry height (standard) : 3 ft 10.85 in (1.19 m)

Pilot entry mean width : 2 ft 3.6 in (0.70 m)

Pilot entry mean height : 3 ft 2.16 in (0.97 m)

## **Specific loadings**

Wing loading : 38.16 lbs / sq.ft (186.3 kg / m<sup>2</sup>)

Power loading : 8.7 lbs / SHP (3.95 kg / SHP)

## 1.4 - Abbreviations and terminology

### Meteorological terminology

|     |   |  |
|-----|---|--|
| ISA | : | International standard atmosphere                        |
| OAT | : | Outside air temperature                                  |
| SAT | : | Static air temperature                                   |
| QFE | : | Atmospheric pressure at the airport reference point.     |
| QNH | : | Atmospheric pressure at sea level, at airplane position. |

• NOTE •

On the ground, the altimeter will indicate zero if it is set to QFE. It will indicate airport altitude if it is set to QNH.

•

### Standard Temperature :

Is 15°C (59°F) at sea level pressure altitude and decreases by 2°C (3.6°F) for each 1000 ft of altitude.

### Pressure altitude :

Is the altitude read from an altimeter when the altimeter's barometric scale has been set to 29.92 inches of mercury (1013.2 hPa).

### General airspeed terminology and symbols

|                 |   |   |
|-----------------|---|---|
| KCAS            | : | Knots Calibrated Airspeed is the indicated airspeed expressed in knots corrected for position and instrument error. Knots calibrated airspeed is equal to KTAS in standard atmosphere at sea level. |
| KIAS            | : | Knots Indicated Airspeed is the speed shown on the airspeed indicator and expressed in knots.   |
| KTAS            | : | Knots True Airspeed is the airspeed expressed in knots relative to undisturbed air which is KCAS corrected for altitude and temperature.  |
| V <sub>A</sub>  | : | Maneuvering Speed is the maximum speed at which full or abrupt control movements may be used.   |
| V <sub>FE</sub> | : | Maximum Flap Extended Speed is the highest speed permissible with wing flaps in a prescribed extended position.   |
| V <sub>LE</sub> | : | Maximum Landing Gear Extended Speed is the maximum speed at which an airplane can be safely flown with the landing gear extended.   |

- V<sub>LO</sub>** : Maximum Landing Gear Operating Speed is the maximum speed at which the landing gear can be safely extended or retracted.
- V<sub>MO</sub>** : Maximum Operating Speed is the speed limit that may not be deliberately exceeded in normal flight operations.
- V<sub>R</sub>** : Rotation Speed is the speed at which rotation is initiated during takeoff to achieve takeoff safety speed at screen height.
- V<sub>SO</sub>** : Stalling Speed or the minimum steady flight speed at which the airplane is controllable in the landing configuration.
- V<sub>S1</sub>** : Stalling Speed or the minimum steady flight speed obtained in a specific configuration.
- V<sub>x</sub>** : Best Angle of Climb Speed is the airspeed which delivers the greatest gain of altitude in the shortest possible horizontal distance.
- V<sub>y</sub>** : Best Rate of Climb Speed is the airspeed which delivers the greatest gain in altitude in the shortest possible time.

## Power terminology

### Recovery altitude :

Maximum altitude at which it is possible, in standard temperature, to maintain a specified power.

### Overheated start :

Engine start or attempt to start which causes the interturbine temperature to be higher than the maximum value permissible during start.

**Flame out** : Involuntary loss of the combustion chamber flame during operation.

**GPU** : Ground power unit.

**Feathering** : Action which reduces the drag of a propeller by positioning blades at the pitch angle allowing minimal drag.

### Maximum Cruise Power :

Power developed corresponding to outside flight level and temperature conditions - refer to chapter 5 Performance.

**Ng** : Gas generator RPM.

**Np** : Propeller rotation speed.

**Reverse** : Drag produced when the propeller blade setting is negative.

**RPM** : Revolutions per minute.



**SHP** : Shaft Horsepower.

**TRQ** : Torque.

## **Airplane performance and flight planning terminology**

### **Climb gradient :**

Is the ratio of the change in height during a portion of climb, to the horizontal distance traversed in the same time interval.

### **Demonstrated crosswind velocity :**

Is the velocity of the crosswind component for which adequate control of the airplane during takeoff and landing was actually demonstrated during certification tests. The value shown is not considered to be limiting.

**g** : Is acceleration due to gravity.

**Usable fuel** : Total fuel which can be effectively consumed by the engine.

## **Weight and balance terminology**

### **Reference datum :**

Datum perpendicular to the longitudinal airplane centerline from which all distances are measured for balance purpose.

**Arm** : Is the distance from the reference datum to the center of gravity (C.G.) of an item.

**Moment** : Is the product of the weight of an item multiplied by its arm.

### **Center of gravity (C.G.) :**

Airplane balance point. Its distance from the reference datum is found by dividing the total moment by the total weight of the airplane.

**C.G. limits** : Center of Gravity Limits are the extreme center of gravity locations within which the airplane must be operated at a given weight.

### **Standard empty weight :**

Weight of a standard airplane including unusable fuel and full operating fluids (oil and hydraulic fluids).

### **Basic empty weight :**

Standard empty weight plus optional equipment.

**Useful load** : Is the difference between maximum ramp weight and the basic empty weight.

**Maximum ramp weight :**

Is the maximum weight approved for ground maneuver. It includes the weight of start, taxi and run up fuel.

**Maximum takeoff weight :**

Is the maximum weight approved at the beginning of the takeoff run.

**Maximum landing weight :**

Is the maximum weight approved for landing touchdown.

**General abbreviations**

|                  |  |
|------------------|--|
| <b>A</b>         | : Ampere or Amber                        |
| <b>ADC</b>       | : Air Data Computer                      |
| <b>AGL</b>       | : Above ground level                     |
| <b>AIL TRIM</b>  | : Aileron TRIM                           |
| <b>ALT. SEL.</b> | : Altitude selector                      |
| <b>ALTI</b>      | : Altimeter                              |
| <b>AMP</b>       | : Ampere                                 |
| <b>AoA</b>       | : Angle of Attack                        |
| <b>AP</b>        | : Autopilot                              |
| <b>ATIS</b>      | : Automatic Terminal Information Service |
| <b>AUTO SEL</b>  | : Automatic selector                     |
| <b>AUX BP</b>    | : Auxiliary boost pump                   |
| <b>BAT</b>       | : Battery                                |
| <b>BRT</b>       | : Brightness                             |
| <b>CAS</b>       | : Crew Alerting System                   |
| <b>°C</b>        | : Celsius degree                         |
| <b>CONT.</b>     | : Control                                |
| <b>DIEGME</b>    | : Diethylene glycol monomethyl ether     |
| <b>DISC</b>      | : Disconnect                             |
| <b>DN</b>        | : Down                                   |
| <b>EDM</b>       | : Emergency Descent Mode                 |
| <b>EGME</b>      | : Ethylene glycol monomethyl ether       |

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|                     |                                   |
|---------------------|-----------------------------------|
| <b>EIS</b>          | : Engine Indication System        |
| <b>EMER</b>         | : Emergency                       |
| <b>ESHP</b>         | : Estimated shaft horsepower      |
| <b>ESP</b>          | : Electronic Stability Protection |
| <b>ESS. BUS TIE</b> | : Essential BUS tie               |
| <b>EXT. LIGHTS</b>  | : Exterior lightings              |
| <b>°F</b>           | : Fahrenheit degree               |
| <b>FCU</b>          | : Fuel control unit               |
| <b>FL</b>           | : Flight level                    |
| <b>FOB</b>          | : Fuel On Board                   |
| <b>FPL</b>          | : Flight Plan                     |
| <b>ft</b>           | : Feet                            |
| <b>ft/min</b>       | : Feet per minute                 |
| <b>G</b>            | : Green                           |
| <b>GIFD</b>         | : Garmin Integrated Flight Deck   |
| <b>HI</b>           | : High                            |
| <b>HP</b>           | : High pressure                   |
| <b>hPa</b>          | : Hectopascal                     |
| <b>hr</b>           | : Hour                            |
| <b>HTR</b>          | : Heater                          |
| <b>IGNIT</b>        | : Ignition                        |
| <b>in</b>           | : Inch / inches                   |
| <b>INERT SEP</b>    | : Inertial separator              |
| <b>INDIC</b>        | : Indicator                       |
| <b>in.Hg</b>        | : Inch of mercury                 |
| <b>INT. LIGHTS</b>  | : Interior lightings              |
| <b>INSTR.</b>       | : Instrument                      |
| <b>ITT</b>          | : Interturbine temperature        |

|                      |   |
|----------------------|---|
| <b>kg</b>            | : Kilogram                              |
| <b>kt</b>            | : Knot (1 nautical mile/hr - 1852 m/hr) |
| <b>kW</b>            | : Kilowatt                              |
| <b>l</b>             | : Litre                                 |
| <b>L or L.H.</b>     | : Left                                  |
| <b>l/h</b>           | : Litre / hour                          |
| <b>lb or lbs</b>     | : Pound(s)                              |
| <b>L / D</b>         | : Lift-to-drag                          |
| <b>LDG</b>           | : Landing                               |
| <b>LDG GR</b>        | : Landing gear                          |
| <b>LDR</b>           | : Lightweight Data Recorder             |
| <b>LFE</b>           | : Landing Field Elevation               |
| <b>LRCR</b>          | : Long Range Cruise                     |
| <b>LO</b>            | : Low                                   |
| <b>LP</b>            | : Low pressure                          |
| <b>LRN</b>           | : Long range navigation                 |
| <b>LTS TEST</b>      | : Lightings test                        |
| <b>m</b>             | : Metre                                 |
| <b>m.a.c. or MAC</b> | : Mean aerodynamic chord                |
| <b>MAIN GEN</b>      | : Main generation                       |
| <b>MAN</b>           | : Manual                                |
| <b>MAN OVRD</b>      | : Manual override                       |
| <b>MAX RPM</b>       | : Maximum revolutions per minute        |
| <b>MDA</b>           | : Minimum Descent Altitude              |
| <b>MFD</b>           | : Multi-function Display                |
| <b>MIN</b>           | : Minimum                               |
| <b>min</b>           | : Minute                                |
| <b>mm</b>            | : Millimetre                            |

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|                  |  |
|------------------|--|
| <b>MLW</b>       | : Maximum Landing Weight                       |
| <b>MRW</b>       | : Maximum Ramp Weight                          |
| <b>msg</b>       | : Message                                      |
| <b>MTOW</b>      | : Maximum Takeoff Weight                       |
| <b>MXCR</b>      | : Maximum Cruise                               |
| <b>MZFW</b>      | : Maximum Zero Fuel Weight                     |
| <b>NM</b>        | : Nautical mile                                |
| <b>NOCR</b>      | : Normal cruise (recommended)                  |
| <b>NORM</b>      | : Normal                                       |
| <b>PFD</b>       | : Primary Flight Display                       |
| <b>PHF</b>       | : Plan Horizontal Fixe (Horizontal stabilizer) |
| <b>PRESS</b>     | : Pressure                                     |
| <b>PROP</b>      | : Propeller                                    |
| <b>psi</b>       | : Pounds per square inch                       |
| <b>PSIG</b>      | : Pounds per Square Inch Gage                  |
| <b>qt</b>        | : Quart ( $\frac{1}{4}$ USG)                   |
| <b>QTY</b>       | : Quantity                                     |
| <b>R or R.H.</b> | : Right  |
| <b>RUD</b>       | : Rudder                                       |
| <b>s or sec</b>  | : Second                                       |
| <b>SEL</b>       | : Selector                                     |
| <b>SIG</b>       | : Signalization                                |
| <b>SL</b>        | : Sea level                                    |
| <b>S/N</b>       | : Serial number                                |
| <b>SPKR</b>      | : Speaker                                      |
| <b>ST - BY</b>   | : Stand-by                                     |
| <b>STALL HTR</b> | : Stall heater                                 |
| <b>Std</b>       | : Standard                                     |

|                   |                          |
|-------------------|--------------------------|
| <b>T°</b>         | : Temperature            |
| <b>TEMP</b>       | : Temperature            |
| <b>TO</b>         | : Takeoff                |
| <b>TURN COORD</b> | : Turn coordinator       |
| <b>USG</b>        | : Gallon U.S             |
| <b>USP</b>        | : Under Speed Protection |
| <b>V</b>          | : Volt or Voltage        |
| <b>WARN</b>       | : Warning                |
| <b>W / S</b>      | : Windshield             |

### **Radio-navigation abbreviations**

|               |   |
|---------------|---|
| <b>ADF</b>    | : Automatic Direction Finder System               |
| <b>ADI</b>    | : Attitude Director Indicator                     |
| <b>ADS-B</b>  | : Automatic Dependent Surveillance-Broadcast      |
| <b>AFCS</b>   | : Automated Flight Control System                 |
| <b>AHRS</b>   | : Attitude and Heading Reference System           |
| <b>AIRAC</b>  | : Aeronautical Information Regulation And Control |
| <b>ATC</b>    | : Air Traffic Control                             |
| <b>B RNAV</b> | : Basic aRea NAVigation                           |
| <b>CDI</b>    | : Course Deviation Indicator                      |
| <b>COM</b>    | : Communications Transceivers                     |
| <b>DME</b>    | : Distance Measuring Equipment                    |
| <b>ELT</b>    | : Emergency Locator Transmitter                   |
| <b>FDE</b>    | : Fault Detection and Exclusion                   |
| <b>FMS</b>    | : Flight Management System                        |
| <b>GNSS</b>   | : Global Navigation Satellite System              |
| <b>GPS</b>    | : Global Positioning System                       |
| <b>HF</b>     | : High Frequency                                  |
| <b>IFR</b>    | : Instrument Flight Rules                         |

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|                  |  |
|------------------|--|
| <b>ILS</b>       | : Instrument Landing System                    |
| <b>IMC</b>       | : Instrument Meteorological Conditions         |
| <b>L NAV</b>     | : Lateral NAVigation                           |
| <b>LPV</b>       | : Localizer Precision Vertical                 |
| <b>MKR</b>       | : Marker Radio Beacon                          |
| <b>NAV</b>       | : Navigation Indicators or Receivers           |
| <b>P RNAV</b>    | : Precision aRea NAVigation                    |
| <b>RAIM</b>      | : Receiver Autonomous Integrity and Monitoring |
| <b>RF Leg</b>    | : Radius to Fix Leg                            |
| <b>R NAV</b>     | : Area NAVigation                              |
| <b>RNP</b>       | : Required Navigation Performance              |
| <b>SAM</b>       | : Standby Attitude Module                      |
| <b>SBAS</b>      | : Satellite Based Augmentation System          |
| <b>STAR</b>      | : Standard Terminal Arrival Route              |
| <b>TAS</b>       | : Traffic Advisory System                      |
| <b>TAWS</b>      | : Terrain Awareness Warning System             |
| <b>VFR</b>       | : Visual Flight Rules                          |
| <b>VHF</b>       | : Very High Frequency                          |
| <b>VMC</b>       | : Visual Meteorological Conditions             |
| <b>V NAV</b>     | : Vertical NAVigation                          |
| <b>VOR</b>       | : VHF Omnidirectional Range                    |
| <b>VOR / LOC</b> | : VHF Omnidirectional Range LOCALizer          |
| <b>WAAS</b>      | : Wide Area Augmentation System                |
| <b>WFDE</b>      | : WAAS Fault Detection and Exclusion           |
| <b>WGS</b>       | : World Geodetic System                        |
| <b>WXR</b>       | : Weather surveillance radar                   |
| <b>XPDR</b>      | : Transponder                                  |

Intentionally left blank



**1.5 - Conversion factors**

| Imperial and U.S units to metric units |         |           | Metric units to Imperial and U.S units |         |           |
|--|---------|-----------|--|---------|-----------|
| Multiply                               | By      | To obtain | Multiply                               | By      | To obtain |
| feet                                   | 0.3048  | metre     | metre                                  | 3.2808  | feet      |
| inch                                   | 25.4    | mm        | mm                                     | 0.03937 | Inch      |
| Imp.Gal                                | 4.546   | litre     | litre                                  | 0.220   | Imp.Gal   |
| USG                                    | 3.785   | litre     | litre                                  | 0.264   | USG       |
| lb                                     | 0.45359 | kg        | kg                                     | 2.2046  | lb        |

Figure 1.5.1 - Imperial and U.S units to metric units

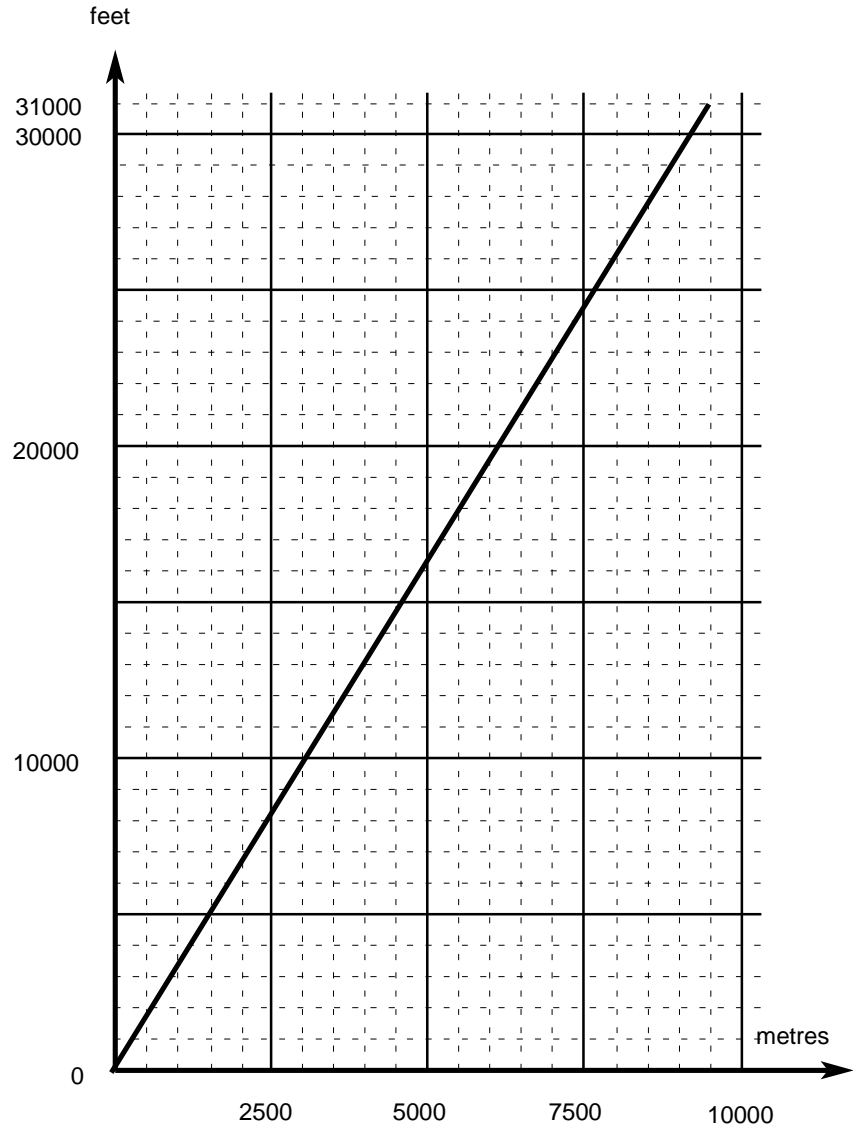


Figure 1.5.2 - Feet versus metres

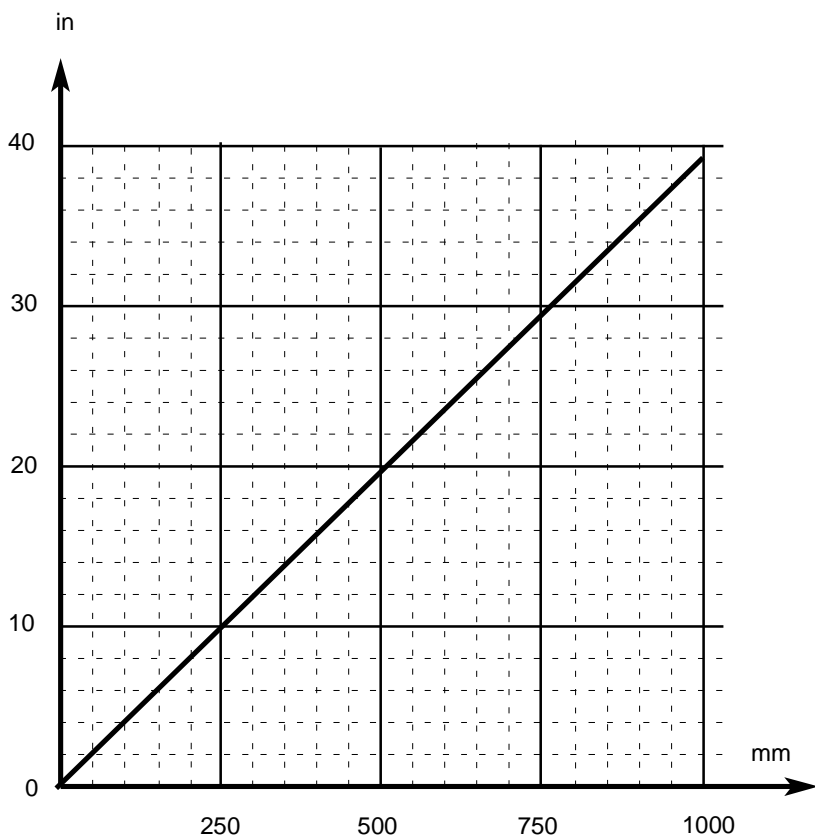


Figure 1.5.3 - Inches versus millimetres

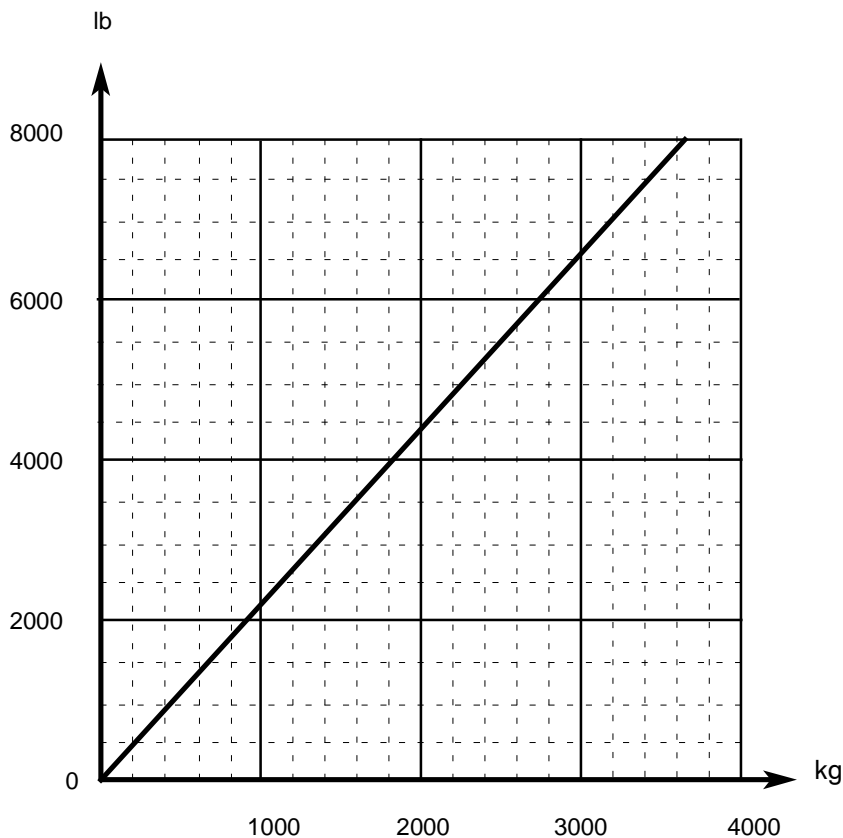


Figure 1.5.4 - Pounds versus kilograms

**1.6 - Pressure and standard atmosphere****Standard atmosphere**

| Pressure<br>altitude<br>(ft) | Pressure<br>(hPa) | °C     | °F     |
|------------------------------|-------------------|--------|--------|
| 0                            | 1013.2            | + 15.0 | + 59.0 |
| 2000                         | 942.1             | + 11.0 | + 51.8 |
| 4000                         | 875.0             | + 7.0  | + 44.6 |
| 6000                         | 811.9             | + 3.1  | + 37.6 |
| 8000                         | 752.6             | - 0.8  | + 30.5 |
| 10000                        | 696.8             | - 4.8  | + 23.4 |
| 12000                        | 644.3             | - 8.7  | + 16.2 |
| 14000                        | 595.2             | - 12.7 | + 9.2  |
| 16000                        | 549.1             | - 16.6 | + 2.2  |
| 18000                        | 505.9             | - 20.6 | - 5.0  |
| 20000                        | 465.6             | - 24.6 | - 12.4 |
| 22000                        | 427.8             | - 28.5 | - 19.3 |
| 24000                        | 392.6             | - 32.5 | - 26.5 |
| 26000                        | 359.8             | - 36.5 | - 33.6 |
| 28000                        | 329.3             | - 40.4 | - 40.7 |
| 30000                        | 300.8             | - 44.4 | - 47.8 |
| 31000                        | 287.4             | - 46.4 | - 51.6 |

Figure 1.6.1 - Standard atmosphere

## Pressure conversion table

• NOTE •

The standard pressure of 1013.2 hPa is equal to 29.92 inches of mercury.

|               |               |               |               |               |               |               |               |               |               |
|---------------|---------------|---------------|---------------|---------------|---------------|---------------|---------------|---------------|---------------|
| 950<br>28.05  | 951<br>28.08  | 952<br>28.11  | 953<br>28.14  | 954<br>28.17  | 955<br>28.20  | 956<br>28.23  | 957<br>28.26  | 958<br>28.29  | 959<br>28.32  |
| 960<br>28.35  | 961<br>28.38  | 962<br>28.41  | 963<br>28.44  | 964<br>28.47  | 965<br>28.50  | 966<br>28.53  | 967<br>28.56  | 968<br>28.58  | 969<br>28.61  |
| 970<br>28.64  | 971<br>28.67  | 972<br>28.70  | 973<br>28.73  | 974<br>28.76  | 975<br>28.79  | 976<br>28.82  | 977<br>28.85  | 978<br>28.88  | 979<br>28.91  |
| 980<br>28.94  | 981<br>28.97  | 982<br>29.00  | 983<br>29.03  | 984<br>29.06  | 985<br>29.09  | 986<br>29.12  | 987<br>29.15  | 988<br>29.18  | 989<br>29.20  |
| 990<br>29.23  | 991<br>29.26  | 992<br>29.29  | 993<br>29.32  | 994<br>29.35  | 995<br>29.38  | 996<br>29.41  | 997<br>29.44  | 998<br>29.47  | 999<br>29.50  |
| 1000<br>29.53 | 1001<br>29.56 | 1002<br>29.59 | 1003<br>29.62 | 1004<br>29.65 | 1005<br>29.68 | 1006<br>29.71 | 1007<br>29.74 | 1008<br>29.77 | 1009<br>29.80 |
| 1010<br>29.83 | 1011<br>29.85 | 1012<br>29.88 | 1013<br>29.91 | 1014<br>29.94 | 1015<br>29.97 | 1016<br>30.00 | 1017<br>30.03 | 1018<br>30.06 | 1019<br>30.09 |
| 1020<br>30.12 | 1021<br>30.15 | 1022<br>30.18 | 1023<br>30.21 | 1024<br>30.24 | 1025<br>30.27 | 1026<br>30.30 | 1027<br>30.33 | 1028<br>30.36 | 1029<br>30.39 |
| 1030<br>30.42 | 1031<br>30.45 | 1032<br>30.47 | 1033<br>30.50 | 1034<br>30.53 | 1035<br>30.56 | 1036<br>30.59 | 1037<br>30.62 | 1038<br>30.65 | 1039<br>30.68 |
| 1040<br>30.71 | 1041<br>30.74 | 1042<br>30.77 | 1043<br>30.80 | 1044<br>30.83 | 1045<br>30.86 | 1046<br>30.89 | 1047<br>30.92 | 1048<br>30.95 | 1049<br>30.98 |

Figure 1.6.2 - Pressure conversion table

## Section 2

### Limitations

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## 2.1 - General

TBM 930 is the trade name of the TBM 700 N version airplane (TBM 700 type), which is certified in the normal category.

This airplane must be flown in compliance with the limits specified by placards or markings and with those given in this section and throughout the POH.

The GARMIN G3000 Integrated Flight Deck Pilot's Guide, No. 190-02046-00, or any later version as applicable, must be readily available to the pilot and permanently kept in the airplane with the POH.

The Pilot's Guide for MD302 Standby Attitude Module P/N 9017846-00 or any later version as applicable, must be permanently kept in the airplane with the POH.

Departure into IMC is not authorized if the MD302 battery fails its initial capacity check with associated message :

"WARNING  
INTERNAL BATTERY MAY PROVIDE LESS THAN 60 MINUTES OF  
OPERATIONS"

or if there is a red "X" over the battery symbol at MD302 initialization.

This section of the airplane POH presents the various operating limitations, the significance of such limitations, instrument markings, color coding, and basic placards necessary for the safe operation of the airplane, its powerplant and installed equipment.

The limitations included in this section have been approved by the Federal Aviation Administration in accordance with 14 CFR section 21.29.

The limitations for some optional systems are given in section 9, Supplements of the POH.

TBM 700 airplane is certified under EASA.A.010 and FAA N° A60EU Type Certificates.

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## 2.2 - Airspeed limitations

Airspeed limitations and their operational significance are shown in figure 2.2.1.

|                 | Speed                                  | KCAS | KIAS | Remarks  |
|-----------------|--|------|------|--|
| V <sub>MO</sub> | Maximum operating speed                | 271  | 266  | Do not intentionally exceed this speed in normal flight category |
| V <sub>A</sub>  | Maneuvering speed                      | 160  | 158  | Do not make abrupt or full control movements above this speed    |
| V <sub>FE</sub> | Maximum flaps extended speed :         |      |      | Do not exceed these speeds depending on flaps position           |
|                 | landing configuration                  | 120  | 122  |  |
|                 | takeoff configuration                  | 180  | 178  |  |
| V <sub>LO</sub> | Maximum landing gear operating speed : |      |      | Do not extend or retract landing gear above this speed           |
|                 | extension                              | 180  | 178  |  |
|                 | retraction                             | 151  | 150  |  |
|                 | emergency extension                    | 151  | 150  |  |
| V <sub>LE</sub> | Maximum landing gear extended speed    | 180  | 178  | Do not exceed this speed with landing gear extended              |

Figure 2.2.1 - Airspeed limitations

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## 2.3 - Powerplant limitations

### Engine

Number of engines : 1

Engine manufacturer : PRATT & WHITNEY CANADA

Engine model number : PT6A - 66D

#### Maximum power :

100 % at Np = 2000 RPM

#### Ng limitation :

104.1 %

#### Np limitation :

2000 RPM  $\pm$  40 RPM

#### ITT limitations :

- Takeoff : 850°C
- Maximum climb/cruise : 840°C
- During start :
  - < 840°C, no duration limitation
  - < 870°C for 20 seconds max.
  - < 1000°C for 5 seconds max.

#### ▲ CAUTION ▲

When normally operating, refer to chapter 5.8 Engine operation tables.



## Oil

### ▲ CAUTION ▲

Do not mix different viscosities or specifications of oil as their different chemical structure can make them incompatible.



Maximum oil temperature : 104 °C

Oil pressure :

- Minimum : 60 psi
- Maximum : 135 psi, a transient oil pressure up to 170 psi is acceptable for maximum 20 seconds

Normal oil pressure is 105 to 135 psi. Oil pressures under 105 psi are undesirable. Under emergency conditions, to complete a flight, a lower oil pressure of 60 psi is permitted at reduced power level not exceeding 80% torque. Oil pressures below 60 psi are unsafe and require that either the engine be shut down or a landing be made as soon as possible using the minimum power required to sustain flight.

Oil capacity :

- System total capacity : 12.7 Quarts (12 litres), oil cooler included
- Usable capacity : 6 Quarts (5.7 litres)

## Fuel

Fuel limitations :

- 2 tanks : 150.5 USG (570 litres) each
- Total fuel : 301 USG (1140 litres)
- Usable fuel : 292 USG (1106 litres)
- Unusable fuel : 9 USG (34 litres)
- Maximum fuel imbalance : 15 USG (57 litres)

• NOTE •

Usable fuel can be safely used during all normal airplane maneuvers.

•

▲ CAUTION ▲

The fuel used must contain an anti-ice additive, in accordance with specification MIL-I-27686 or MIL-I-85470. Additive concentrations (EGME or DIEGME) shall be comprised between a minimum of 0.06 % and a maximum of 0.15 % by volume. Refer to section 8 Handling, servicing and maintenance for additional information.

▲

▲ CAUTION ▲

The use of aviation gasoline (AVGAS) must be restricted to emergency purposes only. AVGAS shall not be used for more than 150 cumulative hours during any period between engine overhaul periods.

▲

• NOTE •

Use of AVGAS to be recorded in engine module logbook.

•

▲ CAUTION ▲

Maximum time for sideslip condition is 30 seconds.

▲

| US specification<br>(US)                                  | French specification<br>(FR) | English specification<br>(UK) | NATO code                                |
|---|------------------------------|-------------------------------|--|
| ASTM-D1655 JET A<br>ASTM-D1655 JET A1<br>ASTM-D1655 JET B | AIR 3405C Grade F35          | DERD 2494 Issue 9             | F35 without<br>additive                  |
| MIL-DTL-5624<br>Grade JP-4                                | AIR 3407B                    | DERD 2454 Issue 4<br>Amdt 1   | F40 with<br>additive                     |
| MIL-DTL-5624<br>Grade JP-5                                | AIR 3404C Grade F44          | DERD 2452 Issue 2<br>Amdt 1   | F44 with<br>additive when<br>utilization |
| MIL-DTL-83133<br>Grade JP-8                               | AIR 3405C Grade F34          | DERD 2453 Issue 4<br>Amdt 1   | F34 with<br>additive S748                |
|   | AIR 3404C Grade F43          | DERD 2498 Issue 7             | F43 without<br>additive                  |

Figure 2.3.1 - Recommended fuel types  
Reference : Service Bulletin P & W C. No. 14004

## Propeller

Number of propellers : 1

Propeller manufacturer : HARTZELL

Propeller model number : HC-E5N-3C / NC8834K

Propeller diameter :

- Minimum : 90 in (2.286 m)
- Maximum : 91 in (2.311 m)

■ Propeller blade setting at station 30 in :

- Low pitch : 19.5°
- Feathering : 85°
- Maximum reverse : - 9°



## 2.4 - Starter operation limits

Starter operation sequence is limited as follows :

if Ng < 30 % ..... 30 seconds

if Ng > 30 % ..... 60 seconds

Should several sequences be necessary, respect following spacing :

1st sequence

wait ..... 1 minute

2nd sequence

wait ..... 5 minutes

3rd sequence

wait ..... 30 minutes

4th sequence

Intentionally left blank

## 2.5 - Weight and C.G. limits

### Weight limits

Maximum ramp weight (MRW) : 7430 lbs (3370 kg)

Maximum takeoff weight (MTOW) : 7394 lbs (3354 kg)

Maximum landing weight (MLW) : 7024 lbs (3186 kg)

Maximum zero fuel weight (MZFW) : 6032 lbs (2736 kg)

Maximum baggage weight :

- in FWD compartment (non pressurized) : 110 lbs (50 kg)

>> *With 6-seat accommodation*

- in rear part of pressurized cabin : 220 lbs (100 kg)

>> *With 4-seat accommodation*

- in rear part of pressurized cabin : 396 lbs (180 kg), with small or large net, see sketch below

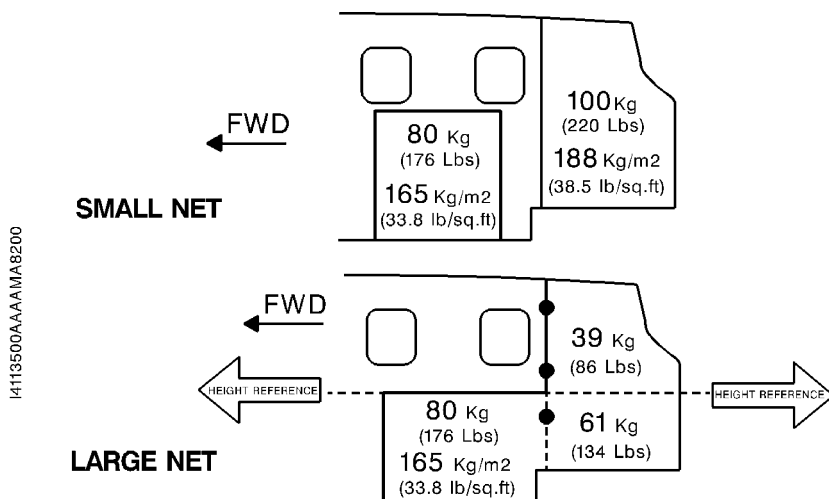


Figure 2.5.1 - Baggage limits

## C.G. limits

Center of gravity range with landing gear down and flaps up, attitude 0° :

Forward limits :

181.3 in (4.604 m) aft of datum at 4409 lbs (2000 kg) or less (14 % of m.a.c)

183.6 in (4.664 m) aft of datum at 6250 lbs (2835 kg) (18 % of m.a.c)

185.3 in (4.707 m) aft of datum at 6579 lbs (2984 kg) (20.85 % of m.a.c)

187 in (4.752 m) aft of datum at all weights above 7024 lbs (3186 kg) (23.8 % of m.a.c)

Aft limits :

193.65 in (4.921 m) aft of datum at 7394 lbs (3354 kg) (35 % of m.a.c)

194 in (4.928 m) aft of datum at 6986 lbs (3169 kg) (35.5 % of m.a.c)

Reference datum : 118.1 in (3 m) in front of the firewall front face.

Straight line variation between points.

Leveling point : Cabin floor rails.

### • NOTE •

It is the responsibility of the pilot to insure that the airplane is properly loaded.

See section 6 Weight and balance for proper loading instructions.

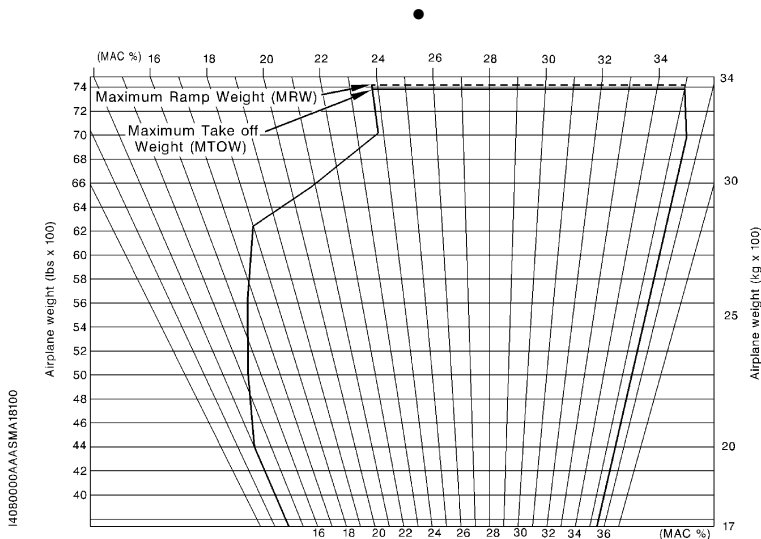


Figure 2.5.2 - C.G. limits

## 2.6 - Operation limits

### Maneuver limits

This airplane is certified in the normal category.

The normal category is applicable to airplanes intended for non-aerobatic operations.

Non-aerobatic operations include any maneuvers incidental to normal flying, stalls (except whip stalls), lazy eights, chandelles, and steep turns in which the angle of bank is no more than 60°.

#### ▲ WARNING ▲

**Aerobatic maneuvers, including spins, are not approved.**



### Temperature limits

Minimum temperature at start and takeoff : - 40°C (- 40°F)

Maximum temperature at start and takeoff :

ISA + 37°C (+ 67°F) from 0 to 8000 ft pressure altitude

Maximum temperature in flight :

ISA + 37°C (+ 67°F) from 0 to 8000 ft pressure altitude

ISA + 30°C (+ 54°F) above 8000 ft pressure altitude

### Flight load factor limits

#### Flaps up

Weight below 6579 lbs (2984 kg) : - 1.5 < n < + 3.8 g

Weight above 6579 lbs (2984 kg) : - 1.5 < n < + 3.5 g

#### Flaps down

- 0 < n < + 2.0 g

#### ▲ CAUTION ▲

Intentional negative load factors prohibited.



## Generator limits

- Generator load must be below 200 A when the airplane is on the ground.

## GFC 700 autopilot limits

- During autopilot operation, a pilot with seat belt fastened must be seated at the left or right position.
- The autopilot and yaw damper must be OFF during takeoff and landing.
- Do not engage autopilot below 1000 ft (300 m) AGL in cruise or climb.
- Do not use autopilot in approach below 200 ft (60 m) AGL.
- Do not use autopilot for airspeeds below 85 KIAS.

● NOTE ●

Do not use the autopilot in descent below 2000 ft (600 m) AGL with a vertical speed in excess of 2000 ft/min.

●

## GNSS (GPS/SBAS) navigation equipment approvals

The GARMIN GNSS navigation system installed in this airplane :

- is a GPS system with a Satellite Based Augmentation System (SBAS) comprised of :
  - two TSO-C145a (or later) Class 3 approved GARMIN GIAs,
  - TSO-C146c Class 3 approved GARMIN GDUs Display Units,
  - GARMIN GA36 and GA37 antennas,
  - GPS software version 5.0 or later approved version.
- is installed in this airplane in accordance with AC 20-138A.
- is, as installed in this airplane, approved for navigation using GPS and GPS/SBAS (within the coverage of a SBAS complying with ICAO Annex 10) for IFR enroute, terminal area, and RNP APCH operations (to LNAV, LNAV/VNAV and LPV minima).
- is, as installed in this airplane, complying with the equipment, performance, and functional requirements to conduct RNAV and RNP operations in accordance with the applicable requirements of the reference documents listed in the following table.

• NOTE •

Depending on the area of navigation, an operational approval may be required to use the navigation performance that are detailed in the table hereafter. The pilot is responsible to ensure compliance with current operational requirements.

•

This table is accurate at the time it was published.

| Phase of flight  | Approved PBN Capability                          | Operational limitations   | Reference Documents                    | ICAO Flight Plan Code |              | Notes   |
|--|--|---|--|-----------------------|--------------|---|
|  |  |   |  | Item 10a Code         | Item 18 PBN/ |   |
| En-route, Oceanic and Remote continental (Class II Navigation) | RNAV 10<br><br>RNP 10                            | GNSS FDE availability must be verified prior to flight. Maximum predicted FDE unavailability is 34 minutes. <sup>1</sup><br><br>Two GNSS systems required to be operational. <sup>2</sup> | FAA<br>AC 90-105A                      | R                     | A1           | Additional communication and surveillance equipment may be required to obtain operational approval to utilize RNP 10 / RNAV 10 performance. |
| En-route continental, Arrival                                  | RNAV 5<br><i>(formerly designated as B-RNAV)</i> | One GNSS system required to be operational.   | JAA<br>AMJ 20X2                        | R                     | B2           |   |
| En-route, Oceanic and Remote continental (Class II Navigation) | RNP 4  | GNSS FDE availability must be verified prior to flight. Maximum predicted FDE unavailability is 25 minutes. <sup>1</sup><br><br>Two GNSS systems required to be operational. <sup>2</sup> | FAA<br>AC 90-105A                      | R                     | L1           | Additional communication and surveillance equipment may be required to obtain operational approval to utilize RNP 4 performance.            |
| Departure<br>En-route continental, Arrival                     | RNAV 2 /<br>RNAV 1                               | One GNSS system required to be operational,   | JAA<br>TGL-10<br><br>FAA<br>AC 90-105A | R                     | C2 /<br>D2   |   |

| Phase of flight                                    | Approved PBN Capability                               | Operational limitations  | Reference Documents                        | ICAO Flight Plan Code |              | Notes   |
|--|---|--|--|-----------------------|--------------|---|
|  |   |  |  | Item 10a Code         | Item 18 PBN/ |   |
| Domestic, Offshore, Oceanic and Remote continental | RNP 2   | GNSS FDE availability must be verified prior to oceanic or remote continental flight. Maximum predicted FDE unavailability is 5 minutes.<br><br>Two GNSS systems required to be operational. <sup>2</sup><br><br>Only one operational GNSS system required for domestic and offshore operations areas. | FAA<br>AC 90-105A                          | R                     | -            | Additional communication and surveillance equipment may be required to obtain operational approval to utilize RNP 2 performance.  |
| Departure, Arrival,                                | RNP 1<br>(with and without RF legs)                   | At a minimum, the flight director must be displayed and utilized when conducting procedures containing RF legs.  | FAA<br>AC 90-105A                          | R                     | O2           | Includes RNP terminal departure and arrival procedures. This includes procedures with Radius-to-Fix legs (RF legs).   |
| Approach   | RNP APCH<br>LNAV minima<br>(with and without RF legs) | At a minimum, the flight director must be displayed and utilized when conducting procedures containing RF legs.  | EASA<br>AMC 20-27<br><br>FAA<br>AC 90-105A | R                     | S1           | Includes non-precision approaches based on conventional navigation aids with "or GPS" in the title and area navigation approaches titled "GPS", "RNAV (GPS)", and "RNAV (GNSS)". This includes procedures with RF legs.<br><br>RF legs may be used in the initial and intermediate legs of the approach procedure or the final leg of the missed approach procedure only. |



| Phase of flight | Approved PBN Capability                                    | Operational limitations   | Reference Documents | ICAO Flight Plan Code |              | Notes  |
|-----------------|--|---|---------------------|-----------------------|--------------|--|
|                 |  |   |                     | Item 10a Code         | Item 18 PBN/ |  |
| Approach        | RNP<br>APCH<br>LPV<br>minima<br>(with and without RF legs) | At a minimum, the flight director must be displayed and utilized when conducting procedures containing RF legs. | EASA<br>AMC 20-28   | B                     |              | RF legs may be used in the initial and intermediate legs of the approach procedure or the final leg of the missed approach procedure only. |

<sup>1</sup> and <sup>2</sup>, see Note 1 and Note 2 hereafter

Table 2.6.1 - GNSS operational requirements

Note 1 FDE/RAIM availability worldwide can be determined using the WFDE Prediction program, part number 006-A0154-01 or later approved version with GARMIN GA36 and GA37 antennas selected, or :

- Within the United States, using the FAA's en-route and terminal RAIM prediction website : [www.raimprediction.net](http://www.raimprediction.net), or by contacting a Flight Service Station.
- Within Europe, using Europe's AUGUR GPS RAIM Prediction Tool at <http://augur.ecacnav.com/augur/app/home>.

Note 2 A **BOTH ON GPS1** or **BOTH ON GPS2** system annunciation does not necessarily mean that one GPS has failed. Refer to the MFD – GPS STATUS page to determine the state of the unused GPS.

### General considerations

The route planning and WFDE prediction program may be downloaded from the GARMIN website on the internet. For information on using the WFDE Prediction Program, refer to GARMIN WAAS FDE Prediction Program, part number 190-00643-01, 'WFDE Prediction Program Instructions'.

Garmin International holds an FAA Type 2 Letter of Acceptance (LOA) in accordance with RTCA/DO-200A and AC 20-153B for database integrity, quality, and database management processes for many of its aviation databases. LOA status and RTCA/DO-200A List of Applicable Avionics (190-01999-00) can be viewed at FlyGarmin.com.

Navigation information is referenced to WGS-84 reference system, and should only be used where the Aeronautical Information Publication (including electronic data and aeronautical charts) conform to WGS-84 or equivalent.

## **GNSS (GPS/SBAS) navigation system limitations**

### **Navigation database limitations**

The pilot must confirm at system initialization that the Navigation database is current.

If the AIRAC cycle will change during flight, the pilot must ensure the accuracy of navigation data, including suitability of navigation facilities used to define the routes and procedures for flight. If an amended chart affecting navigation data is published for the procedure, the database must not be used to conduct the procedure.

GPS/SBAS based IFR enroute, oceanic, and terminal navigation is prohibited unless the pilot verifies and uses a valid, compatible, and current Navigation database or verifies each waypoint for accuracy by reference to current approved data.

Discrepancies that invalidate a procedure must be reported to Garmin International. The affected procedure is prohibited from being flown using data from the Navigation database until a new Navigation database is installed in the airplane and verified that the discrepancy has been corrected.

Contact information to report Navigation database discrepancies can be found at [www.Garmin.com](http://www.Garmin.com)>Support>Contact Garmin Support>Aviation. Pilots and operators can view navigation data base alerts at [www.Garmin.com](http://www.Garmin.com)>In the Air>NavData Alerts.

RNP APCH including "GPS", "or GPS", "RNAV (GPS)" and "RNAV (GNSS)" instrument approaches using the Garmin integrated flight deck are prohibited unless the pilot verifies and uses the current Navigation database. GPS based instrument approaches must be flown in accordance with an approved instrument approach procedure that is loaded from the Navigation database into the flight plan by its name.

Not all published Instrument Approach Procedures (IAP) are in the Navigation database.

Manual entry of waypoints using latitude/longitude or place/bearing is prohibited for published RNP and RNAV routes.

Whenever possible, RNP and RNAV routes including Standard Instrument Departures (SIDs) and Obstacle Departure Procedures (ODPs), Standard Terminal Arrival (STAR), and enroute RNAV Q and RNAV T routes should be loaded into the flight plan from the database in their entirety, rather than loading route waypoints from the database into the flight plan individually. Selecting and inserting individual named fixes from the database is permitted, provided all fixes along the published route to be flown are inserted.

**GNSS integrity limitations**

For flight planning purposes, in areas where SBAS coverage is not available, the pilot must check RAIM availability. The availability of GPS integrity RAIM shall be confirmed for the intended route of flight.

In the event of a predicted continuous loss of RAIM of more than five minutes for any part of the intended route of flight, the flight should be delayed, cancelled, or re-routed on a track where RAIM requirements can be met.

For flight planning purposes, in Remote Continental and Oceanic areas, the pilot must check FDE availability. Refer to the Table 2.6.1 - GNSS operational requirements, to check maximum authorized FDE unavailability and WFDE Prediction program references.

**Approach operations limitations**

LNAV+V feature is a standard LNAV approach with advisory vertical guidance provided for assistance in maintaining a constant vertical glidepath similar to an ILS glideslope on approach. This guidance is displayed on the PFD in the same location as the ILS glideslope using a magenta diamond. In all cases where LNAV+V is indicated by the system during an approach, LNAV minima shall be used.

Use of the GARMIN GPS/SBAS receivers to provide navigation guidance during the final approach segment of an ILS, LOC, LOC-BC, LDA, SDF, MLS or any other type of approach not approved for "or GPS" navigation is prohibited.

When using the VOR/LOC/GS receivers to fly the final approach segment, VOR/LOC/GS navigation data must be selected and presented on the CDI of the pilot flying.

**Procedures with RF legs (Radius to Fix legs)**

At a minimum, the flight director must be displayed and utilized when conducting procedures containing RF legs.

**Icing conditions**

Except for certain phases of flight where the POH specifies that deicing boots should not be used (e.g. take-off, final approach, and landing), compliance with the following is required.

Wing and Tail Leading Edge Pneumatic Deicing Boot System must be activated:

- At the first sign of ice formation anywhere on the aircraft, and
- The system must either be continued to be operated in the automatic cycling mode, if available; or the system must be manually cycled as needed to minimize the ice accretions on the airframe.

The wing and tail leading edge pneumatic deicing boot system may be deactivated only after leaving icing conditions and after the airplane is determined to be clear of ice.

In any case of icing conditions, first refer to Particular procedures described in chapter 4.5 and in case of unforeseen icing conditions, refer in addition to the Emergency procedure described in chapter 3.12.

## Severe icing conditions

### ▲ WARNING ▲

**Severe icing may result from environmental conditions outside of those for which the airplane is certificated. Flight in freezing rain, freezing drizzle, or mixed icing conditions (supercooled liquid water and ice crystals) may result in ice build-up on protected surfaces exceeding the capability of the ice protection system, or may result in ice forming aft of the protected surfaces. This ice may not be shed using the ice protection systems, and may seriously degrade the performance and controllability of the airplane.**



During flight, severe icing conditions that exceed those for which the airplane is certificated shall be determined by the following visual cues. If one or more of these visual cues exists, immediately request priority handling from air traffic control to facilitate a route or an altitude change to exit the icing conditions.

- Unusually extensive ice accumulation on the airframe and windshield in areas not normally observed to collect ice.
- Accumulation of ice on the upper surface of the wing aft of the protected area.

Since the autopilot, when operating, may mask tactile cues that indicate adverse changes in handling characteristics, use of the autopilot is prohibited when any of the visual cues specified above exist, or when unusual lateral trim requirements or autopilot trim warnings are encountered while the airplane is in icing conditions.

All wing icing inspection lights must be operative prior to flight into icing conditions at night.

• NOTE •

This supersedes any relief provided by the Master Minimum Equipment List (MMEL).

•

Refer to the list of equipment required depending on type of operation in this same chapter.

Refer to Particular procedures described in chapter 4.5 and in case of unforeseen icing conditions, refer in addition to the Emergency procedure described in chapter 3.12.

## Flap operating envelope

The use of flaps is not authorized above 15 000 ft.

## Reverse utilization

The use of control reverse BETA ( $\beta$ ) range is prohibited :

- during flight,
- on ground, if the engine is not running.

## Weather radar GWX 70

On ground, the radar radiation is inhibited when the landing gear shock absorbers are compressed. However, it is important to obey the following restrictions :

- Do not operate the radar during refueling operations or in the vicinity of trucks or containers containing flammables or explosives.
- Do not allow personnel within 12 feet of area being scanned by antenna when system is transmitting.

## Equipment required depending on type of operation

The airplane is approved for day & night VFR and day & night IFR operations when appropriate equipment is installed and operating correctly.

The type certification for each use requires the following equipment. The equipment must be installed and operate perfectly according to the indicated type of use.

▲ **CAUTION** ▲

It is the pilot's responsibility to check that the following equipment lists are in accordance with the specific national operation rules of the airplane registration country depending on the type of operation.



▲ **CAUTION** ▲

Systems and equipment mentioned hereafter do not include specific flight and radio-navigation instruments required by decree concerning operation conditions for civil airplanes in general aviation or other foreign regulations (for example FAR PART 91 and 135).



**Day VFR**

1. Pilot instruments
  - Airspeed indicator
  - Sensitive and adjustable altimeter
  - Magnetic compass with built-in compensator
2. CAS warning and caution messages
  - Oil pressure
  - Low fuel pressure
  - Fuel selector OFF
  - Fuel auxiliary pump ON
  - L.H. and R.H fuel tank low level
  - Non functioning of fuel timer
  - Battery stop
  - Main generator OFF
  - Low voltage
  - Ground power unit connected
  - Inertial separator
  - Starter
  - Ignition
  - Flaps
  - Landing gears and doors
3. Aural warning
  - $V_{MO}$  warning
  - Landing gear warning
  - Stall warning
4. Engine instruments
  - Torquemeter
  - Propeller tachometer
  - Interturbine temperature indicator (ITT)
  - Gas generator tachometer (Ng)
  - Oil pressure indicator
  - Oil temperature indicator

5. Various indicators

- Fuel gauge indicators (2)
- Voltmeter
- Ammeter
- Outside air temperature

6. Installations

- Fuel mechanical pump (main)
- Fuel electrical pump (auxiliary)
- Fuel shut-off valve
- Fuel timer
- Starter generator
- Inertial separator
- Stall warning
- Electrical aileron trim
- Electrical rudder trim
- Manual elevator pitch trim
- Engine ignition
- Landing gear electro-hydraulic unit
- Landing gear emergency hydraulic pump (manual)
- Flaps
- Overspeed regulator
- Electrical feathering
- Battery

7. Miscellaneous

- Seats (each occupant)
- Belts (each occupant)
- Straps (each occupant)
- Pilot's operating handbook



**Night VFR**

1. All equipment required for day VFR
2. Attitude display indicator
3. Instrument lighting
4. Instrument panel lighting
5. Emergency lighting
6. Vertical speed indicator
7. Navigation lights (4)
8. Anticollision lights (2)
9. Landing light

**IFR**

1. All equipment required for day VFR
2. All equipment required for night VFR, if flight is performed during night
3. Taxi light, if flight is performed during night
4. Clock
5. 2nd altimeter
6. Emergency static source
7. Pitot static tube deicing

### **Pressurized flight**

1. Cabin altimeter
2. Cabin vertical speed indication
3. Cabin differential pressure indication
4. Pressurization control valve
5. Safety valve
6. Pressurization control
7. Maximum cabin altitude and pressure warning light

### **Flight into icing conditions**

1. All equipment required for IFR flight
2. Propeller deicing
3. L.H. windshield deicing
4. Airframe, stabilizer and elevator horn deicing
5. Wing leading edge inspection light, if night flight
6. Stall warning deicing
7. Inertial separator
8. Garmin annunciation "Airspeed"

## Altitude operating limits

Maximum altitude : 31000 ft (9449 m)

Maximum differential pressure : 6.2 psi

### Operation in RVSM area

This airplane is approved for operations in Reduced Vertical Separation Minimum (RVSM) airspace when required equipment is maintained in accordance with the airplane maintenance manual - refer to section List of equipment, paragraph List of critical RVSM equipment.

This does not constitute operational approval. Individual airplane and operational approval must be obtained in accordance with applicable operating rules.

Each operator must ensure compliance with required crew training and operating practices and procedures.

Moreover, the following equipment must be installed and operating normally upon entering RVSM airspace :

- Pilot and R.H. station primary altimeters
- Autopilot
- Altitude alerter
- ATC transponder

#### • NOTE •

Any changes to the pitot / static, air data computer, autopilot, altitude alerting and / or transponder systems, or other changes that affect operation of these systems must be evaluated for impact on the RVSM approval.

The standby altimeter is not approved for RVSM operations.

•

## In-flight breaker use limits

A tripped breaker should not be reset in flight unless deemed necessary for continued safe flight and landing. Only one reset should be attempted.

## Enhanced mode S

The installed mode S system satisfies the data requirements of ICAO Doc 7030/4, regional supplementary procedures for SSR mode S enhanced surveillance in designated european airspace. The capability to transmit data parameters is shown in column 2 :

| Parameter                   | Available (A) / Not available (NA) |
|-----------------------------|------------------------------------|
| Magnetic heading            | A                                  |
| Indicated airspeed          | A                                  |
| Mach No                     | A                                  |
| Vertical rate               | A                                  |
| Roll angle                  | A                                  |
| True airspeed               | A                                  |
| True track angle            | A                                  |
| Groundspeed                 | A                                  |
| Selected altitude           | A                                  |
| Barometric pressure setting | A                                  |

## Chartview system operating limitations

The geographic-referenced airplane symbol on some charts must not be used for navigation.

• NOTE •

The airplane symbol displayed on some charts provides supplemental airplane situational awareness information. It is not intended as a means for navigation or flight guidance. The airplane symbol is not to be used for conducting instrument approaches or departures, and it should not be relied upon during low visibility taxi operations. Position accuracy, orientation, and related guidance must be assured by other means of required navigation.

•

Operators must have back-up charts available to the flight crew.

Database currency must be verified prior to use via database effectivity page.

The flight crew is responsible for verifying availability of charts for the planned flight.

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## **2.7 - Miscellaneous limits**

### **Seating limits C.G.**

- 2 front seats at 178.5 in (4.534 m)  
>> *With 4-seat accommodation or 6-seat accommodation*
- 2 intermediate seats at 224.8 in (5.710 m)  
>> *With 6-seat accommodation*
- Rear bench (2 seats) at 267.1 in (6.785 m)

### **Baggage limits**

- Baggage in pressurized cabin at 303 in (7.695 m)
- Baggage in non pressurized forward section at 128 in (3.250 m)

### **Minimum crew**

- One pilot at L.H. front seat

### **Maximum occupancy**

The number of persons on board is limited by approved seating configuration installed but must not exceed six, including the pilot.

The number of persons must be less than or equal to the number of seats.

### **Use of doors**

Flight with door open or ajar is prohibited.

### **Chemical toilet cabinet, if installed**

The cabinet must be stowed during takeoff and landing. No baggage on the top of the cabinet for the whole flight.

## Cargo net installation limits

Small cargo net : maximum loading height = 28 in (710 mm)

Large cargo net : maximum loading height = 22 in (565 mm) in cabin, out of baggage compartment.

### ▲ CAUTION ▲

No item may extend forward of the cargo net system to protect door from obstruction.



## Weather radar limitation

Weather radar can be displayed on :

- PFD 1,
- PFD 2,
- MFD, with different indications of mode.

Example :

- PFD 1 : "SBY" MODE
- MFD : "SBY" MODE
- PFD 2 : "ACTIVE" MODE

### ▲ CAUTION ▲

The radar is OFF only when all displays indicate OFF.





## 2.8 - Markings

### Airspeed indicator on PFD(s) and on Standby airspeed indicator

Markings and their color code significance are shown in figure 2.8.1.

| Marking                        | KIAS<br>(Value or range) | Significance   |
|--------------------------------|--------------------------|--|
| Red strip                      | Below 65                 | /  |
| White strip                    | 65 - 122                 | Full flap operating range<br>Lower limit is maximum weight<br>$V_{SO}$ in landing configuration. |
| Green strip                    | 122 - 266                | Normal operating airspeed range  |
| Red/white barber<br>pole strip | Above 266                | 266 = VMO  |

Figure 2.8.1 - Airspeed indicator markings

### Pressurization

| Marking  | Value   | Significance           |
|----------|---------|------------------------|
| Red line | 6.2 psi | Cabin $\Delta P$ limit |

Figure 2.8.2 - Pressurization marking

## Engine instruments

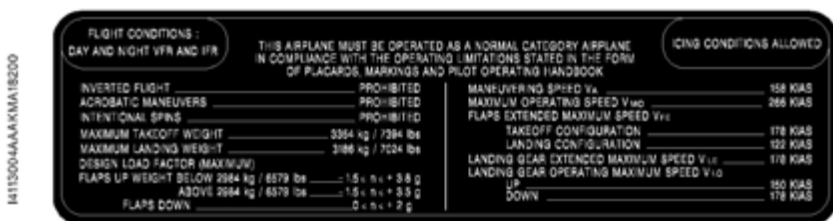
Engine instrument markings and their color code significance are shown in figure 2.8.3.

| Indication                       | Red<br>line or arc<br>-----<br>Minimum<br>limit | Yellow<br>line or arc<br>-----<br>Caution<br>range                     | Green<br>line or arc<br>-----<br>Normal<br>operating | Red line<br>-----<br>Maximum<br>limit   |
|----------------------------------|---|--|--|---|
| Oil<br>temperature               | - 40 °C<br>(- 40 °F)                            | - 40 to 0 °C<br>(- 40 to 32 °F)<br>104 to 110 °C<br>( 219.2 to 230 °F) | 0 to 104 °C<br>(32 to 219.2 °F)                      | 110 °C<br>(230 °F)  |
| Oil pressure                     | 60 psi  | 60 to 105 psi  | 105 to 135 psi                                       | 135 psi<br>(red line)<br>normal limit   |
| Generator RPM<br>(Ng)            | ---   | ---  | 51 to 104 %  | 104 %   |
| Propeller RPM<br>(Np)            | ---   | 450 to 1000 RPM  | 1950 to 2050<br>RPM                                  | 2050 RPM  |
| ITT<br>Engine start<br>or<br>off | ---   | 840 to 1090 °C<br>(1544 to 1994 °F)                                    | 400 to 840 °C<br>(752 to 1544 °F)                    | 840 °C (1544 °F)<br>normal limit<br>-----<br>870 °C (1598 °F)<br>(< 20 seconds limit)<br>-----<br>1090 °C (1994 °F)<br>(red line)<br>absolute limit |
| Engine running                   |   | ---  | 400 to 840 °C<br>(752 to 1544 °F)                    | 840 °C (1544 °F)<br>normal limit  |
| Torque (TRQ)                     | ---   | 100 %  | 0 to 100 %   | 101 %   |

Figure 2.8.3 - Engine instrument markings

## 2.9 - Placards

- (1) Under L.H. front side window



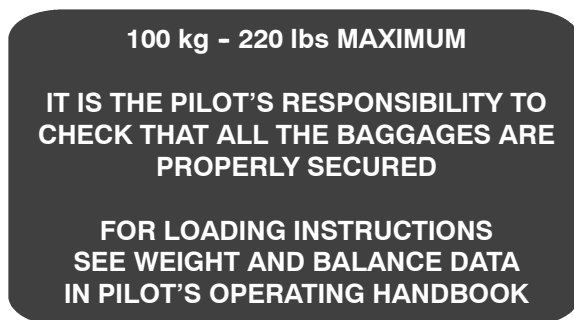
- (2) Calibration chart on compass and on windshield post



|       |   |     |     |   |     |     |
|-------|---|-----|-----|---|-----|-----|
| For   | N | 30  | 60  | E | 120 | 150 |
| Steer |   |     |     |   |     |     |
| For   | S | 210 | 240 | W | 300 | 330 |
| Steer |   |     |     |   |     |     |

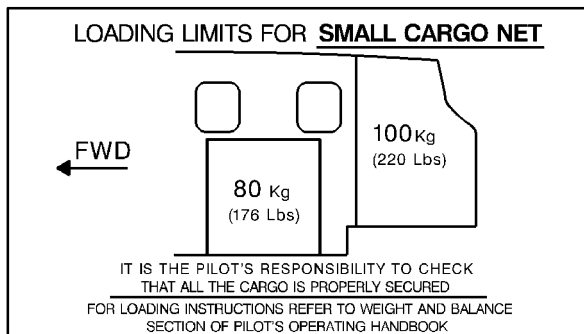
DATE : \_\_\_\_\_ RADIO ON

- (3) On pressurized baggage compartment partition wall



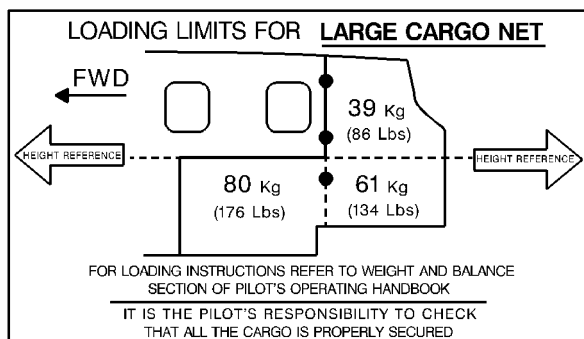
- (3)a For the small cargo net, on frame C13bis or on the right lower upholstery panel

I4113500AAAAA18000



- (3)b For the large cargo net, on R.H. side upholstery panel, in the rear baggage compartment

I4113500AAAAA18100



- (3)c On FWD baggage compartment door frame (non pressurized)

**50 kg - 110 lbs MAXIMUM**

**FOR LOADING INSTRUCTIONS  
SEE WEIGHT AND BALANCE DATA  
IN PILOT'S OPERATING HANDBOOK**

- (4) On pedestal console

14113600A/AA/MA/8001



- (5) On fuel selector

14113006A/AL/MA/8000



- (6) Near fuel tank caps

### JET-A-FUEL

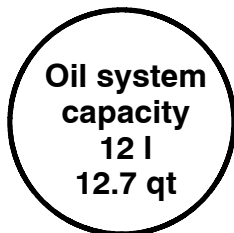
TOTAL CAPACITY 150.5 us gal - 570 l

ANTHICE ADDITIVE REQUIRED.SEE PILOT'S  
OPERATING HANDBOOK FOR OTHER APPROVED  
FUELS QUANTITY AND TYPE OF ADDITIVE

14112004A/AA/MA/8200



- (7) On internal face of L.H. engine cowling



- (8) On landing gear emergency control access door



- (9) Under window, at L.H. Intermediate seat



- (10) On rear passenger's table casing

**TABLE MUST BE STOWED DURING TAKEOFF AND LANDING**

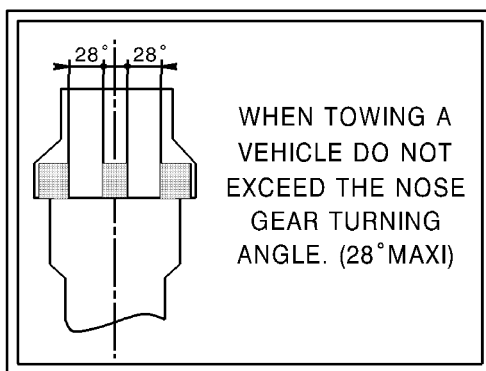
- (11) Under R.H. control wheel

I4351000/AAACMA8000



- (12) On nose gear door

I411200/AAACMA8000



- (13) On nose gear leg

**NOSE LANDING GEAR**  
**TIRE PRESSURE : 6,5 bar**  
**94 psi**

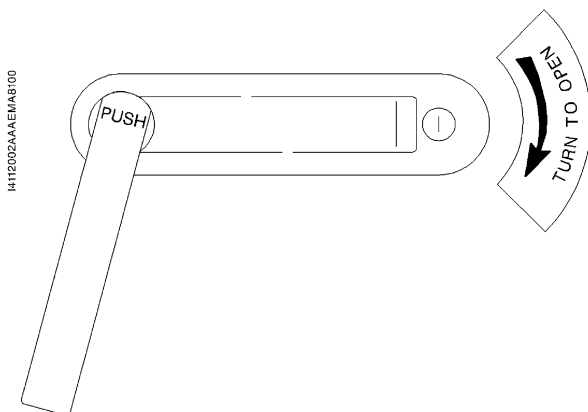
- (14) On main gear leg

**MAIN LANDING GEAR  
TIRE PRESSURE : 8,96 bar  
130 psi**

- (15) On engine cowling, in front of compartment door

**EXTERNAL POWER  
28 VOLTS D.C. NOMINAL  
800 AMP  
STARTING CAPACITY MIN  
DO NOT EXCEED 1000 AMP**

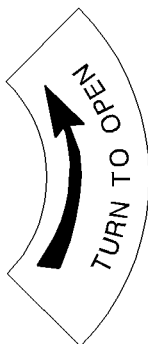
- (16) On pilot door - External side, if installed





- (17) On access door - External side

I4112002AAAEW/A8000



- (18) On outer fuselage skin aft of access door and in the cabin forward of access door



- (19) On access door - Internal side

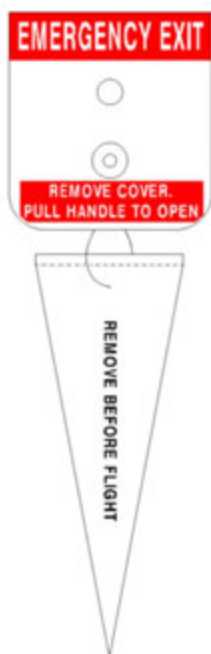


- (20) On pilot door - Internal side, if installed



- (21) On emergency exit handle

Marking on cover



Marking on handle



- (22) On last step of stairs

**STAIRS MAX LOAD : ONE PERSON**

- (23) On R.H. access door jamb

**DO NOT USE  
HAND RAIL  
TO RETRACT  
OR STOW  
STAIRS**

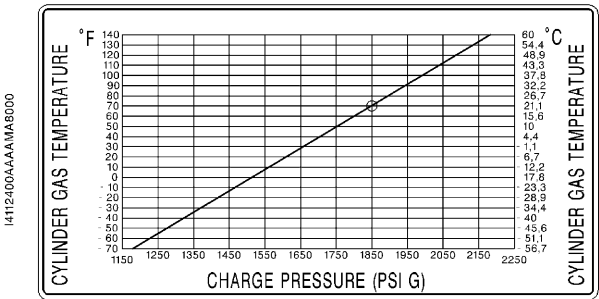
- (24) On R.H. side at front seat level and on the first rear passengers masks container (R.H. side on the ceiling)

**WARNING  
GREASY SUBSTANCES ARE CAPABLE  
OF SPONTANEOUS COMBUSTION  
ON CONTACT WITH OXYGEN  
DO NOT SMOKE WHILE OXYGEN IS IN USE**

- (25) On rear passengers masks containers

**OXYGEN MASKS INSIDE  
PULL MASKS FOR  
OXYGEN SUPPLY ↓**

- (26) On internal face of the oxygen cylinder service door



- (27) On the oxygen service door

I4112400AAAAA/MA8100



- (28) On emergency locator transmitter inspection door

I4112200AAAAA/MA8000



- (29) On the potty seat curtain, if installed, on pilot's side

**CURTAIN MUST BE STOWED FOR TAKE-OFF AND LANDING**

>> Airplane equipped with coat hanger (Post-MOD70-0557-25B)

(30) On the L.H. Rear cargo compartment panel upper edge

14113200AAALMA8000

CLOTHING ONLY

>> Airplane equipped with coat hanger (Post-MOD70-0557-25C)

(31) On the L.H. Rear cargo compartment panel upper edge

14113200AAALMA8300

MAX WEIGHT CAPACITY 4,5kg - 10 lbs

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## Section 3

### CAS messages

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CAS MESSAGES are in alphabetical order by either Chapter, Section, Paragraph or Supplement

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|      |   | Airspeed indicating system failure .....                                | 3.12.11 |
|      |   | Flight into severe icing conditions .....                               | 3.12.12 |
|      |   | <b>FRONT CARGO DOOR</b> .....   | 3.12.13 |
|      |   | <b>GPU DOOR</b> .....   | 3.12.14 |
|      |   | <b>IGNITION</b> .....   | 3.12.15 |
|      |   | <b>AP ON YD OFF</b> .....   | 3.12.16 |
|      |   | Autopilot or electric pitch trim malfunction .....                      | 3.12.17 |
|      |   | Dual GPS/SBAS failure ( <b>DR</b> or <b>LOI</b> annunciation on HSI) .. | 3.12.18 |
|      |   | GPS approach alarm limits exceeded .....                                | 3.12.20 |
|      |   | Left PFD failure .....  | 3.12.21 |
|      |   | AHRS failure .....  | 3.12.22 |
|      |   | ADC failure .....   | 3.12.24 |
|      |   | MFD failure .....   | 3.12.25 |

### 3.1 - General

The recommended procedures for different failures or emergency situations are provided in this section.

Emergency procedures associated with optional or particular equipment that require pilot's operating handbook supplements are provided in section 9 Supplements.

The pilot must know procedures given in this section and be prepared to take appropriate action should an emergency arise.

Some emergency procedures are a part of pilot basic training. Although these emergencies are discussed here, this information is not intended to replace such training, but only to provide a source of reference and review.

It is important for the pilot to be familiar with standard emergency procedures

#### **Alarm system recall**

Main failure or state modification of the different systems are provided by warning or caution messages appearing on CAS display.

The color code philosophy for CAS messages is the following :

- the **red** warning messages indicate a failure or a condition that requires an immediate action from the pilot,
- the **amber** caution messages indicate a failure or a condition that requires an action from the pilot as soon as practical.

Red or amber failure warnings are coupled with the lighting of

- a flashing red indicator/button



or

- a fixed amber indicator/button



Both indicators/buttons are located on the upper part of the L.H. instrument panel. When either one lights up, press it once to reactivate. It will go out and is ready to signal in the event of another failure. On the CAS display, the corresponding failure message remains ON as long as the failed condition exists.

The actions associated to the **red** warning or **amber** caution messages are described in this Section of the POH.

## Procedure format

### PROCEDURE TITLES

| Name of the procedure | 1/X |
|-----------------------|-----|
|-----------------------|-----|

*Procedure introduction or description of symptoms associated with the failure are presented like this at the beginning of the procedure.*

1/X is written if the procedure extends over 2 pages or more.

### MEMORY ITEMS

The memory items are indicated with a grey border box as shown hereafter :

The memory items are written like this.

Memory items are critical steps that must be executed quickly from memory without referring to POH or checklist.

### CONDITIONAL STEPS

*Conditions are presented like this :*

- 1 - With related actions to perform indented inside.

### VALIDITY / EFFECTIVITY

>> *Pre/Post-MOD70-xxxx-xx*

Before procedure title, represents a specific validity / effectivity for the entire procedure below. If nothing is specified, the procedure applies to all airplanes.

>> *Validity inside a procedure is presented like this*

- 1 - With actions related to this validity listed under.

### CONTINUATION AND ENDING

The end of the entire procedure is indicated by :

*End of procedure.*

Procedure completion within the body of the procedure as a result of a condition is indicated by :

*End of procedure ■*

Continuation of a procedure on several pages is indicated by :

► *Continuing*

*Continue* ►

## LANDING DIRECTIVES

- Land as soon as possible ◀ means land on the nearest suitable runway.
- Land as soon as practical ◀ means land on the nearest suitable runway with convenient facilities.

## CAS MESSAGES

Indicated as displayed in the MFD CAS window :

- **FUEL PRESS** means FUEL PRESS warning CAS message,
- **MAIN GEN** means MAIN GEN caution CAS message.

## ANNUNCIATIONS ON PFDs or MFD

Indicated as displayed in the PFD or MFD with specifying “annunciation” next to the message :

- **BOTH ON AHRS1** annunciation,
- **HDG NO COMP** annunciation.



### 3.2 - Rejected takeoff

#### Engine failure at takeoff before rotation

- |     |                |             |
|-----|----------------|-------------|
| 1 - | THROTTLE ..... | Flight IDLE |
| 2 - | Brakes .....   | As required |

*If the airplane cannot be stopped on the runway :*

- |     |                          |           |
|-----|--------------------------|-----------|
| 3 - | THROTTLE .....           | CUT OFF   |
| 4 - | FUEL TANK SELECTOR ..... | OFF       |
| 5 - | Crash lever .....        | Pull down |

*If necessary :*

- |     |   |
|-----|---|
| 6 - | Evacuate after the airplane has come to a stop. |
|-----|---|

*End of procedure.*

Rejected takeoff for any other reason

- |     |                |             |
|-----|----------------|-------------|
| 1 - | THROTTLE ..... | Flight IDLE |
| 2 - | Reverse .....  | As required |
| 3 - | Brakes .....   | As required |

*If the airplane cannot be stopped on the runway :*

- |     |                          |           |
|-----|--------------------------|-----------|
| 4 - | THROTTLE .....           | CUT OFF   |
| 5 - | FUEL TANK SELECTOR ..... | OFF       |
| 6 - | Crash lever .....        | Pull down |

*If necessary :*

- 7 - Evacuate after the airplane has come to a stop.

*End of procedure.*

### 3.3 - Engine failures

#### Engine failure before rotation

Perform procedure ..... Engine failure at takeoff before rotation  
Refer to chapter 3.2

*End of procedure.*

#### Engine failure after rotation

1/2

► Fly the airplane ◀

1 - MAN OVRD control ..... Full forward

*If power recovery successful :*

► Fly the airplane using the MAN OVRD control for power ◀

2 - THROTTLE ..... Flight IDLE

► Land as soon as possible ◀

*End of procedure* ■

*If power recovery unsuccessful :*

3 - MAN OVRD control ..... Full backward

*If height does not allow to choose a suitable landing surface :*

► Land straight ahead without changing LANDING GEAR position ◀

4 - FLAPS lever ..... TO

5 - Airspeed ..... Maintain above 100 KIAS

6 - THROTTLE ..... CUT OFF

*Continue* ►

## Engine failure after rotation

2/2

### ► Continuing

7 - FUEL TANK SELECTOR ..... OFF

*Before touch down :*

8 - FLAPS lever ..... LDG

9 - Crash lever ..... Pull down

10 - Evacuate airplane after coming to complete stop.

Do not unfasten seat belts before complete stop.

*End of procedure* ■

*If height allows to reach a suitable landing surface :*

11 - LANDING GEAR lever ..... DN

12 - FLAPS lever ..... As required

#### Maintain airspeeds

Flaps UP

105 < KIAS < 178

Flaps TO

100 < KIAS < 178

Flaps LDG

85 < KIAS < 122

13 - THROTTLE ..... CUT OFF

14 - FUEL TANK SELECTOR ..... OFF

*Before touch down :*

15 - Crash lever ..... Pull down

16 - Evacuate airplane after coming to complete stop.

Do not unfasten seat belts before complete stop.

*End of procedure.*

## Engine failure in flight

■ *Symptoms : Loss of power*

► Fly the airplane ◀

- |     |                          |              |
|-----|--------------------------|--------------|
| 1 - | FUEL TANK SELECTOR ..... | Switch tanks |
| 2 - | AUX BP switch .....      | ON           |
| 3 - | Autopilot .....          | Disconnect   |

■ *If power recovery successful :*

- 4 - Remaining fuel ..... Check

► Land as soon as possible ◀

*End of procedure* ■

■ *If power recovery unsuccessful :*

- |     |                    |         |
|-----|--------------------|---------|
| 5 - | THROTTLE .....     | CUT OFF |
| 6 - | Oxygen masks ..... | Use     |

- 7 - Air start envelope ..... Check  
Refer to chapter 3.4

*End of procedure.*

**OIL PRESS** or **OIL PRESS**

■ Indicates that oil pressure is below 105 psi.

► Fly the airplane ◀

► Land as soon as possible ◀

- 1 - Oil pressure ..... Monitor
- 2 - TRQ ..... Minimum necessary

▲ **CAUTION** ▲

Due to the oil pressure drop, the propeller blade angle may go towards high pitch and therefore lead to a Np propeller rotation speed decrease.



If engine power decreases :

- 3 - THROTTLE ..... CUT OFF
- 4 - Perform procedure ..... Emergency descent  
Refer to chapter 3.6
- 5 - Perform procedure ..... Forced landing  
Refer to chapter 3.7  
*End of procedure.*

Engine regulation discrepancy, power loss,  
throttle control loss

1/3

*Symptoms :*

- power fluctuations, or
- uncommanded power loss, or
- bad response to *THROTTLE* movements.

► Fly the airplane ◀

*If circumstances and obtained minimum power allow :*

▲ **CAUTION** ▲

In manual override mode, engine is neither protected against slam accelerations, nor against maximum speed overshooting. Avoid rapid control movements and manage engine parameters.



- |     |  |   |
|-----|--|---|
| 1 - | THROTTLE .....   | Flight IDLE                                   |
| 2 - | Confirm engine still running.                          |   |
| 3 - | FUEL TANK SELECTOR .....                               | Switch tanks                                  |
| 4 - | Check that no engine indication exceeds allowed value. |   |
| 5 - | MAN OVRD control .....                                 | Actuate<br>Progressively to minimum necessary |
| 6 - | Continue the flight.                                   |   |

► Land as soon as possible ◀

*If the available power is weak :*

- |     |                          |                                     |
|-----|--------------------------|-------------------------------------|
| 7 - | LANDING GEAR lever ..... | DN                                  |
|     |                          | Only on a final approach glide path |
| 8 - | FLAPS lever .....        | LDG                                 |
|     |                          | Only in short final                 |

Engine regulation discrepancy, power loss,  
throttle control loss

2/3

► Continuing

▲ CAUTION ▲

When MAN OVRD control is used, the available power may not be sufficient to ensure a go-around in landing configuration, particularly if the weight is near the maximum weight.



► Do not perform a go around ◀

► Do not use the reverse ◀

9 - Land normally.

10 - Brakes ..... As required

End of procedure ■

If minimum power obtained is excessive :

11 - Airspeed ..... Reduce below 178 KIAS  
By setting nose-up attitude

12 - INERT SEP switch ..... ON

If ITT > 840°C :

13 - INERT SEP switch ..... OFF

14 - LANDING GEAR lever ..... DN

15 - FLAPS lever ..... TO

16 - Long final or ILS approach ..... Establish  
At IAS < 178 KIAS

Continue ►



**Engine regulation discrepancy, power loss,  
throttle control loss****3/3**► *Continuing**When runway is assured :*

- 17 - THROTTLE ..... CUT OFF
- 18 - FLAPS lever ..... LDG as required  
At IAS < 122 KIAS
- 19 - Land normally.
- 20 - Brakes ..... As required

*End of procedure.*

### Governor control not operating

► Fly the airplane ◀

1 - Continue the flight.

*If  $N_p < 1960$  RPM :*

► Do not perform a go around ◀

► Do not use the reverse ◀

In that case, the go-around performance and the reverse efficiency might be lower than expected.

Repair before further flight.

*End of procedure.*

### Excessive propeller rotation speed

► Fly the airplane ◀

1 - Reduce power and airplane speed to avoid propeller rotation speeds higher than 2050 RPM.

► Land as soon as possible ◀

► Do not perform a go around ◀

In that case, the go-around may damage the gear reduction box and the reverse efficiency might be lower than expected.

Repair before further flight.

*End of procedure.*

**Engine does not stop on ground**

*If the engine does not stop when the THROTTLE is set to CUT OFF :*

1 - FUEL TANK SELECTOR ..... OFF

2 - Wait for engine stop due to lack of fuel in the pipes.

3 - GENERATOR selector ..... OFF

4 - SOURCE selector ..... OFF

5 - Crash lever ..... Pull down

Inform maintenance department.

*End of procedure.*

**ITT**

*During engine start :*

- |  |       |
|--|-------|
| 1 - Starting procedure . . . . .             | Abort |
| Refer to procedure Engine start, chapter 4.4 |       |

- 2 - Cancel the flight.  
Inform maintenance department.

*End of procedure ■*

*After engine start :*

*On ground :*

- 3 - Cancel the flight.  
Inform maintenance department.

*End of procedure ■*

*In flight :*

► Fly the airplane ◀

- |                              |        |
|------------------------------|--------|
| 4 - TRQ . . . . .            | Reduce |
| To get ITT < 840°C           |        |
| ► Land as soon as possible ◀ |        |

Inform maintenance department.

*End of procedure.*

**CHIP**

■ Indicates that metallic chips have been detected in the engine oil.

*In flight :*

► Fly the airplane ◀

► Land as soon as practical ◀

Inform maintenance department.

*End of procedure ■*

*On ground :*

► Do not take off ◀

Airplane is grounded.

Inform maintenance department.

*End of procedure.*

**NG HI**

*Indicates that Ng speed is more than 103 %.*

1 - TRQ ..... Reduce  
To get Ng below 103 %

*End of procedure.*

**OIL TEMP**

Indicates that oil temperature is below 0°C or above 104°C (possibly with **OIL PRESS**)

Indicates that oil temperature is below 0°C or above 104°C

1 - Oil temperature indicator ..... Check

*If the indicated temperature is in the green sector :*

► Land as soon as possible ◀

► Fly the airplane ◀

2 - Oil temperature ..... Monitor

*End of procedure ■*

*If the indicated temperature is not in the green sector :*

Failure is confirmed, you can expect an oil pressure failure shortly.

▲ **CAUTION** ▲

Due to the oil pressure drop, the propeller blade angle may go towards high pitch and therefore lead to a Np propeller rotation speed decrease.



▲ **CAUTION** ▲

Prepare for an engine stop shortly.



3 - TRQ ..... Minimum necessary

► Land as soon as possible ◀

*If engine power decreases :*

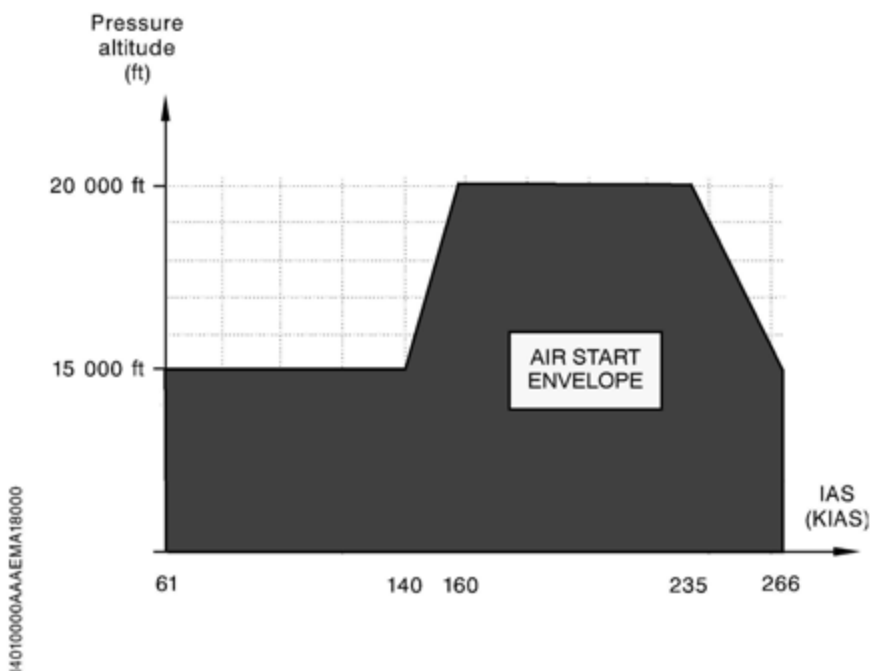
4 - THROTTLE ..... CUT OFF

5 - Perform procedure ..... Forced landing  
Refer to chapter 3.7

*End of procedure.*

### 3.4 - Air start

#### Air start envelope



• NOTE •

Air start may be attempted outside of the envelope. However, above 20000 ft or at lower speeds, ITT tends to increase during start and prudence is recommended.

•

*End of procedure.*

## Air start procedures

1/2

- 1 - Oxygen masks ..... Use

### ▲ CAUTION ▲

The starter cannot operate if the GENERATOR selector is on ST-BY.



- 2 - GENERATOR selector ..... MAIN

### ▲ CAUTION ▲

BLEED switch set to AUTO may cause overtemperature or abnormal acceleration.



- 3 - BLEED switch ..... OFF

- 4 - A/C switch ..... OFF

- 5 - Electric consumption ..... Reduce

- 6 - FUEL TANK SELECTOR ..... L or R

- 7 - AUX BP switch ..... ON

- 8 - IGNITION switch ..... AUTO or ON

- 9 - THROTTLE ..... CUT OFF

- 10 - STARTER switch ..... ON, start timer

*If there is no start after 5 seconds :*

- 11 - STARTER switch ..... ABORT

*When Ng around 13 % :*

- 12 - THROTTLE ..... LO-IDLE

- 13 - ITT and Ng ..... Monitor

*Continue ►*



Air start procedures

2/2

► *Continuing*

*When Ng > 50 % :*

14 - Starter ..... Check OFF automatically

*If starter has not turned off automatically :*

15 - STARTER switch ..... ABORT

16 - THROTTLE ..... Flight IDLE

17 - THROTTLE ..... As required

18 - BLEED switch ..... As required

19 - AUX BP switch ..... AUTO

20 - Electrical equipment ..... As required

*If necessary :*

21 - Perform procedure ..... Emergency descent  
Refer to chapter 3.6

*If air start is not successful :*

22 - Perform procedure ..... Forced landing  
Refer to chapter 3.7

*End of procedure.*

Intentionally left blank

### 3.5 - Fire and smoke

#### Engine fire on ground

Symptoms : ITT increasing, **ITT** , smoke, ...

- |     |                    |         |
|-----|--------------------|---------|
| 1 - | THROTTLE .....     | CUT OFF |
| 2 - | BLEED switch ..... | OFF     |
| 3 - | A/C switch .....   | OFF     |
- 4 - Brakes ..... As required
- 5 - FUEL TANK SELECTOR ..... OFF

If necessary :

- 6 - Warn ground assistance.
- 7 - Crash lever ..... Pull down
- Evacuate as soon as possible ◀

*End of procedure.*

Cabin fire on ground

1 - THROTTLE ..... CUT OFF

2 - Brakes ..... As required

*If necessary :*

3 - Warn ground assistance.

4 - Crash lever ..... Pull down

5 - Cabin extinguisher ..... As required

► Evacuate as soon as possible ◀

*End of procedure.*

## Engine fire in flight

Symptoms : ITT increasing, **ITT** , smoke, ...

### ▲ WARNING ▲

**No air start attempt after an engine fire.**



► Fly the airplane ◀

- |     |                          |         |
|-----|--------------------------|---------|
| 1 - | Oxygen masks .....       | Use     |
| 2 - | THROTTLE .....           | CUT OFF |
| 3 - | AUX BP switch .....      | OFF     |
| 4 - | FUEL TANK SELECTOR ..... | OFF     |

- |     |                    |     |
|-----|--------------------|-----|
| 5 - | BLEED switch ..... | OFF |
| 6 - | A/C switch .....   | OFF |

If necessary :

- |     |                         |   |
|-----|-------------------------|---|
| 7 - | Perform procedure ..... | Emergency descent<br>Refer to chapter 3.6 |
| 8 - | Perform procedure ..... | Forced landing<br>Refer to chapter 3.7    |

*End of procedure.*

Cabin electrical fire or smoke during flight

► Fly the airplane ◀

- 1 - Oxygen masks and goggles ..... Use

*If the origin is known :*

- 2 - Defective equipment breaker ..... Pull

- 3 - Cabin extinguisher ..... Use

*If the origin is unknown :*

- 4 - A/C switch ..... OFF

- 5 - All unnecessary equipment ..... OFF

- 6 - Perform procedure ..... Emergency descent  
Refer to chapter 3.6

*If necessary :*

- 7 - Perform procedure ..... Smoke elimination  
Refer to procedure hereafter

► Land as soon as possible ◀

*End of procedure.*

## Smoke elimination

- 1 - Oxygen masks and goggles ..... Use
- 2 - BLEED switch ..... OFF
- 3 - A/C switch ..... OFF
- 4 - DUMP switch ..... Actuate
- 5 - Wait until the differential pressure drops.
- 6 - EMERGENCY RAM AIR control knob ..... Pull

*If smoke decreases :*

► Land as soon as possible ◀

*End of procedure ■*

*If smoke increases :*

- 7 - EMERGENCY RAM AIR control knob ..... Push

► Land as soon as possible ◀

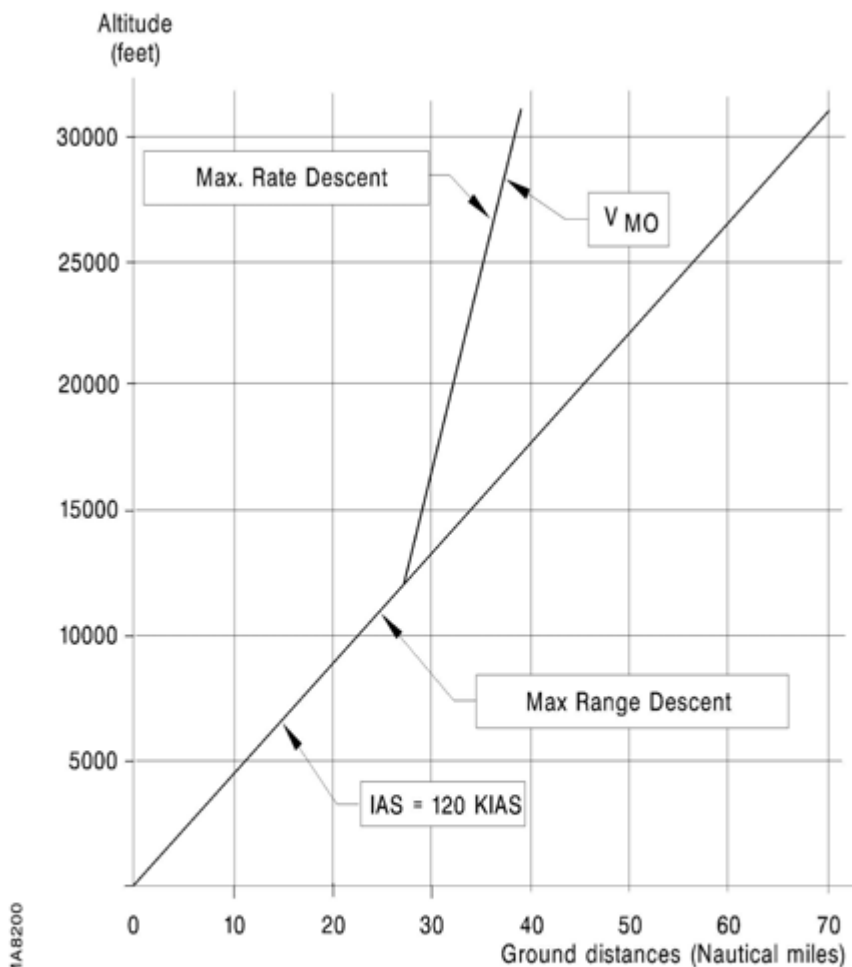
*End of procedure.*

Intentionally left blank



### 3.6 - Emergency descents

#### Emergency descents profiles



14010000AAAAA18200

Maximum rate descent

- |     |                      |                |
|-----|----------------------|----------------|
| 1 - | THROTTLE .....       | Flight IDLE    |
| 2 - | Oxygen masks .....   | Use            |
| 3 - | Pitch attitude ..... | – 10° to – 20° |

*If smooth air :*

- |     |                                     |                     |
|-----|-------------------------------------|---------------------|
| 4 - | FLAPS and LANDING GEAR levers ..... | UP                  |
| 5 - | Airspeed .....                      | $V_{MO} = 266$ KIAS |

*If rough air or in case of structure problem :*

- |     |                          |                |
|-----|--------------------------|----------------|
| 6 - | Airspeed .....           | Below 178 KIAS |
| 7 - | FLAPS lever .....        | UP             |
| 8 - | LANDING GEAR lever ..... | DN             |

*End of procedure.*

## Maximum range descent

1/2

- |     |                    |     |
|-----|--------------------|-----|
| 1 - | Oxygen masks ..... | Use |
|-----|--------------------|-----|
- 2 - THROTTLE ..... CUT OFF
  - 3 - FLAPS and LANDING GEAR levers ..... UP
  - 4 - Airspeed ..... 120 KIAS
  - 5 - DUMP switch ..... Actuate
  - 6 - EMERGENCY RAM AIR control knob ..... Pull

*If VMC and non icing conditions are possible :*

- 7 - ESS BUS TIE switch ..... EMER
- 8 - Prepare for ..... Forced landing  
Refer to chapter 3.7

*End of procedure ■*

*If VMC and non icing conditions are not possible :*

Breakers :

- 9 - PFD 2 ..... Pull
- 10 - ADC 2 ..... Pull
- 11 - XPDR 2 ..... Pull

Switches :

- 12 - DE ICE SYSTEM ..... All OFF
- 13 - Lights ..... All OFF
- 14 - BLEED ..... OFF
- 15 - A/C ..... OFF
- 16 - AUX BP ..... OFF
- 17 - FUEL SEL ..... MAN
- 18 - AP / TRIMS ..... OFF
- 19 - DIMMER / CABIN / ACCESS ..... OFF

*Continue ►*

Maximum range descent

2/2

► *Continuing*

*If icing conditions :*

- 20 - PITOT L HTR switch ..... ON
- 21 - WINDSHIELD switch ..... ON
- 22 - Airspeed ..... Above 135 KIAS  
Configuration flaps UP

*If time permits :*

- 23 - PLUGS breakers ..... Pull
- 24 - AIR COND breaker ..... Pull
- 25 - Prepare for ..... Forced landing  
Refer to chapter 3.7

*End of procedure.*

### 3.7 - Emergency landings, flaps, gear

#### Forced landing

- |     |                          |  |
|-----|--------------------------|--|
| 1 - | THROTTLE .....           | CUT OFF  |
| 2 - | FUEL TANK SELECTOR ..... | OFF  |
| 3 - | AUX BP switch .....      | OFF  |
| 4 - | BLEED switch .....       | OFF  |
| 5 - | A/C switch .....         | OFF  |
| 6 - | DUMP switch .....        | Actuate  |
| 7 - | Gliding airspeed .....   | Maintain 120 KIAS<br>Until favorable ground approach |
| 8 - | ESS BUS TIE switch ..... | NORM<br>To have GEAR and FLAPS available             |

■ *If landing surface is suitable :*

- 9 - LANDING GEAR lever ..... DN

■ *If landing surface is not suitable :*

- 10 - LANDING GEAR lever ..... Keep UP

■ *If night conditions :*

- 11 - OFF/TAXI/LDG switch ..... LDG

*When chosen landing surface is assured :*

- 12 - FLAPS lever ..... LDG
- 13 - Crash lever ..... Pull down
- 14 - Airspeed on final approach ..... 85 KIAS
- 15 - Land flaring out.
- 16 - Evacuate after stop.

*End of procedure.*

Tire blowout during landing

- 1 - Control direction with brakes and nose wheel steering.
- 2 - Reverse ..... As required
- 3 - Stop airplane to minimize damages.
- 4 - Perform procedure ..... Shutdown

Refer to chapter 4.3

*End of procedure.*

**FLAPS ASYM**

*Indicates a dissymmetry of flap deflection. This immediately stops the flap motor and prevents further operation of the flaps.*

► Fly the airplane ◀

- |     |                     |      |
|-----|---------------------|------|
| 1 - | FLAPS breaker ..... | Pull |
| 2 - | FLAPS lever .....   | UP   |

► Land as soon as possible ◀

- 3 - Maintain airspeeds :
  - IAS < 178 KIAS for deflections between UP and TO positions
  - IAS < 122 KIAS for deflections greater than TO position
- 4 - For landing, refer to procedure ..... Landing with flaps malfunction  
 Refer to procedure on following page  
*End of procedure.*

### Flaps malfunction

*In case of blockage of flaps or inoperative flaps control lever between UP and LDG positions, without **FLAPS ASYM** :*

- |     |                     |      |
|-----|---------------------|------|
| 1 - | FLAPS breaker ..... | Pull |
| 2 - | FLAPS lever .....   | UP   |

► Land as soon as possible ◀

- 3 - Maintain airspeeds
  - IAS < 178 KIAS for deflections between UP and TO positions
  - IAS < 122 KIAS for deflections greater than TO position
- 4 - For landing, refer to procedure ..... Landing with flaps malfunction  
Refer to procedure hereafter  
*End of procedure.*

### Landing with flaps malfunction

*For flaps deflections between UP and TO :*

Proceed as for a normal landing with 105 KIAS of approach airspeed.  
Provide for a landing distance increased by 60 %.

*For flaps deflections greater than TO :*

Proceed as for a normal landing with 100 KIAS of approach airspeed.  
Provide for a landing distance increased by 50 %.

*End of procedure.*



## Landing gear retraction discrepancy

### • NOTE •

Symptoms have to be considered at the end of the sequence.

•

*Symptoms :*

- **GEAR UNSAFE** CAS msg and GEAR UNSAFE red warning light are ON, or
- amber light flashing and 3 green lights are OFF.

1 - Airspeed ..... Maintain below 150 KIAS

2 - LDG GEAR breaker ..... Pull

If **GEAR UNSAFE** CAS msg and GEAR UNSAFE red warning light are OFF :

- 3 - The flight may be continued without any restriction.

*For landing gear extension :*

- 4 - Perform procedure ..... Emergency gear extension  
Refer to following procedures

*End of procedure ■*

*If not :*

- 5 - LDG GEAR breaker ..... Push
- 6 - Perform procedure ..... Emergency gear extension  
Refer to following procedures

*End of procedure.*

### Landing gear extension discrepancy

• NOTE •

Symptoms have to be considered at the end of the sequence.

•

Symptoms :

- **GEAR UNSAFE** CAS msg and GEAR UNSAFE red warning light are ON, or
- amber light flashing and 0 to 3 green lights are OFF.

1 - Airspeed ..... Maintain below 150 KIAS

- 2 - Perform procedure ..... Emergency gear extension  
Refer to procedure on following page

*End of procedure.*

## Emergency gear extension

1/3

### • NOTE •

Follow this procedure in case of any doubt about the gear extension.

### ▲ CAUTION ▲

Do not enter icing conditions. This could adversely increase drag and weight due to ice accumulation, and lock wheels and struts.

Climb performance will be degraded by 50 %.

Cruise IAS speed will be reduced compared to a clean airplane, because of the drag.

This should be taken into account when calculating the airplane range.



- |     |                          |                         |
|-----|--------------------------|-------------------------|
| 1 - | Airspeed .....           | Maintain below 150 KIAS |
| 2 - | LANDING GEAR lever ..... | DN                      |
| 3 - | LDG GEAR breaker .....   | Pull                    |
| 4 - | Floor hatch .....        | Open                    |
| 5 - | By-pass selector .....   | Fully pull / Locked     |

### ▲ CAUTION ▲

The entire extension of the landing gear may take up to 110 cycles. It is mandatory to have a clear hardening of the manual control at the end of the maneuver.



- |     |  |  |
|-----|--|--|
| 6 - | Landing gear emergency pump handle ..... | Actuate<br>With maximum amplitude until pump hardening |
| 7 - | MASTER WARNING push-button .....         | Press  |

To reset the **GEAR UNSAFE**

*Continue ►*

Emergency gear extension

2/3

► Continuing

If :

- GEAR UNSAFE red warning light is OFF and
- **GEAR UNSAFE** is OFF and
- 3 green lights are ON :
  - 8 - Exit and / or remain outside icing conditions.

Continue flight at airspeed < 178 KIAS.

► Land as soon as practical ◀

End of procedure ■

If :

- GEAR UNSAFE red warning light is ON and
- **GEAR UNSAFE** is ON and
- 0 to 3 green lights are ON :
  - 9 - LDG GEAR breaker ..... Push
  - 10 - CHECK DOWN push-button ..... Press

If :

- hardening of the pump is marked and
- 3 green lights are ON or
- 3 green lights are ON and flickering while pressing the CHECK DOWN push-button :
  - 11 - Land

End of procedure ■

Continue ►

## Emergency gear extension

3/3

### ► Continuing

If :

- emergency pump remains soft or
- one (or more) green light(s) is(are) not ON and / or flickering while pressing the CHECK DOWN push-button :

A gear unlock condition is confirmed.

Recycle the landing gear as follows :

- 12 - By-pass selector ..... Unlock / Push
- 13 - Wait one minute.
- 14 - LANDING GEAR lever ..... UP  
At airspeed < 150 KIAS
- 15 - Perform landing gear extension attempts in the normal mode while applying positive load factors during the maneuver as well as skidding.

In case of failure :

- 16 - Perform procedure ..... Landing with unlocked  
main landing gear  
or Landing with defective nose landing gear  
Refer to following procedures

### ▲ CAUTION ▲

If one main landing gear is not down, it is recommended to land with landing gear up - refer to procedure Landing with gear up in the following procedures.



*End of procedure.*

Landing with unlocked main landing gear

1/2

▲ **CAUTION** ▲

If one main landing gear is not down, it is recommended to land with landing gear up - refer to procedure Landing with gear up in the following procedures.



- 1 - Ask ATC or another airplane to visually check landing gear position.

*If defective gear is down but unlocked :*

- 2 - BLEED switch ..... OFF
- 3 - DUMP switch ..... Actuate
- 4 - FUEL TANK SELECTOR ..... Maintain on defective LDG gear side  
To lighten corresponding wing (maximum fuel imbalance 15 USG)
- 5 - Choose a runway with headwind or crosswind blowing from defective gear side.
- 6 - Align the airplane to land on the runway edge opposite to the defective landing gear.
- 7 - Perform a normal approach.
- 8 - FLAPS lever ..... LDG  
At airspeed = 90 KIAS
- 9 - Land and set nose gear immediately on ground to assure lateral control.
- 10 - Use full aileron during roll-out to lift the wing with the defective landing gear.

*Continue ►*

**Landing with unlocked main landing gear****2/2**► *Continuing**If landing gear drags during landing :*

- 11 - THROTTLE ..... CUT OFF
- 12 - Crash lever ..... Pull down
- 13 - FUEL TANK SELECTOR ..... OFF
- 14 - Evacuate after airplane comes to a stop.

*End of procedure ■**If landing gear does not drag during landing :*

- 15 - Preferably do not use reverse.
- 16 - Complete taxiing with a slight turn towards defective landing gear.
- 17 - THROTTLE ..... CUT OFF
- 18 - Perform procedure ..... Shutdown  
Refer to chapter 4.3
- 19 - Evacuate.

*End of procedure.*

**Landing with defective nose landing gear  
(down unlocked or not down)**

- 1 - Ask ATC or another airplane to visually check landing gear position.

*If necessary :*

- 1 - Transfer passengers to the rear.
- 2 - Perform a normal approach.
- 3 - FLAPS lever ..... LDG
- 4 - Airspeed ..... Maintain 90 KIAS
- 5 - Land with nose-up attitude. Keep nose high.
- 6 - THROTTLE ..... CUT OFF
- 7 - Touch down slowly with nose wheel and keep elevator at nose-up stop.
- 8 - Brakes ..... Apply moderately
- 9 - Crash lever ..... Pull down
- 10 - FUEL TANK SELECTOR ..... OFF
- 11 - Evacuate after airplane comes to a stop.

*End of procedure.*



**Landing with gear up**

- 1 - Do a standard final approach.
- 2 - FLAPS lever ..... LDG
- 3 - Airspeed ..... Maintain 85 KIAS
- 4 - BLEED switch ..... OFF
- 5 - DUMP switch ..... Actuate

*When runway is assured :*

- 6 - THROTTLE ..... CUT OFF
- 7 - FUEL TANK SELECTOR ..... OFF
- 8 - Flare out.

*After touch-down :*

- 9 - Crash lever ..... Pull down
- 10 - Evacuate after airplane comes to a stop.

*End of procedure.*

Ditching

▲ CAUTION ▲

In heavy swell with light wind, land parallel to the swell (rollers).  
In heavy wind, land facing wind.



- 1 - LANDING GEAR lever ..... UP
- 2 - FLAPS lever ..... LDG
- 3 - Airspeed ..... Maintain above 85 KIAS
- 4 - Maintain a descent rate as low as possible when approaching the water.
- 5 - BLEED switch ..... OFF
- 6 - DUMP switch ..... Actuate
- 7 - Crash lever ..... Pull down
- 8 - Maintain attitude without flaring out until touch-down.
- 9 - Evacuate through EMERGENCY EXIT.

*End of procedure.*

**Landing without elevator control**

- 1 - LANDING GEAR lever ..... DN
- 2 - FLAPS lever ..... LDG
- 3 - Airspeed ..... Maintain 95 KIAS
- 4 - TRQ ..... As required  
To maintain airspeed according to an easy approach slope  $\approx$  300 ft/min
- 5 - Adjust elevator by using manual pitch trim wheel.

*When ground approaches :*

- 6 - Slope ..... Decrease progressively
- 7 - TRQ ..... Reduce progressively

*End of procedure.*

Intentionally left blank

### 3.8 - Fuel system

#### **FUEL PRESS**

1/2

*Indicates a fuel pressure drop at HP engine pump inlet.*

► Fly the airplane ◀

- |     |                          |              |
|-----|--------------------------|--------------|
| 1 - | Remaining fuel .....     | Check        |
| 2 - | FUEL TANK SELECTOR ..... | Switch tanks |
| 3 - | AUX BP switch .....      | AUTO         |

If **FUEL PRESS** remains ON :

- |     |                     |    |
|-----|---------------------|----|
| 4 - | AUX BP switch ..... | ON |
|-----|---------------------|----|

- 5 - **AUX BOOST PMP ON** ..... Check ON

If pressure is normal again and **FUEL PRESS** is OFF :

Mechanical pump has failed.

- 6 - AUX BP switch ..... Maintain ON

► Land as soon as practical ◀

*End of procedure ■*

If **FUEL PRESS** remains ON :

- 7 - FUEL TANK SELECTOR ..... Switch tanks

If **FUEL PRESS** is OFF :

A supply problem may have occurred from the tank selected first (air vent, fuel icing, etc...).

*End of procedure ■*

*Continue ►*

**FUEL PRESS**

2/2

► *Continuing*

If **FUEL PRESS** remains ON :

- 8 -    Fullest tank ..... Select
- 9 -    Avoid high power and rapid movements of the THROTTLE.
- 10 -   Altitude ..... Below 18000 ft

► Land as soon as possible ◀

► Fly the airplane ◀

*End of procedure.*

**AUX BOOST PMP ON**

*Indicates the auxiliary booster pump is running.*

► Fly the airplane ◀

*If AUX BP switch is in ON position :*

Indication is normal.

*End of procedure ■*

*If AUX BP switch is in AUTO position :*

- 1 - Reset AUX BP switch to ..... ON
- 2 - Then, AUX BP switch ..... AUTO

*If **AUX BOOST PMP ON** goes OFF :*

- 3 - Continue the flight normally.

*End of procedure ■*

*If **AUX BOOST PMP ON** remains ON :*

Mechanical booster pump has failed.

- 4 - AUX BP switch ..... ON

► Land as soon as possible ◀

*End of procedure.*

**FUEL LOW L-R**

*Indicates a level drop in the corresponding tank.*

- 1 - Corresponding gage ..... Check
- 2 - Check the other tank has been automatically selected.

*If other tank not automatically selected :*

- 3 - FUEL SEL switch ..... MAN
- 4 - Select tank manually ..... As required

► Fly the airplane ◀

- 5 - Minimum fuel ..... Check
- 6 - Take decision.

*If necessary :*

► Land as soon as practical ◀

*End of procedure.*



**AUTO SEL**

*Indicates that there is no more automatic control mode running.*

► Fly the airplane ◀

1 - FUEL SEL switch ..... Check AUTO

*If FUEL SEL switch already on AUTO :*

Failure is confirmed.

2 - FUEL SEL switch ..... MAN

3 - Select tanks manually ..... As required

▲ CAUTION ▲

Maximum fuel imbalance is 15 USG



*End of procedure.*

**FUEL IMBALANCE**

*Indicates that fuel tanks are imbalanced by more than 15 USG for more than 30 seconds.*

*If FUEL SEL switch is on AUTO :*

- 1 -   Fullest tank ..... Select  
By pressing the SHIFT push-button

*If FUEL SEL switch is on MAN :*

- 2 -   Fullest tank ..... Select  
By shifting FUEL TANK SELECTOR manually

► Fly the airplane ◀

▲ **CAUTION** ▲

Maximum fuel imbalance is 15 USG



*End of procedure.*

**LOW LVL FAIL L-R**

*Indicates a failure of fuel low level sensor.*

- 1 - Remaining fuel in tanks ..... Check
- 2 - Take decision.

*If any doubt :*

► Land as soon as practical ◀

► Fly the airplane ◀

*On the ground :*

Inform maintenance department.

*End of procedure.*

Intentionally left blank

### 3.9 - Electrical system

#### BAT AMP

*Indicates that battery current is over 50 A while on ground.*

After starting the engine with airplane power, a battery charge over 50 amperes is normal.

#### ▲ CAUTION ▲

Do not take off if battery charge is over 50 A.



*If this indication remains steady at a high value :*

It may be due to a battery or generation system failure.

*End of procedure.*

#### BAT OFF

*Indicates that :*

- the SOURCE selector has been positioned on OFF or
- the battery plug is disconnected

► Fly the airplane ◀

- |     |                       |      |
|-----|-----------------------|------|
| 1 - | SOURCE selector ..... | OFF  |
| 2 - | SOURCE selector ..... | BATT |

*If warning persists :*

► Land as soon as possible ◀

- |     |                              |         |
|-----|------------------------------|---------|
| 3 - | Airplane mains voltage ..... | Monitor |
|-----|------------------------------|---------|

*End of procedure.*

**MAIN GEN**

*Indicates that GENERATOR selector has been positioned to OFF or ST-BY, or main generator is cut off.*

- 1 - GENERATOR selector ..... Check / Correct  
If necessary

*If warning persists :*

Main generator switching is confirmed.

- 2 - MAIN GENERATOR RESET push-button ..... Press

*In case of failure :*

► Fly the airplane ◀

- 3 - Keep the following systems connected :

- Autopilot system
- Deicing systems
- STROBE and NAV lights
- Cockpit emergency lights
- VHF 1
- NAV/GPS 1
- BLEED
- LDG LIGHTS on short final

This will allow to keep electrical consumption  
below maximum standby capacity.

All other not necessary equipment can be disconnected.

- 4 - GENERATOR selector ..... ST- BY

*If necessary :*

- 5 - ST-BY GENERATOR RESET push-button ..... Press

- 6 - Maintain ST-BY loads below 100 A.

*End of procedure.*

**LOW VOLTAGE**

*Normal functioning with GENERATOR selector on MAIN.*

1 - Voltmeter voltages ..... Check

*If voltages are < 26 V :*

2 - Monitor a possible voltage drop or any indication of battery discharge.

► Fly the airplane ◀

3 - Keep the following systems connected :

- Autopilot system
- Deicing systems
- STROBE and NAV lights
- Cockpit emergency lights
- VHF 1
- NAV/GPS 1
- BLEED
- LDG LIGHTS on short final

This will allow to keep electrical consumption  
below maximum standby capacity.

All other not necessary equipment can be disconnected.

4 - GENERATOR selector ..... ST-BY

*If necessary :*

5 - ST-BY GENERATOR RESET push-button ..... Press

6 - Maintain ST-BY loads below 100 A.

*End of procedure.*

**MAIN GEN** and **LOW VOLTAGE**

1/3

*With GENERATOR selector on ST-BY (after MAIN generator failure), functioning on ST-BY generator.*

- 1 - GENERATOR selector ..... MAIN
- 2 - MAIN GENERATOR RESET push-button ..... Press

► Fly the airplane ◀

*If MAIN GENERATOR successfully connected :*

- 3 - Disconnect non-essential systems.
- 4 - Voltmeter and ammeter ..... Monitor

► Land as soon as possible ◀

*End of procedure* ■

*If MAIN GENERATOR not successfully connected :*

- 5 - GENERATOR selector ..... ST-BY
- 6 - ST-BY GENERATOR RESET push-button ..... Press

*If ST-BY GENERATOR successfully connected :*

- 7 - Disconnect non-essential systems.
- 8 - Voltmeter and ammeter ..... Monitor

► Land as soon as possible ◀

*End of procedure* ■

*If ST-BY GENERATOR not successfully connected :*

Both generators failure is confirmed.

Return to VMC conditions, if possible.

*Continue* ►



**MAIN GEN** and **LOW VOLTAGE**

2/3

► *Continuing*

9 - GENERATOR selector ..... OFF

*If altitude > 10000 ft :*

10 - OXYGEN switch ..... ON

*If VMC and non-icing conditions are possible :*

11 - ESS BUS TIE switch ..... EMER  
In this configuration, only both ESS BUS bars and  
BATT BUS bar are directly supplied by the battery.

► Land as soon as possible ◀

*If use of other systems is required :*

12 - ESS BUS TIE switch ..... NORM

*End of procedure* ■

*If VMC and non-icing conditions are not possible :*

13 - Manually disconnect systems as follows :

Breakers :

- PFD 2 ..... Pull
- ADC 2 ..... Pull
- TAS ..... Pull
- DATA LINK ..... Pull
- XPDR 2 ..... Pull

Switches :

- AIRFRAME DE ICE ..... OFF
- ICE LIGHT ..... OFF
- PROP DE ICE ..... OFF
- WINDSHIELD ..... OFF

*Continue* ►

**MAIN GEN** and **LOW VOLTAGE**

3/3

► Continuing

- PITOT R & STALL HTR ..... OFF
- OFF/LDG/TAXI light ..... OFF
- PULSE ..... OFF
- STROBE ..... OFF
- BLEED ..... OFF
- A/C ..... OFF
- AUX BP ..... OFF
- FUEL SEL ..... MAN
- AP / TRIMS ..... OFF
- DIMMER / CABIN / ACCESS ..... OFF

*If icing conditions :*

- 14 - PITOT L HTR switch ..... Check ON
- 15 - WINDSHIELD switch ..... ON
- 16 - Maintain minimum recommended airspeeds into known icing conditions.

|           |            |
|-----------|------------|
| Flaps UP  | > 135 KIAS |
| Flaps TO  | > 115 KIAS |
| Flaps LDG | > 95 KIAS  |

*If time permits :*

- 17 - PLUGS breakers ..... Pull
  - 18 - AIR COND breaker ..... Pull
- Land as soon as possible ◀

*End of procedure.*

**ELEC FEATH FAULT**

*Indicates a propeller feathering system malfunction.*

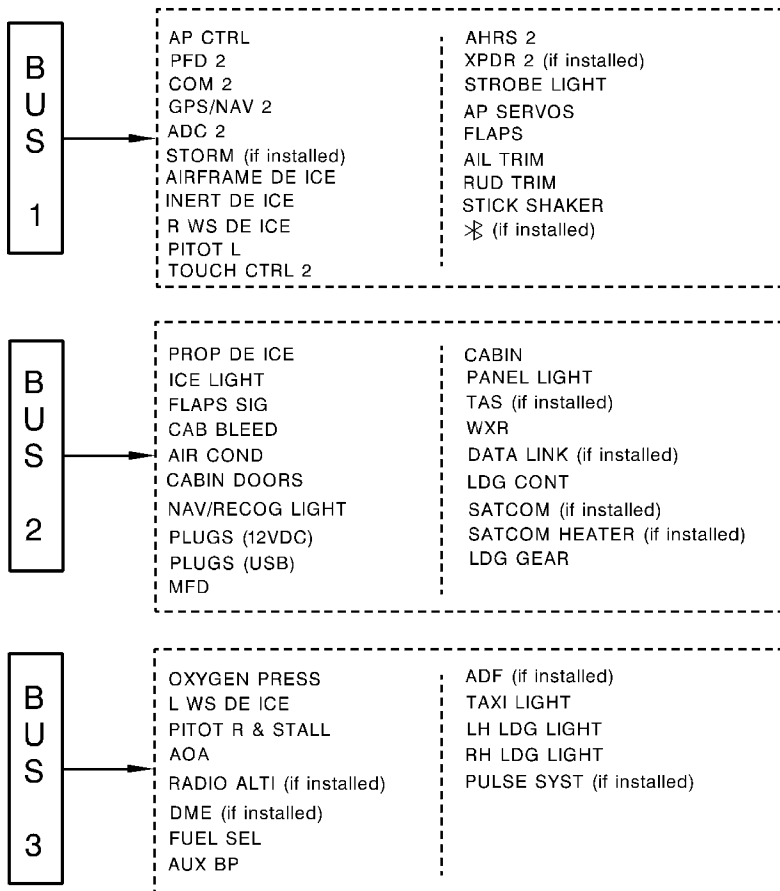
► Fly the airplane ◀

1 - FEATHER breaker ..... Pull

► Land as soon as possible ◀

*End of procedure.*

Bus bar 1/3



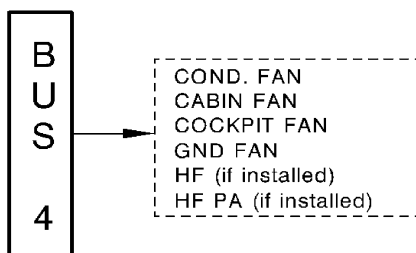
I4246000AAANMA8400

Figure 3.9.1 (1/3) - Electrical distribution of bus bars

Bus bar

2/3

I4246000AANMA8101



NOTE: CIRCUIT BREAKERS ON C13 BIS FRAME

I4246000AAGMA8300

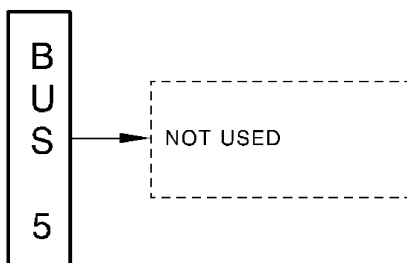
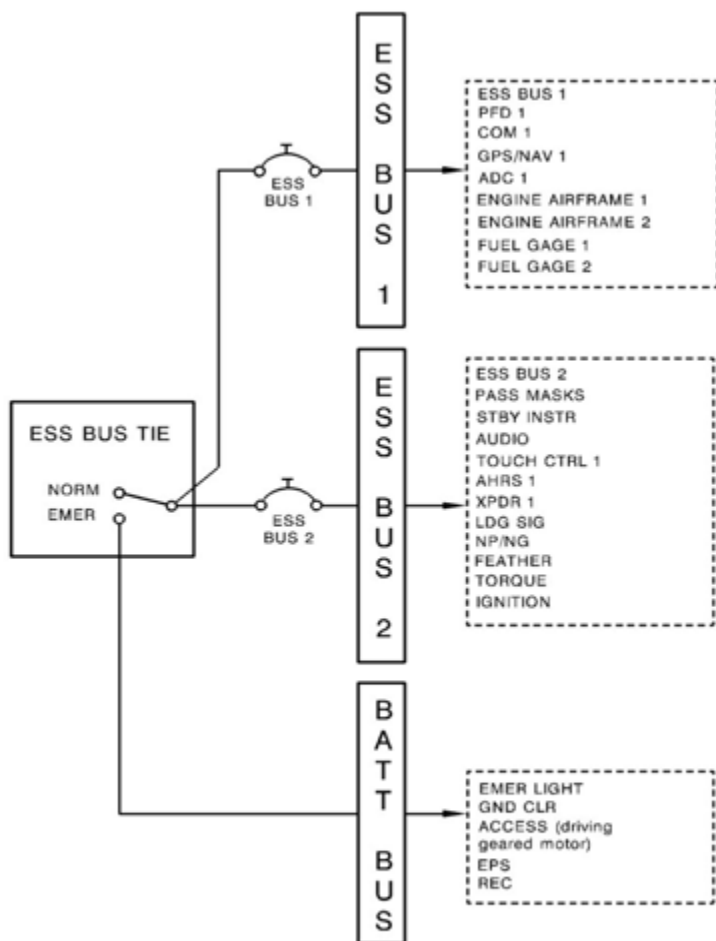


Figure 3.9.1 (2/3) - Electrical distribution of bus bars

Bus bar

3/3



14246000AAGMA18200

Figure 3.9.1 (3/3) - Electrical distribution of bus bars

**Total loss of electrical power**

- 1 - Maintain airplane control.
- 2 - Use the MD302 for
  - attitude,
  - airspeed and/or
  - altitude

► Fly the airplane ◀

► Land as soon as possible ◀

• NOTE •

Airplane power is provided to the MD302 display for normal operation. Operation of the basic system is automatic. The system is powered ON while airplane power is ON.

If airplane power is lost, the internal battery will provide power to the MD302.

*End of procedure.*

Intentionally left blank



### 3.10 - Pressurization and air conditioning

#### **BLEED TEMP**

Indicates overheat of bleed air system. Normally this leads to **BLEED** cut-off and to **BLEED OFF** appearance.

► Fly the airplane ◀

Should automatic cut-off occur or not :

If possible :

- 1 - TRQ ..... Reduce
- 2 - HOT AIR FLOW distributor ..... Turn to the right
- 3 - A/C switch ..... PILOT
- 4 - TEMP selector ..... Mini
- 5 - BLEED switch ..... OFF
- 6 - BLEED switch ..... AUTO

If **BLEED TEMP** and **BLEED OFF** are still ON :

- 7 - Perform procedure ..... **BLEED OFF**

Refer to procedure hereafter

End of procedure ■

If **BLEED TEMP** is ON and **BLEED OFF** is OFF :

- 8 - Shorten the flight.
- Inform maintenance department.

End of procedure.

**BLEED OFF**

1/2

*Possibly due to :*

- system malfunction
- BLEED switch on OFF position

*If in flight :*

- |                      |                    |                 |
|----------------------|--------------------|-----------------|
| 1 -                  | Oxygen masks ..... | Use             |
| 2 -                  | BLEED switch ..... | Check / Correct |
| <i>If possible :</i> |                    |                 |
| 3 -                  | TRQ .....          | Reduce          |
| ▶ Fly the airplane ◀ |                    |                 |
| 4 -                  | BLEED switch ..... | OFF             |
| 5 -                  | BLEED switch ..... | AUTO            |

*If **BLEED OFF** is still ON :*

*If altitude > 10000 ft :*

*If necessary :*

- |     |                         |   |
|-----|-------------------------|---|
| 6 - | Perform procedure ..... | Emergency descent<br>Refer to chapter 3.6 |
| 7 - | Continue the flight.    |   |

• NOTE •

If  $Z_p > 10000 \text{ ft} \pm 500 \text{ ft}$ , it may be followed by **CABIN ALTITUDE** and

**USE OXYGEN MASK**

•

Inform maintenance department.

*End of procedure ■*

*Continue ▶*

**BLEED OFF**

2/2

► *Continuing**If on ground :*

- 8 - BLEED switch ..... OFF
  - 9 - Taxi back to apron.
  - 10 - Perform procedure ..... Shutdown  
Refer to chapter 4.3
- Inform maintenance department.

*End of procedure.*

**CPCS BACKUP MODE**

*Indicates a GASC system malfunction. The GASC cannot compute optimal cabin altitude and is automatically set to 9800 ft default value as cabin altitude reference.*

► Fly the airplane ◀

1 - Continue the flight.

Inform maintenance department before next flight.

▲ **CAUTION** ▲

When the airplane descends below 9800 ft, cabin descent rate coincides with airplane descent rate. The pilot should take into account the airplane descent profile in order to avoid pressure annoyance.



*End of procedure.*

**CABIN ALTITUDE** and **USE OXYGEN MASK**

• NOTE •

**CABIN ALTITUDE** is followed by **USE OXYGEN MASK** and 3 voice alerts  
"Use oxygen mask / Use oxygen mask".

*Indicates a cabin altitude over 10000 ft  $\pm$  500 ft.*

- |  |                                |       |
|--|--------------------------------|-------|
| 1 -  | Pressurization indicator ..... | Check |
| <i>If cabin altitude &gt; 10000 ft <math>\pm</math> 500 ft :</i> |                                |       |
| 2 -  | Oxygen masks .....             | Use   |

► Fly the airplane ◀

- 3 - BLEED switch ..... Check AUTO
- 4 - DUMP switch ..... Check NORM / Guarded
- 5 - EMERGENCY RAM AIR control knob ..... Check pushed

*If necessary :*

- 6 - Perform procedure ..... Emergency descent  
Refer to chapter 3.6

- 7 - Limit flight altitude to maintain cabin altitude below 10000 ft.

■ Inform maintenance department before next flight.

*End of procedure.*

>> Without EDM evolution patch (Pre-MOD70-0657-34)

**CABIN ALTITUDE**

and

**USE OXYGEN MASK**

and

**EDM**

• NOTE •

**CABIN ALTITUDE** is followed by **USE OXYGEN MASK** and 3 voice alerts  
"Use oxygen mask / Use oxygen mask".

EDM makes a 90° left heading change and descent to 15000 ft.  
EDM override is possible by pressing twice the AP / TRIM DISC push-button,  
and other AP modes are usable.  
Power reduction to speed up the descent is recommended.

*Indicates a cabin altitude over 10000 ft ± 500 ft.*

1 - Pressurization indicator ..... Check

*If cabin altitude > 10000 ft ± 500 ft :*

2 - Oxygen masks ..... Use

► Fly the airplane ◀

3 - BLEED switch ..... Check AUTO

4 - DUMP switch ..... Check NORM / Guarded

5 - EMERGENCY RAM AIR control knob ..... Check pushed

*If necessary :*

6 - Perform procedure ..... Emergency descent  
Refer to chapter 3.6

7 - Limit flight altitude to maintain cabin altitude below 10000 ft.

■ Inform maintenance department before next flight.

*End of procedure.*

>> With EDM evolution patch (Post-MOD70-0657-34)

**EDM**

• NOTE •

**EDM** may come on 45 s after **CABIN ALTITUDE** and

**USE OXYGEN MASK**.

EDM makes a 90° left heading change and descent to 15000 ft.  
EDM override is possible by pressing twice the AP / TRIM DISC push-button,  
and other AP modes are usable.

Power reduction to speed up the descent is recommended.

•

*Indicates a cabin altitude over 10000 ft ± 500 ft.*

1 - Pressurization indicator ..... Check

*If cabin altitude > 10000 ft ± 500 ft :*

2 - Oxygen masks ..... Use

► Fly the airplane ◀

3 - BLEED switch ..... Check AUTO

4 - DUMP switch ..... Check NORM / Guarded

5 - EMERGENCY RAM AIR control knob ..... Check pushed

*If necessary :*

6 - Perform procedure ..... Emergency descent  
Refer to chapter 3.6

7 - Limit flight altitude to maintain cabin altitude below 10000 ft.

*End of procedure.*

**EDM OVERRIDE**

*Indicates that Emergency Descent Mode has been overridden by the crew, and is not available again until **EDM OVERRIDE** is OFF.*

► Fly the airplane ◀

*End of procedure.*



**CABIN DIFF PRESS**

*Indicates a cabin pressure differential over 6.4 PSI  $\pm$  0.2 PSI.*

- 1 - Pressurization indicator ..... Check

*If  $\Delta P > 6.4 \text{ PSI} \pm 0.2 \text{ PSI}$  :*

- 2 - BLEED switch ..... OFF

- 3 - Oxygen masks ..... Use

► Fly the airplane ◀

*If necessary :*

- 4 - Perform procedure ..... Emergency descent  
Refer to chapter 3.6

*End of procedure.*

Cabin not depressurized after landing

*If  $\Delta P$  cabin remains  $> 0$  :*

- 1 - DUMP switch ..... Actuate
- 2 - BLEED switch ..... OFF

*If necessary :*

- 3 - EMERGENCY RAM AIR control knob ..... Pull
- 4 - Wait for complete cabin depressurization before opening any door.

*End of procedure.*

**OXYGEN**

*Indicates that the oxygen cylinder isolation valve is closed.*

**▲ WARNING ▲**

**Flight is prohibited with oxygen cylinder closed.**



1 - Oxygen cylinder ..... Open

*End of procedure.*

**DOOR**

*Indicates that one of the door latches of the door(s) is not correctly locked.*

*On ground :*

- 1 - Check the correct locking, as well as the latches position of the door(s).

*If **DOOR** is still ON :*

- 2 - Do not take off.

*End of procedure ■*

*In flight :*

► Fly the airplane ◀

- 3 - Start a slow descent.

- 4 - Decrease cabin pressure differential . . . . . By selecting a higher LFE  
(LFE between 9500 ft and 10000 ft)

*If a real failure of one of the doors is noticed :*

- 5 - Oxygen masks . . . . . Use
- 6 - BLEED switch . . . . . OFF
- 7 - DUMP switch . . . . . Actuate

*If necessary :*

- 8 - Perform procedure . . . . . Emergency descent  
Refer to chapter 3.6

*End of procedure.*

**VACUUM LOW**

Low vacuum may lead to malfunctioning of leading edge deicing and pressurization.

- 1 - Monitor the normal functioning of leading edge deicing and pressurization.

*If necessary :*

- 2 - Altitude ..... Below 10000 ft

- 3 - Return to VMC conditions as soon as possible.

► Fly the airplane ◀

- 4 - BLEED switch ..... OFF

*End of procedure.*

### Defog malfunction

*If moisture starts to quickly cover the inside of the windscreen with the HOT AIR FLOW distributor already turned to the left :*

- 1 - HOT AIR FLOW distributor ..... Set to around  
a 10 o'clock position

*If moisture continues :*

- 2 - HOT AIR FLOW distributor ..... Turn to the left
- 3 - WINDSHIELD switch ..... ON

*If there is no improvement and if the flight safety is engaged :*

- 4 - Altitude ..... Around 10000 ft
- 5 - BLEED switch ..... OFF

#### ▲ CAUTION ▲

In flight, the cabin will quickly depressurize. Therefore, the cabin vertical speed indicator and altimeter indications will rapidly meet those of respectively the airplane VSI and altimeter.



*End of procedure.*

### 3.11 - Deicing system

#### Leading edges deicing failure

*Symptoms : failure on one of the two pneumatic deicing pulses :*

- *ice on wing outboard sections,*
- *or, ice on wing inboard sections and stabilizers,*
- *one of the two cycling green lights is not lit.*

► Leave icing conditions as soon as possible ◀

1 - AIRFRAME DE ICE switch ..... OFF

*End of procedure.*

**PROP DEICE FAIL**

*Symptoms :*

- *propeller deicing green light is not lit,*
- *propeller vibrations.*

- |     |                                    |         |
|-----|------------------------------------|---------|
| 1 - | TRQ .....                          | Reduce  |
|     | ▶ Fly the airplane ◀               |         |
| 2 - | THROTTLE .....                     | Actuate |
|     | To vary RPM within operating range |         |

- ▶ Leave icing conditions as soon as possible ◀

*End of procedure.*



**INERT SEP FAIL**

Symptoms :

- **INERT SEP ON** does not appear within 50 seconds following INERT SEP switch setting ON,
- inertial separator is not retracted after 50 seconds following INERT SEP switch setting OFF,
- INERT DE ICE breaker triggered.

► Leave icing conditions as soon as possible ◀

► Fly the airplane ◀

End of procedure.

### Windshield deicing failure

*Symptoms :*

- *windshield being covered uniformly by ice,*
- *no perception of heat when touching deiced section,*
- *windshield deicing green light is not lit.*

*If symptoms result from overheat :*

- 1 - WINDSHIELD switch ..... OFF / ON  
When necessary

*In case of total failure :*

- 2 - TEMP selector ..... Max warm
- 3 - HOT AIR FLOW distributor ..... Turn to the left

*Before landing :*

- 4 - Wait for a sufficient visibility.

*End of procedure.*

### Windshield misting or internal icing

*Symptoms : mist or ice on windshield internal face.*

- 1 - TEMP selector ..... Set to 12 o'clock position
- 2 - HOT AIR FLOW distributor ..... Turn to the left
- 3 - WINDSHIELD switch ..... ON

*If unsuccessful, to get sufficient visibility :*

- 4 - HOT AIR FLOW distributor ..... Fully turn to the left
- 5 - Manually clean a sufficient visibility area.

*If necessary :*

#### ▲ CAUTION ▲

In case of sideslip approach with pedal on the right during a long period, select R.H. fuel tank.



- 6 - Clean L.H. side window.
- 7 - Perform a sideslip approach with rudder pedals to the right.  
To get sufficient landing visual references

*For landing :*

- 8 - FLAPS lever ..... LDG
- 9 - Airspeed ..... Maintain above 95 KIAS

*End of procedure.*

**PITOT NO HT L-R**

Indicates that :

- corresponding pitot tube heating has failed or
- PITOT L HTR switch or PITOT R & STALL HTR switch is not ON while the engine is running.

If **PITOT NO HT L** :

Icing conditions may alter airspeed indications provided by ADC1.

1 - Avoid icing conditions.

► Fly the airplane ◀

If not possible :

- 2 - Perform moderate descent or climb attitudes.

$V_{MO}$  overshoot and stall warning system are always operating.

End of procedure ■

If **PITOT NO HT R** :

$V_{MO}$  overshoot warning may be altered by icing conditions.

► Fly the airplane ◀

- 3 - Airspeed ..... Monitor below 266 KIAS

End of procedure.

## STALL NO HEAT

*Indicates that :*

- stall warning vane heating has failed or
- PITOT R & STALL HTR switch is not ON while the engine is running.

Correct operation of the aural stall warning may be altered by severe or prolonged icing.

- |     |                |   |
|-----|----------------|---|
| 1 - | Airspeed ..... | Monitor   |
|     |                | Maintain minimum airspeed according<br>to airplane configuration and icing conditions |

► Fly the airplane ◀

*End of procedure.*

Intentionally left blank

### 3.12 - Miscellaneous

#### Trim Runaway

► Fly the airplane ◀

- 1 - AP / TRIM DISC push-button ..... Press and hold

The three trim tabs are disconnected and runaway stops.

- 2 - AP / TRIMS switch ..... OFF

- 3 - AP / TRIM DISC push-button ..... Release

- 4 - Pitch trim may be used manually.

*If necessary :*

- 5 - Airspeed ..... Reduce  
To reduce control forces

*In case of pitch trim runaway :*

- 6 - AP / TRIMS switch ..... AP OFF

The pitch trim may be used manually, the two other trim tabs may be used again electrically.

*End of procedure ■*

*In case of rudder or aileron trim runaway :*

- 7 - RUD TRIM or AIL TRIM breaker ..... Pull  
According to the defective trim

- 8 - AP / TRIMS switch ..... ON

The two other trim tabs may be used again electrically.

*End of procedure.*

Crack in cockpit window or window panel

► Fly the airplane ◀

- 1 - Descend slowly.
- 2 - Cabin  $\Delta P$  ..... Reduce  
By setting Landing Field Elevation to 10000 ft

*End of procedure.*



**Emergency exit use**

- 1 - Check that the anti-theft safety pin has been removed.
- 2 - Lift up the opening handle.
- 3 - Pull emergency exit assembly towards oneself to release it from its recess.
- 4 - Put the emergency exit door inside fuselage or throw it away from the fuselage through the opening.
- 5 - Evacuate airplane.

*End of procedure.*

Emergency beacon (ELT) use

*Before a forced landing :*

*If possible :*

- 1 - Transmit a MAY DAY signal on COM VHF 121.5 MHz or on a known ATC frequency.

*After landing :*

- 2 - ELT remote control switch ..... ON  
Maintain ON until aid arrives

*End of procedure.*

## Inadvertent spins

### ▲ WARNING ▲

**Voluntary spins are prohibited.**



- 1 - AP / TRIM DISC push-button ..... Press and Hold until recovery
- 2 - Control wheel ..... Neutral  
Pitch and roll axis
- 3 - Rudder ..... Fully opposed to the spin
- 4 - THROTTLE ..... Flight IDLE
- 5 - FLAPS lever ..... UP

*When rotation is stopped :*

- 6 - Level the wings and ease out of the dive.

► Fly the airplane ◀

*End of procedure.*

**AP off, stall warning**

• NOTE •

Shaker will vibrate simultaneously with stall warning aural alert.

•

1 - Fly the airplane, wings levelled and nose down until stall warning stops.

2 - TRQ ..... As required

3 - Return to the desired flight path.

*End of procedure.*

**USP ACTIVE**

1 - Do not disconnect AP

2 - Increase power up to 50 % minimum

3 - Manage the flight

• NOTE •

Stall warning may be triggered but AP will remain ON

•

*End of procedure.*

**AURAL WRN FAIL**

*Indicates that no aural warning alerts are available.*

▲ **CAUTION** ▲

No aural stall warning.

No aural overspeed warning.

No landing gear warning.



1 - Maintain airspeeds

|           |                  |
|-----------|------------------|
| Flaps UP  | 105 < KIAS < 266 |
| Flaps TO  | 100 < KIAS < 178 |
| Flaps LDG | 85 < KIAS < 122  |

*End of procedure.*

**AURAL WRN 1 CHNL**

• NOTE •

System switches automatically on the remaining valid aural channel.

•

*End of procedure.*

## Oxygen use

1/2

With or without **USE OXYGEN MASK**.

### ▲ WARNING ▲

**Smoking is strictly prohibited any time oxygen system is used. Before using oxygen, remove any trace of oil, grease, soap and other fatty substances (including lipstick, make-up, etc...)**



*For front seats :*

- 1 - Take a mask on the opposite seat side (pilot : R.H. side ; R.H. Front passenger : L.H. side). Draw it out of the stowage cup and uncoil tube totally.
- 2 - Press on the red side vanes to inflate the harness.
- 3 - Put the mask onto the face.

*If no smokes :*

- 4 - 3-position selector ..... NORMAL  
100 % as required

*In case of smokes :*

- 5 - 3-position selector ..... EMERGENCY
- 6 - Don the smoke goggles onto the face.
- 7 - PASSENGER OXYGEN switch ..... DEPLOY
- 8 - Check oxygen flow indicator for the front seats (the blinker is transparent) and for the rear passengers (the blinker is green).
- 9 - MICRO/MASK switch ..... MASK
- 10 - Perform an emergency descent ..... To the minimum  
enroute altitude

*If possible :*

- 11 - Perform an emergency descent ..... To an altitude  
below 10000 ft

*Continue ►*

Oxygen use

2/2

► *Continuing*

*For rear passengers :*

- 1 - Take a mask.
- 2 - Uncoil tube totally.
- 3 - Pull on the lanyard cord to take out the lanyard pin.
- 4 - Put the mask onto the face.

*End of procedure.*



**Airspeed indicating system failure**

*Symptoms : erroneous indication in flight.*

- 1 - PITOT L HTR switch ..... Check ON
- 2 - PITOT R & STALL HTR switch ..... Check ON

*If symptoms persist :*

- 3 - ALTERNATE STATIC SOURCE selector ..... Pull thoroughly
- 4 - Use standby instrument only.

*If symptoms persist, as well as on the electronic standby instrument on the L.H instrument panel :*

- 5 - Perform a precautionary approach maintaining an adequate  
airspeed.

*End of procedure.*

### Flight into severe icing conditions

*Severe icing conditions, particularly freezing rain and freezing drizzle, can be identified by :*

- *unusually extensive ice accumulation on the airframe and windshield in areas not normally observed to collect ice,*
- *accumulation of ice on the upper surface of the wing aft of the protected area.*

Procedures for exiting freezing rain or freezing drizzle conditions :

- 1 - Inform ATC to exit severe icing conditions by changing the route or the altitude.
- 2 - Avoid any sudden maneuver on flight controls.

► Do not engage the autopilot ◀

*If the autopilot is engaged :*

- 3 - Hold the control wheel firmly and disengage the autopilot.

*If an unusual roll response or uncommanded roll control movement is observed :*

- 4 - Angle of Attack ..... Reduce

► Do not extend flaps when holding in icing conditions ◀

Operation with extended flaps can result in a reduced wing Angle of Attack, with the possibility of ice forming on the upper surface further aft on the wing than normal, possibly aft of the protected area.

*If the flaps are extended :*

- 5 - Do not retract them until the airframe is clear of ice.

*End of procedure.*

**FRONT CARGO DOOR**

*Indicates that front cargo door is open.*

*On ground :*

- 1 - Check and close the door.

*In flight :*

► Fly the airplane ◀

- 2 - Airspeed ..... Reduce  
To minimum available

► Land as soon as practical ◀

*End of procedure.*

**GPU DOOR**

*Indicates that GPU door is open.*

*On ground :*

- 1 - Check and close the door.

*In flight :*

► Fly the airplane ◀

- 2 - Airspeed ..... Reduce  
To minimum available

► Land as soon as practical ◀

*End of procedure.*

**IGNITION**

*Indicates that ignition exciter is running.*

1 - IGNITION switch ..... Check position

*If weather permits :*

2 - IGNITION switch ..... AUTO

► Fly the airplane ◀

• NOTE •

IGNITION switch may be left ON for a long period.

•

*End of procedure.*

**AP ON YD OFF**

*Indicates that the autopilot is ON while Yaw Damper is OFF, so no automatic rudder trim is available.*

1 - Yaw Damper status ..... Check

*If necessary :*

2 - Yaw Damper status ..... Correct

*End of procedure.*

**Autopilot or electric pitch trim malfunction****▲ CAUTION ▲**

When disconnecting the autopilot after a pitch trim malfunction, hold the control wheel firmly. Up to 30 pounds of force on the control wheel may be necessary to hold the airplane level.



- 1 - AP / TRIM DISC push-button ..... Press and hold
- 2 - AP / TRIMS switch ..... OFF
- 3 - AP / TRIM DISC push-button ..... Release

*If necessary :*

- 4 - Control wheel ..... Retrim

*End of procedure.*

Dual GPS/SBAS failure  
( **DR** or **LOI** annunciation on HSI) 1/2

Loss of GPS/SBAS navigation data

When both GPS/SBAS receivers are inoperative or GPS navigation information is not available or invalid, the GARMIN system will enter one of two modes : Dead Reckoning mode (DR) or Loss Of Integrity mode (LOI). The mode is indicated on the HSI by an amber DR or LOI.

Which mode is active depends on the distance from the destination airport in the active flight plan.

If the LOI annunciation is displayed, revert to an alternate means of navigation appropriate to the route and phase of flight.

In Dead Reckoning mode, the MAP – NAVIGATION MAP will continue to be displayed with a ghosted airplane icon in the center and an amber 'DR' overwriting the icon. Airplane position will be based upon the last valid GPS position, then estimated by Dead Reckoning methods. Changes in true airspeed, altitude, or winds aloft can affect the estimated position substantially. Dead Reckoning is only available in Enroute mode; Terminal and Approach modes do not support DR. Course deviation information will be displayed as an amber CDI on both PFDs and will remain for up to 20 minutes after GPS position data has been lost. The autopilot and/or flight director may be coupled in GPS mode while the system is in Dead Reckoning mode.

■ Refer to the GARMIN Pilot's Guide for further information.

Revert to an alternate means of navigation appropriate to the route and phase of flight.

If alternate navigation sources (ILS, LOC, VOR, DME, ADF) are available :

- 1 - Navigation ..... Use alternate sources

Continue ►



Dual GPS/SBAS failure  
( **DR** or **LOI** annunciation on HSI)

2/2

► *Continuing*

If no alternate navigation sources are available :

Dead Reckoning (DR) Mode - Active when the airplane is greater than 30 NM from the destination airport :

- 1 - Navigation ..... Use the airplane symbol, magenta course line on the map display and the amber CDI for course information

• NOTE •

All information normally derived from GPS turns amber. All of this information will become less accurate over time.

TAWS is inoperative.

DR mode uses heading, true airspeed, last known wind data, and the last known GPS position to estimate the airplane's current position. DR information will be available for a maximum of 20 minutes.

MAP – TRAFFIC MAP display is not dependent on GPS information.

The position of displayed traffic relative to the airplane symbol on the map is still accurate.

•

Loss Of Integrity (LOI) Mode - Active when the airplane is within 30 NM of departure airport (as calculated from the previous GPS or DR position)

- 1 - Navigation ..... Fly towards known visual conditions, use ATC or other information sources as possible

• NOTE •

All information derived from GPS or DR will be removed from the displays.

TAWS is inoperative.

The airplane symbol is removed from all maps. The map will remain centered at the last known position. NO GPS POSITION will be annunciated in the center of the map.

•

*End of procedure.*

### GPS approach alarm limits exceeded

During a GPS LPV, LNAV/VNAV, or LNAV+V approach, if the Horizontal or Vertical alarm limits are exceeded, the GARMIN System will downgrade the approach. This will be annunciated in the ALERTS window and by an annunciation change on the HSI from LPV, L/VNAV, or LNAV+V to LNAV. GPS glide path vertical guidance will be removed from the PFD.

The approach may be continued using the LNAV only minimums.

During any GPS approach in which both precision and non-precision alarm limits are exceeded, the GARMIN System will flag the lateral guidance and display a system message ABORT APPROACH loss of navigation.

Immediately upon viewing the message, the unit will revert to Terminal navigation mode alarm limits. If the position integrity is within these limits, lateral guidance will be restored and the GPS may be used to execute the missed approach, otherwise alternate means of navigation must be utilized.

## Left PFD failure

► Fly the airplane ◀

*At takeoff :*

- 1 - Fly the airplane manually ..... Using stand-by instruments
- 2 - AP / TRIM DISC push-button ..... Press  
To mute aural tone

*In flight :*

### ▲ CAUTION ▲

In case of ILS approach, don't forget to select LOC2 on CDI source.

Use of reversionary mode will report left PFD information on MFD and disable supplementary functions as stormscope,...



- 3 - Fly the airplane manually ..... Using stand-by instruments
- 4 - AP / TRIM DISC push-button ..... Press  
To mute aural tone
- 5 - DISPLAY BACKUP mode ..... Engage on PFD2  
Pressurization switches to CPCS BACKUP MODE,  
with **CPCS BACKUP MODE**
- 6 - PFD 1 breaker ..... Check pushed
- 7 - XFR button (on AFCS) ..... Press / to right side
- 8 - Autopilot ..... Use normally

Following systems are lost :

- COM 1, NAV 1, DME 1, XPDR 1
- Radio altimeter, TAS, if installed

► Land as soon as possible ◀

- 9 - COM 2, NAV 2, DME 2, XPDR 2 ..... Use
- 10 - COM 2 MIC ..... Select

*End of procedure.*

## AHRS failure

1/2

*Symptoms : Autopilot is disconnected*

- On PFD(s) : Comparator window

**HDG**

and/or

**PIT**

and/or

**ROL**

annunciation(s)

- On PFD(s) : Reversionary sensor window

**BOTH ON AHRS1**

or

**BOTH ON AHRS2**

annunciation

*Lost systems :*

- AHRS1 or AHRS2
- Autopilot (AP)

*Systems still operative :*

- Flight Director (FD), when engaged again.

*Actions :*

Autopilot is not operative.

- 1 - AHRS1 and/or AHRS2 breaker ..... Check pushed

If **BOTH ON AHRS1** or **BOTH ON AHRS2** annunciation is associated to

**HDG**

and/or

**PIT**

and/or

**ROL**

annunciation(s) :

- 2 - Fly the airplane manually.

*If pilot wishes :*

- 3 - FD default mode ..... Engage  
PIT and ROL

- 4 - FD specific modes ..... Engaged as desired  
HDG, NAV, ALT, ...

- 5 - Fly the airplane manually to follow Command Bars.

*End of procedure ■*

*Continue ►*

AHRS failure

2/2

► Continuing

If all annunciations **HDG** and/or **PIT** and/or **ROL** go off,  
refer to following condition.

If **BOTH ON AHRS1** or **BOTH ON AHRS2** annunciation  
not associated to

**HDG** and/or **PIT** and/or **ROL** annunciation(s) :

- 6 - PFD1 and PFD2 SENSOR softkeys ..... Press
- 7 - AHRS1 on PFD1 and/or AHRS2 on PFD2 ..... Reset
- 8 - **BOTH ON AHRS1** or **BOTH ON AHRS2**  
annunciation ..... Check OFF
- 9 - Autopilot ..... Use normally  
As desired

*End of procedure.*

## ADC failure

### Symptoms :

- On PFD(s) : Comparator window

**IAS**

and/or

**ALT**

annunciation(s)

- On PFD(s) : Reversionary sensor window

**BOTH ON ADC1**

or

**BOTH ON ADC2**

annunciation

### Lost systems :

- ADC1 or ADC2

### Actions :

Autopilot is still operative.

- 1 - ADC 1 and/or ADC 2 breaker ..... Check pushed

If **BOTH ON ADC1** or **BOTH ON ADC2** annunciation

is associated to **IAS** and/or **ALT** annunciation(s)

- 2 - No action required.

End of procedure ■

If all annunciations

**IAS**

, **ALT**

go off, refer to following condition.

If **BOTH ON ADC1** or **BOTH ON ADC2** annunciation

not associated to **IAS** and/or **ALT** annunciation(s)

- 3 - PFD1 and PFD2 SENSOR softkeys ..... Press

- 4 - ADC1 on PFD1 and/or ADC2 on PFD2 ..... Reset

- 5 - **BOTH ON ADC1** or **BOTH ON ADC2**

annunciation ..... Check OFF

End of procedure.

**MFD failure**

## • NOTE •

If PFD2 is in fullscreen mode, it will be splitted a few seconds after MFD failure.

•

*Lost system :*

- MFD

*Actions :*

- 1 - L.H. DISPLAY BACKUP button ..... Press
- 2 - MFD breaker ..... Check pushed

*End of procedure.*

Intentionally left blank



## Section 4

### Normal procedures

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## **4.1 - General**

This section provides procedures for the conduct of normal operation of TBM airplane.

The first part of this section lists the normal procedures required as a check list.

The amplified procedures are developed in the second part of the section.

The normal procedures for optional systems are given in section 9, Supplements of the POH.

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## 4.2 - Airspeeds for normal operation

Conditions :

- Takeoff weight : 7394 lbs (3354 kg)
- Landing weight : 7024 lbs (3186 kg)

Rotation airspeed ( $V_R$ ) :

- Flaps TO ..... 90 KIAS

Best rate of climb speed ( $V_Y$ ) :

- Landing gear and flaps UP ..... 124 KIAS

Best angle of climb speed ( $V_X$ ) :

- Landing gear and flaps UP ..... 100 KIAS

Maximum speed :

- Flaps TO ..... 178 KIAS
- Flaps LDG ..... 122 KIAS

Maximum airspeed with landing gear down ..... 178 KIAS

Maximum landing gear operating airspeeds

- Extension ..... 178 KIAS
- Retraction ..... 150 KIAS

Approach airspeed :

- Flaps LDG ..... 85 KIAS

Maximum operating speed ( $V_{MO}$ ) ..... 266 KIAS

Glide speed (maximum L / D ratio)

- Landing gear and flaps UP ..... 120 KIAS

Intentionally left blank

## 4.3 - Check-list procedures

Initial inside inspection and outside inspection performed.  
OXYGEN cylinder open.

| Inside inspection |   | 1/2             |
|-------------------|---|-----------------|
| 1 -               | Cabin door and pilot door, if installed | Closed / Locked |
| 2 -               | Baggage                                 | Stowed          |
| 3 -               | EMERGENCY EXIT pin                      | Removed         |
| 4 -               | Seat, pedals, harness                   | Adjust / Lock   |
| 5 -               | PASSENGER OXYGEN                        | STBY            |
| 6 -               | OXYGEN                                  | ON              |
| 7 -               | Crew oxygen masks                       | Test            |
| 8 -               | EXT LIGHTS                              | All OFF         |
| 9 -               | INT LIGHTS                              | All OFF         |
| 10 -              | Crash lever                             | Down            |
| 11 -              | STARTER                                 | OFF             |
| 12 -              | IGNITION                                | AUTO            |
| 13 -              | AUX BP                                  | OFF             |
| 14 -              | FUEL SEL                                | MAN             |
| 15 -              | AP / TRIMS                              | OFF             |
| 16 -              | A/C                                     | OFF             |
| 17 -              | CB LIGHTS                               | OFF             |
| 18 -              | MICRO / MASK                            | MICRO / Guarded |
| 19 -              | DE ICE SYSTEM                           | All OFF         |
| 20 -              | INERT SEP                               | OFF             |
| 21 -              | PARK BRAKE                              | Reset / ON      |
| 22 -              | LANDING GEAR                            | DN              |

*Continue ►*

Inside inspection

2/2

► *Continuing*

- 23 - DUMP ..... NORM / Guarded
- 24 - BLEED ..... OFF
- 25 - HOT AIR FLOW ..... Fully turned to the right
- 26 - MAN OVRD ..... Full backward (notched)
- 27 - THROTTLE ..... CUT OFF
- 28 - FUEL TANK SELECTOR ..... Open / L or R
- 29 - ALTERNATE STATIC SOURCE ..... Pushed
- 30 - EMERGENCY RAM AIR ..... Pushed
- 31 - ESS BUS TIE ..... NORM / Guarded
- 32 - Breakers ..... All pushed
- 33 - Landing gear emergency pump handle ..... Check

*End of procedure.*



**Before starting engine**

- 1 - Crash lever ..... Up
- 2 - ATIS ..... Copied
- 3 - Start clearance ..... As required
- 4 - SOURCE ..... BATT or GPU
- 5 - GENERATOR ..... MAIN
- 6 - TEST ..... Press
- 7 - LANDING GEAR LIGHTS / CHECK DOWN ..... Test
- 8 - MFD ..... Initialize
- 9 - Fuel on board ..... Check
- 10 - Residual ITT ..... Check

*If residual ITT > 150°C :*

- 11 - Perform procedure ..... Motoring  
Refer to procedure hereafter
- 12 - VOLTS : BATT > 24.5 V / GPU ~ 28 V ..... Check
- 13 - CAS ..... Check

*End of procedure.*

**Motoring (if residual ITT > 150°C)**

- 1 - IGNITION ..... OFF
- 2 - AUX BP ..... ON
- 3 - **AUX BOOST PMP ON** ..... Check ON
- 4 - Propeller area ..... Clear
- 5 - STARTER ..... ON  
2 sec then OFF

*After 30 seconds maximum :*

- 6 - STARTER ..... ABORT  
Then OFF
- 7 - AUX BP ..... OFF

*End of procedure.*

## Engine start

### ▲ CAUTION ▲

After aborted engine starts, wait :

1 min / 5 min / 30 min before 2<sup>nd</sup> / 3<sup>rd</sup> / 4<sup>th</sup> new engine start.



- 1 - IGNITION ..... AUTO
- 2 - AUX BP ..... ON
- 3 - **AUX BOOST PMP ON** ..... Check ON
- 4 - Propeller area ..... Clear
- 5 - STARTER ..... ON  
2 sec then OFF

When Ng around 13 % :

- 6 - THROTTLE ..... LO-IDLE
- 7 - ITT, Ng, OIL °C and OIL PSI ..... Monitor

|     |   |
|-----|---|
| ITT | Maximum<br>1000°C for 5 sec<br>870°C for 20 sec |
| Ng  | 30 % before 30 sec<br>50 % before 1 min         |

When

- Ng > 50 % and,
  - 1 minute max :
- 8 - Starter ..... Check OFF automatically  
*End of procedure.*

After engine start with GPU

- 1 - SOURCE ..... BATT
- 2 - GPU ..... Disconnect
- 3 - **GPU DOOR** ..... Check OFF

*End of procedure.*

After engine start

- 1 - THROTTLE ..... LO-IDLE ►Flight IDLE
- 2 - Ng ..... Check 70 %  $\pm$  2 %
- 3 - OIL °C and OIL PSI ..... Check
- 4 - AUX BP ..... AUTO
- 5 - FUEL SEL ..... AUTO
- 6 - SHIFT ..... Test
- 7 - AP / TRIMS ..... ON

*If BATT < 80 amps :*

- 8 - GENERATOR ..... ST-BY / Test
- 9 - GENERATOR ..... MAIN
- 10 - CAS ..... Check
- 11 - A/C ..... As required
- 12 - BLEED ..... AUTO or MAX DIFF

*End of procedure.*

**Before taxiing**

- 1 - Stand-by instruments ..... Check
- 2 - DE ICE SYSTEM ..... Test
- 3 - INERT SEP ..... ON
- 4 - Flight controls ..... Check
- 5 - TRIMS ..... Test
- 6 - FLAPS ..... UP
- 7 - MFD
  - FPL ..... Set
  - LFE ..... Set / Check
  - WX RADAR ..... STBY
- 8 - THROTTLE ..... Feather twice
- 9 - EIS ..... Check
- 10 - CAS ..... Check
- 11 - TAXI lights ..... ON

*End of procedure.*

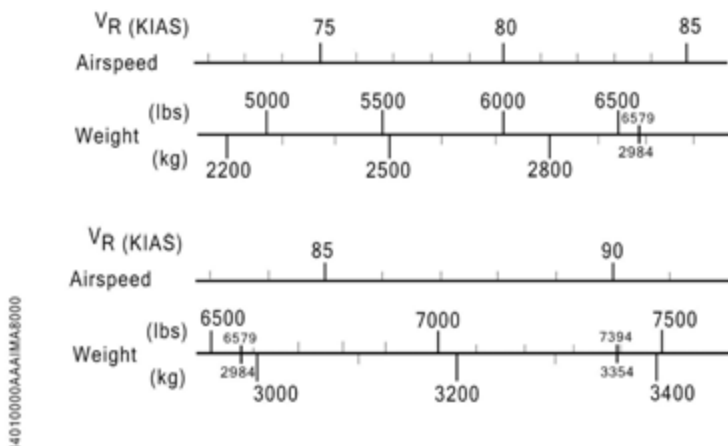
Before line up

- 1 - LDG lights ..... ON
- 2 - NAV ..... ON
- 3 - STROBE ..... ON
- 4 - IGNITION ..... As required  
AUTO or ON
- 5 - AUX BP ..... AUTO
- 6 - FUEL SEL ..... AUTO
- 7 - DE ICE SYSTEM ..... As required
- 8 - PITOT L HTR / PITOT R & STALL HTR ..... ON
- 9 - INERT SEP ..... ON
- 10 - TRIMS ..... TO
- 11 - FLAPS ..... TO
- 12 - A/C ..... As required
- 13 - BLEED ..... AUTO or MAX DIFF
- 14 - LFE ..... Check
- 15 - FUEL gages ..... Check imbalance
- 16 - BATT ..... Check below **50 amps**
- 17 - EIS ..... Check
- 18 - CAS ..... Check
- 19 - Altimeters setting ..... As required
- 20 - Instruments departure setting ..... Check
- 21 - SID (PROC) ..... Set
- 22 - ALT SEL ..... Set
- 23 - XPDR squawk ..... Set

*End of procedure.*

### Normal takeoff

- 1 - ADI, HSI, headings ..... Check
- 2 - PROP RPM ..... Green sector
- 3 - Brakes ..... Release
- 4 - TRQ ..... 100 %
- 5 - Rotation airspeed



- 6 - Attitude ..... 10° Up

*When vertical speed is positive :*

- 7 - Brakes ..... Apply
- 8 - LANDING GEAR ..... UP

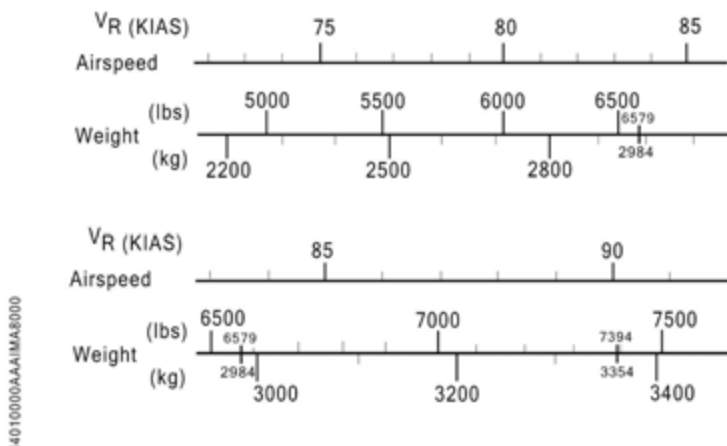
*When airspeed above 115 KIAS :*

- 9 - FLAPS ..... UP

*End of procedure.*

## Short takeoff

- 1 - ADI, HSI, headings ..... Check
- 2 - PROP RPM ..... Green sector
- 3 - TRQ ..... 100 %
- 4 - Brakes ..... Release
- 5 - Rotation airspeed



*Weight < 6579 lbs (2984 kg) :*

- 6 - Attitude ..... 15° Up

*Weight > 6579 lbs (2984 kg) :*

- 7 - Attitude ..... 12.5° Up

*When vertical speed is positive :*

- 8 - Brakes ..... Apply
- 9 - LANDING GEAR ..... UP

*When airspeed above 115 KIAS :*

- 10 - FLAPS ..... UP

*End of procedure.*



**After takeoff**

- 1 - LANDING GEAR ..... Check UP
- 2 - FLAPS ..... Check UP
- 3 - TRQ ..... Check max 100 %
- 4 - EIS ..... Check
- 5 - CAS ..... Check
- 6 - DE ICE SYSTEM ..... As required
- 7 - INERT SEP ..... As required

*End of procedure.*

Climb

- 1 - ALT SEL ..... Check
- 2 - Altimeters setting ..... As required
- 3 - Autopilot ..... Check
- 4 - TRQ adjustment / ITT / Ng ..... Check
- 5 - EIS ..... Check
- 6 - CAS ..... Check
- 7 - WX RADAR ..... As required
- 8 - Pressurization ..... Check
- 9 - FUEL gages ..... Check
- 10 - AMPS / VOLTS ..... Check
- 11 - DE ICE SYSTEM ..... As required
- 12 - INERT SEP ..... As required
- 13 - LDG lights ..... As required

*End of procedure.*

|        |
|--------|
| Cruise |
|--------|

- |      |                                 |             |
|------|---------------------------------|-------------|
| 1 -  | Altimeters setting .....        | Check       |
| 2 -  | Autopilot .....                 | Check       |
| 3 -  | TRQ adjustment / ITT / Ng ..... | Check       |
| 4 -  | EIS .....                       | Check       |
| 5 -  | CAS .....                       | Check       |
| 6 -  | Pressurization .....            | Check       |
| 7 -  | FUEL gages .....                | Check       |
| 8 -  | AMPS / VOLTS .....              | Check       |
| 9 -  | DE ICE SYSTEM .....             | As required |
| 10 - | INERT SEP .....                 | As required |
| 11 - | LDG lights .....                | OFF         |
| 12 - | Top of descent .....            | Computed    |

*End of procedure.*

Before descent

- 1 - Briefing before approach ..... Completed
- 2 - Altimeters setting ..... Check
- 3 - Pressurization ..... Check
- 4 - LFE ..... Check
- 5 - FUEL gages ..... Check
- 6 - AMPS / VOLTS ..... Check
- 7 - DE ICE SYSTEM ..... As required
- 8 - INERT SEP ..... As required

*End of procedure.*

## Approach

- 1 - Altimeters setting (QNH) ..... Set / Check
- 2 - Minimums ..... Set / Check
- 3 - COM / NAV / GPS ..... Set / Check
- 4 - Pressurization ..... Check
- 5 - LFE ..... Check
- 6 - FUEL gages ..... Check
- 7 - AMPS / VOLTS ..... Check
- 8 - DE ICE SYSTEM ..... As required
- 9 - INERT SEP ..... ON

*Below FL 100 :*

- 10 - LDG lights ..... ON

*End of procedure.*

Final approach (in GS) or downwind leg (VMC)

- 1 - LDG lights ..... ON
- 2 - LANDING GEAR ..... DN  
Check 3 green
- 3 - FLAPS ..... TO  
*End of procedure.*

**Short final ( $\approx$  500 ft)**

- 1 - LANDING GEAR ..... Check  
DN and 3 green
- 2 - FLAPS ..... LDG
- 3 - AP / YD ..... Disconnect  
*End of procedure.*

Runway clear

- 1 - TAXI light ..... ON
- 2 - NAV ..... As required
- 3 - STROBE ..... As required
- 4 - DE ICE SYSTEM ..... As required
- 5 - TRIMS ..... Reset to TO
- 6 - FLAPS ..... UP
- 7 - A/C ..... As required
- 8 - XPDR ..... Check
- 9 - WX RADAR ..... Check

*End of procedure.*



## Shutdown

- 1 - PARK BRAKE ..... Set / ON
- 2 - EXT LIGHTS ..... All OFF
- 3 - INT LIGHTS ..... As required
- 4 - OXYGEN ..... OFF
- 5 - FUEL SEL ..... MAN
- 6 - AP / TRIMS ..... OFF
- 7 - A/C ..... OFF
- 8 - BLEED ..... OFF
- 9 - THROTTLE ..... Flight IDLE  
For 2 min
- 10 - THROTTLE ..... LO-IDLE  
For 15 sec
- 11 - THROTTLE ..... CUT OFF
- 12 - INERT SEP ..... OFF
- 13 - **AUX BOOST PMP ON** ..... Check ON
- 14 - AUX BP ..... OFF
- 15 - GENERATOR ..... OFF

*When inertial separator is retracted, after approximately 40 sec :*

- 16 - SOURCE ..... OFF
- 17 - Crash lever ..... Pull down
- 18 - Stand-by instruments ..... OFF
- 19 - Oxygen cylinder (R.H. Karman) ..... Close

### • NOTE •

Within 10 minutes following the engine shutdown, check engine oil level.  
Refer to chapter 8.7 Oil level check.

•

*End of procedure.*

Intentionally left blank

## 4.4 - Amplified procedures

### Preflight inspection

1/16

The preflight inspection procedure is based on a scanning method.

It is divided in 6 subparts to cover all items of the preflight - see figure 4.4.1

- I Initial inside inspection
- II Cabin
- III L.H. Wing
- IV Fuselage forward section
- V R.H. Wing
- VI Fuselage rear section / Empennages

#### ▲ WARNING ▲

**During outside inspection, visually check inspection doors and airplane general condition. Check for systems and parts attachments / deflections / leaks / cracks / deteriorations / non-obstructions / nicks / numbers / free movements / position.**

**In cold weather, remove even small accumulations of frost, ice or snow from wing, tail and control surfaces.**

**In case of night flight, check good operation of all navigation lights, landing lights, strobe lights and make sure that an emergency lamp is on board.**

**If icing conditions are foreseen, particularly check good functioning of all electrical and pneumatic ice protection systems.**

**Check that type and quantity of fuel used for refueling are correct.**

**Remove covers on : pitots (2), static ports (2), static dischargers (2), engine air inlet (1), air inlets (2), exhaust cover and propeller locks (2).**

*Continue ►*

Preflight inspection

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► Continuing

**▲ WARNING ▲**  
**Remove tie-downs.**

Refer to section 8 for quantities, products and specifications of products and materials currently used.

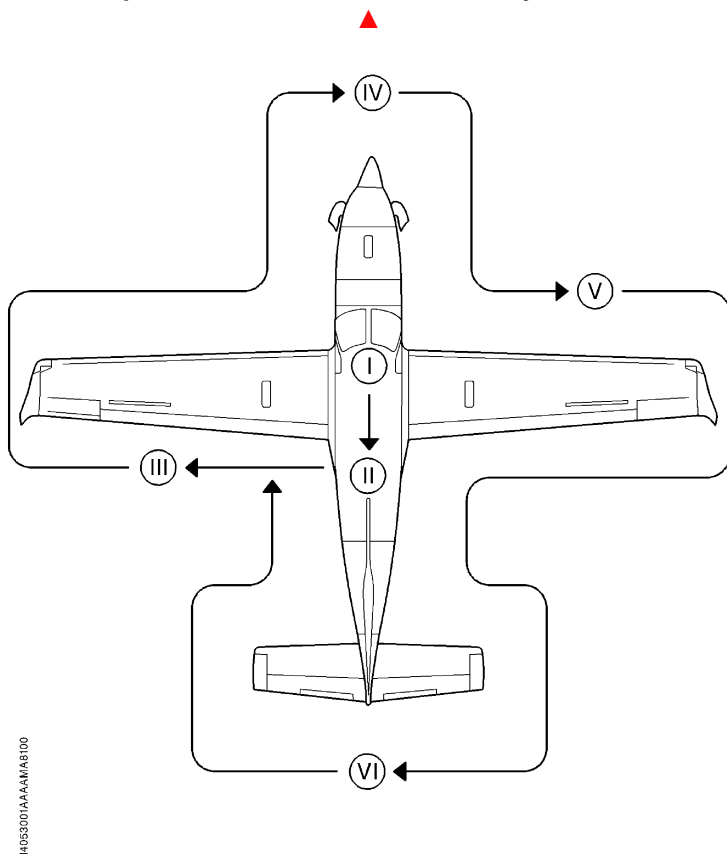


Figure 4.4.1 - Preflight inspection

Continue ►

Preflight inspection

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► *Continuing*

Initial inside inspection

Cockpit (I)

▲ **CAUTION** ▲

When engine is shut down, do not set the PROP DE ICE switch to ON for more than 10 seconds, damage to the propeller blades could result.



- 1 - A/C switch ..... OFF
- 2 - DE ICE SYSTEM panel ..... All OFF
- 3 - MICRO/MASK switch ..... MICRO / Guarded
- 4 - Flight controls lock ..... Removed / Stowed

● NOTE ●

The flight controls lock is normally stowed in the front cargo compartment with the towing bar and the blanking covers.



- 5 - Flight controls deflections ..... Check
- 6 - PARK BRAKE ..... ON
- 7 - LANDING GEAR lever ..... DN
- 8 - DUMP switch ..... NORM / Guarded
- 9 - BLEED switch ..... OFF

Engine controls

- 10 - MAN OVRD control ..... Backward

*Continue* ►

Preflight inspection

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► *Continuing*

▲ **CAUTION** ▲

When the engine is shut down, the throttle must not be moved into the reverse area as a lack of hydraulic pressure prevents movement into reverse range. Trying to force the mechanism will cause damage.



- 11 - THROTTLE ..... CUT OFF
  - 12 - FLAPS lever ..... UP
  - 13 - FUEL TANK SELECTOR ..... L or R
- Open door of emergency landing compartment to check LANDING GEAR emergency control.
- 14 - Lever ..... Pushed down
  - 15 - By-pass selector ..... Fully depressed
  - 16 - Door ..... In place

● **NOTE** ●

By-pass selector must be pushed at its maximum stop, so as to have the door in place.



- 17 - ALTERNATE STATIC SOURCE selector ..... Pushed
  - 18 - EMERGENCY RAM AIR control knob ..... Pushed
  - 19 - Breakers panel ..... All breakers checked
  - 20 - ELT switch ..... ARM / OFF
  - 21 - AP / TRIMS switch ..... OFF
- FUEL panel
- 22 - FUEL SEL switch ..... MAN
  - 23 - AUX BP switch ..... OFF

*Continue* ►

# Preflight inspection

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## ▶ Continuing

### ENGINE START panel

24 - IGNITION switch ..... AUTO or OFF

#### • NOTE •

The IGNITION switch is normally selected to AUTO. This ensures ignition, whenever the STARTER switch is set to ON.

•

25 - STARTER switch ..... OFF

#### • NOTE •

If not, starter is going to operate as soon as SOURCE selector is moved to BATT or GPU (if connected).

•

### ELECTRIC POWER panel

26 - Crash lever ..... Up

27 - GENERATOR selector ..... MAIN

28 - SOURCE selector ..... OFF

29 - ACCESS lighting ..... Check  
To ensure that the fuse of the BATT BUS operates correctly

30 - INT LIGHTS panel ..... All OFF

31 - EXT LIGHTS panel ..... All OFF

32 - OXYGEN switch ..... OFF

33 - PASSENGER OXYGEN switch ..... STBY

34 - Emergency lighting ..... Check

#### ▲ CAUTION ▲

Before selecting source, check position of ignition and starter switches.

▲

35 - IGNITION switch ..... AUTO or OFF

36 - STARTER switch ..... OFF

Continue ▶

Preflight inspection

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► *Continuing*

- 37 - LANDING GEAR lever ..... DN
- 38 - SOURCE selector ..... BATT or GPU
- 39 - Standby instrument battery indicator symbol ..... Not displayed

If a battery symbol appears on the standby instrument display, airplane takeoff is not allowed until the situation is resolved. Refer to the battery details in the standby instrument Pilot's guide for further information.

*If BATT source :*

- 40 - Voltage ..... Check > 24.5 volts

• NOTE •

If not, use a GPU or charge battery. This minimum voltage is not an absolute guarantee for a correctly charged battery. It is recommended to use a GPU in cold weather, when airplane has been stopped more than 3 hours at a temperature below - 10°C (+ 14°F).

•

*If GPU source :*

▲ CAUTION ▲

Low voltage (around 24.5 V) may indicate that only the battery is powering the airplane and not the pair GPU + battery.  
Make sure that a GPU is connected and powering the airplane.

▲

- 41 - Voltage ..... Check ≈ 28 volts

• NOTE •

If using a GPU, ensure that it provides a 28-volt regulated voltage, with negative on earth, as well as it supplies 800 amps minimum and 1000 amps maximum. See placard located near ground power receptacle door.

•

EXT LIGHTS panel

- 42 - OFF/TAXI/LDG switch ..... OFF
- 43 - STROBE switch ..... ON
- 44 - NAV switch ..... ON

*Continue ►*



Preflight inspection

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► *Continuing*

DE ICE SYSTEM panel

45 - All switches ..... OFF

46 - ICE LIGHT switch ..... ON

47 - From outside the airplane, check operation of all lights and stall warning alert.

Reentering the airplane

48 - EXT LIGHTS panel ..... All OFF

49 - CAS display ..... Check

50 - Left and right FUEL quantities ..... Check

51 - FLAPS lever ..... LDG

LANDING GEAR panel

52 - Warning lights ..... Check 3 green ON

53 - LIGHT TEST push-button ..... Press  
Check all lights flashing

DE ICE SYSTEM panel

▲ **WARNING** ▲

**Do not touch pitots nor stall warning vane. They could be hot enough to burn skin.**



54 - PITOT L HTR switch ..... ON

55 - **PITOT HT ON L** ..... Check ON

*Continue* ►

## Preflight inspection

8/16

### ► Continuing

56 - PITOT R & STALL HTR switch ..... ON

#### • NOTE •

Correct operation of pitot (PITOT L and R) tube heating elements and of stall aural warning system (STALL HTR) is indicated by display of corresponding CAS message, when control switches are ON.

57 - **PITOT HT ON L-R** ..... Check ON

58 - **STALL HEAT ON** ..... Check ON

59 - PITOT L HTR switch ..... OFF

60 - PITOT R & STALL HTR switch ..... OFF

61 - Crash lever ..... Pull down

### Cabin (II)

62 - Cabin fire extinguisher ..... Pressure / Attachment

63 - Seats / belts ..... Check

64 - Windows ..... General condition / No crack

65 - Emergency exit ..... Closed / Locked

66 - Anti-theft safety pin ..... Removed / Stowed

Continue ►

## Preflight inspection

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### ► Continuing

- 67 - Baggage compartment ..... Straps in place  
 >> 6-seat accommodation
- 68 - Partition net ..... General condition / In place  
 >> 4-seat accommodation and baggage transportation
- 69 - Large net or small net ..... General condition / In place  
 >> All
- 70 - Doors operation ..... Check
- 71 - Stairs condition ..... Condition / Play

### Outside inspection

The preflight inspection described in figure 4.4.1 is recommended before each flight.

#### ● NOTE ●

If a preflight inspection is performed just after the engine shutdown, be careful because the leading edge of engine air inlet, as well as exhaust stubs may be very hot.

●

If the airplane was in long term storage or if it has undergone major maintenance or if it has been used from emergency airfields, a thorough outside inspection is recommended.

When the airplane is stored outside, the use of the flight control lock and blanking covers is recommended. Propeller should be tied down to prevent rotation without oil pressure.

When the airplane is stored for extended periods of time, a thorough preflight inspection is recommended. Particular attention should be paid to possible blockages in airspeed sensing lines, foreign objects in engine intake and exhaust stubs and water contamination of the fuel system.

*Continue ►*

**Preflight inspection**

**10/16**

► *Continuing*

L.H. wing (III)

- 72 - Flap ..... Condition / Play

Also inspect the lower surface, as well as flap fairing, where pebbles (and even ice in case of slush on the runway) may have accumulated.

- 73 - Aileron and trim / Spoiler ..... Condition / Free movement / Deflection

• NOTE •

Ensure there are no foreign objects in the spoiler recess. When ailerons are in the neutral position, it is normal that spoilers are lightly extended at upper surface.

•

- 74 - Trailing edge static discharger ..... Condition / Number / Attachment

- 75 - Winglet / nav. lights / strobe / landing light /  
recognition light / taxi light ..... Condition

- 76 - OAT probe ..... Condition

- 77 - Fuel tank cap ..... Closed / Locked

• NOTE •

Fuel tank caps must be tight (which is characterized by a consequent exertion to lock and unlock them) to avoid water infiltration in case of rain on ground, and to avoid fuel loss in flight.

•

- 78 - Fuel tank air vent ..... Unobstructed

• NOTE •

Air vent is not likely to be obstructed by ice or water, as it is located in a wing lower surface recess.

•

*Continue ►*

**Preflight inspection**

**11/16**

► *Continuing*

- 79 - Left pitot ..... Condition
- 80 - Wing lower surface ..... No leak
- 81 - Check fuel tank access doors for leaks.
- 82 - Check for surface damage.
- 83 - Wing deicer boots ..... Condition / Attachment

• NOTE •

Care must be taken when refuelling the airplane to avoid damaging the wing deicer boots. A protective apron should be used if possible.

•

- 84 - Fuel tank drain (two on each wing) ..... Drain  
Fuel free of water and contamination

• NOTE •

In case of water in fuel system, drain it carefully using the four drain valves of tank sumps, and the fuel filter drain valve, till every trace of water or deposit has disappeared.

A long term storage of the airplane causes water accumulation in fuel, which absorbs additive. This phenomenon occurs when an excessive quantity of water accumulates in fuel tank sumps. Refer to section 8 for servicing operations relative to fuel additives.

•

L.H. main LANDING GEAR

- 85 - Shock absorber ..... Check
- 86 - Doors ..... Check
- 87 - Tire ..... Check
- 88 - Wheel well ..... Check

*Continue ►*

Preflight inspection

12/16

► *Continuing*

• NOTE •

If airplane has been used from muddy airfields or in snow, check wheel wells to make sure they are clean and not obstructed.

Check frequently all landing gear retraction mechanism components, shock-absorbers, tires and brakes. This is particularly important for airplanes used from hilly fields.

Improperly serviced or worn shock-absorbers may result in excessive loads being transmitted to the airplane structure during ground operations. Without passengers and baggages on board, the unpainted surface of the main gear shock absorber tube must be visible about :

55 mm (2.17 in) of minimum height with half tank,

40 mm (1.57 in) of minimum height with full tanks.

•

Fuselage forward section (IV)

Forward compartment

- |      |                          |   |
|------|--------------------------|---|
| 89 - | Inside .....             | Check   |
| 90 - | Door .....               | Close / Lock                                  |
| 91 - | GPU door .....           | Closed<br>If not used                         |
| 92 - | Fuel circuit drain ..... | Drain<br>Fuel free of water and contamination |

▲ **WARNING** ▲

**If the clogging indicator is extended, red collar visible, the flight is not authorized.**

▲

- |      |   |                       |
|------|---|-----------------------|
| 93 - | Filter contamination indicator (clogging indicator) ..... | Check                 |
| 94 - | L.H. exhaust stub .....                                   | Condition / No cracks |

*Continue ►*

## Preflight inspection

13/16

### ► Continuing

#### • NOTE •

Inspect if possible pressure port located inside exhaust stub. A missing port or a cracked port may hinder correct operation of continuous heating of air inlet lip.

•

95 - Upper engine cowl ..... Open

For the first flight of the day :

96 - Oil cap ..... Closed / Locked

97 - Engine oil level ..... Check

98 - Fuel pipes ..... No leak, deterioration, wear

99 - Engine cowl ..... Condition  
Closed / Locked

Air inlets

100 - Main ..... No cracks - Unobstructed

#### • NOTE •

Check for no cracks, which are sometimes put in evidence by traces of soot resulting from exhaust gases.

•

101 - Lateral / upper ..... Unobstructed

#### • NOTE •

Lateral air inlets, which supply air conditioning system and oil cooler, are provided with blanking covers. It is not the case for upper air inlets of RAM AIR system (circular grille located in front of R.H. windshield) and of vapor cycle cooling system (two rectangular grilles located forward of the circular grille).

•

Continue ►

Preflight inspection

14/16

► *Continuing*

102 - Propeller and spinner . . . . . No nicks, cracks or oil leaks / Attachment

• NOTE •

In case of operation from contaminated runways, it is necessary to carefully examine propeller blades, where traces of abrasion may be found. Propeller damage may reduce blade life time and degrade performance. Any propeller damage should be referred to maintenance personnel.

•

Nose gear

103 - Shock absorber . . . . . Check

104 - Doors . . . . . Check

105 - Tire . . . . . Check

106 - Wheel well . . . . . Check

• NOTE •

Without passengers and baggages on board, the unpainted surface of the nose gear shock absorber tube must be visible about :  
57 mm (2.22 in) of minimum height with full tanks,  
63 mm (2.46 in) of minimum height with half tank.

•

• NOTE •

Crush or relieve the shock absorber one time or twice before the inspection to remove possible sticking.

•

In case of doubt, request a check of the shock absorber pressure.

107 - R.H. exhaust stub . . . . . Condition / No cracks

R.H. wing (V)

Additional remarks are identical to those of L.H. wing.

108 - Fuel tank drain (two on each wing) . . . . . Drain  
Fuel free of water and contamination

*Continue* ►



Preflight inspection

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► *Continuing*

R.H. main LANDING GEAR

- 109 - Shock absorber ..... Check
- 110 - Doors ..... Check
- 111 - Tire ..... Check
- 112 - Wheel well ..... Check
- 113 - Wing deicer boots ..... Condition / Attachment
- 114 - Stall warning ..... Condition / Deflection
- 115 - Wing lower surface ..... No leaks
- 116 - Fuel tank cap ..... Closed / Locked
- 117 - Fuel tank air vent ..... Unobstructed
- 118 - Right pitot ..... Condition
- 119 - Winglet / nav. light / strobe / landing light /  
recognition light / taxi light ..... Condition
- 120 - Trailing edge static discharger ..... Condition / Number / Attachment
- 121 - Aileron / spoiler ..... Condition / Free movement / Deflection
- 122 - Flap ..... Condition / Play

Rear R.H. karman

- 123 - Oxygen cylinder ..... Open
- 124 - Oxygen pressure ..... Check
- 125 - Confirm OXYGEN quantity in regards with the expected flight.
- 126 - Oxygen pressure ..... Check

Fuselage rear section / empennages (VI)

Check that outside handle of emergency exit is flush with door skin.

- 127 - ELT ..... ARM / OFF

*Continue* ►

## Preflight inspection

16/16

### ► Continuing

128 - ELT door ..... Closed / Locked

#### • NOTE •

Access to ELT is possible through an inspection door located on R.H. side of fuselage rear section.

129 - Static pressure ports ..... Clean

130 - Ventral fins ..... Condition / Attachments

#### • NOTE •

Ventral fins are made of two parts (one fixed part and one removable part with rear lower inspection door). Check that these two parts are connected by the locking roller.

131 - Inspection door under fuselage ..... Attachments - Closed

132 - Horizontal stabilizer deicer boots (R.H. side) ..... Condition / Attachments

133 - Elevator and trim ..... Condition / Deflection free movement / Trim position

#### • NOTE •

To check the deflection, hold the two half-elevators near fuselage, inside both elevator trims to avoid stresses.

134 - Static dischargers ..... Condition

135 - Vertical stabilizer deicer boots ..... Condition / Attachments

136 - Rudder and trim ..... Condition / Trim position

137 - Static dischargers ..... Condition

138 - Tail cone / nav. lights / strobe ..... Condition

139 - Static pressure ports ..... Clean

*End of procedure.*

## Inside inspection

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After completion of preflight inspection. Initial inside inspection and outside inspection performed.

- 1 - Cabin door and pilot door, if installed ..... Closed / Locked
- 2 - Baggage ..... Stowed
- 3 - EMERGENCY EXIT pin ..... Removed
- 4 - Seats, pedals, harness ..... Adjust / Lock

### ▲ CAUTION ▲

It is mandatory to adjust seats in fore-aft movement when seat is in maximum high permissible position, to avoid interference between side upholstery panel and seat housing in low and intermediate positions.



Pilot seat and R.H. front seat, if occupied ..... Adjust

### ● NOTE ●

Adjust seats and harnesses, so as to permit access to flight controls. The pilot at L.H. station must be able to easily reach A/C and PRESSURIZATION panel.



- 5 - Height adjustment ..... Max. UP
- 6 - Fore and aft adjustment ..... Adjust and check locking
- 7 - Height adjustment ..... Adjust
- 8 - L.H and R.H. pedals ..... Adjust
- 9 - Pilot and passengers belts and harnesses ..... Fasten

### ● NOTE ●

Check for pilot and passengers correct locking of belt buckles, as well as automatic locking of shoulder harness by exerting a rapid pull on the latter. If airbags installed, unoccupied seat belts have to be strapped. It is forbidden to fly with these belts unstrapped.



*Continue ►*

Inside inspection

2/4

► *Continuing*

- 10 - PASSENGER OXYGEN switch ..... STBY  
11 - OXYGEN switch ..... ON

• NOTE •

Make sure to set on STBY the PASSENGER OXYGEN switch before setting the OXYGEN switch to ON to avoid passengers mask deployment.

•

- 12 - Crew oxygen masks ..... Test

• NOTE •

Press push-button PRESS TO TEST : the blinker shall turn red momentarily, then turns transparent.

•

- 13 - EXT LIGHTS panel ..... All OFF  
14 - INT LIGHTS panel ..... All OFF  
15 - DIMMER switch ..... OFF  
16 - CABIN switch ..... OFF  
17 - ACCESS switch ..... OFF  
18 - PANEL rheostat ..... Fully turned to the left  
19 - All lights ..... OFF  
20 - Crash lever ..... Down  
21 - STARTER switch ..... OFF

• NOTE •

If not, starter is going to operate as soon as SOURCE selector is positioned on BATT or GPU.

•

*Continue* ►

## Inside inspection

3/4

▶ *Continuing*

■ 22 - IGNITION switch ..... AUTO

### • NOTE •

The IGNITION switch is normally selected to AUTO. This ensures ignition,  
whenever the starter is activated.

■ 23 - AUX BP switch ..... OFF

■ 24 - FUEL SEL switch ..... MAN

■ 25 - AP / TRIMS switch ..... OFF

■ 26 - A/C switch ..... OFF

■ 27 - CB LIGHTS switch ..... OFF

■ 28 - MICRO / MASK switch ..... MICRO / Guarded

■ 29 - DE ICE SYSTEM panel ..... All OFF

■ 30 - INERT SEP switch ..... OFF

■ 31 - PARK BRAKE ..... Reset / ON

■ 32 - LANDING GEAR lever ..... DN

■ 33 - DUMP switch ..... NORM / Guarded

■ 34 - BLEED switch ..... OFF

■ 35 - HOT AIRFLOW distributor ..... Fully turned to the right

■ 36 - Pitch trim wheel ..... Check

### ▲ CAUTION ▲

Make sure that MAN OVRD control is backward to avoid  
overtemperature risks at start.

■ 37 - MAN OVRD control ..... Full backward (notched)

*Continue ▶*

Inside inspection4/4

► Continuing

▲ CAUTION ▲

When the engine is shut down, the THROTTLE must not be moved into the reverse area.



- 38 - THROTTLE ..... CUT OFF
- 39 - FUEL TANK SELECTOR ..... Open / L or R
- 40 - ALTERNATE STATIC SOURCE selector ..... Normal / Pushed
- 41 - EMERGENCY RAM AIR ..... Closed / Pushed
- 42 - ESS BUS TIE switch ..... NORM / Guarded
- 43 - Breakers ..... All pushed
- 44 - EMERGENCY LANDING GEAR lever ..... Check

End of procedure.

Before starting engine

1/3

Check that the weight and balance are within the correct limits. Brief passengers about use of seat belts and the emergency oxygen system, as well as opening the access door and the emergency exit.

- 1 - Preflight inspection ..... Completed
- 2 - Crash lever ..... Up
- 3 - ATIS ..... Copied
- 4 - Start clearance ..... As required
- 5 - SOURCE selector ..... BATT (battery start) or GPU (GPU start)

*If one screen (L or R PFD, or MFD) is missing :*

- 6 - SOURCE selector ..... OFF
- 7 - Wait for 30 seconds
- 8 - SOURCE selector ..... BATT (battery start) or GPU (GPU start)

*If GPU use :*

- 9 - **GPU DOOR** ..... Check ON
- 10 - Voltmeter ..... Check 28 Volts  $\pm$  0.5 Volt

• NOTE •

Voltage is higher than 24.5 Volts which corresponds to the voltage in case of battery use.

•

*If battery use :*

- 11 - Battery voltage ..... Check > 24.5 V

*If battery voltage < 24.5 V :*

- 12 - Ask for a GPU and be ready to a GPU start.

- 13 - GENERATOR selector ..... MAIN
- 14 - **MAIN GEN** ..... Check ON

Continue ►

Before starting engine

2/3

► Continuing

15 - **OXYGEN** ..... Check OFF

If **OXYGEN** is ON :

16 - Open isolation valve of the oxygen cylinder in R.H. Karman.

17 - TEST push-button ..... Press  
Audio / DE ICE SYSTEM lights / stick shaker

18 - DUMP switch ..... NORM / Guarded

19 - LANDING GEAR light / CHECK DOWN ..... Test

20 - MFD ..... Initialize

21 - Fuel onboard ..... Check

- Quantity ..... Check

- FUEL TANK SELECTOR ..... L or R

22 - Residual ITT ..... Check

If residual ITT > 150°C :

23 - Perform procedure ..... Motoring  
Refer to this chapter

• NOTE •

A start up procedure with an engine residual ITT above 150°C may generate an ITT exceedance.

Particular monitoring of ITT will have to be performed during start up to ensure to keep the temperature within ITT envelope.

•

24 - VOLTS : BAT > 24.5 V / GPU ≈ 28 V ..... Check

25 - CAS display ..... Check

26 - PARK BRAKE ..... Check ON  
Last check before proceeding to engine start

27 - **PARK BRAKE** ..... Check ON

Continue ►



## Before starting engine

2/3

► *Continuing*

• NOTE •

**PARK BRAKE** appearance does not indicate that parking brake is set. For that, press on brake pedals before turning brake selector to the right.

•

## Engine start

1/3

- 1 - STROBE switch ..... ON
- 2 - G3000 ..... DISPLAY BACKUP  
Composite mode

### • NOTE •

If there is a loss of MFD during start up sequence, that sequence will be ended using the left PFD in composite mode.

- 3 - IGNITION switch ..... AUTO
- 4 - AUX BP switch ..... ON
- 5 - **AUX BOOST PMP ON** ..... Check ON
- 6 - **FUEL PRESS** ..... Check OFF
- 7 - Propeller area ..... Clear

### ▲ CAUTION ▲

If 5 seconds after having positioned the STARTER switch in ON position there is no start, interrupt starting attempt using the ABORT position of the STARTER switch.

### ▲ CAUTION ▲

Starter operation is bound by limitations in chapter 2.4 Starter operating limits.

- 8 - STARTER switch ..... ON  
2 sec then OFF

*Simultaneously :*

- 9 - Timer clock ..... Start  
To check startup acceleration
- 10 - **STARTER** ..... Check ON
- 11 - **MAIN GEN** ..... Check ON

*Continue ►*

## Engine start

2/3

### ► Continuing

#### ▲ CAUTION ▲

When THROTTLE is positioned on LO-IDLE before having obtained 13 % of Ng, there is a risk of overtemperature further to an excessive accumulation of fuel inside the combustion chamber before ignition.



#### • NOTE •

In case of starting with high residual ITT, an ITT decrease below 150°C (within starter operation limits) may allow to stay within the allowed ITT envelope during startup sequence.



### When

- Ng about 13 % and,
- ITT below 150°C and,
- time below 20 seconds :

12 - THROTTLE ..... LO-IDLE

#### Abort starting procedure if :

- No ignition 10 seconds after having positioned THROTTLE to LO-IDLE,
- **ITT** lights on (max ITT < 870°C for more than 20 seconds, < 1000°C for more than 5 seconds),
- Ng < 30 % after 30 seconds of starter use,
- Ng < 50 % after 60 seconds of starter use,

13 - THROTTLE ..... CUT OFF

14 - IGNITION switch ..... OFF or AUTO

#### When ITT < 850°C :

15 - STARTER switch ..... ABORT

End of procedure ■

Continue ►

Engine start

3/3

► *Continuing*

*When*

- *Ng > 50 % and,*
- *1 minute max :*

▲ **CAUTION** ▲

If the starter does not go off automatically, disengage it using the  
ABORT position of the STARTER switch.



- 16 - Starter ..... Check OFF automatically
- 17 - **STARTER** ..... Check OFF
- 18 - Engine parameters ..... Check  
Check  $54 \% \leq Ng \leq 58 \%$ , oil pressure and ITT in green sector

*End of procedure.*

## Motoring

1/3

To drain fuel accumulated inside the combustion chamber, a motoring procedure is required following an aborted start.

A 15-second dry motoring run is sufficient to clear any fuel pooled in the engine. The fuel is removed in liquid or vapor form, through an airflow intended to dry combustion chamber, turbines and exhaust nozzles.

To improve cooling of the bearing cavities and prevent oil coking after shutdown in high OAT [above 35°C (95°F)] environment, it is recommended to perform a 30-second dry motoring run.

It is possible that no trace of drainage be observed under engine, due to the drainage collector intended to prevent parking area from contamination.

### ▲ CAUTION ▲

After any starting interrupt procedure, wait for engine total shutdown and wait at least 30 seconds before initiating a motoring.



#### Engine controls

- 1 - MAN OVRD control ..... Full backward (notched)

### ▲ CAUTION ▲

When the engine is shut down, the THROTTLE must not be moved into the reverse area.



- 2 - THROTTLE ..... CUT OFF
- 3 - IGNITION switch ..... OFF
- 4 - **IGNITION** ..... Check OFF

#### Fuel

- 5 - FUEL TANK SELECTOR ..... L or R
- 6 - AUX BP switch ..... ON
- 7 - **AUX BOOST PMP ON** ..... Check ON

Continue ►

Motoring

2/3

► Continuing

- 8 - **FUEL PRESS** ..... Check OFF

Fuel pressure is necessary for lubrication of HP pump.

- 9 - Propeller area ..... Clear

*To clear fuel and vapor internally trapped :*

- 10 - STARTER switch ..... ON  
2 sec then OFF

*Simultaneously :*

- 11 - Timer clock ..... Start

- 12 - **STARTER** ..... Check ON

- 13 - Motor ..... For 15 sec. max

- 14 - STARTER switch ..... ABORT  
Then OFF

- 15 - **STARTER** ..... Check OFF

*To cool engine following shutdown in high temperature environment :*

- 16 - STARTER switch ..... ON  
2 sec then OFF

*Simultaneously :*

- 17 - Timer clock ..... Start

- 18 - **STARTER** ..... Check ON

- 19 - Motor ..... For 30 sec. max

*If ignition symptoms occur (ITT increasing) :*

- 20 - IGNITION switch ..... Check OFF

- 21 - THROTTLE ..... Check CUT OFF

- 22 - Continue motoring.

Continue ►

Motoring

3/3

► *Continuing*

23 - STARTER switch ..... ABORT  
Then OFF

24 - **STARTER** ..... Check OFF

FUEL panel

25 - AUX BP switch ..... OFF

26 - **AUX BOOST PMP ON** ..... Check OFF

27 - **FUEL PRESS** ..... Check ON

*End of procedure.*

**Motoring followed by an engine start**

**1/3**

Amplified procedures stated in starting engine sequences using airplane power or with GPU are also to be applied to hereunder procedure.

Within starter operating limits (continuous max. 1 minute), it is possible to initiate a starting procedure from a motoring procedure.

This procedure will conserve the battery by taking advantage of first Ng acceleration.

**Engine controls**

- 1 - MAN OVRD control ..... Full backward (notched)

**▲ CAUTION ▲**

When the engine is shut down, the THROTTLE must not be moved into the reverse area.



- 2 - THROTTLE ..... CUT OFF  
3 - IGNITION switch ..... OFF  
4 - **IGNITION** ..... Check OFF

**Fuel**

- 5 - FUEL TANK SELECTOR ..... L or R  
6 - AUX BP switch ..... ON  
7 - **AUX BOOST PMP ON** ..... Check ON  
8 - **FUEL PRESS** ..... Check OFF  
Fuel pressure is necessary for lubrication of HP pump.  
9 - Propeller area ..... Clear  
10 - STARTER switch ..... ON  
2 sec then OFF

**Simultaneously :**

- 11 - Timer clock ..... Start  
12 - **STARTER** ..... Check ON

**Continue ►**



2/3

13 - Motor ..... For 30 sec. max

*After 20 seconds and if ITT < 150°C :*

14 - IGNITION switch ..... AUTO

15- Ng ..... Check > 13 %

16 - THROTTLE ..... LO-IDLE

Monitor increase of :

[illegible]

• NOTE •

No action is required for the following conditions :

- ITT from 850°C to 870°C limited to 20 seconds,
- ITT from 870°C to 1000°C limited to 5 seconds

●

18 - Nq

19 - Oil pressure

20 - **OIL PRESS** ..... Check OFF

*When  $N_g > 50\%$  :*

**▲ CAUTION ▲**

If the starter does not go off automatically, disengage it using the **ABORT** position of the **STARTER** switch.



21 - Starter ..... Check OFF automatically

22 - **STARTER** ..... Check OFF

23 - Engine parameters ..... Check  
Check  $54 \% \leq Ng \leq 58 \%$ , oil pressure and ITT in green sector

Continue ►

Motoring followed by an engine start

3/3

► Continuing

Fuel panel

24 - AUX BP switch ..... AUTO

25 - **AUX BOOST PMP ON** ..... Check OFF

Electric power

26 - **MAIN GEN** ..... Check OFF

Reset if necessary

• NOTE •

**MAIN GEN** normally goes off as soon as **STARTER** goes off.

•

If **MAIN GEN** does not go off :

27 - Ng ..... Increase over 70 %  
To start main generator

28 - Generator and battery AMPS ..... Check charge  
On EIS of MFD

29 - Battery and ESS. bus VOLTS ..... Check voltage ≈ 28 Volts  
On EIS of MFD

End of procedure.

### After engine start with GPU

- 1 - SOURCE selector ..... BATT
- 2 - Electrical network ..... Check
- 3 - GPU ..... Disconnect  
Performed by ground personnel
- 4 - **GPU DOOR** ..... Check OFF
- 5 - GENERATOR selector ..... MAIN
- 6 - **MAIN GEN** ..... Check OFF

• NOTE •

**MAIN GEN** normally goes off as soon as **STARTER** goes off.

•

If **MAIN GEN** does not go off :

- 7 - Ng ..... Increase over 70 %  
To start main generator
- 8 - Generator and battery AMPS ..... Check charge  
On EIS of MFD
- 9 - Battery and ESS. bus VOLTS ..... Check voltage ≈ 28 Volts  
On EIS of MFD
- 10 - CAS display ..... Check
- 11 - A/C switch ..... As required
- 12 - BLEED switch ..... AUTO or MAX DIFF

When ground personnel is cleared from propeller area :

- 13 - Perform procedure ..... After engine start  
Refer to procedure hereafter  
*End of procedure.*

After engine start

1/2

▲ **CAUTION** ▲

Generator load < 200 amps



- 1 - THROTTLE ..... LO-IDLE ►Flight IDLE
- 2 - Ng ..... Check 70 % ± 2 %
- 3 - OIL °C and OIL PSI ..... Check
- 4 - AUX BP switch ..... AUTO
- 5 - FUEL SEL switch ..... AUTO
- 6 - SHIFT push-button ..... Test  
Verify rotation of FUEL TANK SELECTOR
- 7 - AP / TRIMS switch ..... ON  
This initializes the A/P system
- 8 - PFD 1, MFD and PFD 2 ..... NORMAL mode

*Perform generator test :*

- 9 - BLEED switch ..... OFF  
To unload the generator circuit
- 10 - GENERATOR selector ..... Check MAIN
- 11 - AMPS / VOLTS ..... Check

*When MAIN LOAD < 80 amps :*

- 12 - GENERATOR selector ..... ST-BY
- 13 - AMPS / VOLTS ..... Check

*If the voltage on the ST-BY generator is low (close to 27 volts) :*

- 14 - GENERATOR RESET ST-BY push-button ..... Press  
To reset ST-BY generator
- 15 - AMPS / VOLTS ..... Check  
The indicated voltage should be in the green range
- 16 - GENERATOR selector ..... MAIN

*Continue ►*

## After engine start

2/2

## ► Continuing

- 17 - Oxygen ..... Check  
Verify quantity available for the planned flight.  
See tables of paragraph In-flight available oxygen quantity  
in this chapter and chapter 7.10 for a FAR 135 type operation

PFD 1, MFD and PFD 2

## • NOTE •

Detailed control procedures of avionics system are described in the GARMIN  
Integrated Flight Deck Pilot's Guide.

•

- 18 - Brightness ..... Adjust  
19 - DISPLAY BACKUP push-button ..... Check  
Then return to NORMAL mode  
20 - CAS ..... Check  
Check engine parameters  
21 - A/C switch ..... As required

## • NOTE •

A good cabin temperature regulation will only be obtained if A/C switch is set to  
PILOT or PLT + PAX.

•

- 22 - TEMP selector ..... Adjust  
23 - BLEED switch ..... AUTO or MAX DIFF  
24 - HOT AIR FLOW distributor ..... As required

## • NOTE •

HOT AIR FLOW distributor is usually set fully turned to the right. However, if  
canopy misting is evident, set it fully turned to the left.

•

*End of procedure.*

In-flight available oxygen quantity

- 1 - Oxygen pressure ..... Read
- 2 - Outside air temperature (OAT) ..... Read
- 3 - Determine the usable oxygen percent using the chart figure 4.4.2

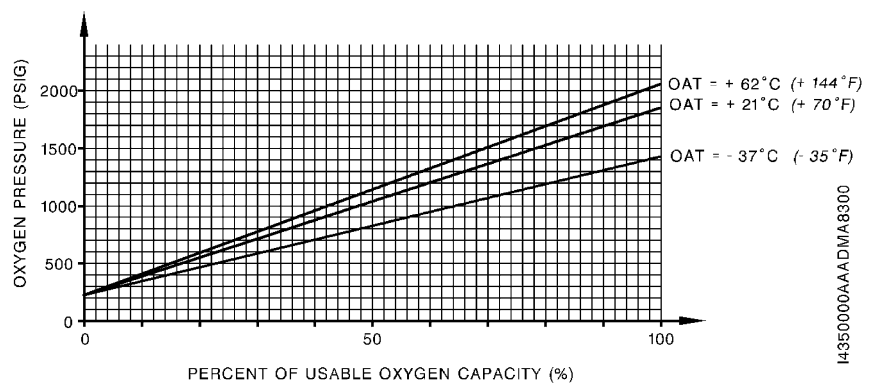


Figure 4.4.2 - Usable oxygen

- 4 - Determine the oxygen duration in minutes by multiplying the values read on table figure 4.4.3 by the percent obtained with the chart figure 4.4.2

| Number of passengers | Duration :<br>Passengers, plus 1 pilot | Duration :<br>Passengers, plus 2 pilots |
|----------------------|--|---|
| 0                    | 226                                    | 113                                     |
| 1                    | 162                                    | 94                                      |
| 2                    | 127                                    | 81                                      |
| 3                    | 104                                    | 71                                      |
| 4                    | 88                                     | 65                                      |

Figure 4.4.3 - Oxygen duration

End of procedure.

Before taxiing

1/4

- 1 - Stand-by instruments ..... Check

Check de-ice system

• NOTE •

Flight into known icing conditions is authorized only when all ice protection equipment are operating correctly. This equipment may be activated before takeoff, even during taxiing, in case of icing conditions on ground. Refer to chapter 4.5 Particular procedures of this section.

•

- 2 - PROP DE ICE switch ..... ON

- 3 - Check illumination of the green light located above the switch.

• NOTE •

Illumination of the green light shows that electric power is supplied to blade root electric resistors. It is advised to wait at least a whole half cycle (90 seconds) to check that both blade heating systems are correctly supplied with electric power.

•

- 4 - PROP DE ICE switch ..... OFF

- 5 - WINDSHIELD switch ..... ON

- 6 - Check illumination of the green lights located above the switch (except if hot conditions).

• NOTE •

These lights may remain OFF if cabin temperature is very high, for example after a prolonged parking in hot conditions (see chapter 7.13 for operational principle).

•

- 7 - WINDSHIELD switch ..... OFF

*Continue ►*

Before taxiing

2/4

► *Continuing*

- 8 - Ng ..... Increase > 80 %  
To check AIRFRAME DE ICE

• NOTE •

Theoretically, necessary air bleed to inflate wing and empennage leading edges, as well as depression necessary to their deflation are sufficient when THROTTLE is positioned on Flight IDLE. However, it is advised for check to choose a Ng power > 80 % in order to obtain operation design pressure, which enables illuminating surely the two green lights and avoiding **VACUUM LOW** untimely alarms.

•

- 9 - AIRFRAME DE ICE switch ..... ON

- 10 - Visually check functioning of deicer boots during 1 total cycle and illumination of the two green lights located above the switch.

• NOTE •

The cycle lasts 67 seconds. Check both inflation impulses and illumination of each corresponding green light :

- the first impulse inflates the external and middle wing boots,
- the second impulse inflates the leading edge boots of empennages and inner wing.

•

- 11 - AIRFRAME DE ICE switch ..... OFF

- 12 - INERT SEP switch ..... ON  
Keep ON while taxiing in order to avoid ingestion of particles by the engine

- 13 - Flight controls ..... Check  
Proper operation from stop to stop, full deflection

Check autopilot and electrical pitch trim :

- 14 - AP / TRIMS ..... Check

• NOTE •

Detailed control procedures of autopilot and electrical pitch trim are described in the GARMIN Integrated Flight Deck Pilot's Guide.

•

*Continue ►*



## Before taxiing

3/4

## ► Continuing

- 15 - Pitch trim ..... UP / DN
  - 16 - Pitch trim ..... Adjust in green range  
Graduated from 12 to 37 %
  - 17 - Yaw trim ..... L / R
  - 18 - Yaw trim ..... Adjust in green range  
Takeoff range
  - 19 - Roll trim ..... L / R
  - 20 - Roll trim ..... Adjust at neutral position
  - 21 - FLAPS lever ..... UP
- Perform MFD flight management
- 22 - Weight computing ..... Set / Check
  - 23 - FOB (fuel on board) synchronization ..... Set
- If requested :*
- 24 - FPL ..... Set
- 25 - LFE selection ..... Done
- Landing Field Elevation selection is done on the touchscreen controller using either :
- automatically the destination airport of the flight plan,
  - or
  - a manual entry by pressing : HOME, AIRCRAFT SYSTEMS, LFE and then MANUAL.
- 26 - VHF/VOR/GPS ..... Adjust / Test
  - 27 - Radar ..... Adjust / Test
  - 28 - Stormscope/TAS/TAWS/Radio altimeter, if installed ..... Adjust / Test
  - 29 - ADI/HSI on PFD1 / PFD2 ..... Check
  - 30 - Altimeter setting ..... Set / Check

Continue ►

Before taxiing

4/4

► *Continuing*

▲ **CAUTION** ▲

During feathering test, keep the spent time with the propeller RPM in the caution (yellow) range at a minimum.



- 31 - THROTTLE ..... FEATHER twice  
Flight IDLE to LO-IDLE then Flight IDLE, twice
- 32 - EIS ..... Check
- 33 - CAS display ..... Check
- 34 - Passenger briefing ..... As required
- 35 - TAXI lights ..... ON
- 36 - PARK BRAKE ..... OFF  
Make sure that chocks are removed, if used
- 37 - **PARK BRAKE** ..... Check OFF

*End of procedure.*

## Taxiing

### ▲ CAUTION ▲

Generator load < 200 amps.



### ▲ CAUTION ▲

Avoid using reverse during taxiing.



### ● NOTE ●

Operation in the Beta ( $\beta$ ) range / reverse is not restricted during ground operations. However, foreign particles (dust, sand, grass, gravel, etc...) may be blown into the air, ingested by the engine (above all if INERT SEP switch is turned OFF) and cause damage to the propeller.



- 1 - TAXI lights ..... Check ON
- 2 - THROTTLE ..... As required

### ● NOTE ●

After initial acceleration, THROTTLE may be in the TAXI range sector, avoiding excessive movements in order to keep a constant ground speed.



- 3 - Brakes ..... Test
- 4 - Nose wheel steering ..... Check  
Check that the control wheel moves (roll) in the same direction  
as the rudder pedals due to the rudder / aileron interconnect.
- 5 - Flight instruments ..... Check  
Check navigation and communication systems  
before or during taxiing, check gyroscopic instruments  
on PFDs 1 / 2 and stand-by indicator during ground turns.

*End of procedure.*

Before line up

1/4

▲ **CAUTION** ▲

Generator load < 200 amps.



- 1 - PARK BRAKE ..... ON
- 2 - **PARK BRAKE** ..... Check ON
- 3 - THROTTLE ..... Flight IDLE  
Ng = 69 % ± 2 %
- 4 - LDG lights ..... ON
- 5 - NAV switch ..... ON
- 6 - STROBE switch ..... ON
- 7 - IGNITION ..... As required  
AUTO or ON
- 8 - AUX BP switch ..... AUTO
- 9 - FUEL SEL switch ..... AUTO

DE ICE SYSTEM panel

- 10 - AIRFRAME DE ICE switch ..... As required
- 11 - PROP DE ICE switch ..... As required
- 12 - WINDSHIELD switch ..... As required
- 13 - PITOT L HTR switch ..... ON
- 14 - PITOT R & STALL HTR switch ..... ON
- 15 - INERT SEP switch ..... ON

*If icing conditions are foreseen :*

- 16 - Perform procedure ..... Flight into known icing conditions  
Refer to chapter 4.5

*Continue ►*

Before line up

2/4

► *Continuing*

*Adjust trims for takeoff*

- 17 - Pitch ..... TO  
Adjust inside green index sector,  
depending on the current balance condition
- 18 - Yaw ..... TO  
Adjust inside green index sector
- 19 - Roll ..... TO  
Adjust at neutral position
- 20 - FLAPS lever ..... TO
- 21 - Flight controls ..... Check  
Check again for proper operation from stop to stop,  
full deflection
- 22 - A/C switch ..... As required
- 23 - BLEED switch ..... AUTO or MAX DIFF
- 24 - LFE ..... Check
- 25 - FUEL gages ..... Check quantity and imbalance

▲ **CAUTION** ▲

Do not take off if battery charge > 50 amps ± 4 amps.



• NOTE •

After starting engine with airplane power, a battery charge above 50 amps is normal. If this indication remains steady at a high value, it may be then a battery or generation system failure. Do not take off in these conditions.



- 26 - AMPS ..... Check below 50 amps
- 27 - **BAT AMP** ..... Check OFF
- 28 - EIS ..... Check

*Continue ►*

Before line up

3/4

► *Continuing*

- 29 - CAS display ..... Check  
All messages OFF,  
except **PARK BRAKE** and, if used **INERT SEP ON**
- 30 - Altimeter setting ..... Set / Check
- 31 - Instruments departure setting ..... Check
- 32 - SID ..... Set
- 33 - ALT SEL ..... Set
- 34 - XPDR ..... Set
- 35 - VHF/VOR/GPS/XPDR ..... Adjust / Check
- 36 - Stormscope/TAS/TAWS/ADF, if installed ..... Adjust / Check
- 37 - Radar ..... Adjust / Check  
On ground, maintain radar on STANDBY  
in order not to generate radiations prejudicial to outside persons.
- 38 - Radio altimeter, if installed ..... Adjust / Check
- 39 - Transponder code ..... Adjust / Check
- 40 - Takeoff distances ..... Check  
Refer to chapter 5.9

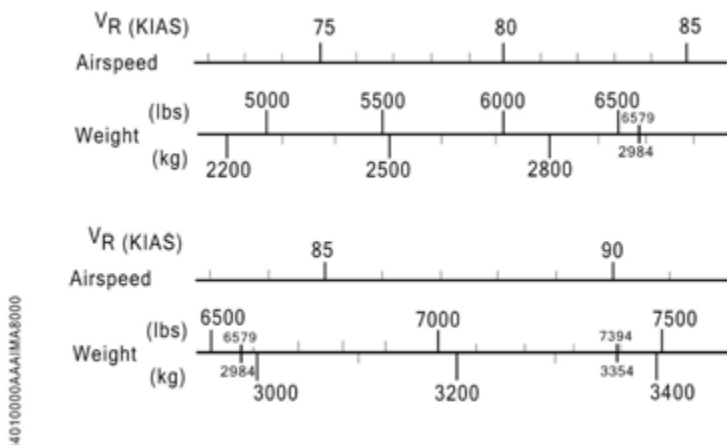
*Continue* ►

Before line up

4/4

► Continuing

41 - Rotation airspeed ( $V_R$ ) ..... Check



42 - Pilot's / Passengers' belts ..... Check

43 - Passengers' table ..... Stowed

44 - Engine instruments ..... Check

All engine parameters must be in green range,  
except propeller RPM, which will be about 1000 RPM or more  
with THROTTLE at Flight IDLE.

45 - PARK BRAKE ..... OFF

46 - **PARK BRAKE** ..... Check OFF

*End of procedure.*

Normal takeoff

1/2

*When lined up, on brakes :*

▲ **CAUTION** ▲

If heavy precipitation, turn IGNITION and INERT SEP switches to ON.  
If icing conditions are foreseen, refer to chapter 4.5, paragraph Flight into known icing conditions.



- 1 - ADI / HSI / headings ..... Check
- 2 - Horizon ..... Check attitude  $\approx + 2^\circ$

● NOTE ●

Horizon has been set so as to indicate a  $2^\circ$  nose up attitude, when airplane center of gravity is at a middle average.



- 3 - HSI - Heading - Stand-by compass ..... Check

● NOTE ●

The indication of the stand-by compass is disturbed when windshield deice systems are activated.



- 4 - OFF/TAXI/LDG switch ..... LDG
- 5 - Engine instruments ..... Check  
ITT in green sector
- 6 - CAS display ..... Check  
All messages OFF, except **IGNITION** and **INERT SEP ON**, if used

- 7 - Apply brakes and increase power.
- 8 - PROP RPM ..... Check green sector
- 9 - Brakes ..... Release
- 10 - TRQ ..... 100 %

● NOTE ●

Torque will be about 40 % to 60 % before brake release. For a normal takeoff, maximum torque (100 %) will be applied after brakes release.



*Continue ►*



Normal takeoff

2/2

► *Continuing*

11 - Rotation airspeed

12 - Attitude ..... 10° Up

*When vertical speed is positive :*

13 - Brakes ..... Apply  
Briefly

14 - LANDING GEAR lever ..... UP  
Airspeed < 150 KIAS

• NOTE •

During the sequence :

- The amber caution light flashes. It indicates that the landing gear pump is running. It goes off when the 3 landing gears are up locked. GEAR UNSAFE red warning light ON and **GEAR UNSAFE** indicate an anomaly (refer to chapter 3.7 Emergency procedures).
- It is possible that the 3 landing gear position green indicator lights flash unevenly then go off at the end of the sequence.

•

15 - GEAR UNSAFE red warning light  
and **GEAR UNSAFE** ..... Check OFF  
At the end of the sequence

*In case of initial climb at Vx :*

▲ WARNING ▲

**It is recommended not to retract FLAPS to UP before 500 ft AGL.**

▲

16 - Airspeed ..... 100 KIAS

*When airspeed above 115 KIAS :*

17 - FLAPS lever ..... UP  
*End of procedure.*

## Short takeoff

1/3

*When lined up, on brakes :*

### ▲ CAUTION ▲

If heavy precipitation, turn IGNITION and INERT SEP switches to ON.  
If icing conditions are foreseen, refer to chapter 4.5, paragraph Flight into known icing conditions.



- 1 - ADI / HSI / headings ..... Check
- 2 - Horizon ..... Check attitude  $\approx + 2^\circ$

#### ● NOTE ●

Horizon has been set so as to indicate a  $2^\circ$  nose up attitude, when airplane center of gravity is at a middle average.



- 3 - HSI - Heading - Stand-by compass ..... Check

#### ● NOTE ●

The indication of the stand-by compass is disturbed when windshield deice systems are activated.



- 4 - OFF/TAXI/LDG switch ..... LDG
- 5 - Engine instruments ..... Check  
ITT in green sector
- 6 - CAS display ..... Check  
All messages OFF, except **IGNITION** and **INERT SEP ON**, if used
- 7 - Apply brakes and increase power.
- 8 - PROP RPM ..... Check green sector
- 9 - TRQ ..... 100 %

*Continue ►*

Short takeoff

2/3

► *Continuing*

10 - Brakes ..... Release

• NOTE •

On short runway, maximum torque will be applied before brakes release.

•

11 - Rotation airspeed

*Weight < 6579 lbs (2984 kg) :*

12 - Attitude ..... 15° Up

*Weight > 6579 lbs (2984 kg) :*

13 - Attitude ..... 12.5° Up

*When vertical speed is positive :*

14 - Brakes ..... Apply  
Briefly

15 - LANDING GEAR lever ..... UP  
Airspeed < 150 KIAS

• NOTE •

During the sequence :

- The amber caution light flashes. It indicates that the landing gear pump is running. It goes off when the 3 landing gears are up locked. GEAR UNSAFE red warning light ON and **GEAR UNSAFE** indicate an anomaly (refer to chapter 3.7 Emergency procedures).
- It is possible that the 3 landing gear position green indicator lights flash unevenly then go off at the end of the sequence.

•

16 - GEAR UNSAFE red warning light  
and **GEAR UNSAFE** ..... Check OFF  
At the end of the sequence

*Continue ►*

Short takeoff

3/3

► *Continuing*

*In case of initial climb at V<sub>x</sub> :*

▲ **WARNING** ▲

**It is recommended not to retract FLAPS to UP before 500 ft AGL.**



17 - Airspeed ..... 100 KIAS

*When airspeed above 115 KIAS :*

18 - FLAPS lever ..... UP

*End of procedure.*

**After takeoff**

- 1 - LANDING GEAR lever ..... Check UP
- 2 - FLAPS lever ..... Check UP
- 3 - TRQ ..... Check 100 % max
- 4 - Climb airspeed ..... 124 KIAS
- 5 - EIS ..... Check
- 6 - CAS display ..... Check
- 7 - DE ICE SYSTEM panel ..... As required
- 8 - INERT SEP switch ..... As required

*End of procedure.*

| Climb | 1/2 |
|-------|-----|
|-------|-----|

- 1 - ALT SEL ..... Check
- 2 - Altimeters setting ..... As required
- 3 - Autopilot ..... Check  
Autopilot status and yaw damper ON

▲ **CAUTION** ▲

Observe TRQ / Ng / Np / ITT / OIL T° and PSI limitations.  
Use optimum torque and / or refer to tables in chapter 5.8.



- 4 - TRQ adjustment / ITT / Ng ..... Check

● NOTE ●

Torque setting during climb must be adjusted according to engine operation tables in chapter 5.8. These tables give the max. climb power torque setting (MXCL). For each engine, when torque is reduced below 100 % at high altitude according to the tables, during the final climb, reaching the maximum permitted Ng (104 %) is possible and the ITT will be approximately constant, giving a particular value of ITT.

For a simplified engine operation during climb, power may be set first of all by torque, using 100 %, then, when the ITT typical value for climb is reached, by indicated ITT, using this particular value. The margin between this indicated ITT and 790°C (recommended ITT limit during continuous operation) will gradually reduce as flight time is performed.



- 5 - Climb airspeed ..... 124 KIAS  
Performance tables concerning climb at 124 KIAS and 170 KIAS / M 0.40 are given in chapter 5.10
- 6 - EIS ..... Check
- 7 - CAS display ..... Check
- 8 - Weather radar ..... As required
- 9 - Pressurization ..... Check

*Continue ►*

Climb

2/2

► *Continuing*

A/C panel

- 10 - TEMP selector ..... Adjust
- 11 - FUEL gages ..... Check  
Verify fuel quantity and imbalance, correct if necessary.
- 12 - AMPS / VOLTS ..... Check

▲ **CAUTION** ▲

If heavy precipitation, turn IGNITION and INERT SEP switches to ON.



- 13 - DE ICE SYSTEM panel ..... As required  
Refer to chapter 4.5
  - 14 - INERT SEP switch ..... As required
  - 15 - LDG lights ..... As required
- End of procedure.*

| Cruise | 1/2 |
|--------|-----|
|--------|-----|

- 1 - Altimeters setting ..... Check
- 2 - Autopilot ..... Check  
Autopilot status and yaw damper ON

▲ **CAUTION** ▲

Observe TRQ / Ng / Np / ITT / OIL T° and PSI limitations.  
Use optimum torque and / or refer to tables in chapter 5.8.



- 3 - TRQ adjustment / ITT / Ng ..... Check  
Adjust according to engine operation tables - chapter 5.8  
or to cruise index on the PFDs

● **NOTE** ●

Engine operation tables (chapter 5.8) give torque to be applied according to  
OAT, in order not to exceed authorized maximum power.  
When INERT SEP switch is OFF, a more accurate setting of torque must then  
be performed according to cruise performance tables presented in  
chapter 5.11.



- 4 - EIS ..... Check
- 5 - CAS display ..... Check
- 6 - Pressurization ..... Check

Regularly check fuel gages for :

- 7 - Consumption
- 8 - Expected fuel at destination
- 9 - Tank automatic change every 5 minutes
- 10 - Imbalance  
Max. imbalance 15 USG

*When the cruise parameters are stabilized, after 4 min minimum :*

- 11 - AMPS / VOLTS ..... Check

*Continue ►*



Cruise

2/2

► *Continuing*

▲ **CAUTION** ▲

If heavy precipitation, turn IGNITION and INERT SEP switches to ON.



- 12 - DE ICE SYSTEM panel ..... As required  
Refer to chapter 4.5
- 13 - INERT SEP switch ..... As required
- 14 - LDG lights ..... As required

*End of procedure.*

**Before descent**

- 1 - Briefing before approach ..... Completed
- 2 - Altimeters settings ..... Check
- 3 - Pressurization ..... Check
- 4 - LFE ..... Check
- 5 - FUEL gages ..... Check  
Check for quantity and imbalance
- 6 - Fullest tank ..... Select
- 7 - AMPS / VOLTS ..... Check

**▲ CAUTION ▲**

If heavy precipitation, turn IGNITION and INERT SEP switches to ON.



- 8 - DE ICE SYSTEM panel ..... As required  
Refer to chapter 4.5
- 9 - Windshield misting protection system ..... As required

*Prior to descent in moist conditions and to avoid canopy misting :*

- 10 - HOT AIR FLOW distributor ..... Set to 12 o'clock position
- 11 - WINDSHIELD switch ..... ON

*If misting continues :*

- 12 - HOT AIR FLOW distributor ..... Turn to the left  
Or refer to chapter 3.11  
paragraph Windshield misting or internal icing
- 13 - INERT SEP switch ..... As required  
*End of procedure.*

## Approach

- 1 - Altimeters settings (QNH) ..... Set / Check
- 2 - Minimums ..... Set / Check
- 3 - COM / NAV / GPS ..... Set / Check
- 4 - Pressurization ..... Check
- 5 - LFE ..... Check
- 6 - FUEL gages ..... Check  
Check for quantity and imbalance
- 7 - Fullest tank ..... Select
- 8 - AMPS / VOLTS ..... Check

### ▲ CAUTION ▲

If heavy precipitation, turn IGNITION and INERT SEP switch to ON.



- 9 - DE ICE SYSTEM panel ..... As required  
Refer to chapter 4.5

- 10 - Windshield misting protection system ..... As required

*Prior to descent in moist conditions and to avoid canopy misting :*

- 11 - HOT AIR FLOW distributor ..... Set to 12 o'clock position
- 12 - WINDSHIELD switch ..... ON

*If misting continues :*

- 13 - HOT AIR FLOW distributor ..... Turn to the left  
Or refer to chapter 3.11  
paragraph Windshield misting or internal icing

- 14 - INERT SEP switch ..... ON

*When below FL 100 :*

- 15 - LDG lights ..... ON
- 16 - Passenger's briefing ..... As required
- 17 - Seats, belts, harnesses ..... Locked
- 18 - Passenger's table ..... Stowed

*End of procedure.*

**Final approach (in GS) or downwind leg (VMC)**

*Long final :*

- 1 - Altimeters ..... Check
- 2 - FUEL gages ..... Check  
Check for quantity and imbalance
- 3 - Fullest tank ..... Select  
Maximum tolerated imbalance is 15 USG

*When below FL 100 :*

- 4 - LDG lights ..... ON
- 5 - INERT SEP switch ..... ON

*When airspeed is below 178 KIAS :*

- 6 - LANDING GEAR lever ..... DN
- 7 - 3 green indicator lights ..... Check ON
- 8 - GEAR UNSAFE red warning light ..... Check OFF
- 9 - **GEAR UNSAFE** ..... Check OFF
- 10 - Amber light ..... Check OFF

• NOTE •

During the sequence :

- The amber caution light flashes. It indicates that the landing gear pump is running. It goes off when the 3 landing gears are down locked. GEAR UNSAFE red warning light ON and **GEAR UNSAFE** indicate an anomaly (refer to chapter 3.7 Emergency procedures).
- It is possible that the 3 landing gear position green indicator lights flash unevenly then come ON at the end of the sequence.

•

- 11 - FLAPS lever ..... TO  
Airspeed < 178 KIAS
- 12 - Radar Mode softkey ..... STANDBY

*End of procedure.*

Short final ( $\approx$  500 ft)

*Stabilized approach*

- 1 - LANDING GEAR lever ..... Check  
DN and 3 green

*When airspeed is below 122 KIAS :*

- 2 - FLAPS lever ..... LDG

• NOTE •

However, when autopilot is engaged, in APR mode, with coupled GS, FLAPS must be extended in landing position before crossing the OUTER MARKER.

•

*Without AP engaged :*

- 3 - Approach airspeed ..... 85 KIAS

*With AP engaged :*

- 4 - Approach airspeed ..... Above 85 KIAS

• NOTE •

This is to avoid any vertical deviation in case of late FLAPS extension to LDG position in short final.

•

- 5 - AP / YD ..... Disconnect  
Before 200 ft

• NOTE •

The pilot effort required to use the rudder pedals is reduced if the yaw damper is turned off. This is particularly significant when landing in a crosswind.

•

*End of procedure.*

Landing

▲ **WARNING** ▲

**Reduce power smoothly.**

**Quickly reducing the power to idle during the flare may induce a pronounced deceleration which may lead to a drop down of the airplane.**



1 - THROTTLE ..... Flight IDLE

● NOTE ●

Avoid three-point landings. Adopt a positive flight attitude in order to touch runway first with main landing gear.



*After wheels touch :*

▲ **CAUTION** ▲

On snowy or dirty runway, it is better not to use reverse below 40 KIAS.



2 - Reverse ..... As required  
Reverse may be applied as soon as the wheels touch the ground

● NOTE ●

To avoid ingestion of foreign objects, come out of the reverse range as speed reduces and use the brakes if necessary for further deceleration.



● NOTE ●

High power reverse at low speed can throw loose material into the air, and can cause control problems and decrease the comfort of crew and passengers. If permitted by the runway length, it is better to adopt a moderate reverse.



3 - Brakes ..... As required

● NOTE ●

It is advised not to brake energetically, as long as speed has not reached 40 KIAS, as otherwise wheels may be locked.



*End of procedure.*

## Go-around with AP OFF

1/2

- 1 - GO AROUND push-button ..... Press  
It provides the moving up of the flight director to + 10°.

*Simultaneously :*

- 2 - THROTTLE ..... T/O power

• NOTE •

The airplane will tend to yaw to the left when power is applied. Right rudder pressure will be required to maintain coordinated straight flight until the rudder trim can be adjusted.

•

- 3 - Attitude ..... 10° Up

- 4 - FLAPS lever ..... TO

>> *Weight below 6579 lbs (2984 kg)*

If airspeed has been maintained at 80 KIAS or more and TRQ 100 %, select flaps to TO position as soon as the 10° Up attitude has been attained.

*When the vertical speed is positive and when airspeed is at or above 85 KIAS :*

- 5 - LANDING GEAR lever ..... UP  
All warning lights OFF

*When airspeed is at or above 110 KIAS :*

- 6 - FLAPS lever ..... UP  
7 - Climb airspeed ..... As required

>> *Weight above 6579 lbs (2984 kg)*

If airspeed has been maintained at 85 KIAS or more and TRQ 100 %, select flaps to TO position as soon as the 10° Up attitude has been attained.

*When the vertical speed is positive and when airspeed is at or above 90 KIAS :*

- 8 - LANDING GEAR lever ..... UP  
All warning lights OFF

*Continue ►*

Go-around with AP OFF

2/2

► *Continuing*

*When airspeed is at or above 115 KIAS :*

9 - FLAPS lever ..... UP

10 - Climb airspeed ..... As required

>> *All*

11 - TRQ ..... As required

*End of procedure.*



# Go-around with AP ON

- 1 - GO AROUND push-button ..... Press  
AP remains ON with the flight director moving up to + 10°.

*Simultaneously :*

- 2 - THROTTLE ..... T/O power  
3 - FLAPS lever ..... TO

>> *Weight below 6579 lbs (2984 kg)*

If airspeed has been maintained at 80 KIAS or more and TRQ 100 %, select flaps to TO position as soon as the 10° Up attitude has been attained.

*When the vertical speed is positive and when airspeed is at or above 85 KIAS :*

- 4 - LANDING GEAR lever ..... UP  
All warning lights OFF

*When airspeed is at or above 110 KIAS :*

- 5 - FLAPS lever ..... UP  
6 - Climb airspeed ..... As required

>> *Weight above 6579 lbs (2984 kg)*

If airspeed has been maintained at 85 KIAS or more and TRQ 100 %, select flaps to TO position as soon as the 10° Up attitude has been attained.

*When the vertical speed is positive and when airspeed is at or above 90 KIAS :*

- 7 - LANDING GEAR lever ..... UP  
All warning lights OFF

*When airspeed is at or above 115 KIAS :*

- 8 - FLAPS lever ..... UP  
9 - Climb airspeed ..... As required

>> *All*

- 10 - TRQ ..... As required

*End of procedure.*

## Touch and go

1/2

*Before wheels touch :*

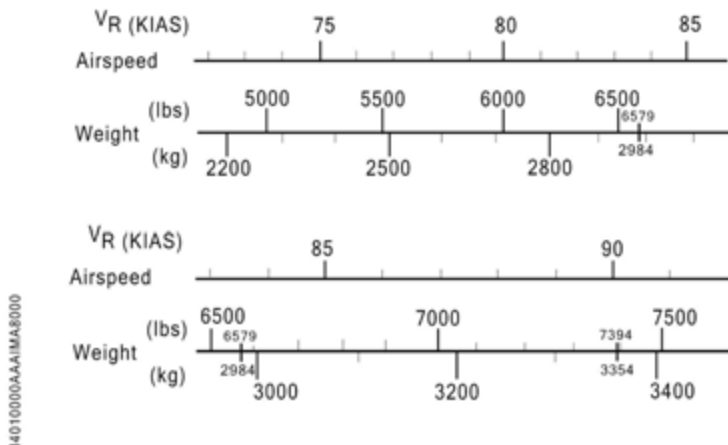
### ▲ WARNING ▲

**Reduce power smoothly.**

**Quickly reducing the power to idle during the flare may induce a pronounced deceleration which may lead to a drop down of the airplane.**



- 1 - Takeoff distances . . . . . Checked  
Refer to chapter 5.9
- 2 - Rotation airspeed ( $V_R$ ) . . . . . Checked



*After wheels touch :*

- 3 - FLAPS lever . . . . . TO
- 4 - Elevator trim . . . . . Green sector  
It is faster to use manual elevator trim control than electric one.  
Ensure that runway length is sufficient to complete this sequence.

*Continue ►*

Touch and go

2/2

► Continuing

▲ **WARNING** ▲

**Check that flaps have well reached the TO position before increasing power. Do not increase power with full flaps, as airplane may lift off prematurely at low speed.**



5 - THROTTLE ..... T/O power

*If normal takeoff :*

6 - Attitude ..... 10° Up

*If short takeoff :*

*Weight < 6579 lbs (2984 kg)*

7 - Attitude ..... 15° Up

*Weight > 6579 lbs (2984 kg)*

8 - Attitude ..... 12°5 Up

● NOTE ●

However, the POH does not supply distances concerning touch and go. These distances are let to pilot's initiative.



*End of procedure.*

## Runway clear

*Runway clear - airplane stopped*

### ▲ CAUTION ▲

Generator load < 200 amps



- 1 - TAXI lights ..... ON
- 2 - NAV switch ..... OFF
- 3 - STROBE switch ..... OFF
- DE ICE SYSTEM panel :
- 4 - AIRFRAME DE ICE switch ..... OFF
- 5 - PROP DE ICE switch ..... OFF
- 6 - WINDSHIELD switch ..... As required
- 7 - PITOT L HTR switch ..... OFF
- 8 - PITOT R & STALL HTR switch ..... OFF
- 9 - INERT SEP switch ..... Check ON
- 10 - Trims ..... Reset to takeoff position
- 11 - FLAPS lever ..... UP
- 12 - A/C switch ..... As required
- 13 - XPDR ..... Check GND
- 14 - WX radar ..... Check

Maintain WX radar on standby in order  
not to generate radiations prejudicial to outside persons.  
The WX radar is automatically set to standby after the touchdown.

*End of procedure.*

## Shutdown

1/3

- 1 - PARK BRAKE ..... Set ON
- 2 - **PARK BRAKE** ..... Check ON
- 3 - EXT LIGHTS panel ..... All OFF
- 4 - INT LIGHTS panel ..... As required
- 5 - OXYGEN switch ..... OFF
- 6 - FUEL SEL switch ..... MAN
- 7 - AP / TRIMS switch ..... OFF
- 8 - A/C switch ..... OFF
- 9 - BLEED switch ..... OFF
- 10 - Check for cabin depressurization ( $\Delta p = 0$  Psi).
- 11 - THROTTLE ..... Flight IDLE  
For 2 min

## • NOTE •

This allows the engine to stabilize at minimum obtainable ITT in order to minimize the likelihood of oil coking in the #3 bearing area.

•

- 12 - THROTTLE ..... LO-IDLE  
For 15 sec

## • NOTE •

Keep THROTTLE on LO-IDLE position for 15 sec minimum before shutting down engine.

•

- 13 - THROTTLE ..... CUT OFF
- 14 - INERT SEP switch ..... OFF
- 15 - Radar Mode Softkey ..... OFF

Continue ►

Shutdown

2/3

► *Continuing*

Fuel system check

16 - **AUX BOOST PMP ON** ..... Check ON

Wait for AUX BP operation, an audible operation of the auxiliary booster pump should be heard, it confirms the proper functioning of the system

17 - AUX BP switch ..... OFF

18 - GENERATOR selector ..... OFF

*When inertial separator is retracted, after approximately 40 sec :*

19 - SOURCE selector ..... OFF

20 - Crash lever ..... Pull down

21 - FUEL TANK SELECTOR ..... OFF

22 - PARK BRAKE ..... As required

▲ **CAUTION** ▲

In case of high OAT [above 35°C (95°F)], it is required to perform 30 sec dry motoring run after shutdown to improve cooling of the bearing cavities and minimize oil coking - refer to procedure Motoring.



Shutdown stand-by instruments

*MD302 normal shutdown procedure :*

- 23 - No pilot action required for normal shutdown. The MD302 will shut down automatically within 60 seconds following electrical power shutdown.

*MD302 manual shutdown procedure :*

● **NOTE** ●

The MD302 can be manually shut down when in the discharge mode to conserve battery power.



- 24 - Press and hold the control knob for approximately 2 seconds.

*Continue* ►

## Shutdown

3/3

► *Continuing*

- 25 - Turn the control knob to select POWER OFF on the menu and press the control knob to shut down the standby attitude module.

*End of procedure.*

Outside check after shutdown

- 1 - Oxygen cylinder (R.H. Karman) ..... Close

• NOTE •

Within 10 minutes following the engine shutdown, check engine oil level.  
Refer to chapter 8.7 Oil level check.

•

*End of procedure.*



## 4.5 - Particular procedures

### • NOTE •

The procedures and procedure elements given in this chapter Particular procedures supplement the normal procedures or complete certain elements of the normal procedures described in chapter(s) 4.3 and/or 4.4.

•

### Flight into known icing conditions

1/5

### ▲ CAUTION ▲

The stall warning system does not function properly in icing conditions and should not be relied upon to provide adequate stall warning in icing conditions and after leaving icing conditions, if ice accretion remains on the airplane.

Moreover, the ESP and USP functions may not be correctly engaged.

▲

### General

Icing conditions exist when the OAT on the ground or in flight is + 5°C or below, and visible moisture in any form is present (clouds, fog with visibility of one mile (1.6 km) or less, rain, snow, sleet or ice crystals).

Icing conditions also exist when the OAT on the ground is + 5°C or below and when operating on ramps, taxiways or runways where surface snow, ice, standing water or slush may be ingested by the engine or freeze on engine or cowlings.

### • NOTE •

Refer to figure 5.5.1 to convert OAT to SAT in flight.

SAT = OAT - 2°C on the ground.

•

Flight into known icing conditions is authorized when all airplane equipment provided for ice protection is operating correctly. This includes :

- Pneumatic deice system for inboard and outboard wing, for stabilizers and for elevator horns.
- Propeller electrical deice system.
- Electrical heating system for both pitots and for the stall warning incidence sensor.

*Continue ►*

Flight into known icing conditions

2/5

► *Continuing*

- Windshield electrical deice system.
- Inertial separator.

Description of deice systems is presented in chapter 7.13.

Ice accumulation thickness is monitored by the pilot on the L.H. wing leading edge.

At night, a leading edge icing inspection light located on the fuselage L.H. side, activated by the ICE LIGHT switch, is provided.

Boots are automatically cycling at the optimum time to assure proper ice removal. Correct operation of the system can be checked observing the corresponding green advisory light illumination at each boot inflation impulse. If correct operation cannot be confirmed, do not enter or leave as soon as possible icing conditions.

Perform emergency procedure Leading edges deicing failure, paragraph 3.11.

**Ice protection procedures**

▲ **CAUTION** ▲

Should conditions require it, apply these directives from beginning of taxi onwards.



*Prior to entering IMC, as a preventive and if OAT < 5°C :*

▲ **CAUTION** ▲

Inertial separator position affects engine parameters, particularly TRQ and ITT. Care must be exercised when operating the inertial separator or when increasing power with the inertial separator ON, to avoid exceeding engine limitations.



- 1 - INERT SEP switch ..... ON
- 2 - IGNITION switch ..... ON

● **NOTE** ●

IGNITION switch may be left ON for a long period.



*Continue ►*

Flight into known icing conditions

3/5

► *Continuing*

- 3 - **INERT SEP ON** ..... Check ON
- 4 - AIRFRAME DE ICE switch ..... ON
- 5 - PROP DE ICE switch ..... ON
- 6 - WINDSHIELD switch ..... ON

• NOTE •

Standby compass indications are altered when windshield deicing system(s) operate(s).

•

*When operating under IMC :*

▲ CAUTION ▲

Inertial separator position affects engine parameters, particularly TRQ and ITT. Care must be exercised when operating the inertial separator or when increasing power with the inertial separator ON, to avoid exceeding engine limitations.

▲

- 7 - INERT SEP switch ..... ON
- 8 - IGNITION switch ..... ON

• NOTE •

IGNITION switch may be left ON for a long period.

•

- 9 - **INERT SEP ON** ..... Check ON
- 10 - AIRFRAME DE ICE switch ..... ON
- 11 - PROP DE ICE switch ..... ON

*Continue ►*

## Flight into known icing conditions

4/5

### ► Continuing

12 - WINDSHIELD switch ..... ON

• NOTE •

Standby compass indications are altered when windshield deicing system(s) operate(s).

•

• NOTE •

The INERT SEP switch must be left ON while the airplane remains in icing conditions.

•

▲ CAUTION ▲

If airplane leaves icing conditions, maintain INERT SEP switch to ON as long as ice thickness on non-deiced visible parts exceeds 15 mm (or ½ in).

▲

This will avoid ice fragments coming from propeller spinner and being ingested by engine.

Procedures for holding, approach and landing in icing conditions :

- Minimum recommended airspeeds are :

|           | Weight               |                      |
|-----------|----------------------|----------------------|
|           | < 6579 lbs (2984 kg) | > 6579 lbs (2984 kg) |
| FLAPS UP  | 130 KIAS             | 135 KIAS             |
| FLAPS TO  | 110 KIAS             | 115 KIAS             |
| FLAPS LDG | 90 KIAS              | 95 KIAS              |

- If there is ice on the unprotected surfaces of the airplane, during flight end phase, conduct holding with the flaps up. Use flaps as required for final approach and landing at minimum airspeeds noted above.

Continue ►

## Flight into known icing conditions

5/5

► *Continuing***Ice accumulation effects**

When ice has accumulated on the unprotected surfaces of the airplane, aerodynamic characteristics may be changed.

Particularly stall airspeeds may increase by up to :

|           |         |
|-----------|---------|
| FLAPS UP  | 20 KIAS |
| FLAPS TO  | 15 KIAS |
| FLAPS LDG | 10 KIAS |

In case of severe or prolonged icing, an ice concretion due to refreezing around the heated stall warning may appear. Above-recommended airspeeds take into account, on one side, the stall airspeed increase due to profile shape deterioration and, on the other side, the weight increase of the iced-up airplane, taking as a basis the airplane maximum weight when not iced-up.

Rate of climb values with ice accumulation on the unprotected surfaces are to be decreased by 10 %.

Cruise airspeeds may be decreased by 10 %, if cruise power is not changed, or more, if cruise power setting should be decreased due to the additional inertial separator limitations (ITT limitation).

Because of the higher landing airspeed, landing distances will be increased. In the landing configuration, using 90 KIAS approach airspeed increases landing distance by 20 % - refer to chapter 5.14 Landing distances.

*End of procedure.*

### Flight into severe icing conditions

The following weather conditions may be conducive to severe in-flight icing :

- Visible rain at temperatures below 0°C ambient air temperature,
- Droplets that splash or splatter on impact at temperatures below 0°C ambient air temperature.

#### Procedures for exiting the severe icing environment

• NOTE •

These procedures are applicable to all flight phases from takeoff to landing.



Monitor the ambient air temperature. While severe icing may form at temperatures as cold as - 18°C, increased vigilance is warranted at temperatures around freezing with visible moisture present. If the visual cues specified in section 2 Limitations for identifying severe icing conditions are observed, accomplish the following :

- 1 - Immediately request priority handling from Air Traffic Control to facilitate a route or an altitude change to exit the severe icing conditions in order to avoid extended exposure to flight conditions more severe than those for which the airplane has been certificated.
- 2 - Avoid abrupt and excessive maneuvering that may exacerbate control difficulties.
- 3 - Do not engage the autopilot.

*If the autopilot is engaged :*

- 4 - Hold the control wheel firmly and disengage the autopilot.

*If an unusual roll response or uncommanded roll control movement is observed :*

- 5 - Angle-of-attack ..... Reduce
- 6 - Do not extend flaps when holding in icing conditions. Operation with flaps extended can result in a reduced wing angle-of-attack, with the possibility of ice forming on the upper surface further aft on the wing than normal, possibly aft of the protected area.

*If the flaps are extended :*

- 7 - Do not retract them until the airframe is clear of ice.
- 8 - Report these weather conditions to Air Traffic Control.

*End of procedure.*

**Flight under heavy precipitations**

- 1 - IGNITION switch ..... ON

• NOTE •

This action is intended, in highly improbable case of an engine flame-out further to an important ingestion, to ensure immediate restarting without action of the pilot.

•

- 2 - INERT SEP switch ..... ON

*End of procedure.*

**Utilization on runways covered with water**

*If takeoff or landing must be performed on a runway covered with water :*

- 1 - IGNITION switch ..... ON

- 2 - INERT SEP switch ..... ON

*End of procedure.*

Utilization on runways covered with melting  
or not tamped snow

1/3

*If required :*

Refer to paragraph Utilization by cold weather and very cold weather.

▲ **CAUTION** ▲

When engine is shut down, do not set the PROP DE ICE switch to ON for more than 10 seconds, damage to the propeller blades could result.



Preflight inspection :

- 1 - Remove any snow or ice from the wings, stabilizers and movable surfaces, landing gear wells and gear doors, as well as flap tracks, actuators and their fairings.
- 2 - Spray anti-icing fluid on the wings, stabilizers and movable surfaces (upper and lower surfaces) and in the landing gear wells, shortly before takeoff.

Taxiing :

- 3 - INERT SEP switch ..... ON
- 4 - **INERT SEP ON** ..... Check ON
- 5 - FLAPS lever ..... UP
- 6 - Taxi airspeed ..... Max. 5 KIAS
- 7 - Brakes ..... Apply occasionally  
To maintain the brake pads warm, this will prevent  
any subsequent locking due to freezing after takeoff

Before line up :

*If the runway is long enough :*

- 8 - FLAPS lever ..... UP

*Continue ►*



Utilization on runways covered with melting  
or not tamped snow

2/3

► *Continuing*

9 - Rotation airspeed ..... Increased by 5 KIAS

• NOTE •

Takeoff distances must be increased to take into account the flap position  
(+ 15 % compared to the takeoff position) and the runway condition.  
The ground roll may be multiplied by 3 in some melting or not tamped snow  
cases.

•

10 - IGNITION switch ..... ON

11 - INERT SEP switch ..... ON

12 - **INERT SEP ON** ..... Check ON

Takeoff :

*During takeoff run :*

13 - Lightly lift up nose wheel

In order to reduce the forward resistance due to snow  
accumulation against the wheel.

*After takeoff :*

14 - Normally retract the landing gear, then perform a complete cycle  
(extension / retraction) at IAS < 150 KIAS.

Before landing :

15 - IGNITION switch ..... ON

16 - INERT SEP switch ..... ON

17 - **INERT SEP ON** ..... Check ON

*Continue ►*

Utilization on runways covered with melting  
or not tamped snow

3/3

► *Continuing*

Touch and Go :

**▲ WARNING ▲**  
**Touch and Go is prohibited.**

On the ramp, after landing or taxiing :

- 18 - Do not use the parking brake to prevent brake lock.
- 19 - Use chocks and / or tie-down the airplane.

*End of procedure.*

Utilization on icy or covered with tamped snow runways 1/2

If required :

Refer to paragraph Utilization by cold weather and very cold weather.

▲ **CAUTION** ▲

When engine is shut down, do not set the PROP DE ICE switch to ON for more than 10 seconds, damage to the propeller blades could result.



Preflight inspection

- 1 - Remove any snow or ice from the wings, stabilizers and movable surfaces, landing gear wells and gear doors, as well as flap tracks, actuators and their fairings.
- 2 - Spray anti-icing fluid on the wings, stabilizers and movable surfaces (upper and lower surfaces), shortly before takeoff.

Taxiing :

- 3 - INERT SEP switch ..... ON
- 4 - **INERT SEP ON** ..... Check ON
- 5 - Taxi airspeed ..... Max. 5 KIAS  
Use β area of THROTTLE to adjust airspeed  
Apply very smooth variations using THROTTLE
- 6 - Steer the airplane using the rudder.

● NOTE ●

Make turns at a very low airspeed, engine torque tends to make the airplane turn to the left.



- 7 - Use brakes only at very low airspeed and progressively.

Before line up :

- 8 - IGNITION switch ..... ON
- 9 - INERT SEP switch ..... ON
- 10 - **INERT SEP ON** ..... Check ON

Continue ►

Utilization on icy or covered with tamped snow runways 2/2

► *Continuing*

Takeoff :

- 11 - After takeoff, normally retract the landing gear, then perform a complete cycle (extension / retraction) at IAS < 150 KIAS.

Before landing :

- 12 - IGNITION switch ..... ON  
13 - INERT SEP switch ..... ON  
14 - **INERT SEP ON** ..... Check ON

Landing :

*After wheels touch*

- 15 - Use reverse only if necessary and very progressively by monitoring the airplane behaviour.

• NOTE •

The engine torque tends to make the airplane turn to the left.



- 16 - Taxi airspeed ..... Max. 5 KIAS  
Use  $\beta$  area of THROTTLE to adjust airspeed  
Apply very smooth variations using THROTTLE

- 17 - Steer the airplane using the rudder.

• NOTE •

Make turns at a very low airspeed, engine torque tends to make the airplane turn to the left.



- 18 - Use brakes only at very low airspeed and progressively.

On the ramp, after landing or taxiing :

- 19 - Do not use the parking brake to prevent brake lock.  
20 - Use chocks and / or tie-down the airplane.

*End of procedure.*

Utilization by cold weather (- 0°C to - 25°C) and very cold weather (- 25°C to - 40°C)

• NOTE •

The procedure hereafter supplements the normal procedures for the airplane use when operating under temperatures between 0°C and - 40°C on ground.

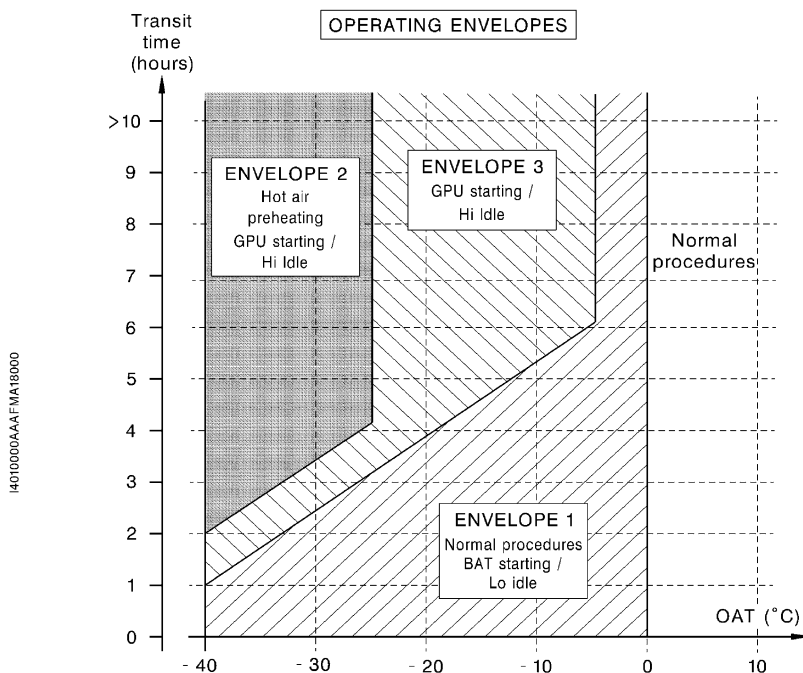


Figure 4.5.1 - Operating envelopes by cold weather (- 0°C to - 25°C) and very cold weather (- 25°C to - 40°C)

*End of procedure.*

Utilization by cold weather (- 0°C to - 25°C) and very cold  
weather (- 25°C to - 40°C) - Envelope 1 1/3

• NOTE •

The procedure hereafter supplements the normal procedures for the airplane  
use when operating in the Envelope 1 defined in figure 4.5.1.

•

Preflight inspection :

- 1 - Remove any snow or ice from the wings, stabilizers and movable surfaces.

According to the condition of runways and taxiways

- 2 - Perform procedure ..... Utilization on runways covered with  
melting or not tamped snow  
Refer to chapter 4.5

or

- 3 - Perform procedure ..... Utilization on icy or covered with  
tamped snow runways  
Refer to chapter 4.5

- 4 - Carry out a complete rotation of the propeller to check its free rotation.
- 5 - Do not perform a fuel draining. If the airplane is operating permanently  
under negative temperatures, drainings will have to be performed once a  
week after having parked the airplane in a heated hangar.
- 6 - Remove chocks and / or release ties from the airplane.
- 7 - Check the free deflection of the flight controls and of the elevator trim.
- 8 - Check the free deflection of THROTTLE.

Before starting engine / Engine start / After engine start :

- 9 - Perform normal procedures defined in chapter(s) 4.3 and / or 4.4.

Before taxiing / Taxiing / Before line up / Takeoff :

DE ICE SYSTEM panel

- 10 - INERT SEP switch ..... ON

- 11 - **INERT SEP ON** ..... Check ON

Continue ►

Utilization by cold weather (- 0°C to - 25°C) and very cold  
weather (- 25°C to - 40°C) - Envelope 1 2/3

► *Continuing*

- 12 - PITOT L HTR switch ..... ON
- 13 - PITOT R & STALL HTR switch ..... ON
- 14 - PROP DE ICE switch ..... ON
- 15 - Perform normal procedures defined in chapter(s) 4.3 and / or 4.4.  
According to the condition of runways and taxiways
- 16 - Perform procedure ..... Utilization on runways covered with  
melting or not tamped snow  
Refer to chapter 4.5
- or
- 17 - Perform procedure ..... Utilization on icy or covered with  
tamped snow runways  
Refer to chapter 4.5

Landing / After landing :

- 18 - Perform normal procedures defined in chapter(s) 4.3 and / or 4.4.  
According to the condition of runways and taxiways
- 19 - Perform procedure ..... Utilization on runways covered with  
melting or not tamped snow  
Refer to chapter 4.5
- or
- 20 - Perform procedure ..... Utilization on icy or covered with  
tamped snow runways  
Refer to chapter 4.5

Shutdown :

- 21 - PARK BRAKE ..... OFF

*Continue ►*

Utilization by cold weather (- 0°C to - 25°C) and very cold  
weather (- 25°C to - 40°C) - Envelope 1 3/3

► *Continuing*

22 - **PARK BRAKE** ..... Check OFF

• NOTE •

It is recommended not to use the parking brake by cold or very cold weather, so  
that the brakes do not stick when cooling.

•

- 23 - Perform normal procedures defined in chapter(s) 4.3 and / or 4.4.
- 24 - Use chocks and / or tie-down the airplane using anchor points on ground.
- 25 - Put blanking caps and plugs on air inlets, exhaust stubs, pitots and static ports.

*End of procedure.*



Utilization by cold weather (- 0°C to - 25°C) and very cold  
weather (- 25°C to - 40°C) - Envelope 2 1/5

• NOTE •

The procedures hereafter supplement or replace the normal procedures for the airplane use when operating in the Envelope 2 defined in figure 4.5.1.

•

Preflight inspection :

- 1 - Preheat the engine and the cabin.

• NOTE •

Preheating during at least 30 minutes is necessary using a heater (70°C mini). Hot air pipes must be installed in the air inlet, on engine rear table by opening the upper cowling and in the cabin by half-opening the door.

•

- 2 - Remove any snow or ice from the wings, stabilizers and movable surfaces.

According to the condition of runways and taxiways

- 3 - Perform procedure ..... Utilization on runways covered with melting or not tamped snow  
Refer to chapter 4.5

or

- 4 - Perform procedure ..... Utilization on icy or covered with tamped snow runways  
Refer to chapter 4.5

- 5 - Spray anti-icing fluid on the wings, stabilizers and movable surfaces (upper and lower surfaces), shortly before takeoff.
- 6 - Carry out a complete rotation of the propeller to check its free rotation.
- 7 - Do not perform a fuel draining. If the airplane is operating permanently under negative temperatures, drainings will have to be performed once a week after having parked the airplane in a heated hangar.
- 8 - Remove chocks and / or release ties from the airplane.
- 9 - Check the free deflection of the flight controls and of the elevator trim.
- 10 - Check the free deflection of THROTTLE.

*Continue ►*

Utilization by cold weather (- 0°C to - 25°C) and very cold  
weather (- 25°C to - 40°C) - Envelope 2 2/5

► *Continuing*

11 - IGNITION switch ..... ON during 30 seconds

12 - **IGNITION** ..... Check ON

Then :

13 - IGNITION switch ..... AUTO

14 - **IGNITION** ..... Check OFF

• NOTE •

This enables to preheat spark igniters before starting the engine.

•

Before starting the engine :

15 - Perform normal procedures defined in Chapter(s) 4.3 and / or 4.4.

Engine start :

▲ CAUTION ▲

The starting must be mandatorily performed using an external power  
source (GPU).

▲

16 - Ground power unit ..... Connected

17 - SOURCE selector ..... GPU

18 - **GPU DOOR** ..... Check ON

19 - Battery and ESS. bus VOLTS ..... Check voltage ≈ 28 Volts  
On EIS of MFD

Engine controls

20 - MAN OVRD control ..... Full backward (notched)

*Continue ►*

Utilization by cold weather (- 0°C to - 25°C) and very cold  
weather (- 25°C to - 40°C) - Envelope 2 3/5

► Continuing

▲ CAUTION ▲

When the engine is shut down, the THROTTLE must not be moved into the reverse area.



21 - THROTTLE ..... CUT OFF

FUEL panel

22 - AUX BP switch ..... ON

23 - **AUX BOOST PMP ON** ..... Check ON

24 - **FUEL PRESS** ..... Check OFF

25 - Propeller area ..... Clear

ENGINE START panel

26 - IGNITION switch ..... ON

27 - **IGNITION** ..... Check ON

28 - STARTER switch ..... ON  
2 sec then OFF

Simultaneously :

29 - Timer ..... Start

30 - **STARTER** ..... Check ON

When Ng ≈ 13 % :

31 - THROTTLE ..... HI-IDLE  
Move directly THROTTLE to HI-IDLE

● NOTE ●

The more the temperature is low, the more the selector is hard to move.  
Starter limits and checks of starting sequence are unchanged.



Continue ►

Utilization by cold weather (- 0°C to - 25°C) and very cold weather (- 25°C to - 40°C) - Envelope 2 4/5

► Continuing

When Ng > 50% :

▲ CAUTION ▲

If the starter does not go off automatically, disengage it using the ABORT position of the STARTER switch.



- 32 - Starter ..... Check OFF automatically
- 33 - **STARTER** ..... Check OFF
- 34 - Engine parameters ..... Check  
Ng = 70 % ± 2 %  
Oil pressure and ITT in green sector
- 35 - SOURCE selector ..... BATT
- 36 - **BAT OFF** ..... Check OFF
- 37 - IGNITION switch ..... AUTO
- 38 - **IGNITION** ..... Check OFF
- 39 - Ground power unit ..... Disconnect
- 40 - GPU door ..... Close
- 41 - **GPU DOOR** ..... Check OFF
- FUEL panel
  - 42 - AUX BP switch ..... AUTO
  - 43 - **AUX BOOST PMP ON** ..... Check OFF
- 44 - GENERATOR selector ..... MAIN
- 45 - **MAIN GEN** ..... Check OFF  
Reset if necessary

Continue ►

Utilization by cold weather (- 0°C to - 25°C) and very cold  
weather (- 25°C to - 40°C) - Envelope 2 5/5

► *Continuing*

■ After engine start :

*As soon as the current flow is lower than 100 A :*

A/C panel

46 - A/C switch ..... PILOT

■ 47 - TEMP selector ..... Max warm

48 - FAN speed selector ..... 0

PRESSURIZATION panel

49 - BLEED switch ..... AUTO or MAX DIFF

*As soon as the oil temperature is greater than 0°C :*

50 - THROTTLE ..... FEATHER twice  
Flight IDLE to LO-IDLE, then Flight IDLE twice

51 - Perform normal procedures defined in chapter(s) 4.3 and / or 4.4.

■ Before taxiing / Taxiing / Before line up / Takeoff :

52 - Perform procedures defined for Envelope 1.

Landing / After landing / Shutdown :

53 - Apply procedures defined for Envelope 1.

*End of procedure.*

Utilization by cold weather (- 0°C to - 25°C) and very cold weather (- 25°C to - 40°C) - Envelope 3 1/2

• NOTE •

The procedures defined for the Envelope 2 are also applicable for the Envelope 3. However it is possible to start the engine using GPU without preheating of the engine and the cabin with a heater. In that case follow the procedure hereafter.

•

■ Preflight inspection / Before starting the engine / Engine start :

- 1 - Apply the procedures defined for the Envelope 2.

■ After engine start :

*As soon as the current flow is lower than 100 A :*

A/C panel

- 2 - A/C switch ..... PILOT

■ 3 - TEMP selector ..... Max warm

- 4 - FAN airspeed selector ..... 0

PRESSURIZATION panel

- 5 - BLEED switch ..... AUTO or MAX DIFF

6 - Preheat the cabin respecting time defined in figure 4.5.2.

Before switching on the navigation and monitoring systems. This allows to respect minimum temperatures necessary for the equipment operation.

*As soon as the oil temperature is greater than 0°C :*

- 7 - THROTTLE ..... FEATHER twice  
Flight IDLE to LO-IDLE, then Flight IDLE twice

- 8 - Perform normal procedures defined in chapter(s) 4.3 and / or 4.4.

■ Taxiing / Before line up / Takeoff /

- 9 - Perform procedures defined for Envelope 1.

Continue ►

Utilization by cold weather (- 0°C to - 25°C) and very cold weather (- 25°C to - 40°C) - Envelope 3

2/2

► Continuing

Landing / After landing / Shutdown /

10 - Perform procedures defined for Envelope 1.

• NOTE •

If landing is foreseen by cold or very cold weather, or in case of prolonged operation of the airplane in such conditions, it is recommended to prepare the airplane as specified in chapter 8.10.

•

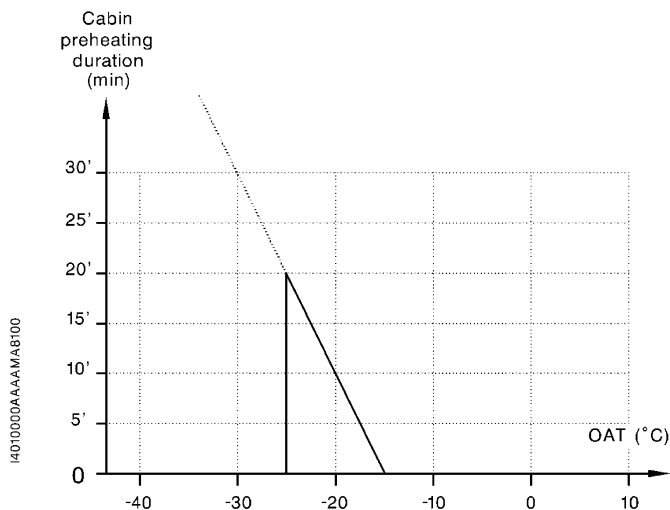


Figure 4.5.2 - Preheating duration

End of procedure.

**Landing procedure with strong headwind or crosswind 1/2**

*If landing must be performed with strong headwind or crosswind :*

- 1 - Increase approach airspeed by the greatest of these 2 following values :

$$\Delta V = \frac{(\text{wind down} - 10)}{2} \text{ (Ex. wind down} = 30 \text{ kt i.e. } \Delta V = 10 \text{ kt)}$$

The wind down is the longitudinal component of the wind.

- Gust amplitude

- 2 - FLAPS lever ..... LDG

• NOTE •

It is not desirable to adopt configuration with flaps in TO position. Lateral control is not improved, and flare phase is lengthened in time and in distance, with increase of piloting difficulties and landing performance.

•

*During approach with crosswind :*

▲ CAUTION ▲

Do not use or select the fuel tank on the low wing side during prolonged sideslips with a fuel low warning or gage indicating low.

▲

▲ CAUTION ▲

Maximum time for sideslip condition is 30 seconds.

▲

- 3 - Maintain airplane in drift correction at the latest until the beginning of flare.

*In short final, on a short runway :*

- 4 - Use normal approach airspeed ..... IAS = 80 KIAS

- 5 - FLAPS lever ..... LDG

To avoid an excessive airspeed

• NOTE •

In this case, landing distance indicated in chapter 5.14, would not be respected.

•

*Continue ►*



**Landing procedure with strong headwind or crosswind 2/2**► *Continuing**Before touch-down :*

- 6 - Generate a slideslip with the rudder in order to align fuselage with the runway (ie left crosswind, left wing low).

*Immediately after landing :***▲ CAUTION ▲**

Do not try to stabilize the airplane by pushing down the elevator control just after the touch ; this operation may provide pitch oscillations while increasing the yaw movement to the wind.

Do not deflect ailerons into wind while taxiing. This will raise spoilers and have a detrimental effect. A good solution is to maintain ailerons to neutral position during taxiing after landing and taxiing before takeoff.



- 7 - FLAPS lever ..... UP

**● NOTE ●**

Flaps travel is slow and will not have an appreciable effect on landing performance.



Maximum demonstrated crosswind for landing is 20 kt.

The most restrictive situation is as follows :

- takeoff with wind coming from the left,
- wet runway,
- aft C.G.

*End of procedure.*

Utilization on grass runway

1/2

▲ CAUTION ▲

The small wheels of the airplane and its weight may lead it to sink in soaked or soft ground.



Before planning the landing, ensure that the field is hard, smooth and dry enough. Landing and moreover takeoff shall not begin if any doubt exists about the condition of such a runway.

**Particular directives**

Taxi / Takeoff :

1 - INERT SEP switch ..... ON

2 - **INERT SEP ON** ..... Check ON

► Do not use the reverse ◀

● NOTE ●

In fact, on a flat runway with grass, it is necessary to adopt a power greater than the one obtained when the THROTTLE is set to Flight IDLE, so the pilot will not be tempted to use the reverse.



*End of procedure* ■

Landing :

3 - INERT SEP switch ..... ON

4 - **INERT SEP ON** ..... Check ON

*After wheels touch down :*

5 - Reverse ..... Only if necessary

▲ CAUTION ▲

Do not maintain reverse at airspeeds below 40 KIAS to avoid ingestion of foreign matter.



*Continue* ►

## Utilization on grass runway

2/2

► *Continuing*

## • NOTE •

Under 40 KIAS, using the reverse makes a cloud of solid particles (dusts, sand, gravels, cut grass, ...) appear around the front face of the airplane. This will damage the propeller and, after ingestion, the engine internal components (compressor and turbine blades).

•

*End of procedure.*

## GPS navigation

### Set up conditions

- 1 - Verify if the data base is current.
- 2 - Verify that altitude data is valid for the GPS prior to flight.

Check the systems availability requirements in the table 2.6.1 in section 2 / GNSS (GPS/SBAS) navigation equipment approvals, depending on the planned navigation performance.

### GPS flight plan

In the active flight plan, addition of a STAR or an approach is always made at the end of the flight plan. In the scope of these additions, the pilot must pay attention not to duplicate points.

*End of procedure.*

## Section 5

### Performance

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|      |   |  |         |
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## **5.1 - General**

This section provides all of the required and additional performance data for airplane operations.

The section 9, Supplements of the POH, provides specific airplane performance associated with optional equipment and systems.

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## 5.2 - Noise level

|  | Maximum noise level<br>permitted | Demonstrated noise<br>level |
|--|----------------------------------|-----------------------------|
| FAR PART 36,<br>Appendix G - Amdt 28                                     | 88 dB(A)                         | 76.4 dB(A)                  |
| ICAO, Annex 16,<br>Vol. 1, 6th edition, Amdt 8<br>Chapter 10, Appendix 6 | 85 dB(A)                         | 76.4 dB(A)                  |

Approved noise levels for TBM airplane are stated in EASA.A.010 Type Certificate Data Sheet.

● NOTE ●

No determination has been made by the Federal Aviation Administration that the noise levels of this airplane are or should be acceptable or unacceptable for operation at, into or out of any airport.

●

Intentionally left blank

### 5.3 - Airspeed calibration

• NOTE •

Indicated airspeeds (IAS) : instrument error supposed to be null (power configuration for cruise condition flight).

•

| Flaps UP<br>LDG GR UP |         | Flaps TO<br>LDG GR DN |         | Flaps LDG<br>LDG GR DN |         |
|-----------------------|---------|-----------------------|---------|------------------------|---------|
| KIAS                  | KCAS    | KIAS                  | KCAS    | KIAS                   | KCAS    |
| 125                   | 128     | 70                    | 69      | 60                     | 58      |
| 150                   | 154     | 80                    | 80      | 70                     | 68      |
| 175                   | 179     | 90                    | 90      | 80                     | 78      |
| 200                   | 205     | 100                   | 101     | 90                     | 88      |
| 225                   | 230     | 120                   | 121     | 100                    | 98      |
| 250                   | 255     | 140                   | 141     | 110                    | 108     |
| 266                   | 271     | 160                   | 162     | 120                    | 118     |
| MPH IAS               | MPH CAS | MPH IAS               | MPH CAS | MPH IAS                | MPH CAS |
| 144                   | 147     | 81                    | 79      | 69                     | 67      |
| 173                   | 177     | 92                    | 92      | 81                     | 78      |
| 201                   | 206     | 104                   | 104     | 92                     | 90      |
| 230                   | 236     | 115                   | 116     | 104                    | 101     |
| 259                   | 264     | 138                   | 139     | 115                    | 113     |
| 288                   | 293     | 161                   | 162     | 127                    | 124     |
| 307                   | 312     | 184                   | 187     | 138                    | 136     |

Figure 5.3.1 - Normal static source

| Flaps UP<br>LDG GR UP |         | Flaps TO<br>LDG GR DN |         | Flaps LDG<br>LDG GR DN |         |
|-----------------------|---------|-----------------------|---------|------------------------|---------|
| KIAS                  | KCAS    | KIAS                  | KCAS    | KIAS                   | KCAS    |
| 125                   | 124     | 70                    | 70      | 60                     | 59      |
| 150                   | 149     | 80                    | 80      | 70                     | 69      |
| 175                   | 174     | 90                    | 90      | 80                     | 79      |
| 200                   | 199     | 100                   | 100     | 90                     | 90      |
| 225                   | 224     | 120                   | 120     | 100                    | 100     |
| 250                   | 249     | 140                   | 139     | 110                    | 110     |
| 271                   | 270     | 160                   | 159     | 120                    | 120     |
| MPH IAS               | MPH CAS | MPH IAS               | MPH CAS | MPH IAS                | MPH CAS |
| 144                   | 142     | 81                    | 81      | 69                     | 68      |
| 173                   | 171     | 92                    | 92      | 81                     | 79      |
| 201                   | 200     | 104                   | 104     | 92                     | 91      |
| 230                   | 229     | 115                   | 115     | 104                    | 104     |
| 259                   | 258     | 138                   | 138     | 115                    | 115     |
| 288                   | 287     | 161                   | 160     | 127                    | 127     |
| 312                   | 311     | 184                   | 183     | 138                    | 138     |

Figure 5.3.2 - Alternate static source (Bleed auto)

## 5.4 - Cabin pressurization envelope

### • NOTE •

The cabin pressurization envelope below characterizes the cabin altitude that could be sustained by the fuselage at different flight levels. The curve shows the minimum cabin altitude as a function of flight level, corresponding to the maximum differential pressure. The maximum differential pressure is limited by the pressurization system protection function.

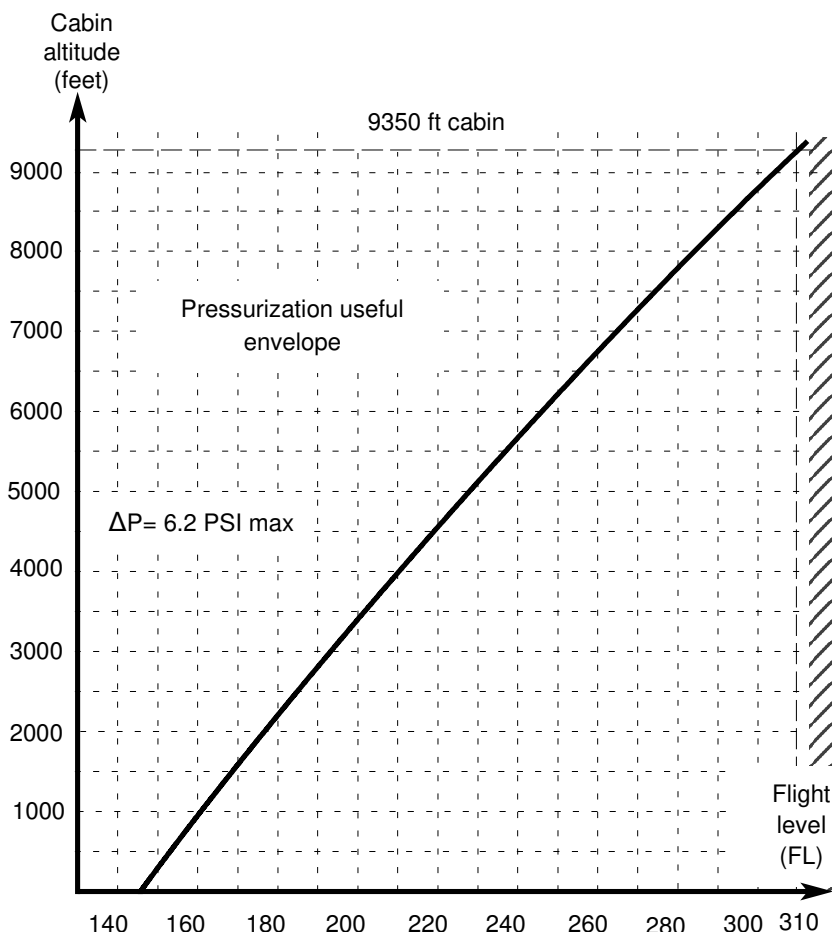


Figure 5.4.1 - Cabin pressurization envelope

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## 5.5 - SAT - OAT conversions

### • NOTE •

These indicated temperatures are available for stabilized cruise at normal operating power.

| Pressure altitude (feet) | ISA - 20°C |      | ISA - 10°C |      | ISA  |      | ISA + 10°C |      | ISA + 20°C |      |
|--------------------------|------------|------|------------|------|------|------|------------|------|------------|------|
|                          | SAT        | OAT  | SAT        | OAT  | SAT  | OAT  | SAT        | OAT  | SAT        | OAT  |
| SL                       | - 05       | - 04 | 05         | 06   | 15   | 16   | 25         | 26   | 35         | 36   |
| 2000                     | - 09       | - 08 | 01         | 02   | 11   | 12   | 21         | 22   | 31         | 32   |
| 4000                     | - 13       | - 12 | - 03       | - 02 | 07   | 08   | 17         | 18   | 27         | 28   |
| 6000                     | - 17       | - 16 | - 07       | - 06 | 03   | 04   | 13         | 14   | 23         | 24   |
| 8000                     | - 21       | - 20 | - 11       | - 10 | - 01 | 00   | 09         | 10   | 19         | 20   |
| 10000                    | - 25       | - 24 | - 15       | - 14 | - 05 | - 04 | 05         | 06   | 15         | 16   |
| 12000                    | - 29       | - 28 | - 19       | - 18 | - 09 | - 08 | 01         | 02   | 11         | 12   |
| 14000                    | - 33       | - 32 | - 23       | - 22 | - 13 | - 12 | - 03       | - 02 | 07         | 08   |
| 16000                    | - 37       | - 36 | - 27       | - 26 | - 17 | - 16 | - 07       | - 06 | 03         | 04   |
| 18000                    | - 41       | - 40 | - 31       | - 30 | - 21 | - 20 | - 11       | - 10 | - 01       | 00   |
| 20000                    | - 45       | - 44 | - 35       | - 34 | - 25 | - 24 | - 15       | - 14 | - 05       | - 04 |
| 22000                    | - 49       | - 48 | - 39       | - 38 | - 29 | - 28 | - 19       | - 18 | - 09       | - 08 |
| 24000                    | - 53       | - 52 | - 43       | - 42 | - 33 | - 32 | - 23       | - 22 | - 13       | - 12 |
| 26000                    | - 57       | - 56 | - 47       | - 46 | - 37 | - 36 | - 27       | - 26 | - 17       | - 16 |
| 28000                    | - 61       | - 60 | - 51       | - 50 | - 41 | - 40 | - 31       | - 30 | - 21       | - 20 |
| 30000                    | - 65       | - 64 | - 55       | - 54 | - 45 | - 44 | - 35       | - 34 | - 25       | - 24 |
| 31000                    | - 67       | - 66 | - 57       | - 56 | - 47 | - 46 | - 37       | - 36 | - 27       | - 26 |

Figure 5.5.1 - SAT - OAT conversions

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## 5.6 - Stall speeds

| Airplane weight       | Config.     |       | Bank |      |         |      |      |         |      |      |         |      |      |         |
|-----------------------|-------------|-------|------|------|---------|------|------|---------|------|------|---------|------|------|---------|
|                       | Flight idle |       | 0°   |      |         | 30°  |      |         | 45°  |      |         | 60°  |      |         |
|                       | LDG GR      | Flaps | KIAS | KCAS | MPH IAS | KIAS | KCAS | MPH IAS | KIAS | KCAS | MPH IAS | KIAS | KCAS | MPH IAS |
| 4850 lbs<br>(2200 kg) | UP          | UP    | 65   | 66   | 75      | 70   | 71   | 81      | 78   | 79   | 90      | 91   | 93   | 105     |
|                       | DN          | TO    | 62   | 63   | 71      | 67   | 68   | 77      | 73   | 75   | 84      | 87   | 89   | 100     |
|                       | DN          | LDG   | 53   | 53   | 61      | 57   | 57   | 66      | 63   | 63   | 73      | 75   | 75   | 86      |
| 5512 lbs<br>(2500 kg) | UP          | UP    | 70   | 71   | 81      | 75   | 76   | 86      | 82   | 84   | 94      | 98   | 100  | 113     |
|                       | DN          | TO    | 66   | 67   | 76      | 71   | 72   | 82      | 78   | 80   | 90      | 93   | 95   | 107     |
|                       | DN          | LDG   | 57   | 57   | 66      | 61   | 61   | 70      | 68   | 68   | 78      | 81   | 81   | 93      |
| 6579 lbs<br>(2984 kg) | UP          | UP    | 75   | 76   | 86      | 80   | 82   | 92      | 88   | 90   | 101     | 105  | 107  | 121     |
|                       | DN          | TO    | 71   | 72   | 82      | 75   | 77   | 86      | 84   | 86   | 97      | 100  | 102  | 115     |
|                       | DN          | LDG   | 61   | 61   | 70      | 66   | 66   | 76      | 73   | 73   | 84      | 86   | 86   | 99      |
| 7394 lbs<br>(3354 kg) | UP          | UP    | 81   | 83   | 93      | 88   | 89   | 101     | 97   | 99   | 112     | 119  | 117  | 137     |
|                       | DN          | TO    | 77   | 77   | 89      | 81   | 83   | 93      | 91   | 92   | 105     | 108  | 109  | 124     |
|                       | DN          | LDG   | 65   | 65   | 75      | 69   | 70   | 79      | 76   | 77   | 88      | 92   | 92   | 106     |

Figure 5.6.1 - Stall speeds

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Figure 5.7.1 - Wind components

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## Pilot's Operating Handbook

**5.8 - Engine operation**

The following tables or/and the optimum torque indicator must be used during normal operation of the airplane.

**▲ CAUTION ▲**

It is the responsibility of the operator to make sure that the required version of GARMIN system software is installed prior to using the hereafter engine operation tables.

The GARMIN system software required for this revision of the engine operation tables is the version 0719.14 or later.

This information is displayed on the MFD power-up page upon system start.

**▲ CAUTION ▲**

The TRQ setting must never exceed 100 %.  
When setting TRQ, Ng must never exceed 104 %.




The following conditions are given for all the tables (pages 5.8.4 to 5.8.11) :

**● NOTE ●**

Inertial separator must be OFF and BLEED HI msg OFF.



- Landing gear and flaps UP.
- BLEED switch on AUTO.
-  represent the ISA conditions at the flight level.

The torque must be set at or below the value corresponding to the local conditions of flight level and temperature.

**● NOTE ●**

The engine ITT limit at 840°C during continuous operation may be used in case of operational need.



Example, for conditions :

- FL = 260
- OAT = - 22°C

the following tables give the maximum torque to be set.

**Maximum climb power**

TRQ setting = 83 % for IAS = 124 KIAS (Add 0.5 % of TRQ for each additional 10 KIAS on climb airspeed), cf. tables figures 5.8.1 and 5.8.1A

**Maximum cruise power**

TRQ setting = 97 %, cf. tables figures 5.8.3 and 5.8.3A

**Recommended cruise power**

TRQ setting = 92 %, cf. tables figures 5.8.4 and 5.8.4A

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## Maximum climb power (FL < 200) - 124 KIAS

Conditions : If BLEED HI msg ON, reduce TRQ by 5 %

- NOTE • : Add 0.5 % of TRQ for each additional 10 KIAS on climb airspeed.

Table not valid if **INERT SEP ON** and/or BLEED HI msg ON.

| T° (°C) | Flight level (FL)                  |     |     |     |     |     |     |     |     |     |     |  |
|---------|------------------------------------|-----|-----|-----|-----|-----|-----|-----|-----|-----|-----|--|
| OAT     | 100                                | 110 | 120 | 130 | 140 | 150 | 160 | 170 | 180 | 190 | 200 |  |
| -24     | ▲ CAUTION ▲ Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -22     | TRQ max 100 %                      |     |     |     |     |     |     |     |     |     |     |  |
| -20     | Ng max 104 %                       |     |     |     |     |     |     |     |     |     |     |  |
| -18     |                                    |     |     |     |     |     |     |     |     |     |     |  |
| -16     |                                    |     |     |     |     |     |     |     |     |     |     |  |
| -14     |                                    |     |     |     |     |     |     |     |     |     |     |  |
| -12     |                                    |     |     |     |     |     |     |     |     |     |     |  |
| -10     |                                    |     |     |     |     |     |     |     |     |     | 100 |  |
| -8      |                                    |     |     |     |     |     |     |     |     |     | 100 |  |
| -6      |                                    |     |     |     |     |     |     |     |     |     | 98  |  |
| -4      |                                    |     |     |     |     |     |     |     |     |     | 98  |  |
| -2      |                                    |     |     |     |     |     |     |     |     |     | 96  |  |
| 0       |                                    |     |     |     |     |     |     |     |     |     | 100 |  |
| 2       |                                    |     |     |     |     |     |     |     |     |     | 99  |  |
| 4       |                                    |     |     |     |     |     |     |     |     |     | 95  |  |
| 6       |                                    |     |     |     |     |     |     |     |     |     | 93  |  |
| 8       |                                    |     |     |     |     |     |     |     |     |     | 91  |  |
| 10      |                                    |     |     |     |     |     |     |     |     |     | 100 |  |
| 12      |                                    |     |     |     |     |     |     |     |     |     | 98  |  |
| 14      |                                    |     |     |     |     |     |     |     |     |     | 93  |  |
| 16      |                                    |     |     |     |     |     |     |     |     |     | 90  |  |
| 18      |                                    |     |     |     |     |     |     |     |     |     | 88  |  |
| 20      |                                    |     |     |     |     |     |     |     |     |     | 85  |  |
| 22      |                                    |     |     |     |     |     |     |     |     |     | 82  |  |
| 24      |                                    |     |     |     |     |     |     |     |     |     | 80  |  |
| 26      |                                    |     |     |     |     |     |     |     |     |     | 100 |  |
| 28      |                                    |     |     |     |     |     |     |     |     |     | 97  |  |
| 30      |                                    |     |     |     |     |     |     |     |     |     | 94  |  |
| 32      |                                    |     |     |     |     |     |     |     |     |     | 89  |  |
| 34      |                                    |     |     |     |     |     |     |     |     |     | 86  |  |
| 36      |                                    |     |     |     |     |     |     |     |     |     | 83  |  |
| 38      |                                    |     |     |     |     |     |     |     |     |     | 80  |  |
| 40      |                                    |     |     |     |     |     |     |     |     |     | 78  |  |
| 42      |                                    |     |     |     |     |     |     |     |     |     | 75  |  |
| 44      |                                    |     |     |     |     |     |     |     |     |     | 72  |  |
| 46      |                                    |     |     |     |     |     |     |     |     |     | 70  |  |
| 48      |                                    |     |     |     |     |     |     |     |     |     | 68  |  |
| 50      |                                    |     |     |     |     |     |     |     |     |     | 66  |  |
| 52      |                                    |     |     |     |     |     |     |     |     |     | 64  |  |
| 54      |                                    |     |     |     |     |     |     |     |     |     | 62  |  |
| 56      |                                    |     |     |     |     |     |     |     |     |     | 60  |  |
| 58      |                                    |     |     |     |     |     |     |     |     |     | 58  |  |
| 60      |                                    |     |     |     |     |     |     |     |     |     | 56  |  |
| 62      |                                    |     |     |     |     |     |     |     |     |     | 54  |  |
| 64      |                                    |     |     |     |     |     |     |     |     |     | 52  |  |
| 66      |                                    |     |     |     |     |     |     |     |     |     | 50  |  |
| 68      |                                    |     |     |     |     |     |     |     |     |     | 48  |  |
| 70      |                                    |     |     |     |     |     |     |     |     |     | 46  |  |
| 72      |                                    |     |     |     |     |     |     |     |     |     | 44  |  |
| 74      |                                    |     |     |     |     |     |     |     |     |     | 42  |  |
| 76      |                                    |     |     |     |     |     |     |     |     |     | 40  |  |
| 78      |                                    |     |     |     |     |     |     |     |     |     | 38  |  |
| 80      |                                    |     |     |     |     |     |     |     |     |     | 36  |  |
| 82      |                                    |     |     |     |     |     |     |     |     |     | 34  |  |
| 84      |                                    |     |     |     |     |     |     |     |     |     | 32  |  |
| 86      |                                    |     |     |     |     |     |     |     |     |     | 30  |  |
| 88      |                                    |     |     |     |     |     |     |     |     |     | 28  |  |
| 90      |                                    |     |     |     |     |     |     |     |     |     | 26  |  |
| 92      |                                    |     |     |     |     |     |     |     |     |     | 24  |  |
| 94      |                                    |     |     |     |     |     |     |     |     |     | 22  |  |
| 96      |                                    |     |     |     |     |     |     |     |     |     | 20  |  |
| 98      |                                    |     |     |     |     |     |     |     |     |     | 18  |  |
| 100     |                                    |     |     |     |     |     |     |     |     |     | 16  |  |
| 102     |                                    |     |     |     |     |     |     |     |     |     | 14  |  |
| 104     |                                    |     |     |     |     |     |     |     |     |     | 12  |  |
| 106     |                                    |     |     |     |     |     |     |     |     |     | 10  |  |
| 108     |                                    |     |     |     |     |     |     |     |     |     | 8   |  |
| 110     |                                    |     |     |     |     |     |     |     |     |     | 6   |  |
| 112     |                                    |     |     |     |     |     |     |     |     |     | 4   |  |
| 114     |                                    |     |     |     |     |     |     |     |     |     | 2   |  |
| 116     |                                    |     |     |     |     |     |     |     |     |     | 0   |  |
| 118     |                                    |     |     |     |     |     |     |     |     |     | -2  |  |
| 120     |                                    |     |     |     |     |     |     |     |     |     | -4  |  |
| 122     |                                    |     |     |     |     |     |     |     |     |     | -6  |  |
| 124     |                                    |     |     |     |     |     |     |     |     |     | -8  |  |
| 126     |                                    |     |     |     |     |     |     |     |     |     | -10 |  |
| 128     |                                    |     |     |     |     |     |     |     |     |     | -12 |  |
| 130     |                                    |     |     |     |     |     |     |     |     |     | -14 |  |
| 132     |                                    |     |     |     |     |     |     |     |     |     | -16 |  |
| 134     |                                    |     |     |     |     |     |     |     |     |     | -18 |  |
| 136     |                                    |     |     |     |     |     |     |     |     |     | -20 |  |
| 138     |                                    |     |     |     |     |     |     |     |     |     | -22 |  |
| 140     |                                    |     |     |     |     |     |     |     |     |     | -24 |  |
| 142     |                                    |     |     |     |     |     |     |     |     |     | -26 |  |
| 144     |                                    |     |     |     |     |     |     |     |     |     | -28 |  |
| 146     |                                    |     |     |     |     |     |     |     |     |     | -30 |  |
| 148     |                                    |     |     |     |     |     |     |     |     |     | -32 |  |
| 150     |                                    |     |     |     |     |     |     |     |     |     | -34 |  |
| 152     |                                    |     |     |     |     |     |     |     |     |     | -36 |  |
| 154     |                                    |     |     |     |     |     |     |     |     |     | -38 |  |
| 156     |                                    |     |     |     |     |     |     |     |     |     | -40 |  |
| 158     |                                    |     |     |     |     |     |     |     |     |     | -42 |  |
| 160     |                                    |     |     |     |     |     |     |     |     |     | -44 |  |
| 162     |                                    |     |     |     |     |     |     |     |     |     | -46 |  |
| 164     |                                    |     |     |     |     |     |     |     |     |     | -48 |  |
| 166     |                                    |     |     |     |     |     |     |     |     |     | -50 |  |
| 168     |                                    |     |     |     |     |     |     |     |     |     | -52 |  |
| 170     |                                    |     |     |     |     |     |     |     |     |     | -54 |  |
| 172     |                                    |     |     |     |     |     |     |     |     |     | -56 |  |
| 174     |                                    |     |     |     |     |     |     |     |     |     | -58 |  |
| 176     |                                    |     |     |     |     |     |     |     |     |     | -60 |  |
| 178     |                                    |     |     |     |     |     |     |     |     |     | -62 |  |
| 180     |                                    |     |     |     |     |     |     |     |     |     | -64 |  |
| 182     |                                    |     |     |     |     |     |     |     |     |     | -66 |  |
| 184     |                                    |     |     |     |     |     |     |     |     |     | -68 |  |
| 186     |                                    |     |     |     |     |     |     |     |     |     | -70 |  |
| 188     |                                    |     |     |     |     |     |     |     |     |     | -72 |  |
| 190     |                                    |     |     |     |     |     |     |     |     |     | -74 |  |
| 192     |                                    |     |     |     |     |     |     |     |     |     | -76 |  |
| 194     |                                    |     |     |     |     |     |     |     |     |     | -78 |  |
| 196     |                                    |     |     |     |     |     |     |     |     |     | -80 |  |
| 198     |                                    |     |     |     |     |     |     |     |     |     | -82 |  |
| 200     |                                    |     |     |     |     |     |     |     |     |     | -84 |  |

Figure 5.8.1 - Maximum climb power (FL < 200) - 124 KIAS

- NOTE •

Refer to page 5.8.1 for general conditions

•



## Pilot's Operating Handbook

**Maximum climb power (FL > 200) - 124 KIAS**

Conditions : If BLEED HI msg ON, reduce TRQ by 5 %

- NOTE : Add 0.5 % of TRQ for each additional 10 KIAS on climb airspeed.

Table not valid if **INERT SEP ON** and/or BLEED HI msg ON.

| T° (°C) | Flight level (FL)   |     |     |                        |     |     |     |     |     |     |     |     |
|---------|---|-----|-----|------------------------|-----|-----|-----|-----|-----|-----|-----|-----|
| OAT     | 200   | 210 | 220 | 230                    | 240 | 250 | 260 | 270 | 280 | 290 | 300 | 310 |
| -66     |   |     |     |                        |     |     |     |     | 99  | 95  | 90  | 86  |
| -64     | <div> <div>▲ CAUTION ▲</div> <div>TRQ max 100 %</div> <div>Ng max 104 %</div> <div>▲</div> </div> |     |     | Recommended Ng < 103 % |     |     |     |     | 98  | 94  | 89  | 85  |
| -62     |   |     |     |                        |     |     |     |     | 97  | 93  | 88  | 84  |
| -60     |   |     |     |                        |     |     |     | 100 | 96  | 92  | 87  | 83  |
| -58     |   |     |     |                        |     |     |     | 100 | 95  | 91  | 86  | 82  |
| -56     |   |     |     |                        |     |     |     | 99  | 94  | 90  | 85  | 81  |
| -54     |   |     |     |                        |     |     |     | 98  | 93  | 89  | 85  | 81  |
| -52     |   |     |     |                        |     |     | 100 | 97  | 92  | 88  | 84  | 80  |
| -50     |   |     |     |                        |     |     | 100 | 95  | 91  | 87  | 83  | 79  |
| -48     |   |     |     |                        |     |     | 99  | 94  | 90  | 86  | 82  | 78  |
| -46     |   |     |     |                        |     |     | 98  | 93  | 89  | 85  | 81  | 77  |
| -44     |   |     |     |                        |     | 100 | 97  | 92  | 88  | 84  | 80  | 77  |
| -42     |   |     |     |                        |     | 100 | 96  | 91  | 87  | 83  | 79  | 75  |
| -40     |   |     |     |                        |     | 99  | 95  | 90  | 86  | 82  | 78  | 74  |
| -38     |   |     |     |                        |     | 98  | 93  | 89  | 85  | 81  | 77  | 73  |
| -36     |   |     |     |                        | 100 | 97  | 92  | 88  | 84  | 80  | 76  | 72  |
| -34     |   |     |     |                        | 99  | 95  | 91  | 87  | 82  | 78  | 75  | 71  |
| -32     |   |     |     |                        | 98  | 94  | 90  | 85  | 81  | 77  | 73  | 70  |
| -30     |   |     |     | 100                    | 97  | 93  | 88  | 84  | 80  | 76  | 72  | 69  |
| -28     |   |     |     | 100                    | 96  | 92  | 87  | 83  | 79  | 75  | 71  | 68  |
| -26     |   |     |     |                        | 98  | 94  | 90  | 86  | 82  | 78  | 74  | 70  |
| -24     |   |     | 100 | 97                     | 93  | 89  | 85  | 80  | 76  | 73  | 69  | 65  |
| -22     |   |     | 100 | 96                     | 92  | 88  | 83  | 79  | 75  | 71  | 67  | 64  |
| -20     |   |     | 99  | 95                     | 90  | 86  | 82  | 78  | 74  | 70  | 66  | 62  |
| -18     |   | 100 | 97  | 93                     | 89  | 85  | 81  | 77  | 72  | 68  | 64  | 60  |
| -16     |   | 100 | 96  | 92                     | 88  | 83  | 79  | 75  | 71  | 66  | 62  | 59  |
| -14     |   | 99  | 94  | 90                     | 86  | 82  | 77  | 73  | 69  | 65  | 61  | 57  |
| -12     | 100   | 97  | 93  | 89                     | 85  | 80  | 75  | 71  | 67  | 63  | 59  | 55  |
| -10     | 100   | 96  | 91  | 87                     | 82  | 78  | 74  | 69  | 65  | 61  | 57  | 53  |
| -8      | 98  | 94  | 89  | 85                     | 81  | 76  | 72  | 67  | 63  | 59  | 55  | 51  |
| -6      | 96  | 92  | 88  | 83                     | 79  | 74  | 70  | 65  | 61  | 57  | 53  |     |
| -4      | 95  | 90  | 85  | 81                     | 77  | 72  | 67  | 63  | 59  | 55  |     |     |
| -2      | 93  | 88  | 83  | 79                     | 74  | 70  | 65  | 61  | 57  |     |     |     |
| 0       | 91  | 85  | 81  | 76                     | 71  | 67  | 63  | 59  |     |     |     |     |
| 2       | 88  | 83  | 78  | 74                     | 69  | 65  | 61  |     |     |     |     |     |
| 4       | 85  | 80  | 76  | 71                     | 67  | 63  |     |     |     |     |     |     |
| 6       | 82  | 78  | 74  | 69                     | 65  |     |     |     |     |     |     |     |
| 8       | 80  | 76  | 71  | 67                     |     |     |     |     |     |     |     |     |
| 10      | 78  | 73  | 69  |                        |     |     |     |     |     |     |     |     |
| 12      | 75  | 70  |     |                        |     |     |     |     |     |     |     |     |

Figure 5.8.1A - Maximum climb power (FL &gt; 200) - 124 KIAS

- NOTE

Refer to page 5.8.1 for general conditions

# **Maximum climb power (FL < 200) - 170 KIAS / M 0.40**

Conditions : If BLEED HI msg ON, reduce TRQ by 5 %

- NOTE • : Add 0.5 % of TRQ for each additional 10 KIAS on climb airspeed.

Table not valid if **INERT SEP ON** and/or BLEED HI msg ON.

| T° (°C) | Flight level (FL) |     |                        |     |     |     |     |     |     |     |     |  |
|---------|-------------------|-----|------------------------|-----|-----|-----|-----|-----|-----|-----|-----|--|
| OAT     | 100               | 110 | 120                    | 130 | 140 | 150 | 160 | 170 | 180 | 190 | 200 |  |
| -24     |                   |     |                        |     |     |     |     |     |     |     |     |  |
| -22     | ▲ CAUTION ▲       |     | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |  |
| -20     | TRQ max 100 %     |     |                        |     |     |     |     |     |     |     |     |  |
| -18     | Ng max 104 %      |     |                        |     |     |     |     |     |     |     |     |  |
| -16     | ▲                 |     |                        |     |     |     |     |     |     |     |     |  |
| -14     |                   |     |                        |     |     |     |     |     |     |     |     |  |
| -12     |                   |     |                        |     |     |     |     |     |     |     |     |  |
| -10     |                   |     |                        |     |     |     |     |     |     |     |     |  |
| -8      |                   |     |                        |     |     |     |     |     |     |     | 100 |  |
| -6      |                   |     |                        |     |     |     |     |     |     |     | 100 |  |
| -4      |                   |     |                        |     |     |     |     |     |     |     | 98  |  |
| -2      |                   |     |                        |     |     |     |     |     |     | 100 | 95  |  |
| 0       |                   |     |                        |     |     |     |     |     | 100 | 98  | 92  |  |
| 2       |                   |     |                        |     |     |     |     |     | 100 | 95  | 90  |  |
| 4       |                   |     |                        |     |     |     |     | 100 | 97  | 92  | 87  |  |
| 6       |                   |     |                        |     |     |     |     | 99  | 94  | 90  | 85  |  |
| 8       |                   |     |                        |     |     |     | 100 | 97  | 92  | 87  | 82  |  |
| 10      |                   |     |                        |     |     |     | 99  | 94  | 89  | 84  | 79  |  |
| 12      |                   |     |                        |     |     | 100 | 96  | 91  | 86  | 81  | 77  |  |
| 14      |                   |     |                        |     | 100 | 98  | 93  | 88  | 83  | 79  | 74  |  |
| 16      |                   |     |                        |     | 100 | 95  | 90  | 85  | 81  | 76  |     |  |
| 18      |                   |     |                        | 100 | 97  | 92  | 87  | 82  | 78  |     |     |  |
| 20      |                   |     |                        | 99  | 94  | 89  | 85  | 80  |     |     |     |  |
| 22      |                   |     | 100                    | 96  | 91  | 86  | 82  |     |     |     |     |  |
| 24      |                   | 100 | 98                     | 93  | 88  | 84  |     |     |     |     |     |  |
| 26      |                   | 99  | 95                     | 90  | 85  |     |     |     |     |     |     |  |
| 28      | 100               | 96  | 92                     | 87  |     |     |     |     |     |     |     |  |
| 30      | 98                | 93  | 89                     |     |     |     |     |     |     |     |     |  |
| 32      | 95                | 90  |                        |     |     |     |     |     |     |     |     |  |
| 34      | 92                |     |                        |     |     |     |     |     |     |     |     |  |

Figure 5.8.2 - Maximum climb power (FL < 200) - 170 KIAS / M 0.40

- NOTE •

Refer to page 5.8.1 for general conditions

•

## Pilot's Operating Handbook

**Maximum climb power (FL > 200) - 170 KIAS / M 0.40**

Conditions : If BLEED HI msg ON, reduce TRQ by 5 %

- NOTE • : Add 0.5 % of TRQ for each additional 10 KIAS on climb airspeed.

Table not valid if **INERT SEP ON** and/or BLEED HI msg ON.

| T° (°C) | Flight level (FL)                                 |     |     |                        |     |     |     |     |     |     |     |     |
|---------|---|-----|-----|------------------------|-----|-----|-----|-----|-----|-----|-----|-----|
| OAT     | 200   | 210 | 220 | 230                    | 240 | 250 | 260 | 270 | 280 | 290 | 300 | 310 |
| -66     |   |     |     |                        |     |     |     |     |     | 98  | 93  | 88  |
| -64     | ▲ CAUTION ▲<br>TRQ max 100 %<br>Ng max 104 %<br>▲ |     |     | Recommended Ng < 103 % |     |     |     |     |     | 97  | 92  | 87  |
| -62     |   |     |     |                        |     |     |     |     | 100 | 96  | 91  | 86  |
| -60     |   |     |     |                        |     |     |     |     | 100 | 95  | 90  | 85  |
| -58     |   |     |     |                        |     |     |     |     | 99  | 94  | 89  | 84  |
| -56     |   |     |     |                        |     |     |     |     | 98  | 93  | 88  | 83  |
| -54     |   |     |     |                        |     |     |     | 100 | 96  | 92  | 87  | 83  |
| -52     |   |     |     |                        |     |     |     | 100 | 95  | 90  | 86  | 82  |
| -50     |   |     |     |                        |     |     |     | 99  | 94  | 89  | 85  | 81  |
| -48     |   |     |     |                        |     |     |     | 98  | 93  | 89  | 84  | 80  |
| -46     |   |     |     |                        |     |     | 100 | 97  | 92  | 88  | 83  | 79  |
| -44     |   |     |     |                        |     |     | 100 | 96  | 91  | 86  | 82  | 78  |
| -42     |   |     |     |                        |     |     | 99  | 94  | 90  | 85  | 81  | 77  |
| -40     |   |     |     |                        |     |     | 98  | 93  | 88  | 84  | 80  | 76  |
| -38     |   |     |     |                        |     | 100 | 97  | 92  | 87  | 83  | 79  | 75  |
| -36     |   |     |     |                        |     | 100 | 95  | 91  | 86  | 82  | 78  | 73  |
| -34     |   |     |     |                        |     | 99  | 94  | 89  | 85  | 81  | 76  | 72  |
| -32     |   |     |     |                        |     | 97  | 93  | 88  | 84  | 79  | 75  | 71  |
| -30     |   |     |     |                        | 100 | 96  | 91  | 87  | 82  | 78  | 74  | 70  |
| -28     |   |     |     |                        | 99  | 95  | 90  | 86  | 81  | 77  | 73  | 69  |
| -26     |   |     |     |                        | 98  | 94  | 89  | 84  | 80  | 76  | 72  | 68  |
| -24     |   |     |     | 100                    | 97  | 92  | 88  | 83  | 79  | 75  | 71  | 66  |
| -22     |   |     |     | 100                    | 96  | 91  | 86  | 82  | 77  | 73  | 69  | 65  |
| -20     |   |     |     | 99                     | 94  | 90  | 85  | 80  | 76  | 72  | 67  | 63  |
| -18     |   |     | 100 | 97                     | 93  | 88  | 83  | 79  | 74  | 70  | 65  | 61  |
| -16     |   |     | 100 | 96                     | 91  | 86  | 82  | 77  | 72  | 68  | 64  | 60  |
| -14     |   |     | 98  | 94                     | 89  | 85  | 80  | 75  | 71  | 66  | 62  | 57  |
| -12     |   | 100 | 96  | 92                     | 87  | 83  | 78  | 73  | 69  | 64  | 60  | 55  |
| -10     |   | 99  | 95  | 90                     | 85  | 81  | 76  | 71  | 66  | 62  | 58  | 54  |
| -8      | 100   | 97  | 93  | 88                     | 83  | 79  | 73  | 68  | 64  | 60  | 56  | 52  |
| -6      | 100   | 95  | 91  | 86                     | 81  | 76  | 71  | 66  | 62  | 58  | 54  |     |
| -4      | 98  | 93  | 88  | 83                     | 78  | 74  | 69  | 64  | 60  | 56  |     |     |
| -2      | 95  | 90  | 85  | 81                     | 76  | 71  | 67  | 62  | 58  |     |     |     |
| 0       | 92  | 88  | 83  | 78                     | 74  | 69  | 65  | 60  |     |     |     |     |
| 2       | 90  | 85  | 81  | 76                     | 72  | 67  | 62  |     |     |     |     |     |
| 4       | 87  | 83  | 78  | 74                     | 69  | 65  |     |     |     |     |     |     |
| 6       | 85  | 80  | 76  | 71                     | 67  |     |     |     |     |     |     |     |
| 8       | 82  | 78  | 73  | 69                     |     |     |     |     |     |     |     |     |
| 10      | 79  | 75  | 71  |                        |     |     |     |     |     |     |     |     |
| 12      | 77  | 72  |     |                        |     |     |     |     |     |     |     |     |

Figure 5.8.2A - Maximum climb power (FL &gt; 200) - 170 KIAS / M 0.40

- NOTE •

Refer to page 5.8.1 for general conditions

•

## Maximum cruise power (FL < 200)

Conditions : If BLEED HI msg ON, reduce TRQ by 5 %

- NOTE • : Use preferably recommended cruise power.

Table not valid if **INERT SEP ON** and/or BLEED HI msg ON.

| T° (°C) | Flight level (FL) |     |     |     |     |     |     |     |     |     |     |     |
|---------|-------------------|-----|-----|-----|-----|-----|-----|-----|-----|-----|-----|-----|
| OAT     | 100               | 110 | 120 | 130 | 140 | 150 | 160 | 170 | 180 | 190 | 200 |     |
| -24     |                   |     |     |     |     |     |     |     |     |     |     |     |
| -22     |                   |     |     |     |     |     |     |     |     |     |     |     |
| -20     |                   |     |     |     |     |     |     |     |     |     |     |     |
| -18     |                   |     |     |     |     |     |     |     |     |     |     |     |
| -16     |                   |     |     |     |     |     |     |     |     |     |     |     |
| -14     |                   |     |     |     |     |     |     |     |     |     |     |     |
| -12     |                   |     |     |     |     |     |     |     |     |     |     |     |
| -10     |                   |     |     |     |     |     |     |     |     |     |     |     |
| -8      |                   |     |     |     |     |     |     |     |     |     |     |     |
| -6      |                   |     |     |     |     |     |     |     |     |     |     |     |
| -4      |                   |     |     |     |     |     |     |     |     |     |     |     |
| -2      |                   |     |     |     |     |     |     |     |     |     |     |     |
| 0       |                   |     |     |     |     |     |     |     |     |     |     |     |
| 2       |                   |     |     |     |     |     |     |     |     |     |     | 100 |
| 4       |                   |     |     |     |     |     |     |     |     |     |     | 98  |
| 6       |                   |     |     |     |     |     |     |     |     |     |     | 95  |
| 8       |                   |     |     |     |     |     |     |     |     |     |     | 92  |
| 10      |                   |     |     |     |     |     |     |     |     |     |     | 89  |
| 12      |                   |     |     |     |     |     |     |     |     |     |     | 87  |
| 14      |                   |     |     |     |     |     |     |     |     |     |     | 84  |
| 16      |                   |     |     |     |     |     |     |     |     |     |     |     |
| 18      |                   |     |     |     |     |     |     |     |     |     |     |     |
| 20      |                   |     |     |     |     |     |     |     |     |     |     |     |
| 22      |                   |     |     |     |     |     |     |     |     |     |     |     |
| 24      |                   |     |     |     |     |     |     |     |     |     |     |     |
| 26      |                   |     |     |     |     |     |     |     |     |     |     |     |
| 28      |                   |     |     |     |     |     |     |     |     |     |     |     |
| 30      |                   |     |     |     |     |     |     |     |     |     |     |     |
| 32      |                   |     |     |     |     |     |     |     |     |     |     |     |
| 34      |                   |     |     |     |     |     |     |     |     |     |     |     |

Figure 5.8.3 - Maximum cruise power (FL < 200)

- NOTE •

Refer to page 5.8.1 for general conditions

•

## Pilot's Operating Handbook

**Maximum cruise power (FL > 200)**

Conditions : If BLEED HI msg ON, reduce TRQ by 5 %

- NOTE • : Use preferably recommended cruise power.

Table not valid if **INERT SEP ON** and/or BLEED HI msg ON.

| T° (°C) | Flight level (FL) |     |     |     |     |     |     |     |     |     |     |     |
|---------|-------------------|-----|-----|-----|-----|-----|-----|-----|-----|-----|-----|-----|
| OAT     | 200               | 210 | 220 | 230 | 240 | 250 | 260 | 270 | 280 | 290 | 300 | 310 |
| -62     |                   |     |     |     |     |     |     |     |     |     |     | 100 |
| -60     |                   |     |     |     |     |     |     |     |     |     |     | 100 |
| -58     |                   |     |     |     |     |     |     |     |     |     |     | 98  |
| -56     |                   |     |     |     |     |     |     |     |     |     |     | 97  |
| -54     |                   |     |     |     |     |     |     |     |     |     | 100 | 96  |
| -52     |                   |     |     |     |     |     |     |     |     |     | 99  | 94  |
| -50     |                   |     |     |     |     |     |     |     |     |     | 98  | 93  |
| -48     |                   |     |     |     |     |     |     |     |     | 100 | 97  | 92  |
| -46     |                   |     |     |     |     |     |     |     |     | 100 | 95  | 90  |
| -44     |                   |     |     |     |     |     |     |     |     | 99  | 94  | 89  |
| -42     |                   |     |     |     |     |     |     |     |     | 97  | 92  | 87  |
| -40     |                   |     |     |     |     |     |     |     | 100 | 96  | 91  | 86  |
| -38     |                   |     |     |     |     |     |     | 99  | 94  | 89  | 85  | 85  |
| -36     |                   |     |     |     |     |     |     | 98  | 93  | 88  | 83  | 83  |
| -34     |                   |     |     |     |     |     | 100 | 96  | 91  | 86  | 82  | 82  |
| -32     |                   |     |     |     |     |     | 100 | 95  | 90  | 85  | 80  | 80  |
| -30     |                   |     |     |     |     |     | 98  | 93  | 88  | 84  | 79  | 79  |
| -28     |                   |     |     |     |     |     | 97  | 92  | 87  | 82  | 78  | 78  |
| -26     |                   |     |     |     |     |     | 100 | 95  | 90  | 85  | 81  | 76  |
| -24     |                   |     |     |     |     |     | 99  | 94  | 89  | 84  | 79  | 74  |
| -22     |                   |     |     |     |     |     | 97  | 92  | 87  | 82  | 77  | 72  |
| -20     |                   |     |     |     |     | 100 | 96  | 90  | 85  | 80  | 75  | 70  |
| -18     |                   |     |     |     |     | 99  | 94  | 88  | 83  | 78  | 73  | 68  |
| -16     |                   |     |     |     | 100 | 97  | 92  | 86  | 81  | 76  | 71  | 67  |
| -14     |                   |     |     |     | 100 | 95  | 89  | 84  | 79  | 74  | 69  | 64  |
| -12     |                   |     |     |     | 98  | 93  | 87  | 82  | 77  | 72  | 67  | 62  |
| -10     |                   |     |     | 100 | 96  | 90  | 85  | 80  | 74  | 69  | 64  | 60  |
| -8      |                   |     |     | 99  | 93  | 88  | 82  | 77  | 72  | 67  | 62  | 58  |
| -6      |                   |     | 100 | 96  | 90  | 85  | 80  | 74  | 69  | 65  | 60  |     |
| -4      |                   |     | 99  | 93  | 88  | 82  | 77  | 72  | 67  | 63  |     |     |
| -2      |                   | 100 | 96  | 90  | 85  | 80  | 75  | 70  | 65  |     |     |     |
| 0       |                   | 98  | 93  | 87  | 82  | 77  | 73  | 68  |     |     |     |     |
| 2       | 100               | 95  | 90  | 85  | 80  | 75  | 70  |     |     |     |     |     |
| 4       | 98                | 93  | 88  | 82  | 77  | 73  |     |     |     |     |     |     |
| 6       | 95                | 90  | 85  | 80  | 75  |     |     |     |     |     |     |     |
| 8       | 92                | 87  | 82  | 77  |     |     |     |     |     |     |     |     |
| 10      | 89                | 84  | 79  |     |     |     |     |     |     |     |     |     |
| 12      | 87                | 81  |     |     |     |     |     |     |     |     |     |     |

Figure 5.8.3A - Maximum cruise power (FL &gt; 200)

- NOTE •

Refer to page 5.8.1 for general conditions

•

## Normal (recommended) cruise power (FL < 200)

Conditions : If BLEED HI msg ON, reduce TRQ by 5 %

• NOTE • : Table not valid if **INERT SEP ON** and/or BLEED HI msg ON.

| T° (°C) | Flight level (FL)      |     |     |     |     |     |     |     |     |     |     |  |
|---------|------------------------|-----|-----|-----|-----|-----|-----|-----|-----|-----|-----|--|
| OAT     | 100                    | 110 | 120 | 130 | 140 | 150 | 160 | 170 | 180 | 190 | 200 |  |
| -24     | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -22     | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -20     | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -18     | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -16     | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -14     | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -12     | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -10     | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -8      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -6      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -4      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| -2      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 0       | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 2       | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 4       | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 6       | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 8       | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 10      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 12      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 14      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 16      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 18      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 20      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 22      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 24      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 26      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 28      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 30      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 32      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |
| 34      | Recommended Ng < 103 % |     |     |     |     |     |     |     |     |     |     |  |

Figure 5.8.4 - Normal (recommended) cruise power (FL < 200)

• NOTE •

Refer to page 5.8.1 for general conditions

•

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**Normal (recommended) cruise power (FL > 200)**

Conditions : If BLEED HI msg ON, reduce TRQ by 5 %

- NOTE • : Table not valid if **INERT SEP ON** and/or BLEED HI msg ON.

| T° (°C) | Flight level (FL) |     |     |     |     |     |     |     |     |     |     |     |
|---------|-------------------|-----|-----|-----|-----|-----|-----|-----|-----|-----|-----|-----|
| OAT     | 200               | 210 | 220 | 230 | 240 | 250 | 260 | 270 | 280 | 290 | 300 | 310 |
| -66     |                   |     |     |     |     |     |     |     |     |     |     | 100 |
| -64     |                   |     |     |     |     |     |     |     |     |     |     | 99  |
| -62     |                   |     |     |     |     |     |     |     |     |     |     | 98  |
| -60     |                   |     |     |     |     |     |     |     |     |     | 100 | 96  |
| -58     |                   |     |     |     |     |     |     |     |     |     | 100 | 95  |
| -56     |                   |     |     |     |     |     |     |     |     |     | 98  | 93  |
| -54     |                   |     |     |     |     |     |     |     |     | 100 | 96  | 92  |
| -52     |                   |     |     |     |     |     |     |     |     | 100 | 95  | 90  |
| -50     |                   |     |     |     |     |     |     |     |     | 98  | 93  | 89  |
| -48     |                   |     |     |     |     |     |     |     | 100 | 97  | 92  | 87  |
| -46     |                   |     |     |     |     |     |     |     | 100 | 95  | 91  | 86  |
| -44     |                   |     |     |     |     |     |     |     | 99  | 94  | 89  | 84  |
| -42     |                   |     |     |     |     |     |     |     | 97  | 92  | 87  | 83  |
| -40     |                   |     |     |     |     |     |     | 100 | 96  | 91  | 86  | 82  |
| -38     |                   |     |     |     |     |     |     | 99  | 94  | 90  | 85  | 80  |
| -36     |                   |     |     |     |     |     |     | 98  | 93  | 88  | 83  | 79  |
| -34     |                   |     |     |     |     |     | 100 | 96  | 92  | 87  | 82  | 78  |
| -32     |                   |     |     |     |     |     | 100 | 95  | 90  | 85  | 81  | 76  |
| -30     |                   |     |     |     |     |     | 99  | 94  | 89  | 84  | 79  | 75  |
| -28     |                   |     |     |     |     |     | 97  | 92  | 87  | 82  | 78  | 73  |
| -26     |                   |     |     |     |     | 100 | 96  | 91  | 86  | 81  | 76  | 72  |
| -24     |                   |     |     |     |     | 99  | 94  | 89  | 84  | 79  | 74  | 70  |
| -22     |                   |     |     |     | 100 | 97  | 92  | 87  | 82  | 77  | 72  | 68  |
| -20     |                   |     |     |     | 100 | 95  | 90  | 85  | 80  | 75  | 70  | 66  |
| -18     |                   |     |     |     | 98  | 93  | 88  | 83  | 78  | 73  | 68  | 64  |
| -16     |                   |     |     | 100 | 96  | 91  | 86  | 81  | 76  | 71  | 66  | 61  |
| -14     |                   |     |     | 99  | 94  | 89  | 84  | 79  | 73  | 68  | 63  | 59  |
| -12     |                   |     | 100 | 97  | 92  | 87  | 81  | 76  | 71  | 66  | 61  | 57  |
| -10     |                   |     | 100 | 95  | 89  | 84  | 78  | 73  | 68  | 64  | 59  | 55  |
| -8      |                   | 100 | 97  | 92  | 86  | 81  | 76  | 71  | 66  | 62  | 57  | 53  |
| -6      |                   | 100 | 94  | 89  | 84  | 79  | 74  | 69  | 64  | 59  | 55  |     |
| -4      | 100               | 97  | 91  | 86  | 81  | 76  | 71  | 66  | 62  | 57  |     |     |
| -2      | 99                | 94  | 89  | 83  | 79  | 74  | 69  | 64  | 59  |     |     |     |
| 0       | 96                | 91  | 86  | 81  | 76  | 71  | 66  | 62  |     |     |     |     |
| 2       | 94                | 88  | 83  | 78  | 73  | 69  | 64  |     |     |     |     |     |
| 4       | 91                | 85  | 80  | 75  | 71  | 66  |     |     |     |     |     |     |
| 6       | 88                | 83  | 78  | 73  | 68  |     |     |     |     |     |     |     |
| 8       | 85                | 80  | 75  | 70  |     |     |     |     |     |     |     |     |
| 10      | 82                | 77  | 72  |     |     |     |     |     |     |     |     |     |
| 12      | 79                | 74  |     |     |     |     |     |     |     |     |     |     |

Figure 5.8.4A - Normal (recommended) cruise power (FL &gt; 200)

- NOTE •

Refer to page 5.8.1 for general conditions

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## 5.9 - Takeoff distances

The following tables give the takeoff distances for several weight configurations.

All common information applicable to tables (pages 5.9.2 to 5.9.4) are listed below.

Associated conditions :

- Landing gear DN and flaps TO
- TRQ = 100 %
- BLEED switch on AUTO
- Hard, dry and level runway

In table headings :

- GR = Ground roll (in ft)
- D<sub>50</sub> = Takeoff distance (clear to 50 ft) (in ft)

● NOTE ●

Between ISA + 30°C and ISA + 37°C, it may be necessary to cut-off the BLEED in order to set TRQ = 100 % during takeoff while respecting the engine limitations. In this case, reduce power after takeoff to set the BLEED switch to AUTO.

In SL ISA conditions, nominal Np is of 1985 RPM.

●

Corrections :

- In case of wind, apply the following corrections :
  - Reduce total distances by 10 % every 10 kts of headwind
  - Increase total distances by 30 % every 10 kts of tail wind
- Other runway surfaces :

Takeoff distances given in the tables are for takeoff from hard, dry and level runway. Other runway surfaces require the following correction factors.

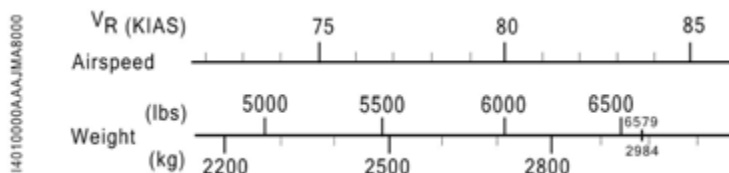
Increase distances by :

- 7 % on hard grass
- 10 % on short grass
- 15 % on wet runway
- 25 % on high grass
- 30 % on slippery runway

## Weight : 5512 lbs (2500 kg)

Associated condition :

- 15° of attitude after rotation
- Rotation speed choice ( $V_R$ )



| Weight : 5512 lbs (2500 kg) At 50 ft = 91 KIAS - 105 MPH IAS |            |      |            |      |            |      |            |      |
|--|------------|------|------------|------|------------|------|------------|------|
| Pressure altitude ft   | ISA - 35°C |      | ISA - 20°C |      | ISA - 10°C |      | ISA        |      |
|  | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0  | 665        | 1085 | 740        | 1190 | 780        | 1255 | 820        | 1295 |
| 2000   | 735        | 1185 | 800        | 1265 | 850        | 1340 | 905        | 1415 |
| 4000   | 800        | 1260 | 885        | 1380 | 935        | 1460 | 990        | 1545 |
| 6000   | 880        | 1375 | 965        | 1505 | 1025       | 1595 | 1090       | 1690 |
| 8000   | 965        | 1500 | 1060       | 1645 | 1140       | 1765 | 1220       | 1880 |
| Pressure altitude ft   | ISA + 10°C |      | ISA + 20°C |      | ISA + 30°C |      | ISA + 37°C |      |
|  | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0  | 865        | 1365 | 920        | 1435 | 965        | 1505 | 1000       | 1555 |
| 2000   | 955        | 1490 | 1005       | 1565 | 1060       | 1645 | 1100       | 1705 |
| 4000   | 1050       | 1625 | 1110       | 1720 | 1180       | 1825 | 1230       | 1895 |
| 6000   | 1165       | 1800 | 1240       | 1910 | 1320       | 2020 | 1380       | 2100 |
| 8000   | 1305       | 2000 | 1390       | 2120 | 1480       | 2245 | 1565       | 2330 |

Figure 5.9.1 - Takeoff distances - 5512 lbs (2500 kg)

### ▲ CAUTION ▲

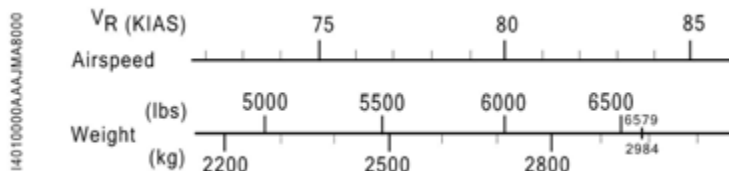
Refer to page 5.9.1 for notes and correction factors.



**Weight : 6579 lbs (2984 kg)**

Associated condition :

- 15° of attitude after rotation
- Rotation speed choice ( $V_R$ )



| Weight : 6579 lbs (2984 kg) At 50 ft = 94 KIAS - 108 MPH IAS |            |      |            |      |            |      |            |      |
|--|------------|------|------------|------|------------|------|------------|------|
| Pressure altitude ft   | ISA - 35°C |      | ISA - 20°C |      | ISA - 10°C |      | ISA        |      |
|  | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0  | 1020       | 1470 | 1115       | 1600 | 1185       | 1680 | 1245       | 1765 |
| 2000   | 1115       | 1595 | 1220       | 1730 | 1285       | 1820 | 1355       | 1915 |
| 4000   | 1215       | 1725 | 1325       | 1875 | 1400       | 1975 | 1475       | 2075 |
| 6000   | 1320       | 1865 | 1445       | 2030 | 1545       | 2160 | 1645       | 2305 |
| 8000   | 1435       | 2020 | 1600       | 2240 | 1715       | 2400 | 1850       | 2570 |
| Pressure altitude ft   | ISA + 10°C |      | ISA + 20°C |      | ISA + 30°C |      | ISA + 37°C |      |
|  | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0  | 1310       | 1855 | 1375       | 1940 | 1440       | 2030 | 1490       | 2090 |
| 2000   | 1425       | 2010 | 1500       | 2110 | 1595       | 2235 | 1660       | 2320 |
| 4000   | 1580       | 2205 | 1675       | 2345 | 1790       | 2485 | 1865       | 2590 |
| 6000   | 1755       | 2455 | 1880       | 2615 | 2005       | 2780 | 2095       | 2895 |
| 8000   | 1980       | 2745 | 2115       | 2925 | 2275       | 3110 | 2380       | 3245 |

Figure 5.9.2 - Takeoff distances - 6579 lbs (2984 kg)

▲ CAUTION ▲

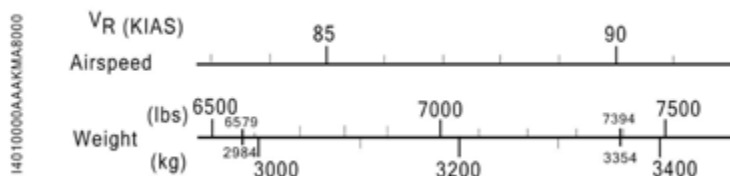
Refer to page 5.9.1 for notes and correction factors.

▲

## Weight : 7394 lbs (3354 kg)

Associated condition :

- 12°5 of attitude after rotation
- Rotation speed choice ( $V_R$ )



| Weight : 7394 lbs (3354 kg) At 50 ft = 99 KIAS - 114 MPH IAS |            |      |            |      |            |      |            |      |
|--|------------|------|------------|------|------------|------|------------|------|
| Pressure altitude ft   | ISA - 35°C |      | ISA - 20°C |      | ISA - 10°C |      | ISA        |      |
|  | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0  | 1440       | 2020 | 1560       | 2175 | 1645       | 2275 | 1725       | 2380 |
| 2000   | 1555       | 2170 | 1690       | 2335 | 1770       | 2445 | 1860       | 2560 |
| 4000   | 1685       | 2325 | 1820       | 2505 | 1910       | 2630 | 2045       | 2785 |
| 6000   | 1810       | 2500 | 1970       | 2710 | 2130       | 2930 | 2290       | 3135 |
| 8000   | 1960       | 2695 | 2220       | 3045 | 2410       | 3265 | 2590       | 3490 |
| Pressure altitude ft   | ISA + 10°C |      | ISA + 20°C |      | ISA + 30°C |      | ISA + 37°C |      |
|  | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0  | 1800       | 2485 | 1880       | 2595 | 1965       | 2705 | 2060       | 2810 |
| 2000   | 1945       | 2675 | 2080       | 2865 | 2215       | 3040 | 2325       | 3160 |
| 4000   | 2185       | 3000 | 2355       | 3200 | 2500       | 3385 | 2610       | 3520 |
| 6000   | 2470       | 3340 | 2640       | 3550 | 2810       | 3765 | 2935       | 3915 |
| 8000   | 2775       | 3720 | 2965       | 3950 | 3180       | 4185 | 3315       | 4350 |

Figure 5.9.3 - Takeoff distances - 7394 lbs (3354 kg)

### ▲ CAUTION ▲

Refer to page 5.9.1 for notes and correction factors.



## 5.10 - Climb performance

### MXCL - Speeds (IAS = 124 KIAS)

Conditions :

- Maximum climb power TRQ = 100 %
- Landing gear and flaps UP
- IAS = 124 KIAS - BLEED switch on AUTO and BLEED HI msg OFF

| Airplane weight       | Pressure altitude (feet) | Rate of climb (ft/min) |            |      |            |            |            |
|-----------------------|--------------------------|------------------------|------------|------|------------|------------|------------|
|                       |                          | ISA - 20°C             | ISA - 10°C | ISA  | ISA + 10°C | ISA + 20°C | ISA + 30°C |
| 5794 lbs<br>(2628 kg) | SL                       | 2885                   | 2870       | 2855 | 2845       | 2830       | 2810       |
|                       | 2000                     | 2860                   | 2845       | 2830 | 2810       | 2795       | 2775       |
|                       | 4000                     | 2840                   | 2820       | 2805 | 2785       | 2765       | 2750       |
|                       | 6000                     | 2810                   | 2790       | 2770 | 2750       | 2735       | 2710       |
|                       | 8000                     | 2775                   | 2755       | 2735 | 2710       | 2690       | 2665       |
| 6594 lbs<br>(2991 kg) | SL                       | 2440                   | 2425       | 2410 | 2400       | 2380       | 2365       |
|                       | 2000                     | 2415                   | 2400       | 2385 | 2365       | 2350       | 2330       |
|                       | 4000                     | 2395                   | 2375       | 2360 | 2340       | 2325       | 2305       |
|                       | 6000                     | 2365                   | 2345       | 2330 | 2310       | 2290       | 2270       |
|                       | 8000                     | 2335                   | 2315       | 2290 | 2270       | 2250       | 2230       |
| 7394 lbs<br>(3354 kg) | SL                       | 2080                   | 2065       | 2050 | 2040       | 2020       | 2005       |
|                       | 2000                     | 2055                   | 2040       | 2025 | 2005       | 1990       | 1975       |
|                       | 4000                     | 2035                   | 2015       | 1995 | 1980       | 1965       | 1945       |
|                       | 6000                     | 2005                   | 1985       | 1970 | 1950       | 1930       | 1910       |
|                       | 8000                     | 1975                   | 1955       | 1935 | 1910       | 1890       | 1870       |

Figure 5.10.1 - MXCL - Speeds (IAS = 124 KIAS)

• NOTE •

In SL ISA conditions, nominal Np is of 1985 RPM.

•

## MXCL - Speeds (IAS = 170 KIAS / M 0.40)

Conditions :

- Maximum climb power TRQ = 100 %
- Landing gear and flaps UP
- IAS = 170 KIAS / M 0.40
- BLEED switch on AUTO and BLEED HI msg OFF

| Airplane weight       | Pressure altitude (feet) | Rate of climb (ft/min) |            |       |            |            |            |
|-----------------------|--------------------------|------------------------|------------|-------|------------|------------|------------|
|                       |                          | ISA - 20°C             | ISA - 10°C | ISA   | ISA + 10°C | ISA + 20°C | ISA + 30°C |
| 5794 lbs<br>(2628 kg) | SL                       | 2 420                  | 2 390      | 2 365 | 2 335      | 2 310      | 2 285      |
|                       | 2000                     | 2 385                  | 2 355      | 2 325 | 2 295      | 2 265      | 2 235      |
|                       | 4000                     | 2 345                  | 2 315      | 2 280 | 2 250      | 2 220      | 2 190      |
|                       | 6000                     | 2 305                  | 2 270      | 2 235 | 2 205      | 2 170      | 2 140      |
|                       | 8000                     | 2 260                  | 2 225      | 2 190 | 2 155      | 2 120      | 2 085      |
| 6594 lbs<br>(2991 kg) | SL                       | 2 075                  | 2 050      | 2 025 | 2 000      | 1 975      | 1 955      |
|                       | 2000                     | 2 045                  | 2 015      | 1 990 | 1 965      | 1 935      | 1 910      |
|                       | 4000                     | 2 010                  | 1 985      | 1 950 | 1 920      | 1 895      | 1 865      |
|                       | 6000                     | 1 975                  | 1 940      | 1 910 | 1 880      | 1 850      | 1 820      |
|                       | 8000                     | 1 930                  | 1 900      | 1 870 | 1 835      | 1 805      | 1 770      |
| 7394 lbs<br>(3354 kg) | SL                       | 1 800                  | 1 775      | 1 755 | 1 730      | 1 710      | 1 685      |
|                       | 2000                     | 1 770                  | 1 745      | 1 720 | 1 695      | 1 670      | 1 645      |
|                       | 4000                     | 1 735                  | 1 710      | 1 685 | 1 655      | 1 630      | 1 605      |
|                       | 6000                     | 1 705                  | 1 670      | 1 645 | 1 615      | 1 590      | 1 560      |
|                       | 8000                     | 1 660                  | 1 635      | 1 605 | 1 575      | 1 545      | 1 515      |

Figure 5.10.2 - MXCL - Speeds (IAS = 170 KIAS / M 0.40)

• NOTE •

In SL ISA conditions, nominal Np is of 1985 RPM.

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## Pilot's Operating Handbook

**MXCL - Time, consumption and climb distance (IAS = 124 KIAS)**

Conditions :

- **ISA - 20°C**
- Maximum climb power
- Landing gear and flaps UP
- IAS = 124 KIAS - BLEED switch on AUTO

- **NOTE** • : Time, consumption and distance from the 50 ft  
If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in :

- Time to climb increased by up to 7 %
- Climb distance increased by up to 7 %
- Fuel consumption increased by up to 5 %

| Pressure altitude (ft) | Weight<br>5794 lbs (2628 kg) |          |    |      |               | Weight<br>6579 lbs (2984 kg) |          |    |      |               | Weight<br>7394 lbs (3354 kg) |          |    |      |               |
|------------------------|------------------------------|----------|----|------|---------------|------------------------------|----------|----|------|---------------|------------------------------|----------|----|------|---------------|
|                        | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) |
|                        |                              | l        | kg | USG  |               |                              | l        | kg | USG  |               |                              | l        | kg | USG  |               |
| SL                     | 00:00                        | 0        | 0  | 0    | 0             | 00:00                        | 0        | 0  | 0    | 0             | 00:00                        | 0        | 0  | 0    | 0             |
| 2000                   | 00:45                        | 4        | 3  | 1.0  | 1             | 00:45                        | 5        | 4  | 1.2  | 2             | 01:00                        | 5        | 4  | 1.4  | 2             |
| 4000                   | 01:30                        | 8        | 6  | 2.0  | 3             | 01:45                        | 9        | 7  | 2.4  | 3             | 02:00                        | 11       | 8  | 2.8  | 4             |
| 6000                   | 02:15                        | 11       | 9  | 3.0  | 4             | 02:30                        | 13       | 10 | 3.5  | 5             | 03:00                        | 16       | 12 | 4.1  | 6             |
| 8000                   | 03:00                        | 15       | 12 | 3.9  | 6             | 03:30                        | 18       | 14 | 4.6  | 7             | 04:00                        | 21       | 16 | 5.5  | 8             |
| 10000                  | 03:30                        | 18       | 14 | 4.9  | 8             | 04:15                        | 22       | 17 | 5.7  | 9             | 05:00                        | 26       | 20 | 6.8  | 11            |
| 12000                  | 04:15                        | 22       | 17 | 5.8  | 9             | 05:15                        | 26       | 20 | 6.8  | 11            | 06:00                        | 30       | 24 | 8.0  | 13            |
| 14000                  | 05:00                        | 25       | 20 | 6.7  | 11            | 06:00                        | 30       | 23 | 7.9  | 13            | 07:15                        | 35       | 28 | 9.3  | 16            |
| 16000                  | 05:45                        | 29       | 23 | 7.6  | 13            | 07:00                        | 34       | 27 | 9.0  | 15            | 08:15                        | 40       | 32 | 10.6 | 18            |
| 18000                  | 06:30                        | 32       | 25 | 8.5  | 15            | 07:45                        | 38       | 30 | 10.0 | 18            | 09:15                        | 45       | 35 | 11.9 | 21            |
| 20000                  | 07:30                        | 35       | 28 | 9.4  | 17            | 08:45                        | 42       | 33 | 11.1 | 20            | 10:30                        | 50       | 39 | 13.2 | 24            |
| 22000                  | 08:15                        | 39       | 30 | 10.3 | 19            | 09:45                        | 46       | 36 | 12.2 | 23            | 11:30                        | 55       | 43 | 14.4 | 27            |
| 24000                  | 09:00                        | 42       | 33 | 11.1 | 21            | 10:45                        | 50       | 39 | 13.2 | 25            | 12:45                        | 60       | 47 | 15.7 | 30            |
| 26000                  | 09:45                        | 46       | 36 | 12.0 | 24            | 11:45                        | 54       | 43 | 14.3 | 28            | 13:45                        | 64       | 51 | 17.0 | 34            |
| 28000                  | 10:30                        | 49       | 38 | 13.0 | 26            | 12:45                        | 58       | 46 | 15.4 | 31            | 15:00                        | 70       | 55 | 18.4 | 38            |
| 30000                  | 11:30                        | 53       | 41 | 13.9 | 29            | 13:45                        | 63       | 49 | 16.6 | 35            | 16:30                        | 75       | 59 | 19.8 | 42            |
| 31000                  | 12:00                        | 54       | 43 | 14.4 | 31            | 14:30                        | 65       | 51 | 17.2 | 37            | 17:15                        | 78       | 61 | 20.6 | 44            |

Figure 5.10.3 - MXCL - Time, consumption and climb distance  
(IAS = 124 KIAS) / ISA - 20°C

## MXCL - Time, consumption and climb distance (IAS = 124 KIAS)

Conditions :

- **ISA**
- Maximum climb power
- Landing gear and flaps UP
- IAS = 124 KIAS - BLEED switch on AUTO

- **NOTE** ● : Time, consumption and distance from the 50 ft  
If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in :
  - Time to climb increased by up to 8 %
  - Climb distance increased by up to 8 %
  - Fuel consumption increased by up to 6 %

| Pressure altitude (ft) | Weight<br>5794 lbs (2628 kg) |          |    |      |               | Weight<br>6579 lbs (2984 kg) |          |    |      |               | Weight<br>7394 lbs (3354 kg) |          |    |      |               |
|------------------------|------------------------------|----------|----|------|---------------|------------------------------|----------|----|------|---------------|------------------------------|----------|----|------|---------------|
|                        | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) |
|                        |                              | l        | kg | USG  |               |                              | l        | kg | USG  |               |                              | l        | kg | USG  |               |
| SL                     | 00:00                        | 0        | 0  | 0    | 0             | 00:00                        | 0        | 0  | 0    | 0             | 00:00                        | 0        | 0  | 0    | 0             |
| 2000                   | 00:45                        | 4        | 3  | 1.0  | 1             | 00:45                        | 5        | 4  | 1.2  | 2             | 01:00                        | 6        | 4  | 1.5  | 2             |
| 4000                   | 01:30                        | 8        | 6  | 2.1  | 3             | 01:45                        | 9        | 7  | 2.4  | 4             | 02:00                        | 11       | 9  | 2.9  | 4             |
| 6000                   | 02:15                        | 12       | 9  | 3.1  | 5             | 02:30                        | 14       | 11 | 3.6  | 5             | 03:00                        | 16       | 13 | 4.3  | 6             |
| 8000                   | 03:00                        | 15       | 12 | 4.1  | 6             | 03:30                        | 18       | 14 | 4.8  | 7             | 04:00                        | 21       | 17 | 5.7  | 9             |
| 10000                  | 03:45                        | 19       | 15 | 5.0  | 8             | 04:15                        | 22       | 18 | 5.9  | 10            | 05:15                        | 27       | 21 | 7.0  | 11            |
| 12000                  | 04:30                        | 23       | 18 | 6.0  | 10            | 05:15                        | 27       | 21 | 7.1  | 12            | 06:15                        | 32       | 25 | 8.4  | 14            |
| 14000                  | 05:15                        | 26       | 21 | 6.9  | 12            | 06:15                        | 31       | 24 | 8.2  | 14            | 07:15                        | 37       | 29 | 9.7  | 17            |
| 16000                  | 06:00                        | 30       | 23 | 7.9  | 14            | 07:00                        | 35       | 28 | 9.3  | 16            | 08:15                        | 42       | 33 | 11.0 | 19            |
| 18000                  | 06:45                        | 33       | 26 | 8.8  | 16            | 08:00                        | 39       | 31 | 10.4 | 19            | 09:30                        | 47       | 37 | 12.4 | 22            |
| 20000                  | 07:30                        | 37       | 29 | 9.7  | 18            | 09:00                        | 44       | 34 | 11.5 | 21            | 10:45                        | 52       | 41 | 13.7 | 26            |
| 22000                  | 08:15                        | 40       | 32 | 10.6 | 20            | 10:00                        | 48       | 38 | 12.7 | 24            | 11:45                        | 57       | 45 | 15.1 | 29            |
| 24000                  | 09:15                        | 44       | 34 | 11.6 | 23            | 11:00                        | 52       | 41 | 13.8 | 27            | 13:00                        | 62       | 49 | 16.5 | 32            |
| 26000                  | 10:00                        | 47       | 37 | 12.5 | 25            | 12:00                        | 57       | 44 | 14.9 | 30            | 14:15                        | 68       | 53 | 17.9 | 37            |
| 28000                  | 11:00                        | 51       | 40 | 13.5 | 28            | 13:15                        | 61       | 48 | 16.2 | 34            | 16:00                        | 73       | 58 | 19.4 | 41            |
| 30000                  | 12:15                        | 55       | 43 | 14.6 | 32            | 14:30                        | 66       | 52 | 17.5 | 39            | 17:45                        | 80       | 63 | 21.1 | 47            |
| 31000                  | 12:45                        | 57       | 45 | 15.1 | 34            | 15:30                        | 69       | 54 | 18.2 | 41            | 18:45                        | 83       | 65 | 21.9 | 51            |

Figure 5.10.4 - MXCL - Time, consumption and climb distance  
(IAS = 124 KIAS) / ISA



## Pilot's Operating Handbook

**MXCL - Time, consumption and climb distance (IAS = 124 KIAS)**

Conditions :

- **ISA + 20°C**
- Maximum climb power
- Landing gear and flaps UP
- IAS = 124 KIAS - BLEED switch on AUTO

- NOTE • : Time, consumption and distance from the 50 ft

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in :

- Time to climb increased by up to 10 %
- Climb distance increased by up to 10 %
- Fuel consumption increased by up to 7 %

| Pressure altitude (ft) | Weight<br>5794 lbs (2628 kg) |          |    |      |               | Weight<br>6579 lbs (2984 kg) |          |    |      |               | Weight<br>7394 lbs (3354 kg) |          |    |      |               |
|------------------------|------------------------------|----------|----|------|---------------|------------------------------|----------|----|------|---------------|------------------------------|----------|----|------|---------------|
|                        | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) |
|                        |                              | l        | kg | USG  |               |                              | l        | kg | USG  |               |                              | l        | kg | USG  |               |
| SL                     | 00:00                        | 0        | 0  | 0    | 0             | 00:00                        | 0        | 0  | 0    | 0             | 00:00                        | 0        | 0  | 0    | 0             |
| 2 000                  | 00:45                        | 4        | 3  | 1.1  | 2             | 00:45                        | 5        | 4  | 1.3  | 2             | 01:00                        | 6        | 4  | 1.5  | 2             |
| 4 000                  | 01:30                        | 8        | 6  | 2.1  | 3             | 01:45                        | 10       | 8  | 2.5  | 4             | 02:00                        | 11       | 9  | 3.0  | 4             |
| 6 000                  | 02:15                        | 12       | 9  | 3.2  | 5             | 02:30                        | 14       | 11 | 3.8  | 6             | 03:00                        | 17       | 13 | 4.5  | 7             |
| 8 000                  | 03:00                        | 16       | 12 | 4.2  | 7             | 03:30                        | 19       | 15 | 5.0  | 8             | 04:15                        | 22       | 17 | 5.9  | 9             |
| 10 000                 | 03:45                        | 20       | 15 | 5.2  | 8             | 04:30                        | 23       | 18 | 6.2  | 10            | 05:15                        | 28       | 22 | 7.3  | 12            |
| 12 000                 | 04:30                        | 23       | 18 | 6.2  | 10            | 05:15                        | 28       | 22 | 7.3  | 12            | 06:15                        | 33       | 26 | 8.7  | 15            |
| 14 000                 | 05:15                        | 27       | 21 | 7.2  | 12            | 06:15                        | 32       | 25 | 8.5  | 15            | 07:30                        | 38       | 30 | 10.1 | 18            |
| 16 000                 | 06:00                        | 31       | 24 | 8.1  | 14            | 07:15                        | 37       | 29 | 9.7  | 17            | 08:30                        | 44       | 34 | 11.5 | 21            |
| 18 000                 | 06:45                        | 34       | 27 | 9.1  | 17            | 08:15                        | 41       | 32 | 10.8 | 20            | 09:45                        | 49       | 38 | 12.9 | 24            |
| 20 000                 | 07:45                        | 38       | 30 | 10.1 | 19            | 09:15                        | 46       | 36 | 12.0 | 23            | 11:00                        | 54       | 43 | 14.4 | 27            |
| 22 000                 | 08:30                        | 42       | 33 | 11.1 | 22            | 10:15                        | 50       | 39 | 13.2 | 26            | 12:15                        | 60       | 47 | 15.9 | 31            |
| 24 000                 | 09:45                        | 46       | 36 | 12.1 | 25            | 11:30                        | 55       | 43 | 14.5 | 30            | 14:00                        | 66       | 52 | 17.5 | 36            |
| 26 000                 | 10:45                        | 50       | 39 | 13.2 | 28            | 13:00                        | 60       | 47 | 15.9 | 34            | 15:45                        | 73       | 57 | 19.2 | 42            |
| 28 000                 | 12:00                        | 54       | 43 | 14.4 | 33            | 14:30                        | 66       | 51 | 17.3 | 40            | 17:45                        | 80       | 63 | 21.0 | 49            |
| 30 000                 | 13:30                        | 59       | 46 | 15.6 | 38            | 16:30                        | 72       | 56 | 18.9 | 46            | 20:15                        | 88       | 69 | 23.2 | 58            |
| 31 000                 | 14:15                        | 62       | 48 | 16.3 | 41            | 17:30                        | 75       | 59 | 19.8 | 50            | 21:45                        | 92       | 72 | 24.4 | 63            |

Figure 5.10.5 - MXCL - Time, consumption and climb distance  
(IAS = 124 KIAS) / ISA + 20°C

## MXCL - Time, consumption and climb distance (IAS = 170 KIAS / M 0.40)

Conditions :

- ISA - 20°C
- Maximum climb power
- Landing gear and flaps UP
- IAS = 170 KIAS / M 0.40 - BLEED switch on AUTO

- NOTE • : Time, consumption and distance from the 50 ft  
If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in :
  - Time to climb increased by up to 10 %
  - Climb distance increased by up to 10 %
  - Fuel consumption increased by up to 7 %

| Pressure altitude (ft) | Weight<br>5794 lbs (2628 kg) |          |    |      |            |  | Weight<br>6579 lbs (2984 kg) |          |    |      |            |  | Weight<br>7394 lbs (3354 kg) |          |    |      |            |  |
|------------------------|------------------------------|----------|----|------|------------|--|------------------------------|----------|----|------|------------|--|------------------------------|----------|----|------|------------|--|
|                        | Time (min. s)                | Consump. |    |      | Dist. (NM) |  | Time (min. s)                | Consump. |    |      | Dist. (NM) |  | Time (min. s)                | Consump. |    |      | Dist. (NM) |  |
|                        |                              | l        | kg | USG  |            |  |                              | l        | kg | USG  |            |  |                              | l        | kg | USG  |            |  |
| SL                     | 00:00                        | 0        | 0  | 0    | 0          |  | 00:00                        | 0        | 0  | 0    | 0          |  | 00:00                        | 0        | 0  | 0    | 0          |  |
| 2000                   | 00:45                        | 4        | 3  | 1.2  | 2          |  | 01:00                        | 5        | 4  | 1.4  | 3          |  | 01:00                        | 6        | 5  | 1.6  | 3          |  |
| 4000                   | 01:45                        | 9        | 7  | 2.3  | 5          |  | 02:00                        | 10       | 8  | 2.7  | 5          |  | 02:15                        | 12       | 9  | 3.1  | 6          |  |
| 6000                   | 02:30                        | 13       | 10 | 3.5  | 7          |  | 03:00                        | 15       | 12 | 4.0  | 8          |  | 03:30                        | 18       | 14 | 4.7  | 10         |  |
| 8000                   | 03:30                        | 17       | 14 | 4.6  | 10         |  | 04:00                        | 20       | 16 | 5.4  | 11         |  | 04:30                        | 23       | 18 | 6.2  | 13         |  |
| 10000                  | 04:15                        | 22       | 17 | 5.7  | 12         |  | 05:00                        | 25       | 20 | 6.7  | 15         |  | 05:45                        | 29       | 23 | 7.7  | 17         |  |
| 12000                  | 05:15                        | 26       | 20 | 6.8  | 15         |  | 06:00                        | 30       | 24 | 7.9  | 18         |  | 07:00                        | 35       | 27 | 9.2  | 21         |  |
| 14000                  | 06:00                        | 30       | 24 | 7.9  | 18         |  | 07:00                        | 35       | 27 | 9.3  | 22         |  | 08:15                        | 41       | 32 | 10.8 | 25         |  |
| 16000                  | 07:00                        | 34       | 27 | 9.1  | 22         |  | 08:15                        | 40       | 31 | 10.6 | 25         |  | 09:30                        | 47       | 37 | 12.3 | 29         |  |
| 18000                  | 08:00                        | 39       | 30 | 10.2 | 25         |  | 09:15                        | 45       | 35 | 11.9 | 29         |  | 11:00                        | 52       | 41 | 13.8 | 34         |  |
| 20000                  | 09:00                        | 43       | 34 | 11.3 | 29         |  | 10:30                        | 50       | 39 | 13.2 | 33         |  | 12:15                        | 58       | 46 | 15.4 | 39         |  |
| 22000                  | 10:00                        | 47       | 37 | 12.4 | 32         |  | 11:45                        | 55       | 43 | 14.6 | 38         |  | 13:45                        | 64       | 50 | 17.0 | 44         |  |
| 24000                  | 11:00                        | 51       | 40 | 13.6 | 36         |  | 13:00                        | 60       | 47 | 15.9 | 43         |  | 15:00                        | 70       | 55 | 18.6 | 50         |  |
| 26000                  | 12:00                        | 55       | 43 | 14.6 | 40         |  | 14:00                        | 65       | 51 | 17.0 | 47         |  | 16:30                        | 76       | 59 | 20.0 | 55         |  |
| 28000                  | 12:45                        | 59       | 46 | 15.5 | 43         |  | 15:00                        | 69       | 54 | 18.2 | 51         |  | 17:30                        | 81       | 63 | 21.3 | 59         |  |
| 30000                  | 13:45                        | 62       | 49 | 16.5 | 46         |  | 16:00                        | 73       | 57 | 19.3 | 55         |  | 19:00                        | 86       | 67 | 22.7 | 64         |  |
| 31000                  | 14:15                        | 64       | 50 | 16.9 | 48         |  | 16:45                        | 75       | 59 | 19.9 | 57         |  | 19:45                        | 89       | 70 | 23.4 | 67         |  |

Figure 5.10.6 - MXCL - Time, consumption and climb distance  
(IAS = 170 KIAS / M 0.40) / ISA - 20°C

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**MXCL - Time, consumption and climb distance**  
**(IAS = 170 KIAS / M 0.40)**

Conditions :

- **ISA**
- Maximum climb power
- Landing gear and flaps UP
- IAS = 170 KIAS / M 0.40 - BLEED switch on AUTO

- **NOTE** • : Time, consumption and distance from the 50 ft
- If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in :
  - Time to climb increased by up to 11 %
  - Climb distance increased by up to 11 %
  - Fuel consumption increased by up to 8 %

| Pressure altitude (ft) | Weight<br>5794 lbs (2628 kg) |          |    |      |            |  | Weight<br>6579 lbs (2984 kg) |          |    |      |            |  | Weight<br>7394 lbs (3354 kg) |          |    |      |            |  |
|------------------------|------------------------------|----------|----|------|------------|--|------------------------------|----------|----|------|------------|--|------------------------------|----------|----|------|------------|--|
|                        | Time (min. s)                | Consump. |    |      | Dist. (NM) |  | Time (min. s)                | Consump. |    |      | Dist. (NM) |  | Time (min. s)                | Consump. |    |      | Dist. (NM) |  |
|                        |                              | l        | kg | USG  |            |  |                              | l        | kg | USG  |            |  |                              | l        | kg | USG  |            |  |
| SL                     | 00:00                        | 0        | 0  | 0    | 0          |  | 00:00                        | 0        | 0  | 0    | 0          |  | 00:00                        | 0        | 0  | 0    | 0          |  |
| 2000                   | 00:45                        | 5        | 4  | 1.2  | 2          |  | 01:00                        | 5        | 4  | 1.4  | 3          |  | 01:15                        | 6        | 5  | 1.7  | 3          |  |
| 4000                   | 01:45                        | 9        | 7  | 2.4  | 5          |  | 02:00                        | 11       | 8  | 2.8  | 6          |  | 02:15                        | 12       | 10 | 3.3  | 7          |  |
| 6000                   | 02:30                        | 14       | 11 | 3.6  | 8          |  | 03:00                        | 16       | 13 | 4.2  | 9          |  | 03:30                        | 19       | 15 | 4.9  | 10         |  |
| 8000                   | 03:30                        | 18       | 14 | 4.8  | 10         |  | 04:00                        | 21       | 17 | 5.6  | 12         |  | 04:45                        | 25       | 19 | 6.5  | 14         |  |
| 10000                  | 04:30                        | 23       | 18 | 6.0  | 13         |  | 05:15                        | 26       | 21 | 7.0  | 16         |  | 06:00                        | 31       | 24 | 8.1  | 18         |  |
| 12000                  | 05:15                        | 27       | 21 | 7.2  | 16         |  | 06:15                        | 32       | 25 | 8.4  | 19         |  | 07:15                        | 37       | 29 | 9.7  | 22         |  |
| 14000                  | 06:15                        | 32       | 25 | 8.4  | 20         |  | 07:15                        | 37       | 29 | 9.8  | 23         |  | 08:30                        | 43       | 34 | 11.4 | 27         |  |
| 16000                  | 07:15                        | 36       | 28 | 9.5  | 23         |  | 08:30                        | 42       | 33 | 11.2 | 27         |  | 10:00                        | 49       | 39 | 13.0 | 32         |  |
| 18000                  | 08:15                        | 41       | 32 | 10.7 | 27         |  | 09:45                        | 48       | 37 | 12.6 | 32         |  | 11:15                        | 56       | 44 | 14.7 | 37         |  |
| 20000                  | 09:15                        | 45       | 36 | 11.9 | 31         |  | 11:00                        | 53       | 42 | 14.0 | 36         |  | 12:45                        | 62       | 49 | 16.4 | 42         |  |
| 22000                  | 10:30                        | 50       | 39 | 13.2 | 35         |  | 12:15                        | 58       | 46 | 15.4 | 41         |  | 14:15                        | 68       | 54 | 18.1 | 48         |  |
| 24000                  | 11:30                        | 54       | 43 | 14.4 | 39         |  | 13:30                        | 64       | 50 | 16.9 | 46         |  | 15:45                        | 75       | 59 | 19.8 | 54         |  |
| 26000                  | 12:30                        | 59       | 46 | 15.5 | 43         |  | 14:45                        | 69       | 54 | 18.2 | 51         |  | 17:15                        | 81       | 63 | 21.3 | 60         |  |
| 28000                  | 13:30                        | 63       | 49 | 16.5 | 48         |  | 16:00                        | 74       | 58 | 19.5 | 56         |  | 18:45                        | 87       | 68 | 22.9 | 66         |  |
| 30000                  | 14:45                        | 67       | 52 | 17.6 | 52         |  | 17:15                        | 79       | 62 | 20.8 | 62         |  | 20:30                        | 93       | 73 | 24.6 | 73         |  |
| 31000                  | 15:15                        | 69       | 54 | 18.2 | 55         |  | 18:15                        | 81       | 64 | 21.5 | 65         |  | 21:30                        | 96       | 76 | 25.5 | 77         |  |

Figure 5.10.7 - MXCL - Time, consumption and climb distance  
(IAS = 170 KIAS / M 0.40) / ISA

## MXCL - Time, consumption and climb distance (IAS = 170 KIAS / M 0.40)

Conditions :

- **ISA + 20°C**
- Maximum climb power
- Landing gear and flaps UP
- IAS = 170 KIAS / M 0.40 - BLEED switch on AUTO

- **NOTE** • : Time, consumption and distance from the 50 ft
- If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in :
  - Time to climb increased by up to 15 %
  - Climb distance increased by up to 16 %
  - Fuel consumption increased by up to 12 %

| Pressure altitude<br>(ft) | Weight<br>5794 lbs (2628 kg) |          |    |      |               | Weight<br>6579 lbs (2984 kg) |          |    |      |               | Weight<br>7394 lbs (3354 kg) |          |    |      |               |
|---------------------------|------------------------------|----------|----|------|---------------|------------------------------|----------|----|------|---------------|------------------------------|----------|----|------|---------------|
|                           | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) | Time<br>(min. s)             | Consump. |    |      | Dist.<br>(NM) |
|                           |                              | l        | kg | USG  |               |                              | l        | kg | USG  |               |                              | l        | kg | USG  |               |
| SL                        | 00:00                        | 0        | 0  | 0    | 0             | 00:00                        | 0        | 0  | 0    | 0             | 00:00                        | 0        | 0  | 0    | 0             |
| 2000                      | 00:45                        | 5        | 4  | 1.3  | 3             | 01:00                        | 6        | 4  | 1.5  | 3             | 01:15                        | 7        | 5  | 1.7  | 3             |
| 4000                      | 01:45                        | 10       | 8  | 2.6  | 5             | 02:00                        | 11       | 9  | 3.0  | 6             | 02:30                        | 13       | 10 | 3.5  | 7             |
| 6000                      | 02:45                        | 14       | 11 | 3.8  | 8             | 03:00                        | 17       | 13 | 4.5  | 9             | 03:30                        | 20       | 15 | 5.2  | 11            |
| 8000                      | 03:30                        | 19       | 15 | 5.1  | 11            | 04:15                        | 22       | 18 | 5.9  | 13            | 05:00                        | 26       | 20 | 6.9  | 15            |
| 10000                     | 04:30                        | 24       | 19 | 6.3  | 14            | 05:15                        | 28       | 22 | 7.4  | 17            | 06:15                        | 33       | 26 | 8.6  | 19            |
| 12000                     | 05:30                        | 29       | 22 | 7.5  | 18            | 06:30                        | 33       | 26 | 8.8  | 21            | 07:30                        | 39       | 31 | 10.3 | 24            |
| 14000                     | 06:30                        | 33       | 26 | 8.8  | 21            | 07:30                        | 39       | 31 | 10.3 | 25            | 09:00                        | 46       | 36 | 12.0 | 29            |
| 16000                     | 07:30                        | 38       | 30 | 10.1 | 25            | 08:45                        | 45       | 35 | 11.8 | 29            | 10:15                        | 52       | 41 | 13.8 | 34            |
| 18000                     | 08:30                        | 43       | 34 | 11.3 | 29            | 10:00                        | 50       | 40 | 13.3 | 34            | 11:45                        | 59       | 46 | 15.6 | 40            |
| 20000                     | 09:45                        | 48       | 38 | 12.7 | 33            | 11:30                        | 56       | 44 | 14.8 | 39            | 13:15                        | 66       | 52 | 17.4 | 46            |
| 22000                     | 11:00                        | 53       | 42 | 14.1 | 38            | 13:00                        | 63       | 49 | 16.5 | 45            | 15:15                        | 74       | 58 | 19.5 | 53            |
| 24000                     | 12:30                        | 59       | 46 | 15.6 | 45            | 14:45                        | 70       | 55 | 18.4 | 53            | 17:15                        | 82       | 64 | 21.7 | 62            |
| 26000                     | 13:45                        | 64       | 50 | 17.0 | 51            | 16:30                        | 76       | 60 | 20.1 | 60            | 19:30                        | 90       | 71 | 23.8 | 72            |
| 28000                     | 15:30                        | 70       | 55 | 18.4 | 57            | 18:15                        | 83       | 65 | 21.9 | 68            | 22:00                        | 99       | 77 | 26.1 | 82            |
| 30000                     | 17:15                        | 75       | 59 | 19.8 | 64            | 20:30                        | 90       | 70 | 23.7 | 77            | 25:00                        | 108      | 85 | 28.5 | 94            |
| 31000                     | 18:00                        | 78       | 61 | 20.6 | 68            | 21:45                        | 93       | 73 | 24.7 | 82            | 26:30                        | 113      | 89 | 29.8 | 101           |

Figure 5.10.8 - MXCL - Time, consumption and climb distance  
(IAS = 170 KIAS / M 0.40) / ISA + 20°C

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## Climb performance after go-around

Conditions :

- Landing gear DN and flaps LDG
- **IAS = 90 KIAS**

| Airplane weight       | Pressure altitude (feet) | Rate of climb (ft/min) |            |            |      |            |            |            |
|-----------------------|--------------------------|------------------------|------------|------------|------|------------|------------|------------|
|                       |                          | ISA - 35°C             | ISA - 20°C | ISA - 10°C | ISA  | ISA + 10°C | ISA + 20°C | ISA + 30°C |
| 6594 lbs<br>(2991 kg) | SL                       | 1635                   | 1610       | 1590       | 1565 | 1545       | 1525       | 1505       |
|                       | 2000                     | 1615                   | 1580       | 1555       | 1535 | 1510       | 1490       | 1470       |
|                       | 4000                     | 1585                   | 1545       | 1525       | 1500 | 1480       | 1455       | 1435       |
|                       | 6000                     | 1555                   | 1515       | 1490       | 1465 | 1440       | 1420       | 1395       |
|                       | 8000                     | 1520                   | 1480       | 1455       | 1430 | 1400       | 1375       | 1345       |

Conditions :

- Landing gear DN and flaps LDG
- **IAS = 95 KIAS**

| Airplane weight       | Pressure altitude (feet) | Rate of climb (ft/min) |            |            |      |            |            |            |
|-----------------------|--------------------------|------------------------|------------|------------|------|------------|------------|------------|
|                       |                          | ISA - 35°C             | ISA - 20°C | ISA - 10°C | ISA  | ISA + 10°C | ISA + 20°C | ISA + 30°C |
| 7394 lbs<br>(3354 kg) | SL                       | 1350                   | 1320       | 1295       | 1275 | 1255       | 1235       | 1215       |
|                       | 2000                     | 1325                   | 1290       | 1265       | 1245 | 1225       | 1205       | 1180       |
|                       | 4000                     | 1295                   | 1255       | 1235       | 1210 | 1190       | 1165       | 1140       |
|                       | 6000                     | 1265                   | 1225       | 1200       | 1175 | 1150       | 1120       | 1095       |
|                       | 8000                     | 1230                   | 1190       | 1160       | 1135 | 1105       | 1075       | 1050       |

Figure 5.10.9 - Climb performance after go-around

## Climb performance - Flaps TO

Conditions :

- Landing gear UP and flaps TO
- **IAS = 110 KIAS**

| Airplane weight       | Pressure altitude (feet) | Rate of climb (ft/min) |            |            |      |            |            |            |
|-----------------------|--------------------------|------------------------|------------|------------|------|------------|------------|------------|
|                       |                          | ISA - 35°C             | ISA - 20°C | ISA - 10°C | ISA  | ISA + 10°C | ISA + 20°C | ISA + 30°C |
| 6594 lbs<br>(2991 kg) | SL                       | 2295                   | 2275       | 2260       | 2250 | 2240       | 2225       | 2215       |
|                       | 2000                     | 2280                   | 2260       | 2245       | 2230 | 2220       | 2210       | 2190       |
|                       | 4000                     | 2265                   | 2245       | 2230       | 2215 | 2200       | 2180       | 2165       |
|                       | 6000                     | 2250                   | 2225       | 2210       | 2190 | 2175       | 2155       | 2135       |
|                       | 8000                     | 2235                   | 2205       | 2185       | 2165 | 2145       | 2130       | 2110       |

Conditions:

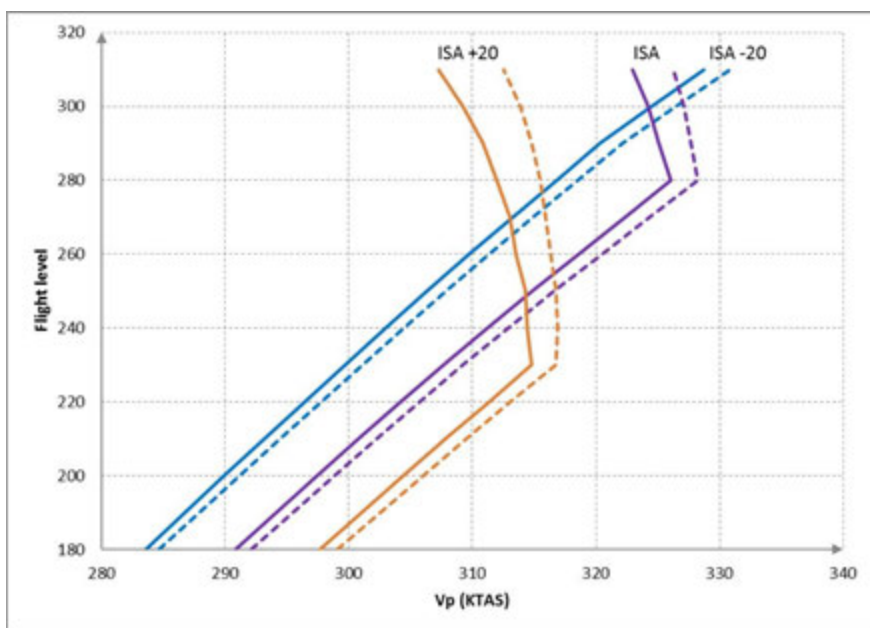
- Landing gear UP and flaps TO
- **IAS = 115 KIAS**

| Airplane weight       | Pressure altitude (feet) | Rate of climb (ft/min) |            |            |      |            |            |            |
|-----------------------|--------------------------|------------------------|------------|------------|------|------------|------------|------------|
|                       |                          | ISA - 35°C             | ISA - 20°C | ISA - 10°C | ISA  | ISA + 10°C | ISA + 20°C | ISA + 30°C |
| 7394 lbs<br>(3354 kg) | SL                       | 1985                   | 1965       | 1955       | 1940 | 1930       | 1915       | 1900       |
|                       | 2000                     | 1970                   | 1950       | 1940       | 1925 | 1910       | 1890       | 1875       |
|                       | 4000                     | 1955                   | 1935       | 1920       | 1900 | 1885       | 1865       | 1850       |
|                       | 6000                     | 1940                   | 1910       | 1895       | 1875 | 1860       | 1840       | 1825       |
|                       | 8000                     | 1915                   | 1890       | 1870       | 1850 | 1835       | 1815       | 1795       |

Figure 5.10.10 - Climb performance - Flaps TO

## 5.11 - Cruise performance

### Maximum cruise



———— 7100 lbs  
 - - - - - 6300 lbs

Figure 5.11.1 - Cruise performance (Maximum cruise)

## Maximum cruise

Conditions :

- **ISA - 20°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

- NOTE ● : Use preferably recommended cruise power.

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 5 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | -4       | 100     | 325       | 255    | 85.9    | 240                | 236 | 239                | 236 | 239                | 235 |
| 5000                     | -14      | 100     | 299       | 234    | 78.9    | 235                | 248 | 235                | 248 | 234                | 247 |
| 10000                    | -24      | 100     | 278       | 218    | 73.3    | 230                | 262 | 230                | 261 | 229                | 260 |
| 15000                    | -34      | 100     | 265       | 208    | 70.1    | 226                | 276 | 225                | 275 | 224                | 275 |
| 18000                    | -40      | 100     | 256       | 201    | 67.7    | 223                | 285 | 222                | 285 | 221                | 284 |
| 20000                    | -44      | 100     | 251       | 197    | 66.2    | 221                | 292 | 220                | 291 | 219                | 290 |
| 21000                    | -46      | 100     | 248       | 195    | 65.6    | 220                | 295 | 219                | 294 | 218                | 293 |
| 22000                    | -48      | 100     | 246       | 193    | 65.0    | 219                | 299 | 218                | 298 | 217                | 296 |
| 23000                    | -50      | 100     | 244       | 192    | 64.5    | 218                | 302 | 217                | 301 | 216                | 300 |
| 24000                    | -52      | 100     | 243       | 190    | 64.1    | 217                | 306 | 216                | 304 | 215                | 303 |
| 25000                    | -54      | 100     | 241       | 189    | 63.7    | 216                | 309 | 215                | 308 | 214                | 306 |
| 26000                    | -56      | 100     | 240       | 188    | 63.3    | 215                | 313 | 214                | 311 | 213                | 310 |
| 27000                    | -57      | 100     | 239       | 188    | 63.2    | 214                | 316 | 213                | 315 | 212                | 313 |
| 28000                    | -59      | 100     | 238       | 187    | 63.0    | 213                | 320 | 212                | 318 | 211                | 317 |
| 29000                    | -61      | 100     | 238       | 187    | 62.9    | 212                | 324 | 211                | 322 | 209                | 320 |
| 30000                    | -63      | 100     | 238       | 187    | 62.8    | 211                | 328 | 210                | 326 | 209                | 324 |
| 31000                    | -65      | 100     | 238       | 187    | 63.0    | 210                | 332 | 209                | 331 | 208                | 329 |

Figure 5.11.2 - Cruise performance  
Maximum cruise / ISA - 20°C



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## Maximum cruise

Conditions :

- **ISA - 10°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Use preferably recommended cruise power.

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 6 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 6        | 100     | 329       | 258    | 86.9    | 238                | 239 | 238                | 239 | 237                | 238 |
| 5000                     | -4       | 100     | 302       | 237    | 79.7    | 234                | 252 | 233                | 251 | 233                | 250 |
| 10000                    | -14      | 100     | 281       | 220    | 74.2    | 229                | 265 | 228                | 265 | 228                | 264 |
| 15000                    | -24      | 100     | 268       | 210    | 70.8    | 224                | 280 | 223                | 279 | 222                | 278 |
| 18000                    | -30      | 100     | 259       | 203    | 68.4    | 221                | 289 | 220                | 288 | 219                | 287 |
| 20000                    | -34      | 100     | 253       | 199    | 66.9    | 219                | 296 | 218                | 295 | 217                | 294 |
| 21000                    | -36      | 100     | 251       | 197    | 66.2    | 218                | 299 | 217                | 298 | 216                | 297 |
| 22000                    | -38      | 100     | 249       | 195    | 65.7    | 217                | 303 | 216                | 302 | 215                | 300 |
| 23000                    | -40      | 100     | 247       | 194    | 65.1    | 216                | 306 | 215                | 305 | 214                | 304 |
| 24000                    | -42      | 100     | 245       | 192    | 64.7    | 215                | 310 | 214                | 309 | 213                | 307 |
| 25000                    | -44      | 100     | 243       | 191    | 64.3    | 214                | 314 | 213                | 312 | 212                | 311 |
| 26000                    | -46      | 100     | 242       | 190    | 63.9    | 213                | 317 | 212                | 316 | 211                | 314 |
| 27000                    | -47      | 100     | 242       | 190    | 63.8    | 212                | 321 | 211                | 320 | 210                | 318 |
| 28000                    | -49      | 100     | 241       | 189    | 63.6    | 211                | 325 | 210                | 323 | 209                | 322 |
| 29000                    | -51      | 100     | 240       | 189    | 63.5    | 210                | 329 | 209                | 328 | 208                | 326 |
| 30000                    | -53      | 100     | 239       | 188    | 63.2    | 209                | 333 | 208                | 332 | 207                | 329 |
| 31000                    | -55      | 97      | 230       | 181    | 60.8    | 205                | 333 | 204                | 331 | 202                | 328 |

Figure 5.11.3 - Cruise performance  
Maximum cruise / ISA - 10°C

## Maximum cruise

Conditions :

- **ISA - 5°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Use preferably recommended cruise power.

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 6 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 11       | 100     | 331       | 259    | 87.3    | 238                | 240 | 237                | 240 | 237                | 240 |
| 5000                     | 1        | 100     | 304       | 238    | 80.2    | 233                | 253 | 232                | 253 | 232                | 252 |
| 10000                    | -9       | 100     | 282       | 221    | 74.5    | 228                | 267 | 227                | 266 | 227                | 265 |
| 15000                    | -19      | 100     | 269       | 211    | 71.2    | 223                | 282 | 222                | 281 | 222                | 280 |
| 18000                    | -25      | 100     | 260       | 204    | 68.7    | 220                | 291 | 219                | 290 | 218                | 289 |
| 20000                    | -29      | 100     | 254       | 200    | 67.2    | 218                | 298 | 217                | 297 | 216                | 296 |
| 21000                    | -31      | 100     | 252       | 198    | 66.5    | 217                | 301 | 216                | 300 | 215                | 299 |
| 22000                    | -33      | 100     | 250       | 196    | 66.0    | 216                | 305 | 215                | 304 | 214                | 302 |
| 23000                    | -35      | 100     | 248       | 195    | 65.5    | 215                | 308 | 214                | 307 | 213                | 306 |
| 24000                    | -37      | 100     | 246       | 193    | 65.0    | 214                | 312 | 213                | 311 | 212                | 309 |
| 25000                    | -39      | 100     | 244       | 192    | 64.6    | 213                | 316 | 212                | 315 | 211                | 313 |
| 26000                    | -41      | 100     | 243       | 191    | 64.2    | 212                | 320 | 211                | 318 | 210                | 316 |
| 27000                    | -42      | 100     | 243       | 191    | 64.1    | 211                | 323 | 210                | 322 | 209                | 320 |
| 28000                    | -44      | 100     | 242       | 190    | 64.0    | 210                | 328 | 209                | 326 | 208                | 324 |
| 29000                    | -46      | 100     | 242       | 190    | 63.8    | 210                | 332 | 209                | 330 | 207                | 328 |
| 30000                    | -48      | 97      | 233       | 183    | 61.5    | 206                | 332 | 205                | 330 | 203                | 327 |
| 31000                    | -50      | 94      | 224       | 176    | 59.3    | 202                | 332 | 200                | 329 | 199                | 326 |

Figure 5.11.4 - Cruise performance  
Maximum cruise / ISA - 5°C

## Maximum cruise

Conditions :

- **ISA**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Use preferably recommended cruise power.

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 6 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 16       | 100     | 333       | 261    | 87.9    | 237                | 242 | 237                | 242 | 236                | 241 |
| 5000                     | 6        | 100     | 305       | 240    | 80.7    | 232                | 255 | 232                | 254 | 231                | 253 |
| 10000                    | -4       | 100     | 284       | 223    | 74.9    | 227                | 268 | 227                | 268 | 226                | 267 |
| 15000                    | -14      | 100     | 271       | 213    | 71.5    | 222                | 283 | 222                | 283 | 221                | 282 |
| 18000                    | -20      | 100     | 261       | 205    | 69.0    | 219                | 293 | 219                | 292 | 218                | 291 |
| 20000                    | -24      | 100     | 256       | 201    | 67.6    | 217                | 300 | 216                | 299 | 215                | 297 |
| 21000                    | -26      | 100     | 253       | 199    | 66.9    | 216                | 303 | 215                | 302 | 214                | 301 |
| 22000                    | -28      | 100     | 251       | 197    | 66.3    | 215                | 307 | 214                | 306 | 213                | 304 |
| 23000                    | -30      | 100     | 249       | 195    | 65.8    | 214                | 310 | 213                | 309 | 212                | 308 |
| 24000                    | -32      | 100     | 247       | 194    | 65.3    | 213                | 314 | 212                | 313 | 211                | 311 |
| 25000                    | -34      | 100     | 246       | 193    | 64.9    | 212                | 318 | 211                | 317 | 210                | 315 |
| 26000                    | -36      | 100     | 244       | 192    | 64.5    | 211                | 322 | 210                | 320 | 209                | 319 |
| 27000                    | -37      | 100     | 244       | 191    | 64.4    | 210                | 326 | 209                | 324 | 208                | 322 |
| 28000                    | -39      | 100     | 242       | 190    | 64.1    | 210                | 330 | 208                | 328 | 207                | 326 |
| 29000                    | -41      | 97      | 234       | 184    | 61.8    | 206                | 330 | 204                | 328 | 203                | 325 |
| 30000                    | -43      | 94      | 226       | 177    | 59.7    | 202                | 329 | 200                | 327 | 199                | 324 |
| 31000                    | -45      | 90      | 218       | 171    | 57.5    | 198                | 329 | 196                | 326 | 194                | 323 |

Figure 5.11.5 - Cruise performance  
Maximum cruise / ISA

## Maximum cruise

Conditions :

- **ISA + 5°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

- **NOTE** ● : Use preferably recommended cruise power.  
If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 5 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 21       | 100     | 334       | 263    | 88.4    | 236                | 243 | 236                | 243 | 235                | 242 |
| 5000                     | 11       | 100     | 307       | 241    | 81.1    | 231                | 256 | 231                | 256 | 230                | 255 |
| 10000                    | 1        | 100     | 285       | 224    | 75.3    | 226                | 270 | 226                | 269 | 225                | 269 |
| 15000                    | -9       | 100     | 272       | 214    | 72.0    | 221                | 285 | 221                | 284 | 220                | 283 |
| 18000                    | -15      | 100     | 263       | 206    | 69.4    | 218                | 295 | 218                | 294 | 217                | 293 |
| 20000                    | -19      | 100     | 257       | 202    | 67.9    | 216                | 302 | 216                | 301 | 215                | 299 |
| 21000                    | -21      | 100     | 254       | 200    | 67.2    | 215                | 305 | 215                | 304 | 213                | 303 |
| 22000                    | -23      | 100     | 252       | 198    | 66.6    | 214                | 309 | 214                | 308 | 212                | 306 |
| 23000                    | -25      | 100     | 250       | 196    | 66.1    | 213                | 312 | 213                | 311 | 211                | 309 |
| 24000                    | -27      | 100     | 248       | 195    | 65.7    | 212                | 316 | 212                | 315 | 210                | 313 |
| 25000                    | -29      | 100     | 247       | 194    | 65.2    | 211                | 320 | 210                | 319 | 209                | 317 |
| 26000                    | -31      | 100     | 245       | 192    | 64.8    | 210                | 324 | 209                | 322 | 208                | 320 |
| 27000                    | -32      | 100     | 244       | 192    | 64.6    | 210                | 328 | 209                | 326 | 207                | 324 |
| 28000                    | -34      | 97      | 236       | 185    | 62.3    | 206                | 328 | 204                | 326 | 203                | 323 |
| 29000                    | -36      | 93      | 227       | 178    | 60.0    | 202                | 327 | 200                | 325 | 199                | 322 |
| 30000                    | -38      | 90      | 219       | 172    | 57.9    | 198                | 327 | 196                | 324 | 194                | 321 |
| 31000                    | -40      | 87      | 211       | 166    | 55.8    | 194                | 326 | 192                | 323 | 190                | 320 |

Figure 5.11.6 - Cruise performance  
Maximum cruise / ISA + 5°C

## Maximum cruise

Conditions :

- **ISA + 10°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Use preferably recommended cruise power.

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 5 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 26       | 100     | 336       | 264    | 88.9    | 236                | 245 | 235                | 244 | 235                | 244 |
| 5000                     | 16       | 100     | 309       | 242    | 81.6    | 231                | 258 | 230                | 257 | 230                | 256 |
| 10000                    | 6        | 100     | 287       | 225    | 75.7    | 226                | 272 | 225                | 271 | 224                | 270 |
| 15000                    | -4       | 100     | 274       | 215    | 72.3    | 221                | 287 | 220                | 286 | 219                | 285 |
| 18000                    | -10      | 100     | 264       | 207    | 69.7    | 218                | 297 | 217                | 296 | 216                | 294 |
| 20000                    | -14      | 100     | 258       | 203    | 68.3    | 216                | 303 | 215                | 302 | 214                | 301 |
| 21000                    | -16      | 100     | 256       | 201    | 67.6    | 215                | 307 | 214                | 306 | 213                | 304 |
| 22000                    | -18      | 100     | 254       | 199    | 67.0    | 214                | 311 | 213                | 309 | 211                | 308 |
| 23000                    | -20      | 100     | 252       | 197    | 66.5    | 212                | 314 | 212                | 313 | 210                | 311 |
| 24000                    | -22      | 100     | 250       | 196    | 66.0    | 212                | 318 | 211                | 317 | 209                | 315 |
| 25000                    | -24      | 100     | 248       | 195    | 65.5    | 211                | 322 | 210                | 320 | 208                | 319 |
| 26000                    | -26      | 100     | 246       | 193    | 65.1    | 210                | 326 | 209                | 325 | 207                | 323 |
| 27000                    | -27      | 97      | 238       | 187    | 62.8    | 206                | 325 | 204                | 324 | 203                | 321 |
| 28000                    | -29      | 93      | 229       | 180    | 60.5    | 202                | 325 | 200                | 323 | 198                | 320 |
| 29000                    | -31      | 90      | 221       | 173    | 58.3    | 198                | 325 | 196                | 322 | 194                | 319 |
| 30000                    | -33      | 86      | 213       | 167    | 56.2    | 194                | 324 | 192                | 321 | 190                | 317 |
| 31000                    | -35      | 83      | 205       | 161    | 54.1    | 190                | 323 | 188                | 320 | 186                | 316 |

Figure 5.11.7 - Cruise performance  
Maximum cruise / ISA + 10°C

## Maximum cruise

Conditions :

- **ISA + 20°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

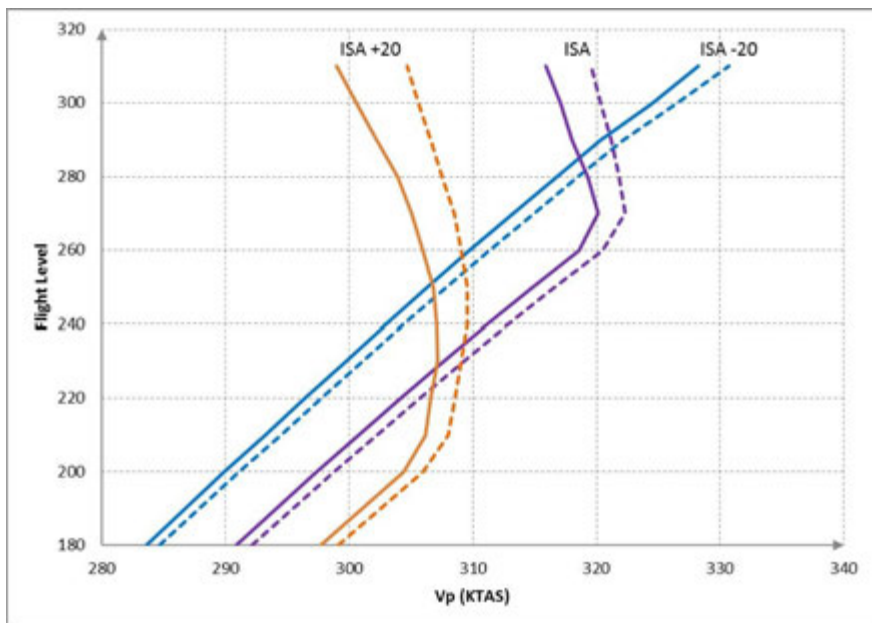
● NOTE ● : Use preferably recommended cruise power.

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 6 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 36       | 100     | 340       | 267    | 89.8    | 234                | 247 | 234                | 247 | 233                | 246 |
| 5000                     | 26       | 100     | 312       | 245    | 82.5    | 229                | 261 | 229                | 260 | 228                | 259 |
| 10000                    | 16       | 100     | 290       | 227    | 76.5    | 224                | 275 | 224                | 274 | 223                | 273 |
| 15000                    | 6        | 100     | 276       | 217    | 73.0    | 219                | 290 | 218                | 289 | 217                | 288 |
| 18000                    | 0        | 100     | 266       | 209    | 70.4    | 216                | 300 | 215                | 299 | 214                | 298 |
| 20000                    | -4       | 100     | 261       | 205    | 69.0    | 214                | 307 | 213                | 306 | 212                | 304 |
| 21000                    | -6       | 100     | 258       | 203    | 68.3    | 213                | 311 | 212                | 309 | 211                | 308 |
| 22000                    | -8       | 100     | 256       | 201    | 67.6    | 212                | 314 | 211                | 313 | 210                | 311 |
| 23000                    | -10      | 100     | 254       | 200    | 67.1    | 211                | 318 | 210                | 317 | 209                | 315 |
| 24000                    | -12      | 98      | 246       | 193    | 65.0    | 208                | 319 | 206                | 317 | 205                | 314 |
| 25000                    | -14      | 95      | 238       | 187    | 62.8    | 204                | 319 | 203                | 317 | 201                | 314 |
| 26000                    | -16      | 92      | 230       | 180    | 60.7    | 200                | 318 | 199                | 316 | 197                | 314 |
| 27000                    | -17      | 88      | 222       | 174    | 58.6    | 197                | 318 | 195                | 316 | 193                | 313 |
| 28000                    | -19      | 85      | 214       | 168    | 56.6    | 193                | 318 | 192                | 316 | 189                | 312 |
| 29000                    | -21      | 82      | 207       | 162    | 54.6    | 190                | 318 | 188                | 315 | 185                | 311 |
| 30000                    | -23      | 79      | 199       | 156    | 52.7    | 186                | 317 | 184                | 314 | 181                | 309 |
| 31000                    | -25      | 76      | 192       | 151    | 50.7    | 182                | 316 | 180                | 313 | 177                | 307 |

Figure 5.11.8 - Cruise performance  
Maximum cruise / ISA + 20°C

## Normal cruise (Recommended)



— 7100 lbs  
- - 6300 lbs

Figure 5.11.9 - Cruise performance (Recommended cruise)

## Normal cruise (Recommended)

Conditions :

- **ISA - 20°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Power recommended by PRATT & WHITNEY CANADA

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 6 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | -4       | 100     | 325       | 255    | 85.9    | 240                | 236 | 239                | 236 | 239                | 235 |
| 5000                     | -14      | 100     | 299       | 234    | 78.9    | 235                | 248 | 235                | 248 | 234                | 247 |
| 10000                    | -24      | 100     | 278       | 218    | 73.3    | 230                | 262 | 230                | 261 | 229                | 260 |
| 15000                    | -34      | 100     | 265       | 208    | 70.1    | 226                | 276 | 225                | 275 | 224                | 275 |
| 18000                    | -40      | 100     | 256       | 201    | 67.7    | 223                | 285 | 222                | 285 | 221                | 284 |
| 20000                    | -44      | 100     | 251       | 197    | 66.2    | 221                | 292 | 220                | 291 | 219                | 290 |
| 21000                    | -46      | 100     | 248       | 195    | 65.6    | 220                | 295 | 219                | 294 | 218                | 293 |
| 22000                    | -48      | 100     | 246       | 193    | 65.0    | 219                | 299 | 218                | 298 | 217                | 296 |
| 23000                    | -50      | 100     | 244       | 192    | 64.5    | 218                | 302 | 217                | 301 | 216                | 300 |
| 24000                    | -52      | 100     | 243       | 190    | 64.1    | 217                | 306 | 216                | 304 | 215                | 303 |
| 25000                    | -54      | 100     | 241       | 189    | 63.7    | 216                | 309 | 215                | 308 | 214                | 306 |
| 26000                    | -56      | 100     | 240       | 188    | 63.3    | 215                | 313 | 214                | 311 | 213                | 310 |
| 27000                    | -57      | 100     | 239       | 188    | 63.2    | 214                | 316 | 213                | 315 | 212                | 313 |
| 28000                    | -59      | 100     | 238       | 187    | 63.0    | 213                | 320 | 212                | 318 | 211                | 317 |
| 29000                    | -61      | 100     | 238       | 187    | 62.9    | 212                | 324 | 211                | 322 | 209                | 320 |
| 30000                    | -63      | 100     | 238       | 187    | 62.8    | 211                | 328 | 210                | 326 | 209                | 324 |
| 31000                    | -65      | 100     | 238       | 187    | 62.9    | 210                | 332 | 209                | 331 | 208                | 328 |

Figure 5.11.10 - Cruise performance  
Normal cruise / ISA - 20°C



## Normal cruise (Recommended)

Conditions :

- **ISA - 10°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Power recommended by PRATT & WHITNEY CANADA

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 5 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 6        | 100     | 329       | 258    | 86.9    | 238                | 239 | 238                | 239 | 237                | 238 |
| 5000                     | -4       | 100     | 302       | 237    | 79.7    | 234                | 252 | 233                | 251 | 233                | 250 |
| 10000                    | -14      | 100     | 281       | 220    | 74.2    | 229                | 265 | 228                | 265 | 228                | 264 |
| 15000                    | -24      | 100     | 268       | 210    | 70.8    | 224                | 280 | 223                | 279 | 222                | 278 |
| 18000                    | -30      | 100     | 259       | 203    | 68.4    | 221                | 289 | 220                | 288 | 219                | 287 |
| 20000                    | -34      | 100     | 253       | 199    | 66.9    | 219                | 296 | 218                | 295 | 217                | 294 |
| 21000                    | -36      | 100     | 251       | 197    | 66.2    | 218                | 299 | 217                | 298 | 216                | 297 |
| 22000                    | -38      | 100     | 249       | 195    | 65.7    | 217                | 303 | 216                | 302 | 215                | 300 |
| 23000                    | -40      | 100     | 247       | 194    | 65.1    | 216                | 306 | 215                | 305 | 214                | 304 |
| 24000                    | -42      | 100     | 245       | 192    | 64.7    | 215                | 310 | 214                | 309 | 213                | 307 |
| 25000                    | -44      | 100     | 243       | 191    | 64.3    | 214                | 314 | 213                | 312 | 212                | 311 |
| 26000                    | -46      | 100     | 242       | 190    | 63.9    | 213                | 317 | 212                | 316 | 211                | 314 |
| 27000                    | -47      | 100     | 242       | 190    | 63.8    | 212                | 321 | 211                | 320 | 210                | 318 |
| 28000                    | -49      | 100     | 241       | 189    | 63.6    | 211                | 325 | 210                | 323 | 209                | 322 |
| 29000                    | -51      | 100     | 238       | 187    | 62.9    | 210                | 328 | 209                | 327 | 207                | 324 |
| 30000                    | -53      | 96      | 230       | 180    | 60.7    | 206                | 328 | 204                | 326 | 203                | 323 |
| 31000                    | -55      | 93      | 222       | 174    | 58.6    | 202                | 328 | 200                | 325 | 198                | 322 |

Figure 5.11.11 - Cruise performance  
Normal cruise / ISA - 10°C

## Normal cruise (Recommended)

Conditions :

- **ISA - 5°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Power recommended by PRATT & WHITNEY CANADA

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 5 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 11       | 100     | 331       | 259    | 87.3    | 238                | 240 | 237                | 240 | 237                | 240 |
| 5000                     | 1        | 100     | 304       | 238    | 80.2    | 233                | 253 | 232                | 253 | 232                | 252 |
| 10000                    | -9       | 100     | 282       | 221    | 74.5    | 228                | 267 | 227                | 266 | 227                | 265 |
| 15000                    | -19      | 100     | 269       | 211    | 71.2    | 223                | 282 | 222                | 281 | 222                | 280 |
| 18000                    | -25      | 100     | 260       | 204    | 68.7    | 220                | 291 | 219                | 290 | 218                | 289 |
| 20000                    | -29      | 100     | 254       | 200    | 67.2    | 218                | 298 | 217                | 297 | 216                | 296 |
| 21000                    | -31      | 100     | 252       | 198    | 66.5    | 217                | 301 | 216                | 300 | 215                | 299 |
| 22000                    | -33      | 100     | 250       | 196    | 66.0    | 216                | 305 | 215                | 304 | 214                | 302 |
| 23000                    | -35      | 100     | 248       | 195    | 65.5    | 215                | 308 | 214                | 307 | 213                | 306 |
| 24000                    | -37      | 100     | 246       | 193    | 65.0    | 214                | 312 | 213                | 311 | 212                | 309 |
| 25000                    | -39      | 100     | 244       | 192    | 64.6    | 213                | 316 | 212                | 315 | 211                | 313 |
| 26000                    | -41      | 100     | 243       | 191    | 64.2    | 212                | 320 | 211                | 318 | 210                | 316 |
| 27000                    | -42      | 100     | 243       | 191    | 64.1    | 211                | 323 | 210                | 322 | 209                | 320 |
| 28000                    | -44      | 100     | 239       | 188    | 63.2    | 210                | 326 | 208                | 324 | 207                | 322 |
| 29000                    | -46      | 96      | 231       | 181    | 61.0    | 206                | 326 | 204                | 324 | 202                | 321 |
| 30000                    | -48      | 93      | 223       | 175    | 58.9    | 202                | 325 | 200                | 323 | 198                | 320 |
| 31000                    | -50      | 89      | 215       | 169    | 56.8    | 198                | 325 | 196                | 322 | 194                | 319 |

Figure 5.11.12 - Cruise performance  
Normal cruise / ISA - 5°C

## Normal cruise (Recommended)

Conditions :

- **ISA**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Power recommended by PRATT & WHITNEY CANADA

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 5 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 16       | 100     | 333       | 261    | 87.9    | 237                | 242 | 237                | 242 | 236                | 241 |
| 5000                     | 6        | 100     | 305       | 240    | 80.7    | 232                | 255 | 232                | 254 | 231                | 253 |
| 10000                    | -4       | 100     | 284       | 223    | 74.9    | 227                | 268 | 227                | 268 | 226                | 267 |
| 15000                    | -14      | 100     | 271       | 213    | 71.5    | 222                | 283 | 222                | 283 | 221                | 282 |
| 18000                    | -20      | 100     | 261       | 205    | 69.0    | 219                | 293 | 219                | 292 | 218                | 291 |
| 20000                    | -24      | 100     | 256       | 201    | 67.6    | 217                | 300 | 216                | 299 | 215                | 297 |
| 21000                    | -26      | 100     | 253       | 199    | 66.9    | 216                | 303 | 215                | 302 | 214                | 301 |
| 22000                    | -28      | 100     | 251       | 197    | 66.3    | 215                | 307 | 214                | 306 | 213                | 304 |
| 23000                    | -30      | 100     | 249       | 195    | 65.8    | 214                | 310 | 213                | 309 | 212                | 308 |
| 24000                    | -32      | 100     | 247       | 194    | 65.3    | 213                | 314 | 212                | 313 | 211                | 311 |
| 25000                    | -34      | 100     | 246       | 193    | 64.9    | 212                | 318 | 211                | 317 | 210                | 315 |
| 26000                    | -36      | 100     | 244       | 192    | 64.5    | 211                | 322 | 210                | 320 | 209                | 319 |
| 27000                    | -37      | 99      | 241       | 189    | 63.6    | 209                | 324 | 208                | 322 | 207                | 320 |
| 28000                    | -39      | 96      | 232       | 182    | 61.4    | 205                | 324 | 204                | 322 | 202                | 319 |
| 29000                    | -41      | 92      | 224       | 176    | 59.2    | 201                | 323 | 200                | 321 | 198                | 318 |
| 30000                    | -43      | 89      | 216       | 170    | 57.0    | 198                | 323 | 196                | 320 | 194                | 317 |
| 31000                    | -45      | 86      | 208       | 164    | 55.0    | 194                | 322 | 192                | 320 | 190                | 316 |

Figure 5.11.13 - Cruise performance  
Normal cruise / ISA

## Normal cruise (Recommended)

Conditions :

- **ISA + 5°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Power recommended by PRATT & WHITNEY CANADA

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 5 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 21       | 100     | 334       | 263    | 88.4    | 236                | 243 | 236                | 243 | 235                | 242 |
| 5000                     | 11       | 100     | 307       | 241    | 81.1    | 231                | 256 | 231                | 256 | 230                | 255 |
| 10000                    | 1        | 100     | 285       | 224    | 75.3    | 226                | 270 | 226                | 269 | 225                | 269 |
| 15000                    | -9       | 100     | 272       | 214    | 72.0    | 221                | 285 | 221                | 284 | 220                | 283 |
| 18000                    | -15      | 100     | 263       | 206    | 69.4    | 218                | 295 | 218                | 294 | 217                | 293 |
| 20000                    | -19      | 100     | 257       | 202    | 67.9    | 216                | 302 | 216                | 301 | 215                | 299 |
| 21000                    | -21      | 100     | 254       | 200    | 67.2    | 215                | 305 | 215                | 304 | 213                | 303 |
| 22000                    | -23      | 100     | 252       | 198    | 66.6    | 214                | 309 | 214                | 308 | 212                | 306 |
| 23000                    | -25      | 100     | 250       | 196    | 66.1    | 213                | 312 | 213                | 311 | 211                | 309 |
| 24000                    | -27      | 100     | 248       | 195    | 65.7    | 212                | 316 | 212                | 315 | 210                | 313 |
| 25000                    | -29      | 100     | 247       | 194    | 65.2    | 211                | 320 | 210                | 319 | 209                | 317 |
| 26000                    | -31      | 99      | 242       | 190    | 64.0    | 209                | 322 | 208                | 320 | 207                | 318 |
| 27000                    | -32      | 96      | 234       | 184    | 61.8    | 205                | 322 | 204                | 320 | 202                | 317 |
| 28000                    | -34      | 92      | 226       | 177    | 59.6    | 202                | 321 | 200                | 319 | 198                | 316 |
| 29000                    | -36      | 89      | 217       | 171    | 57.4    | 198                | 321 | 196                | 319 | 194                | 315 |
| 30000                    | -38      | 85      | 209       | 164    | 55.3    | 194                | 320 | 192                | 318 | 190                | 314 |
| 31000                    | -40      | 82      | 202       | 158    | 53.3    | 190                | 320 | 188                | 317 | 186                | 313 |

Figure 5.11.14 - Cruise performance  
Normal cruise / ISA + 5°C

## Normal cruise (Recommended)

Conditions :

- **ISA + 10°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Power recommended by PRATT & WHITNEY CANADA

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 6 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 26       | 100     | 336       | 264    | 88.9    | 236                | 245 | 235                | 244 | 235                | 244 |
| 5000                     | 16       | 100     | 309       | 242    | 81.6    | 231                | 258 | 230                | 257 | 230                | 256 |
| 10000                    | 6        | 100     | 287       | 225    | 75.7    | 226                | 272 | 225                | 271 | 224                | 270 |
| 15000                    | -4       | 100     | 274       | 215    | 72.3    | 221                | 287 | 220                | 286 | 219                | 285 |
| 18000                    | -10      | 100     | 264       | 207    | 69.7    | 218                | 297 | 217                | 296 | 216                | 294 |
| 20000                    | -14      | 100     | 258       | 203    | 68.3    | 216                | 303 | 215                | 302 | 214                | 301 |
| 21000                    | -16      | 100     | 256       | 201    | 67.6    | 215                | 307 | 214                | 306 | 213                | 304 |
| 22000                    | -18      | 100     | 254       | 199    | 67.0    | 214                | 311 | 213                | 309 | 211                | 308 |
| 23000                    | -20      | 100     | 252       | 197    | 66.5    | 212                | 314 | 212                | 313 | 210                | 311 |
| 24000                    | -22      | 100     | 250       | 196    | 66.0    | 212                | 318 | 211                | 317 | 209                | 315 |
| 25000                    | -24      | 99      | 244       | 192    | 64.6    | 209                | 320 | 208                | 318 | 207                | 316 |
| 26000                    | -26      | 96      | 236       | 185    | 62.3    | 205                | 320 | 204                | 318 | 203                | 316 |
| 27000                    | -27      | 92      | 227       | 178    | 60.1    | 202                | 319 | 200                | 317 | 198                | 315 |
| 28000                    | -29      | 89      | 219       | 172    | 57.9    | 198                | 319 | 196                | 317 | 194                | 313 |
| 29000                    | -31      | 85      | 211       | 166    | 55.8    | 194                | 318 | 192                | 316 | 190                | 312 |
| 30000                    | -33      | 82      | 203       | 160    | 53.7    | 190                | 318 | 188                | 315 | 186                | 311 |
| 31000                    | -35      | 79      | 196       | 154    | 51.7    | 186                | 317 | 184                | 313 | 182                | 309 |

Figure 5.11.15 - Cruise performance  
Normal cruise / ISA + 10°C

## Normal cruise (Recommended)

Conditions :

- **ISA + 20°C**
- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

● NOTE ● : Power recommended by PRATT & WHITNEY CANADA

If BLEED HI msg ON, reduce TRQ by 5 %. This TRQ reduction will result in an airspeed reduction of up to 6 KIAS.

| Pressure altitude (feet) | OAT (°C) | TRQ (%) | Fuel flow |        |         | Airspeeds (kt)     |     |                    |     |                    |     |
|--------------------------|----------|---------|-----------|--------|---------|--------------------|-----|--------------------|-----|--------------------|-----|
|                          |          |         |           |        |         | 5500 lbs (2495 kg) |     | 6300 lbs (2858 kg) |     | 7100 lbs (3220 kg) |     |
|                          |          |         | l / h     | kg / h | USG / h | IAS                | TAS | IAS                | TAS | IAS                | TAS |
| SL                       | 36       | 100     | 340       | 267    | 89.8    | 234                | 247 | 234                | 247 | 233                | 246 |
| 5000                     | 26       | 100     | 312       | 245    | 82.5    | 229                | 261 | 229                | 260 | 228                | 259 |
| 10000                    | 16       | 100     | 290       | 227    | 76.5    | 224                | 275 | 224                | 274 | 223                | 273 |
| 15000                    | 6        | 100     | 276       | 217    | 73.0    | 219                | 290 | 218                | 289 | 217                | 288 |
| 18000                    | 0        | 100     | 266       | 209    | 70.4    | 216                | 300 | 215                | 299 | 214                | 298 |
| 20000                    | -4       | 100     | 261       | 205    | 69.0    | 214                | 307 | 213                | 306 | 212                | 304 |
| 21000                    | -6       | 100     | 256       | 201    | 67.6    | 212                | 309 | 211                | 308 | 210                | 306 |
| 22000                    | -8       | 97      | 248       | 195    | 65.6    | 209                | 310 | 208                | 309 | 206                | 307 |
| 23000                    | -10      | 95      | 241       | 189    | 63.7    | 206                | 311 | 205                | 309 | 203                | 307 |
| 24000                    | -12      | 92      | 234       | 184    | 61.8    | 203                | 311 | 201                | 310 | 200                | 307 |
| 25000                    | -14      | 89      | 226       | 178    | 59.8    | 199                | 312 | 198                | 310 | 196                | 307 |
| 26000                    | -16      | 86      | 219       | 172    | 57.7    | 196                | 311 | 194                | 309 | 192                | 306 |
| 27000                    | -17      | 83      | 211       | 166    | 55.7    | 192                | 311 | 190                | 308 | 188                | 305 |
| 28000                    | -19      | 80      | 203       | 160    | 53.7    | 188                | 310 | 187                | 308 | 184                | 304 |
| 29000                    | -21      | 77      | 196       | 154    | 51.8    | 185                | 310 | 183                | 307 | 180                | 302 |
| 30000                    | -23      | 74      | 189       | 148    | 50.0    | 181                | 309 | 179                | 306 | 176                | 301 |
| 31000                    | -25      | 72      | 183       | 143    | 48.2    | 178                | 309 | 175                | 305 | 172                | 299 |

Figure 5.11.16 - Cruise performance  
Normal cruise / ISA + 20°C

## Pilot's Operating Handbook

## Long range cruise (5500 lbs - 2495 kg)

Conditions :

- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

Legend :

OAT : °C      IAS : KIAS  
FF : USG/h  
FF : kg/h      TAS : KTAS

| Pressure altitude (feet) | TRQ (%) | ISA - 20°C                               | ISA - 10°C                               | ISA                                      | ISA + 10°C                               | ISA + 20°C                               |
|--------------------------|---------|--|--|--|--|--|
| 15000                    | 38      | <b>-34 153</b><br>40.7<br>121 <b>189</b> | <b>-24 152</b><br>41.2<br>122 <b>192</b> | <b>-14 150</b><br>41.4<br>123 <b>193</b> | <b>-4 148</b><br>41.6<br>124 <b>194</b>  | <b>6 147</b><br>42.2<br>125 <b>197</b>   |
| 18000                    | 39      | <b>-40 150</b><br>38.2<br>113 <b>194</b> | <b>-30 149</b><br>38.7<br>115 <b>197</b> | <b>-20 148</b><br>39.2<br>116 <b>200</b> | <b>-10 147</b><br>39.7<br>118 <b>203</b> | <b>0 146</b><br>40.2<br>119 <b>205</b>   |
| 19000                    | 39      | <b>-42 149</b><br>37.4<br>111 <b>196</b> | <b>-32 148</b><br>37.9<br>113 <b>199</b> | <b>-22 147</b><br>38.4<br>114 <b>202</b> | <b>-12 145</b><br>38.7<br>115 <b>203</b> | <b>-2 143</b><br>38.9<br>116 <b>204</b>  |
| 20000                    | 39      | <b>-44 150</b><br>37.0<br>110 <b>201</b> | <b>-34 148</b><br>37.3<br>111 <b>202</b> | <b>-24 147</b><br>37.9<br>112 <b>205</b> | <b>-14 146</b><br>38.4<br>114 <b>208</b> | <b>-4 144</b><br>38.7<br>115 <b>209</b>  |
| 21000                    | 39      | <b>-46 148</b><br>36.0<br>107 <b>201</b> | <b>-36 147</b><br>36.6<br>109 <b>204</b> | <b>-26 146</b><br>37.1<br>110 <b>207</b> | <b>-16 145</b><br>37.6<br>112 <b>210</b> | <b>-6 144</b><br>38.2<br>113 <b>213</b>  |
| 22000                    | 39      | <b>-48 147</b><br>35.3<br>105 <b>203</b> | <b>-38 146</b><br>35.8<br>106 <b>206</b> | <b>-28 145</b><br>36.4<br>108 <b>209</b> | <b>-18 143</b><br>36.6<br>109 <b>211</b> | <b>-8 142</b><br>37.2<br>111 <b>214</b>  |
| 23000                    | 39      | <b>-50 146</b><br>34.5<br>103 <b>205</b> | <b>-40 145</b><br>35.1<br>104 <b>209</b> | <b>-30 144</b><br>35.6<br>106 <b>212</b> | <b>-20 142</b><br>35.9<br>107 <b>213</b> | <b>-10 141</b><br>36.4<br>108 <b>216</b> |
| 24000                    | 40      | <b>-52 146</b><br>34.1<br>101 <b>209</b> | <b>-42 145</b><br>34.6<br>103 <b>212</b> | <b>-32 144</b><br>35.2<br>104 <b>215</b> | <b>-22 142</b><br>35.4<br>105 <b>217</b> | <b>-12 141</b><br>36.0<br>107 <b>219</b> |

Figure 5.11.17 (1/2) - Cruise performance  
Long range cruise (5500 lbs - 2495 kg) (Altitude < 24000 ft)

## Long range cruise (5500 lbs - 2495 kg)

Conditions :

- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

Legend :

OAT : °C      IAS : KIAS  
FF : USG/h  
FF : kg/h      TAS : KTAS

| Pressure altitude (feet) | TRQ (%) | ISA - 20°C                               | ISA - 10°C                               | ISA                                      | ISA + 10°C                               | ISA + 20°C                               |
|--------------------------|---------|--|--|--|--|--|
| 24 000                   | 40      | <b>-52 146</b><br>34.1<br>101 <b>209</b> | <b>-42 145</b><br>34.6<br>103 <b>212</b> | <b>-32 144</b><br>35.2<br>104 <b>215</b> | <b>-22 142</b><br>35.4<br>105 <b>217</b> | <b>-12 141</b><br>36.0<br>107 <b>219</b> |
| 25 000                   | 41      | <b>-54 148</b><br>34.1<br>101 <b>215</b> | <b>-44 146</b><br>34.4<br>102 <b>217</b> | <b>-34 145</b><br>34.9<br>104 <b>220</b> | <b>-24 144</b><br>35.5<br>105 <b>223</b> | <b>-14 142</b><br>35.8<br>106 <b>225</b> |
| 26 000                   | 43      | <b>-56 151</b><br>34.6<br>103 <b>223</b> | <b>-46 150</b><br>35.1<br>104 <b>226</b> | <b>-36 148</b><br>35.4<br>105 <b>228</b> | <b>-26 146</b><br>35.6<br>106 <b>230</b> | <b>-16 145</b><br>36.2<br>108 <b>233</b> |
| 27 000                   | 45      | <b>-57 152</b><br>34.6<br>103 <b>228</b> | <b>-47 151</b><br>35.1<br>104 <b>232</b> | <b>-37 150</b><br>35.7<br>106 <b>235</b> | <b>-27 148</b><br>36.0<br>107 <b>237</b> | <b>-17 147</b><br>36.5<br>108 <b>241</b> |
| 28 000                   | 46      | <b>-59 153</b><br>34.5<br>103 <b>233</b> | <b>-49 152</b><br>35.1<br>104 <b>237</b> | <b>-39 151</b><br>35.7<br>106 <b>241</b> | <b>-29 149</b><br>36.0<br>107 <b>243</b> | <b>-19 147</b><br>36.3<br>108 <b>245</b> |
| 29 000                   | 46      | <b>-61 153</b><br>34.3<br>102 <b>237</b> | <b>-51 151</b><br>34.6<br>103 <b>240</b> | <b>-41 150</b><br>35.2<br>104 <b>244</b> | <b>-31 148</b><br>35.5<br>105 <b>246</b> | <b>-21 146</b><br>35.7<br>106 <b>248</b> |
| 30 000                   | 46      | <b>-63 153</b><br>34.2<br>101 <b>241</b> | <b>-53 151</b><br>34.4<br>102 <b>244</b> | <b>-43 149</b><br>34.7<br>103 <b>246</b> | <b>-33 148</b><br>35.3<br>105 <b>250</b> | <b>-23 146</b><br>35.6<br>106 <b>252</b> |
| 31 000                   | 46      | <b>-65 152</b><br>33.7<br>100 <b>244</b> | <b>-55 150</b><br>34.0<br>101 <b>247</b> | <b>-45 148</b><br>34.3<br>102 <b>249</b> | <b>-35 147</b><br>34.8<br>103 <b>253</b> | <b>-25 145</b><br>35.1<br>104 <b>255</b> |

Figure 5.11.17 (2/2) - Cruise performance  
Long range cruise (5500 lbs - 2495 kg) (Altitude > 24000 ft)



## Pilot's Operating Handbook

## Long range cruise (6300 lbs - 2858 kg)

Conditions :

- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

Legend :

**OAT** : °C      **IAS** : KIAS  
**FF** : USG/h  
**FF** : kg/h      **TAS** : KTAS

| Pressure altitude (feet) | TRQ (%) | ISA - 20°C                               | ISA - 10°C                               | ISA                                      | ISA + 10°C                               | ISA + 20°C                               |
|--------------------------|---------|--|--|--|--|--|
| 15 000                   | 42      | <b>-34 156</b><br>42.3<br>126 <b>193</b> | <b>-24 155</b><br>42.9<br>128 <b>195</b> | <b>-14 154</b><br>43.5<br>129 <b>198</b> | <b>-4 153</b><br>44.0<br>131 <b>201</b>  | <b>6 152</b><br>44.6<br>133 <b>203</b>   |
| 18 000                   | 42      | <b>-40 154</b><br>40.0<br>119 <b>199</b> | <b>-30 152</b><br>40.4<br>120 <b>201</b> | <b>-20 151</b><br>41.0<br>122 <b>204</b> | <b>-10 150</b><br>41.6<br>124 <b>207</b> | <b>0 149</b><br>42.1<br>125 <b>209</b>   |
| 19 000                   | 43      | <b>-42 156</b><br>40.0<br>119 <b>205</b> | <b>-32 154</b><br>40.3<br>120 <b>207</b> | <b>-22 152</b><br>40.7<br>121 <b>209</b> | <b>-12 151</b><br>41.3<br>123 <b>211</b> | <b>-2 150</b><br>41.9<br>124 <b>214</b>  |
| 20 000                   | 43      | <b>-44 154</b><br>38.9<br>116 <b>206</b> | <b>-34 153</b><br>39.5<br>117 <b>209</b> | <b>-24 151</b><br>39.9<br>118 <b>211</b> | <b>-14 150</b><br>40.5<br>120 <b>214</b> | <b>-4 149</b><br>41.1<br>122 <b>216</b>  |
| 21 000                   | 44      | <b>-46 153</b><br>38.2<br>113 <b>208</b> | <b>-36 152</b><br>38.7<br>115 <b>211</b> | <b>-26 151</b><br>39.4<br>117 <b>214</b> | <b>-16 150</b><br>39.9<br>119 <b>217</b> | <b>-6 149</b><br>40.6<br>121 <b>220</b>  |
| 22 000                   | 44      | <b>-48 152</b><br>37.4<br>111 <b>210</b> | <b>-38 151</b><br>38.0<br>113 <b>213</b> | <b>-28 150</b><br>38.6<br>115 <b>216</b> | <b>-18 149</b><br>39.2<br>117 <b>219</b> | <b>-8 148</b><br>39.8<br>118 <b>222</b>  |
| 23 000                   | 44      | <b>-50 152</b><br>36.9<br>110 <b>213</b> | <b>-40 151</b><br>37.5<br>111 <b>217</b> | <b>-30 149</b><br>37.9<br>113 <b>219</b> | <b>-20 148</b><br>38.5<br>114 <b>222</b> | <b>-10 147</b><br>39.1<br>116 <b>225</b> |
| 24 000                   | 44      | <b>-52 150</b><br>36.0<br>107 <b>214</b> | <b>-42 149</b><br>36.6<br>109 <b>218</b> | <b>-32 148</b><br>37.2<br>111 <b>221</b> | <b>-22 147</b><br>37.8<br>112 <b>224</b> | <b>-12 146</b><br>38.4<br>114 <b>227</b> |

Figure 5.11.18 (1/2) - Cruise performance  
Long range cruise (6300 lbs - 2858 kg) (Altitude < 24000 ft)

## Long range cruise (6300 lbs - 2858 kg)

Conditions :

- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

Legend :

OAT : °C      IAS : KIAS  
FF : USG/h  
FF : kg/h      TAS : KTAS

| Pressure altitude (feet) | TRQ (%) | ISA - 20°C                 | ISA - 10°C                 | ISA                        | ISA + 10°C                 | ISA + 20°C                 |
|--------------------------|---------|----------------------------|----------------------------|----------------------------|----------------------------|----------------------------|
| 24 000                   | 44      | -52 150<br>36.0<br>107 214 | -42 149<br>36.6<br>109 218 | -32 148<br>37.2<br>111 221 | -22 147<br>37.8<br>112 224 | -12 146<br>38.4<br>114 227 |
| 25 000                   | 44      | -54 149<br>35.4<br>105 216 | -44 148<br>36.0<br>107 220 | -34 147<br>36.6<br>109 223 | -24 145<br>36.9<br>110 225 | -14 143<br>37.2<br>111 226 |
| 26 000                   | 45      | -56 152<br>35.9<br>107 224 | -46 150<br>36.2<br>108 226 | -36 148<br>36.6<br>109 228 | -26 147<br>37.2<br>111 232 | -16 146<br>37.8<br>112 235 |
| 27 000                   | 47      | -57 154<br>36.2<br>107 231 | -47 152<br>36.5<br>108 233 | -37 150<br>36.9<br>109 235 | -27 148<br>37.2<br>111 237 | -17 147<br>37.8<br>112 241 |
| 28 000                   | 49      | -59 156<br>36.5<br>108 238 | -49 154<br>36.8<br>109 240 | -39 152<br>37.2<br>111 243 | -29 151<br>37.8<br>112 246 | -19 149<br>38.2<br>113 248 |
| 29 000                   | 49      | -61 155<br>36.1<br>107 240 | -51 153<br>36.4<br>108 243 | -41 151<br>36.8<br>109 245 | -31 149<br>37.1<br>110 247 | -21 147<br>37.4<br>111 249 |
| 30 000                   | 50      | -63 155<br>35.9<br>107 244 | -53 153<br>36.2<br>108 247 | -43 151<br>36.6<br>109 250 | -33 149<br>37.0<br>110 252 | -23 147<br>37.3<br>111 254 |
| 31 000                   | 50      | -65 154<br>35.5<br>105 247 | -55 152<br>35.8<br>106 250 | -45 150<br>36.2<br>108 252 | -35 148<br>36.6<br>109 255 | -25 146<br>37.0<br>110 257 |

Figure 5.11.18 (2/2) - Cruise performance  
Long range cruise (6300 lbs - 2858 kg) (Altitude > 24000 ft)

## Pilot's Operating Handbook

## Long range cruise (7100 lbs - 3220 kg)

Conditions :

- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

Legend :

OAT : °C      IAS : KIAS  
FF : USG/h  
FF : kg/h      TAS : KTAS

| Pressure altitude (feet) | TRQ (%) | ISA - 20°C                               | ISA - 10°C                               | ISA                                      | ISA + 10°C                               | ISA + 20°C                               |
|--------------------------|---------|--|--|--|--|--|
| 15 000                   | 48      | <b>-34 164</b><br>45.4<br>135 <b>202</b> | <b>-24 163</b><br>46.0<br>137 <b>205</b> | <b>-14 162</b><br>46.7<br>139 <b>208</b> | <b>-4 161</b><br>47.4<br>141 <b>211</b>  | <b>6 160</b><br>48.0<br>143 <b>213</b>   |
| 18 000                   | 49      | <b>-40 161</b><br>42.7<br>127 <b>208</b> | <b>-30 160</b><br>43.5<br>129 <b>211</b> | <b>-20 159</b><br>43.9<br>130 <b>214</b> | <b>-10 158</b><br>44.8<br>133 <b>217</b> | <b>0 157</b><br>45.5<br>135 <b>220</b>   |
| 19 000                   | 49      | <b>-42 160</b><br>42.0<br>125 <b>210</b> | <b>-32 159</b><br>42.6<br>127 <b>213</b> | <b>-22 158</b><br>43.3<br>129 <b>217</b> | <b>-12 157</b><br>44.0<br>131 <b>219</b> | <b>-2 156</b><br>44.6<br>133 <b>222</b>  |
| 20 000                   | 49      | <b>-44 160</b><br>41.4<br>123 <b>214</b> | <b>-34 159</b><br>42.1<br>125 <b>217</b> | <b>-24 157</b><br>42.5<br>126 <b>219</b> | <b>-14 156</b><br>43.2<br>128 <b>222</b> | <b>-4 155</b><br>43.9<br>130 <b>225</b>  |
| 21 000                   | 49      | <b>-46 158</b><br>40.4<br>120 <b>214</b> | <b>-36 157</b><br>41.1<br>122 <b>218</b> | <b>-26 156</b><br>41.8<br>124 <b>221</b> | <b>-16 155</b><br>42.4<br>126 <b>224</b> | <b>-6 154</b><br>43.1<br>128 <b>227</b>  |
| 22 000                   | 49      | <b>-48 157</b><br>39.8<br>118 <b>217</b> | <b>-38 156</b><br>40.4<br>120 <b>220</b> | <b>-28 155</b><br>41.0<br>122 <b>223</b> | <b>-18 153</b><br>41.4<br>123 <b>225</b> | <b>-8 152</b><br>42.1<br>125 <b>228</b>  |
| 23 000                   | 49      | <b>-50 155</b><br>38.9<br>116 <b>217</b> | <b>-40 154</b><br>39.5<br>117 <b>221</b> | <b>-30 153</b><br>40.1<br>119 <b>224</b> | <b>-20 150</b><br>40.3<br>120 <b>225</b> | <b>-10 148</b><br>40.7<br>121 <b>226</b> |
| 24 000                   | 49      | <b>-52 154</b><br>38.3<br>114 <b>220</b> | <b>-42 153</b><br>38.9<br>116 <b>223</b> | <b>-32 152</b><br>39.6<br>118 <b>227</b> | <b>-22 150</b><br>40.0<br>119 <b>228</b> | <b>-12 149</b><br>40.6<br>121 <b>231</b> |

Figure 5.11.19 (1/2) - Cruise performance  
Long range cruise (7100 lbs - 3220 kg) (Altitude < 24000 ft)

## Long range cruise (7100 lbs - 3220 kg)

Conditions :

- Landing gear and flaps UP
- BLEED switch on AUTO and BLEED HI msg OFF

Legend :

OAT : °C      IAS : KIAS  
FF : USG/h  
FF : kg/h      TAS : KTAS

| Pressure altitude (feet) | TRQ (%) | ISA - 20°C                               | ISA - 10°C                               | ISA                                      | ISA + 10°C                               | ISA + 20°C                               |
|--------------------------|---------|--|--|--|--|--|
| 24 000                   | 49      | <b>-52 154</b><br>38.3<br>114 <b>220</b> | <b>-42 153</b><br>38.9<br>116 <b>223</b> | <b>-32 152</b><br>39.6<br>118 <b>227</b> | <b>-22 150</b><br>40.0<br>119 <b>228</b> | <b>-12 149</b><br>40.6<br>121 <b>231</b> |
| 25 000                   | 49      | <b>-54 153</b><br>37.7<br>112 <b>222</b> | <b>-44 152</b><br>38.3<br>114 <b>226</b> | <b>-34 151</b><br>39.0<br>116 <b>229</b> | <b>-24 149</b><br>39.4<br>117 <b>231</b> | <b>-14 147</b><br>39.8<br>118 <b>232</b> |
| 26 000                   | 51      | <b>-56 153</b><br>37.4<br>111 <b>226</b> | <b>-46 151</b><br>37.9<br>113 <b>228</b> | <b>-36 150</b><br>38.5<br>114 <b>231</b> | <b>-26 149</b><br>39.2<br>117 <b>235</b> | <b>-16 148</b><br>39.8<br>118 <b>238</b> |
| 27 000                   | 52      | <b>-57 155</b><br>37.7<br>112 <b>232</b> | <b>-47 153</b><br>38.1<br>113 <b>235</b> | <b>-37 151</b><br>38.5<br>114 <b>237</b> | <b>-27 149</b><br>39.0<br>116 <b>239</b> | <b>-17 148</b><br>39.6<br>118 <b>242</b> |
| 28 000                   | 53      | <b>-59 157</b><br>38.1<br>113 <b>239</b> | <b>-49 154</b><br>38.2<br>114 <b>240</b> | <b>-39 152</b><br>38.7<br>115 <b>243</b> | <b>-29 150</b><br>39.1<br>116 <b>245</b> | <b>-19 149</b><br>39.8<br>118 <b>248</b> |
| 29 000                   | 53      | <b>-61 156</b><br>37.7<br>112 <b>242</b> | <b>-51 154</b><br>38.1<br>113 <b>244</b> | <b>-41 152</b><br>38.6<br>115 <b>247</b> | <b>-31 150</b><br>39.0<br>116 <b>249</b> | <b>-21 148</b><br>39.5<br>117 <b>251</b> |
| 30 000                   | 53      | <b>-63 155</b><br>37.3<br>111 <b>244</b> | <b>-53 153</b><br>37.8<br>112 <b>247</b> | <b>-43 151</b><br>38.2<br>113 <b>250</b> | <b>-33 149</b><br>38.7<br>115 <b>252</b> | <b>-23 147</b><br>39.1<br>116 <b>254</b> |
| 31 000                   | 49      | <b>-65 155</b><br>37.3<br>111 <b>249</b> | <b>-55 153</b><br>37.7<br>112 <b>251</b> | <b>-45 150</b><br>37.9<br>113 <b>252</b> | <b>-35 148</b><br>38.3<br>114 <b>255</b> | <b>-25 146</b><br>38.8<br>115 <b>257</b> |

Figure 5.11.19 (2/2) - Cruise performance  
Long range cruise (7100 lbs - 3220 kg) (Altitude > 24000 ft)

## 5.12 - Time, consumption and descent distance

Conditions :

- Power as required to maintain constant Vz
- Landing gear and flaps UP
- CAS = 230 KCAS - BLEED switch on AUTO

| Pressure altitude (feet) | Vz = 1500 ft/min |          |    |        |            | Vz = 2000 ft/min |          |    |        |            | Vz = 2500 ft/min |          |    |        |            |
|--------------------------|------------------|----------|----|--------|------------|------------------|----------|----|--------|------------|------------------|----------|----|--------|------------|
|                          | Time (min. s)    | Consump. |    |        | Dist. (NM) | Time (min. s)    | Consump. |    |        | Dist. (NM) | Time (min. s)    | Consump. |    |        | Dist. (NM) |
|                          |                  | l        | kg | us gal |            |                  | l        | kg | us gal |            |                  | l        | kg | us gal |            |
| 31000                    | 20:40            | 70       | 55 | 18.5   | 101        | 15:30            | 47       | 37 | 12.4   | 75         | 12:25            | 34       | 27 | 9.0    | 60         |
| 30000                    | 20:00            | 68       | 53 | 17.9   | 97         | 15:00            | 45       | 36 | 12.0   | 72         | 12:00            | 33       | 26 | 8.8    | 58         |
| 28000                    | 18:40            | 64       | 50 | 16.8   | 89         | 14:00            | 43       | 34 | 11.3   | 66         | 11:10            | 31       | 25 | 8.3    | 53         |
| 26000                    | 17:20            | 59       | 47 | 15.7   | 81         | 13:00            | 40       | 31 | 10.6   | 61         | 10:25            | 29       | 23 | 7.8    | 48         |
| 24000                    | 16:00            | 55       | 43 | 14.5   | 73         | 12:00            | 37       | 29 | 9.8    | 55         | 09:35            | 28       | 22 | 7.3    | 44         |
| 22000                    | 14:40            | 51       | 40 | 13.4   | 66         | 11:00            | 34       | 27 | 9.1    | 50         | 08:50            | 26       | 20 | 6.8    | 40         |
| 20000                    | 13:20            | 47       | 37 | 12.3   | 59         | 10:00            | 32       | 25 | 8.4    | 44         | 08:00            | 24       | 19 | 6.3    | 35         |
| 18000                    | 12:00            | 42       | 33 | 11.1   | 53         | 09:00            | 29       | 23 | 7.6    | 39         | 07:10            | 22       | 17 | 5.8    | 31         |
| 16000                    | 10:40            | 38       | 30 | 10.0   | 46         | 08:00            | 26       | 20 | 6.8    | 34         | 06:25            | 20       | 15 | 5.2    | 27         |
| 14000                    | 09:20            | 33       | 26 | 8.8    | 40         | 07:00            | 23       | 18 | 6.1    | 30         | 05:35            | 18       | 14 | 4.6    | 24         |
| 12000                    | 08:00            | 29       | 23 | 7.6    | 33         | 06:00            | 20       | 16 | 5.3    | 25         | 04:50            | 15       | 12 | 4.1    | 20         |
| 10000                    | 06:40            | 24       | 19 | 6.4    | 27         | 05:00            | 17       | 13 | 4.5    | 21         | 04:00            | 13       | 10 | 3.4    | 16         |
| 8000                     | 05:20            | 20       | 15 | 5.2    | 22         | 04:00            | 14       | 11 | 3.7    | 16         | 03:10            | 11       | 8  | 2.8    | 13         |
| 6000                     | 04:00            | 15       | 12 | 3.9    | 16         | 03:00            | 11       | 8  | 2.8    | 12         | 02:25            | 8        | 6  | 2.2    | 10         |
| 4000                     | 02:40            | 10       | 8  | 2.7    | 10         | 02:00            | 7        | 6  | 1.9    | 8          | 01:35            | 6        | 4  | 1.5    | 6          |
| 2000                     | 01:20            | 5        | 4  | 1.4    | 5          | 01:00            | 4        | 3  | 1.0    | 4          | 00:50            | 3        | 2  | 0.8    | 3          |
| SL                       | 00:00            | 0        | 0  | 0      | 0          | 00:00            | 0        | 0  | 0      | 0          | 00:00            | 0        | 0  | 0      | 0          |

Figure 5.12.1 - Time, consumption and descent distance

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### 5.13 - Holding time

Conditions :

- Landing gear and flaps UP
- IAS = 120 KIAS - BLEED switch on AUTO
- TRQ  $\approx$  26 %

| Pressure altitude (feet) | Fuel used during holding time |    |     |        |    |      |                           |    |     |        |    |      |
|--------------------------|-------------------------------|----|-----|--------|----|------|---------------------------|----|-----|--------|----|------|
|                          | Weight 5500 lbs (2495 kg)     |    |     |        |    |      | Weight 6300 lbs (2858 kg) |    |     |        |    |      |
|                          | 10 min                        |    |     | 30 min |    |      | 10 min                    |    |     | 30 min |    |      |
|                          | l                             | kg | USG | l      | kg | USG  | l                         | kg | USG | l      | kg | USG  |
| SL                       | 30                            | 23 | 7.8 | 89     | 70 | 23.5 | 30                        | 24 | 8.0 | 91     | 71 | 24.1 |
| 5000                     | 26                            | 21 | 6.9 | 79     | 62 | 20.8 | 27                        | 21 | 7.1 | 81     | 64 | 21.4 |
| 10000                    | 24                            | 18 | 6.2 | 71     | 55 | 18.7 | 24                        | 19 | 6.5 | 73     | 58 | 19.4 |
| 15000                    | 22                            | 17 | 5.8 | 66     | 51 | 17.3 | 23                        | 18 | 6.0 | 69     | 54 | 18.1 |
| 20000                    | 20                            | 16 | 5.3 | 60     | 47 | 15.9 | 21                        | 17 | 5.6 | 63     | 50 | 16.7 |

Figure 5.13.1 - Holding time

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## 5.14 - Landing distances

The following tables give the landing distances for several weight configurations.

All common information applicable to tables (pages 5.14.2 to 5.14.4) are listed below.

Associated conditions :

- Landing gear DN and flaps LDG
- Maximum braking without reverse
- Hard, dry and level runway

In table headings :

- GR = Ground roll (in ft)
- D<sub>50</sub> = Landing distance (clear to 50 ft) (in ft)

Corrections :

- In case of wind, apply the following corrections :
  - Reduce total distances by 10 % every 10 kts of headwind
  - Increase total distances by 30 % every 10 kts of tail wind
- Other runway surfaces :

Landing distances given in the tables are for landing on hard, dry and level runway. Other runway surfaces require the following correction factors.

Increase distances by :

7 % on hard grass

10 % on short grass

15 % on wet runway

25 % on high grass

30 % on slippery runway

## Weight : 7024 lbs (3186 kg)

Associated conditions :

- Approach speed IAS = 85 KIAS
- Touch-down speed IAS = 78 KIAS

| Pressure<br>altitude ft | ISA - 35°C |      | ISA - 20°C |      | ISA - 10°C |      | ISA        |      |
|-------------------------|------------|------|------------|------|------------|------|------------|------|
|                         | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0                       | 1575       | 2135 | 1675       | 2265 | 1740       | 2330 | 1840       | 2430 |
| 2000                    | 1675       | 2265 | 1805       | 2395 | 1870       | 2495 | 1970       | 2590 |
| 4000                    | 1805       | 2395 | 1940       | 2560 | 2035       | 2660 | 2135       | 2790 |
| 6000                    | 1940       | 2560 | 2100       | 2725 | 2200       | 2855 | 2300       | 2955 |
| 8000                    | 2100       | 2725 | 2265       | 2920 | 2360       | 3020 | 2495       | 3180 |
| Pressure<br>altitude ft | ISA + 10°C |      | ISA + 20°C |      | ISA + 30°C |      | ISA + 37°C |      |
|                         | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0                       | 1905       | 2530 | 2000       | 2625 | 2070       | 2690 | 2135       | 2790 |
| 2000                    | 2070       | 2690 | 2135       | 2790 | 2230       | 2890 | 2300       | 2955 |
| 4000                    | 2230       | 2890 | 2330       | 2985 | 2430       | 3085 | 2495       | 3185 |
| 6000                    | 2395       | 3050 | 2530       | 3215 | 2625       | 3315 | 2690       | 3380 |
| 8000                    | 2590       | 3280 | 2725       | 3410 | 2855       | 3570 | 2920       | 3640 |

Figure 5.14.1 - Landing distances - 7024 lbs (3186 kg)

### ▲ CAUTION ▲

Refer to page 5.14.1 for correction factors.



**Weight : 6250 lbs (2835 kg)**

Associated conditions :

- Approach speed IAS = 80 KIAS
- Touch-down speed IAS = 65 KIAS

| Pressure altitude ft | ISA - 35°C |      | ISA - 20°C |      | ISA - 10°C |      | ISA        |      |
|----------------------|------------|------|------------|------|------------|------|------------|------|
|                      | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0                    | 1050       | 1900 | 1115       | 2000 | 1180       | 2070 | 1215       | 2135 |
| 2000                 | 1115       | 2000 | 1215       | 2100 | 1245       | 2200 | 1310       | 2265 |
| 4000                 | 1180       | 2100 | 1280       | 2230 | 1345       | 2330 | 1410       | 2395 |
| 6000                 | 1280       | 2230 | 1380       | 2360 | 1445       | 2460 | 1510       | 2525 |
| 8000                 | 1380       | 2360 | 1475       | 2490 | 1540       | 2590 | 1610       | 2690 |
| Pressure altitude ft | ISA + 10°C |      | ISA + 20°C |      | ISA + 30°C |      | ISA + 37°C |      |
|                      | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0                    | 1280       | 2200 | 1310       | 2300 | 1380       | 2360 | 1445       | 2430 |
| 2000                 | 1345       | 2330 | 1410       | 2430 | 1475       | 2495 | 1540       | 2560 |
| 4000                 | 1445       | 2460 | 1510       | 2560 | 1575       | 2655 | 1640       | 2755 |
| 6000                 | 1575       | 2645 | 1640       | 2720 | 1705       | 2820 | 1770       | 2920 |
| 8000                 | 1705       | 2790 | 1770       | 2885 | 1835       | 2985 | 1900       | 3085 |

Figure 5.14.2 - Landing distances - 6250 lbs (2835 kg)

▲ **CAUTION** ▲

Refer to page 5.14.1 for correction factors.



## Weight : 5071 lbs (2300 kg)

Associated conditions :

- Approach speed IAS = 80 KIAS
- Touch-down speed IAS = 60 KIAS

| Pressure<br>altitude ft | ISA - 35°C |      | ISA - 20°C |      | ISA - 10°C |      | ISA        |      |
|-------------------------|------------|------|------------|------|------------|------|------------|------|
|                         | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0                       | 885        | 1900 | 950        | 2000 | 1000       | 2070 | 1030       | 2135 |
| 2000                    | 950        | 2000 | 1030       | 2100 | 1065       | 2200 | 1115       | 2265 |
| 4000                    | 1000       | 2100 | 1080       | 2230 | 1150       | 2330 | 1200       | 2395 |
| 6000                    | 1080       | 2230 | 1180       | 2360 | 1230       | 2460 | 1280       | 2525 |
| 8000                    | 1180       | 2360 | 1245       | 2490 | 1310       | 2590 | 1360       | 2690 |
| Pressure<br>altitude ft | ISA + 10°C |      | ISA + 20°C |      | ISA + 30°C |      | ISA + 37°C |      |
|                         | GR         | D50  | GR         | D50  | GR         | D50  | GR         | D50  |
| 0                       | 1080       | 2200 | 1115       | 2300 | 1180       | 2360 | 1230       | 2430 |
| 2000                    | 1150       | 2330 | 1200       | 2430 | 1245       | 2495 | 1310       | 2560 |
| 4000                    | 1230       | 2460 | 1280       | 2560 | 1345       | 2655 | 1395       | 2755 |
| 6000                    | 1345       | 2645 | 1395       | 2720 | 1445       | 2820 | 1510       | 2920 |
| 8000                    | 1445       | 2790 | 1510       | 2885 | 1560       | 2985 | 1610       | 3085 |

Figure 5.14.3 - Landing distances - 5071 lbs (2300 kg)

### ▲ CAUTION ▲

Refer to page 5.14.1 for correction factors.



## Section 6

### Weight and balance

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## 6.1 - General

This section is intended to provide the pilot with the procedure to determine the weight and balance of the airplane.

**▲ WARNING ▲**

**It is the pilot's responsibility to ensure that the airplane is properly loaded and the weight and balance limits are adhered to.**



■ This airplane allows multiple cabin seat configurations between 2 seats and 6 seats, as required by the operator - refer to chapter 7.3.

A list of equipment available for this airplane is referenced at the end of this POH - refer to chapter 6.5.

The list of specific optional equipment installed on your airplane as delivered from the factory can be found in the records carried in the airplane.

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## **6.2 - Airplane weighing procedures**

Refer to maintenance manual for the procedures to use.

• NOTE •

Weighing carried out at the factory takes into account all equipment installed on the airplane. The list of this equipment and the total weight is noted in the individual inspection record.

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### 6.3 - Baggage loading

There are two baggage compartments :

- one in fuselage non pressurized forward section, between firewall and cockpit with maximum baggage capacity of 110 lbs (50 kg),
- one located in the rear of the pressurized cabin with following characteristics :

>> *With 6-seat configuration*

- in the baggage compartment, behind the rear seats, with maximum loading capacity of 220 lbs (100 kg).
- stowing straps are provided for securing parcels and baggage on compartment floor. A partition net separating the cabin from the baggage compartment is attached to frame C14.

>> *With other allowed seat accommodations*

There are two loading areas :

- one in place of the 2 removed rear seats, with maximum loading capacity of 176 lbs (80 kg),
- one, in the baggage compartment, behind the rear seats area, with maximum loading capacity of 220 lbs (100 kg).

Two types of baggage securing nets can be used :

- the small cargo net is attached through nine anchoring points on seat rails, between frame C11 and frame C13bis - refer to section 2 for limitations, Figure 7.2.1B.
- the large cargo net is attached through seven anchoring points on seat rails, between frame C11 and frame C13bis and six anchoring points on fuselage sides, at frame C14 - refer to section 2 for limitations, Figure 7.2.1A.

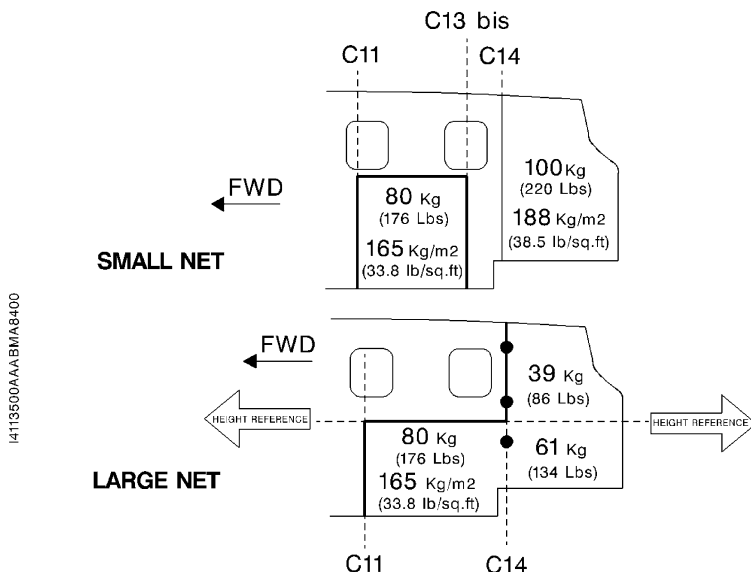


Figure 6.3.1 - Baggage limits

Authorized anchoring points are identified with green self-adhesive labels affixed to the inside of the rail.

A placard indicates loading limits for each securing net.

Evenly distribute the load within the cargo zone and ensure that overall weight is centered.

When using the large net, distribute the weight in each zone, delineated by the step in the floor, according to the zone limits.

&gt;&gt; All

**▲ WARNING ▲**

**It is the pilot's responsibility to check that all parcels and baggages are properly secured in the cabin.**

**Transport of dangerous product is normally prohibited, however if transport of such product is necessary, it must be performed in compliance with regulations concerning transport of dangerous product and any other applicable regulation.**



Baggage compartments loading must be done in accordance with the weight and balance limits of the airplane - refer to section 2 for limitations.

Generally, if rear seats are not used or are removed, first load AFT compartment, then, if required, FWD compartment. If rear seats are used, first load FWD compartment, then, if required, AFT compartment.

Compute and check the weight and balance diagram to ensure the airplane is within the allowable limits.

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## **6.4 - Determining weight and balance**

### **General**

This section is intended to provide the pilot with the procedure to determine the weight and balance of the airplane.

#### **▲ WARNING ▲**

**It is the pilot's responsibility to ensure that the airplane is properly loaded and that the weight and balance limits are adhered to.**



The procedure requires the following data related to the basic characteristics of the empty airplane to be obtained from the last airplane weight and balance report :

- the empty weight, expressed in kg or lbs,
- the moment, expressed in m.kg or in.lbs,
- the CG, expressed in MAC %

If the airplane empty weight has varied since last weight and balance report (for example, due to installation of optional equipment), refer to paragraph Determining empty airplane characteristics to determine new empty weight and the corresponding moment.

## Utilization of weight and balance graph

This procedure determines the airplane weight and balance characteristics for flight.

Select the units for the weight and balance determination, either m and kg, or lb and in, and use the dedicated form - see figures 6.4.3 or 6.4.4, appropriate to the chosen units.

- 1) Record the basic empty weight (1a), moment (1b) and CG (MAC %) (1c) from the last weight and balance report - see sample of weight and balance report, figures 6.4.1 and 6.4.2.
- 2) Record the expected loading (2a) and compute each associated moment (2b).
- 3) Compute zero fuel weight (3a) and moment (3b) as sum of all the above weights (1a)+(2a) and moments (1b) + (2b).
- 4) Check value (3a) to be below maximum zero fuel weight.
- 5) Compute zero fuel weight arm (5) and CG (MAC %) (5c) using given formulas.
- 6) Record the loaded fuel (6a) and compute associated moment (6b).
- 7) Compute ramp weight (7a) and moment (7b) as sum of zero fuel weight (3a) + loaded fuel (6a) and moments (3b) + (6b).
- 8) Check value (7a) to be below maximum ramp weight.
- 9) Compute ramp weight arm (9) and CG (MAC %) (9c) using given formulas.
- 10) Record the expected taxi fuel (negative value) (10a) and compute associated moment (10b).
- 11) Compute takeoff weight (11a) and moment (11b) as sum of ramp weight (7a) + taxi fuel (10a) and moments (7b) + (10b).
- 12) Check value (11a) to be below maximum takeoff weight.
- 13) Compute takeoff weight arm (13) and CG (MAC %) (13c) using given formulas.
- 14) Record the expected trip fuel (negative value) (14a) and compute associated moment (14b).
- 15) Compute landing weight (15a) and moment (15b) as sum of takeoff weight (11a) + trip fuel (14a) and moments (11b) + (14b).
- 16) Check value (15a) to be below maximum landing weight.
- 17) Compute landing weight arm (17) and CG (MAC %) (17c) using given formulas.
- 18) Plot zero fuel weight, takeoff weight and landing weight on weight and balance diagram.



- 19) Check that all points are within the weight and balance limits and check that they are vertically aligned.
- 20) Record these data on your navigation log.

## Airplane loading form (m, kg)

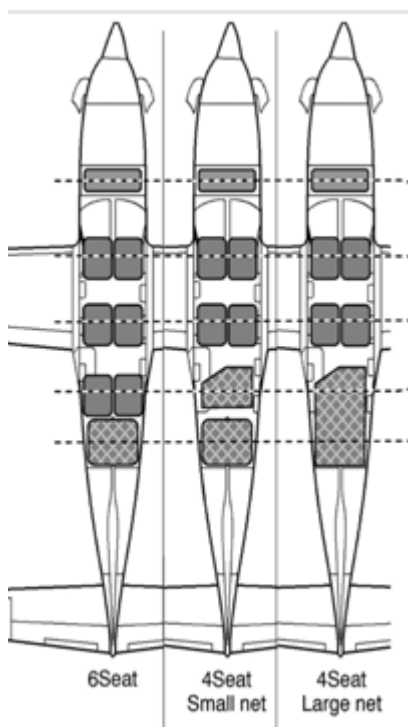
Moment = Weight x Arm

$$CG (MAC \%) = \frac{(Arm (m) - 4.392)}{1.51} \times 100$$

■ >> Up to S/N 1159

| Item                         |                           | Weight<br>(kg) | Arm<br>(m) | Moment<br>(m.kg) | CG<br>(MAC %) |
|------------------------------|---------------------------|----------------|------------|------------------|---------------|
| Empty weight (kg)            |                           | (1a)           |            | (1b)             | (1c)          |
| Baggage FWD (< 50 kg)        |                           | (2a)           | 3.250      | (2b)             |               |
| Front seats (kg)             |                           | (2a)           | 4.534      | (2b)             |               |
| Inter. seats                 | -17.5 kg per seat removed | (2a)           | 5.710      | (2b)             |               |
|                              | Pax                       | (2a)           |            | (2b)             |               |
| Rear bench/net               | -24 kg per seat removed   | (2a)           | 6.785      | (2b)             |               |
|                              | Pax                       | (2a)           |            | (2b)             |               |
|                              | Cargo (< 80 kg)           | (2a)           |            | (2b)             |               |
| Baggage AFT (< 100 kg)       |                           | (2a)           | 7.695      | (2b)             |               |
| Zero fuel weight (< 2736 kg) |                           | (3a)           | (5)        | (3b)             | (5c)          |
| Fuel (kg)                    |                           | (6a)           | 4.820      | (6b)             |               |
| Ramp weight (< 3370 kg)      |                           | (7a)           | (9)        | (7b)             | (9c)          |
| Taxi fuel (kg)               |                           | (10a)          | 4.820      | (10b)            |               |
| Takeoff weight (< 3354 kg)   |                           | (11a)          | (13)       | (11b)            | (13c)         |
| Trip fuel (kg)               |                           | (14a)          | 4.820      | (14b)            |               |
| Landing weight (< 3186 kg)   |                           | (15a)          | (17)       | (17)             | (17c)         |

■ >> Up to S/N 1159



| Item           |                           | Weight (kg) |
|----------------|---------------------------|-------------|
| Empty weight   |                           | (1a)        |
| Baggage FWD    | (< 50 kg)                 | (2a)        |
| Front seats    |                           | (2a)        |
| Inter. seats   | -17.5 kg per seat removed | (2a)        |
|                | Pax                       | (2a)        |
| Rear bench/net | -24 kg per seat removed   | (2a)        |
|                | Pax                       | (2a)        |
|                | Cargo (< 80 kg)           | (2a)        |
| Baggage AFT    | (< 100 kg)                | (2a)        |

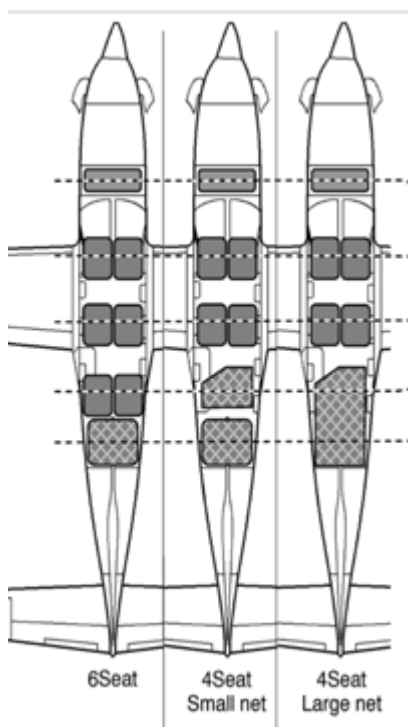
Moment = Weight x Arm

$$CG (MAC \%) = \frac{(Arm (m) - 4.392)}{1.51} \times 100$$

>> From S/N 1160

| Item                         |                           | Weight<br>(kg) | Arm<br>(m) | Moment<br>(m.kg) | CG<br>(MAC %) |
|------------------------------|---------------------------|----------------|------------|------------------|---------------|
| Empty weight (kg)            |                           | (1a)           |            | (1b)             | (1c)          |
| Baggage FWD (< 50 kg)        |                           | (2a)           | 3.250      | (2b)             |               |
| Front seats (kg)             |                           | (2a)           | 4.534      | (2b)             |               |
| Inter. seats                 | -15 kg per seat removed   | (2a)           | 5.710      | (2b)             |               |
|                              | Pax                       | (2a)           |            | (2b)             |               |
| Rear bench/net               | -20.5 kg per seat removed | (2a)           | 6.785      | (2b)             |               |
|                              | Pax                       | (2a)           |            | (2b)             |               |
|                              | Cargo (< 80 kg)           | (2a)           |            | (2b)             |               |
| Baggage AFT (< 100 kg)       |                           | (2a)           | 7.695      | (2b)             |               |
| Zero fuel weight (< 2736 kg) |                           | (3a)           | (5)        | (3b)             | (5c)          |
| Fuel (kg)                    |                           | (6a)           | 4.820      | (6b)             |               |
| Ramp weight (< 3370 kg)      |                           | (7a)           | (9)        | (7b)             | (9c)          |
| Taxi fuel (kg)               |                           | (10a)          | 4.820      | (10b)            |               |
| Takeoff weight (< 3354 kg)   |                           | (11a)          | (13)       | (11b)            | (13c)         |
| Trip fuel (kg)               |                           | (14a)          | 4.820      | (14b)            |               |
| Landing weight (< 3186 kg)   |                           | (15a)          | (17)       | (17)             | (17c)         |

■ >> From S/N 1160



| Item           |                           | Weight (kg) |
|----------------|---------------------------|-------------|
| Empty weight   |                           | (1a)        |
| Baggage FWD    | (< 50 kg)                 | (2a)        |
| Front seats    |                           | (2a)        |
| Inter. seats   | -15 kg per seat removed   | (2a)        |
|                | Pax                       | (2a)        |
| Rear bench/net | -20.5 kg per seat removed | (2a)        |
|                | Pax                       | (2a)        |
|                | Cargo (< 80 kg)           | (2a)        |
| Baggage AFT    | (< 100 kg)                | (2a)        |

## Example of airplane weight and balance report

• NOTE •

Airplane original report shall be kept with airplane POH

•

| REGISTRE INDIVIDUEL DE CONTROLE<br>INDIVIDUAL INSPECTION RECORD  |               | TBM 700  |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
|--|---------------|--|---------------|-----------|---------------|--------------|---------|-------------|----------|--------------|-------------|-------|--------|------------------|-------|------|-------|-------------------------|---------------|-----|--------|--|--|
| FICHE DE PESEE<br>CONFIGURATION 6 PLACES   |               | S/N :<br>F : 8   |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| <p>Mise de niveau :</p> <p>Rais de sièges</p>  |               | <p>SEULEMENT POUR INFORMATION, SE<br/>REFERER A LA LIMITATION DU MANUEL DE<br/>VOL SECTION 2</p> <p><b>LIMITATION</b></p> <p>Masse maxi au décollage : 3354 Kg H</p> <p>Masse maxi à l'atterrissage : 3186 Kg H</p> <p><b>Centrage</b> (trains sortis et volets rentrés)</p> <table border="1"> <tr> <td>2000 Kg</td> <td>14 à 35,5 %</td> <td>H</td> </tr> <tr> <td>2835 Kg</td> <td>18 à 35,5 %</td> <td></td> </tr> <tr> <td>2954 Kg</td> <td>20 à 35,5 %</td> <td></td> </tr> </table> |               | 2000 Kg   | 14 à 35,5 %   | H            | 2835 Kg | 18 à 35,5 % |          | 2954 Kg      | 20 à 35,5 % |       |        |                  |       |      |       |                         |               |     |        |  |  |
| 2000 Kg  | 14 à 35,5 %   | H  |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| 2835 Kg  | 18 à 35,5 %   |  |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| 2954 Kg  | 20 à 35,5 %   |  |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| <p><b>PESEE EFFECTUEE SUR POINTS DE LEVAGE</b></p> <p>Avion non peint<br/>Sans huile moteur</p> <table border="1"> <thead> <tr> <th></th> <th>Masse (Kg)</th> <th>Armée (m)</th> <th>Moment (Kg.m)</th> </tr> </thead> <tbody> <tr> <td>Point gauche</td> <td>793,5</td> <td>P1</td> <td></td> </tr> <tr> <td>Point droit</td> <td>825</td> <td>P2</td> <td></td> </tr> <tr> <td>Point avant</td> <td>499,5</td> <td>P0</td> <td></td> </tr> <tr> <td><b>Masse P0 (Kg) =</b></td> <td><b>2119,0</b></td> <td></td> <td></td> </tr> </tbody> </table> <p><math>d0 = \frac{(P1 \times d1) + (P2 \times d2)}{P0}</math></p> <p><math>d0 = \frac{(499,5 \times 2,085) + (793,5 + 825) \times 5,297}{2119,0} = 4,752</math></p> |               |  | Masse (Kg)    | Armée (m) | Moment (Kg.m) | Point gauche | 793,5   | P1          |          | Point droit  | 825         | P2    |        | Point avant      | 499,5 | P0   |       | <b>Masse P0 (Kg) =</b>  | <b>2119,0</b> |     |        | <p><b>DISTANCE DU C.G.<br/>A LA REFERENCE</b></p>  |  |
|  | Masse (Kg)    | Armée (m)  | Moment (Kg.m) |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| Point gauche   | 793,5         | P1   |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| Point droit  | 825           | P2   |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| Point avant  | 499,5         | P0   |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| <b>Masse P0 (Kg) =</b>   | <b>2119,0</b> |  |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| <p><b>CORRECTION</b></p> <table border="1"> <thead> <tr> <th></th> <th>Masse (Kg)</th> <th>Long (m)</th> <th>Moment (Kg.m)</th> </tr> </thead> <tbody> <tr> <td>Masse P0</td> <td>2119,0</td> <td>4,752</td> <td>10064,74</td> </tr> <tr> <td>Peinture (2)</td> <td>30</td> <td>4,673</td> <td>140,19</td> </tr> <tr> <td>Huile moteur (3)</td> <td>11,2</td> <td>2,09</td> <td>23,41</td> </tr> <tr> <td>Carburant inconnu. (1a)</td> <td>27,2</td> <td>4,7</td> <td>127,84</td> </tr> </tbody> </table>  |               |  | Masse (Kg)    | Long (m)  | Moment (Kg.m) | Masse P0     | 2119,0  | 4,752       | 10064,74 | Peinture (2) | 30          | 4,673 | 140,19 | Huile moteur (3) | 11,2  | 2,09 | 23,41 | Carburant inconnu. (1a) | 27,2          | 4,7 | 127,84 | <p><b>CENTRAGE A VIDE<br/>(EN % DE LA CMA)</b></p> <p><math>CG = \frac{d0 - 4,392}{1,51} \times 100</math></p> <p><math>CG = \frac{4,751 - 4,392}{1,51} \times 100 = 23,8</math></p> |  |
|  | Masse (Kg)    | Long (m)   | Moment (Kg.m) |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| Masse P0   | 2119,0        | 4,752  | 10064,74      |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| Peinture (2)   | 30            | 4,673  | 140,19        |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| Huile moteur (3)   | 11,2          | 2,09   | 23,41         |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| Carburant inconnu. (1a)  | 27,2          | 4,7  | 127,84        |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| <p><b>RESULTATS</b> W: 2145,2 D0: 4,751 M: 10192,58 (1b)</p>   |               |  |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| <p><b>CALCUL DE L'INDEX DE BASE</b><br/>Cf section 6 du manuel de vol</p> <p>Masse à vide (W) : 2145,2 Kg (1a)</p> <p>Centrage à Vide (CG) : 23,8 % (1c)</p> <p>Moment (M) : 10192,58</p>  |               |  |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| <p>1 - Rayer la mention inutile.<br/>2 - Valeur non prise en compte si l'avion était peint lors de la pesée.<br/>3 - Valeur non prise en compte si l'avion a été pesé avec huile.</p>  |               |  |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |
| DATE :   |               | VISA DE CONTRÔLE :   |               |           |               |              |         |             |          |              |             |       |        |                  |       |      |       |                         |               |     |        |  |  |

Figure 6.4.1 - Example of weight and balance report and basic airplane characteristics, in kg and m

## • NOTE •

Airplane original report shall be kept with airplane POH

| REGISTRE INDIVIDUEL DE CONTROLE<br>INDIVIDUAL INSPECTION RECORD   |                  | TBM 700        |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
|---|------------------|----------------|----------------|----------------|----------------|------------|--------|-----------|---------|-------------|----------------|-------|---------|---------------|--------|-----|----------------|--------------------------|----------------|--|--|---|--|-------------|----------|----------------|------|----|--------|------|----|--------|------|----|--------|
| WEIGHT AND BALANCE REPORT   |                  | S/N :          |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| 6-SEAT CONFIGURATION  |                  | F : 6          |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| <p>Leveling : </p> <p>Front wheel      Left and right wheel points</p> <p><b>WEIGHING CARRIED OUT ON JACK POINTS</b></p> <p>Not painted airplane<br/>Without engine oil</p> <table border="1"> <thead> <tr> <th>Point</th> <th>Weight (lb)</th> <th>Arm (in)</th> <th>Moment (lb-in)</th> </tr> </thead> <tbody> <tr> <td>Left point</td> <td>1749.4</td> <td>117.5</td> <td>205,583</td> </tr> <tr> <td>Right point</td> <td>1818.8</td> <td>117.5</td> <td>214,164</td> </tr> <tr> <td>Front point</td> <td>1101.2</td> <td>0</td> <td>0</td> </tr> <tr> <td><b>Weight PO (lbs) =</b></td> <td><b>4669.4</b></td> <td></td> <td></td> </tr> </tbody> </table> <p>Weight PO (lbs) = 4669.4</p> |                  | Point          | Weight (lb)    | Arm (in)       | Moment (lb-in) | Left point | 1749.4 | 117.5     | 205,583 | Right point | 1818.8         | 117.5 | 214,164 | Front point   | 1101.2 | 0   | 0              | <b>Weight PO (lbs) =</b> | <b>4669.4</b>  |  |  | <p><b>LIMITATIONS</b> ONLY FOR INFORMATION, REFER TO LIMITATIONS SECTION 2 OF POH</p> <p>Maximum take off weight : 7384.3 lbs H</p> <p>Maximum landing weight : 7024 lbs H</p> <p><b>Balance ( landing gear down and flaps up )</b></p> <table border="1"> <thead> <tr> <th>Weight (lb)</th> <th>Arm (in)</th> <th>Percentage (%)</th> </tr> </thead> <tbody> <tr> <td>4400</td> <td>14</td> <td>35.5 %</td> </tr> <tr> <td>6250</td> <td>18</td> <td>35.5 %</td> </tr> <tr> <td>6579</td> <td>20</td> <td>35.5 %</td> </tr> </tbody> </table> <p><b>DISTANCE FROM C.G. TO REFERENCE</b></p> <p><math>d0 = \frac{(P1 \times d1) + (P2 \times P3) \times d2}{P0}</math></p> <p><math>d0 = \frac{(1101.2 \times 117.5) + (1749.4 + 1818.8) \times 208.5}{4669.4} = 187.0</math></p> |  | Weight (lb) | Arm (in) | Percentage (%) | 4400 | 14 | 35.5 % | 6250 | 18 | 35.5 % | 6579 | 20 | 35.5 % |
| Point   | Weight (lb)      | Arm (in)       | Moment (lb-in) |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| Left point  | 1749.4           | 117.5          | 205,583        |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| Right point   | 1818.8           | 117.5          | 214,164        |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| Front point   | 1101.2           | 0              | 0              |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| <b>Weight PO (lbs) =</b>  | <b>4669.4</b>    |                |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| Weight (lb)   | Arm (in)         | Percentage (%) |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| 4400  | 14               | 35.5 %         |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| 6250  | 18               | 35.5 %         |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| 6579  | 20               | 35.5 %         |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| <p><b>CORRECTIONS</b></p> <table border="1"> <thead> <tr> <th>Weight (lb)</th> <th>Arm (in)</th> <th>Moment (lb-in)</th> </tr> </thead> <tbody> <tr> <td>Weight PO</td> <td>4669.4</td> <td>187.0</td> </tr> <tr> <td>Paint (2)</td> <td>66</td> <td>191.8</td> </tr> <tr> <td>Engine oil (3)</td> <td>62.3</td> <td>2032.8</td> </tr> <tr> <td>Unusable fuel</td> <td>60</td> <td>165</td> </tr> <tr> <td><b>RESULTS</b></td> <td><b>W: 4729.4</b></td> <td><b>D0: 187</b></td> </tr> </tbody> </table>  |                  | Weight (lb)    | Arm (in)       | Moment (lb-in) | Weight PO      | 4669.4     | 187.0  | Paint (2) | 66      | 191.8       | Engine oil (3) | 62.3  | 2032.8  | Unusable fuel | 60     | 165 | <b>RESULTS</b> | <b>W: 4729.4</b>         | <b>D0: 187</b> | <p><b>BALANCE</b></p> <p>MAC ( % )</p> <p><math>CG = \frac{D0 - 173}{59.5} \times 100</math></p> <p><math>CG = \frac{187 - 173}{59.5} \times 100 = 23.8</math></p> |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| Weight (lb)   | Arm (in)         | Moment (lb-in) |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| Weight PO   | 4669.4           | 187.0          |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| Paint (2)   | 66               | 191.8          |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| Engine oil (3)  | 62.3             | 2032.8         |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| Unusable fuel   | 60               | 165            |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| <b>RESULTS</b>  | <b>W: 4729.4</b> | <b>D0: 187</b> |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| <p><b>BASIC INDEX CALCULATION</b></p> <p>See section 6 of Pilot's Operating Handbook</p> <p>Empty weight ( W ) : <b>4729.4 lb (1a)</b></p> <p>Balance ( CG ) : <b>23.8 % (1c)</b></p> <p>Moment ( M ) : 884460.7</p>  |                  |                |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |
| <p>1 - Scratch useless mention.</p> <p>2 - Values not taken into account if the airplane was painted when weighed.</p> <p>3 - Values not taken into account if the oil tank was full when the airplane was weighed.</p> <p>DATE :      INSPECTION VISA :</p>  |                  |                |                |                |                |            |        |           |         |             |                |       |         |               |        |     |                |                          |                |  |  |   |  |             |          |                |      |    |        |      |    |        |      |    |        |

Figure 6.4.2 - Example of weight and balance report and basic airplane characteristics, in lb and in

## Weight and balance form and diagram (m, kg)

Moment = Weight x Arm

$$CG \text{ (MAC \%)} = \frac{(Arm \text{ (m)} - 4.392)}{1.51} \times 100$$

■ >> Up to S/N 1159

| Item                         |                            | Weight<br>(kg) | Arm<br>(m) | Moment<br>(m.kg) | CG<br>(MAC %) |
|------------------------------|----------------------------|----------------|------------|------------------|---------------|
| Empty weight (kg)            |                            |                |            |                  |               |
| Baggage FWD (< 50 kg)        |                            |                | 3.250      |                  |               |
| Front seats (kg)             |                            |                | 4.534      |                  |               |
| Inter. seats                 | - 17.5 kg per seat removed |                | 5.710      |                  |               |
|                              | Pax                        |                |            |                  |               |
| Rear bench/net               | - 24 kg per seat removed   |                | 6.785      |                  |               |
|                              | Pax                        |                |            |                  |               |
|                              | Cargo (< 80 kg)            |                |            |                  |               |
| Baggage AFT (< 100 kg)       |                            |                | 7.695      |                  |               |
| Zero fuel weight (< 2736 kg) |                            |                |            |                  |               |
| Fuel (kg)                    |                            |                | 4.820      |                  |               |
| Ramp weight (< 3370 kg)      |                            |                |            |                  |               |
| Taxi fuel (kg)               |                            |                | 4.820      |                  |               |
| Takeoff weight (< 3354 kg)   |                            |                |            |                  |               |
| Trip fuel (kg)               |                            |                | 4.820      |                  |               |
| Landing weight (< 3186 kg)   |                            |                |            |                  |               |



Moment = Weight x Arm

$$CG (MAC \%) = \frac{(Arm (m) - 4.392)}{1.51} \times 100$$

>> From S/N 1160

| Item                         |                            | Weight<br>(kg) | Arm<br>(m) | Moment<br>(m.kg) | CG<br>(MAC %) |
|------------------------------|----------------------------|----------------|------------|------------------|---------------|
| Empty weight (kg)            |                            |                |            |                  |               |
| Baggage FWD (< 50 kg)        |                            |                | 3.250      |                  |               |
| Front seats (kg)             |                            |                | 4.534      |                  |               |
| Inter. seats                 | - 15 kg per seat removed   |                | 5.710      |                  |               |
|                              | Pax                        |                |            |                  |               |
| Rear bench/net               | - 20.5 kg per seat removed |                | 6.785      |                  |               |
|                              | Pax                        |                |            |                  |               |
|                              | Cargo (< 80 kg)            |                |            |                  |               |
| Baggage AFT (< 100 kg)       |                            |                | 7.695      |                  |               |
| Zero fuel weight (< 2736 kg) |                            |                |            |                  |               |
| Fuel (kg)                    |                            |                | 4.820      |                  |               |
| Ramp weight (< 3370 kg)      |                            |                |            |                  |               |
| Taxi fuel (kg)               |                            |                | 4.820      |                  |               |
| Takeoff weight (< 3354 kg)   |                            |                |            |                  |               |
| Trip fuel (kg)               |                            |                | 4.820      |                  |               |
| Landing weight (< 3186 kg)   |                            |                |            |                  |               |

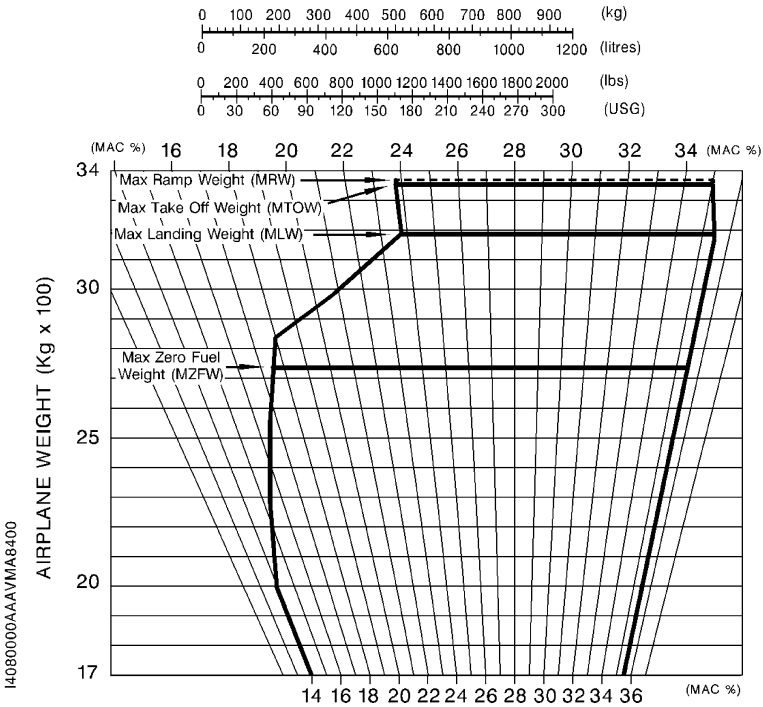


Figure 6.4.3 - Weight and balance diagram

## Weight and balance form and diagram (in, lbs)

$$\text{Moment} = \text{Weight} \times \text{Arm} \qquad \text{CG (MAC \%)} = \frac{(\text{Arm (in)} - 172.93)}{59.45} \times 100$$

■ >> Up to S/N 1159

| Item                          |                             | Weight<br>(lbs) | Arm<br>(in) | Moment<br>(in.lbs) | CG<br>(MAC %) |
|-------------------------------|-----------------------------|-----------------|-------------|--------------------|---------------|
| Empty weight (lbs)            |                             |                 |             |                    |               |
| Baggage FWD (< 110 lbs)       |                             |                 | 128.0       |                    |               |
| Front seats (lbs)             |                             |                 | 178.5       |                    |               |
| Inter. seats                  | - 38.6 lbs per seat removed |                 | 224.8       |                    |               |
|                               | Pax                         |                 |             |                    |               |
| Rear bench/net                | - 52.9 lbs per seat removed |                 | 267.1       |                    |               |
|                               | Pax                         |                 |             |                    |               |
|                               | Cargo (< 176 lbs)           |                 |             |                    |               |
| Baggage AFT (< 220 lbs)       |                             |                 | 303.0       |                    |               |
| Zero fuel weight (< 6032 lbs) |                             |                 |             |                    |               |
| Fuel (lbs)                    |                             |                 | 189.8       |                    |               |
| Ramp weight (< 7430 lbs)      |                             |                 |             |                    |               |
| Taxi fuel (lbs)               |                             |                 | 189.8       |                    |               |
| Takeoff weight (< 7394 lbs)   |                             |                 |             |                    |               |
| Trip fuel (lbs)               |                             |                 | 189.8       |                    |               |
| Landing weight (< 7024 lbs)   |                             |                 |             |                    |               |

Moment = Weight x Arm

$$CG (MAC \%) = \frac{(Arm (in) - 172.93)}{59.45} \times 100$$

■ >> From S/N 1160

| Item                          |                             | Weight<br>(lbs) | Arm<br>(in) | Moment<br>(in.lbs) | CG<br>(MAC %) |
|-------------------------------|-----------------------------|-----------------|-------------|--------------------|---------------|
| Empty weight (lbs)            |                             |                 |             |                    |               |
| Baggage FWD                   | (< 110 lbs)                 |                 | 128.0       |                    |               |
| Front seats (lbs)             |                             |                 | 178.5       |                    |               |
| Inter. seats                  | - 33.1 lbs per seat removed |                 | 224.8       |                    |               |
|                               | Pax                         |                 |             |                    |               |
| Rear bench/net                | - 45.2 lbs per seat removed |                 | 267.1       |                    |               |
|                               | Pax                         |                 |             |                    |               |
|                               | Cargo (< 176 lbs)           |                 |             |                    |               |
| Baggage AFT (< 220 lbs)       |                             |                 | 303.0       |                    |               |
| Zero fuel weight (< 6032 lbs) |                             |                 |             |                    |               |
| Fuel (lbs)                    |                             |                 | 189.8       |                    |               |
| Ramp weight (< 7430 lbs)      |                             |                 |             |                    |               |
| Taxi fuel (lbs)               |                             |                 | 189.8       |                    |               |
| Takeoff weight (< 7394 lbs)   |                             |                 |             |                    |               |
| Trip fuel (lbs)               |                             |                 | 189.8       |                    |               |
| Landing weight (< 7024 lbs)   |                             |                 |             |                    |               |

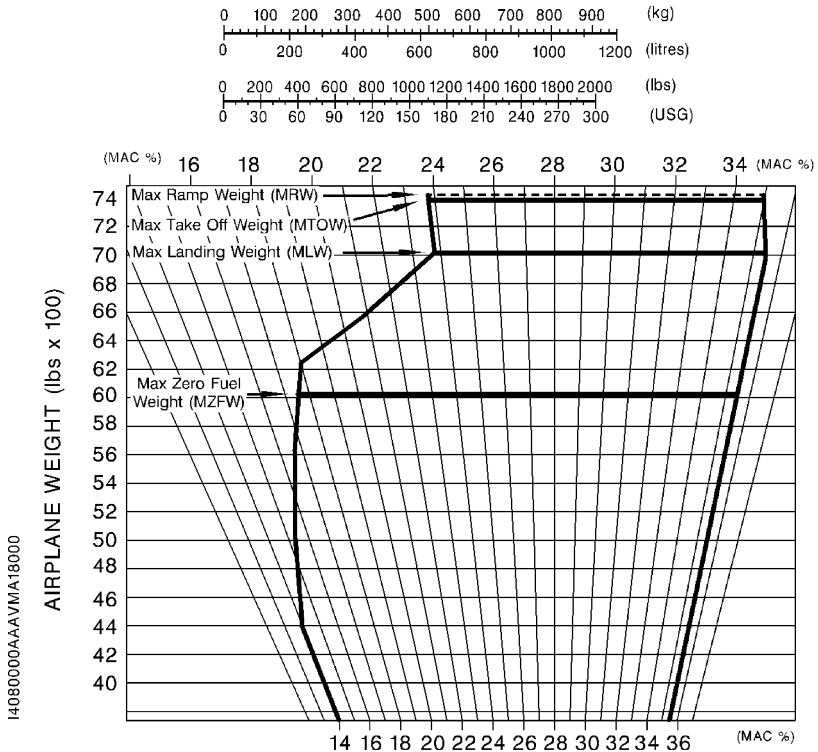


Figure 6.4.4 - Weight and balance diagram

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## Weight and balance samples (m, kg)

>> Up to S/N 1159

### ▲ CAUTION ▲

Loading samples - see figure 6.4.5 or 6.4.6 - are only given as an example ; for calculation concerning your airplane, refer to the diagram corresponding to its validity.



|  | Fig. 6.4.5  |
|--|---|
| <b>1 - Airplane basic characteristics :</b><br><br><div>W = Empty weight</div> <div>Moment</div> <div>Balance arm</div> <div>CG (MAC %)</div>                          | <div>2 126 kg</div> <div>10 073 m.kg</div> <div>4.738 m</div> <div>22.9 %</div> |
| <b>2 - Foreseen loading :</b><br><br><div>1 Pilot and 1 front passenger</div> <div>2 Rear passengers</div> <div>AFT Cargo in baggage compartment</div> <div>Fuel</div> | <div>200 kg</div> <div>160 kg</div> <div>50 kg</div> <div>820 kg</div>          |
| <b>3 - Foreseen fuel :</b><br><br><div>Taxi fuel</div> <div>Trip fuel</div>  | <div>- 16 kg</div> <div>- 600 kg</div>  |

Moment = Weight x Arm

$$CG (MAC \%) = \frac{(Arm (m) - 4.392)}{1.51} \times 100$$

■ >> Up to S/N 1159

| Item                         |                            | Weight<br>(kg) | Arm<br>(m) | Moment<br>(m.kg) | CG<br>(MAC %) |
|------------------------------|----------------------------|----------------|------------|------------------|---------------|
| Empty weight (kg)            |                            | 2 126          | 4.738      | 10 073           | 22.9          |
| Baggage FWD (< 50 kg)        |                            | 0              | 3.250      | 0                |               |
| Front seats (kg)             |                            | 200            | 4.534      | 907              |               |
| Inter. seats                 | - 17.5 kg per seat removed | 0              | 5.710      | 0                |               |
|                              | Pax                        | 0              |            | 0                |               |
| Rear bench/net               | - 24 kg per seat removed   | 0              | 6.785      | 0                |               |
|                              | Pax                        | 160            |            | 1 086            |               |
|                              | Cargo (< 80 kg)            | 0              |            | 0                |               |
| Baggage AFT (< 100 kg)       |                            | 50             | 7.695      | 385              |               |
| Zero fuel weight (< 2736 kg) |                            | 2 536          | 4.910      | 12 451           | 34.3          |
| Fuel (kg)                    |                            | 820            | 4.820      | 3 952            |               |
| Ramp weight (< 3370 kg)      |                            | 3 356          | 4.888      | 16 403           | 32.8          |
| Taxi fuel (kg)               |                            | - 16           | 4.820      | - 77             |               |
| Takeoff weight (< 3354 kg)   |                            | 3 340          | 4.888      | 16 326           | 32.8          |
| Trip fuel (kg)               |                            | - 600          | 4.820      | - 2 892          |               |
| Landing weight (< 3186 kg)   |                            | 2 740          | 4.903      | 13 434           | 33.8          |



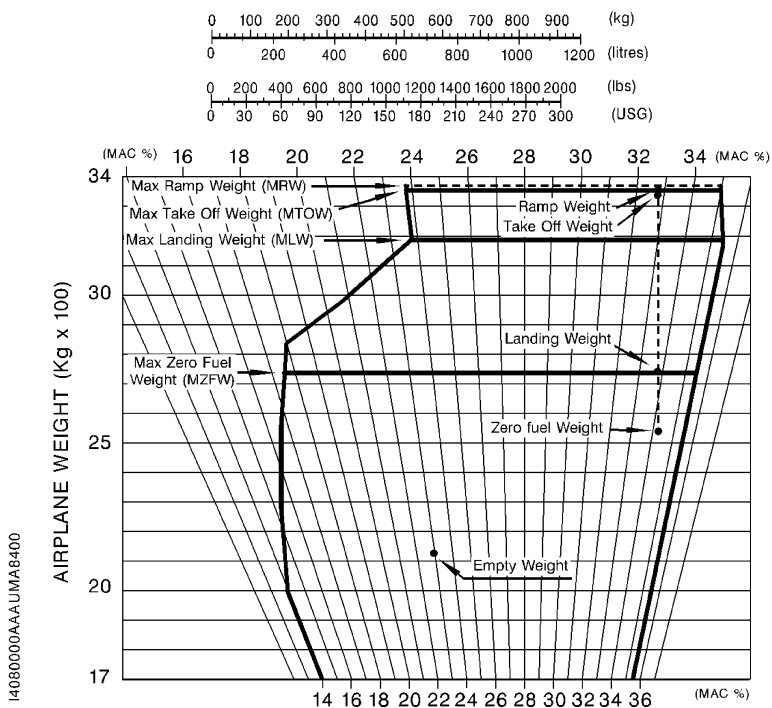


Figure 6.4.5 - Loading sample (in kg and m)

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## Weight and balance samples (in, lbs)

>> From S/N 1160

### ▲ CAUTION ▲

Loading samples - see figure 6.4.5 or 6.4.6 - are only given as an example ; for calculation concerning your airplane, refer to the diagram corresponding to its validity.



|                                      | Fig. 6.4.6     |
|--------------------------------------|----------------|
| 1 - Airplane basic characteristics : |                |
| W = Empty weight                     | 4 638 lbs      |
| Moment                               | 864 173 in.lbs |
| Balance arm                          | 186.3 in       |
| CG (MAC %)                           | 22.6 %         |
| 2 - Foreseen loading :               |                |
| FWD compartment                      | 0 lbs          |
| 1 Pilot and 1 front passenger        | 400 lbs        |
| 1 Intermediate passenger             | 220 lbs        |
| 2 Rear seats removed                 | - 90.4 lbs     |
| Rear cargo                           | 176 lbs        |
| AFT Cargo in baggage compartment     | 220 lbs        |
| Fuel                                 | 1 850 lbs      |
| 3 - Foreseen fuel :                  |                |
| Taxi fuel                            | - 36 lbs       |
| Trip fuel                            | - 1 400 lbs    |

Moment = Weight x Arm

$$CG \text{ (MAC \%)} = \frac{(Arm \text{ (in)} - 172.93)}{59.45} \times 100$$

>> From S/N 1160

| Item                          |                             | Weight<br>(lbs) | Arm<br>(in) | Moment<br>(in.lbs) | CG<br>(MAC %) |
|-------------------------------|-----------------------------|-----------------|-------------|--------------------|---------------|
| Empty weight (lbs)            |                             | 4 638           | 186.3       | 864 173            | 22.6          |
| Baggage FWD (< 110 lbs)       |                             | 0               | 128.0       | 0                  |               |
| Front seats (lbs)             |                             | 400             | 178.5       | 71 400             |               |
| Inter. seats                  | - 33.1 lbs per seat removed | 0               | 224.8       | 0                  |               |
|                               | Pax                         | 220             |             | 49 456             |               |
| Rear bench/net                | - 45.2 lbs per seat removed | - 90.4          | 267.1       | - 24 680           |               |
|                               | Pax                         | 0               |             | 0                  |               |
|                               | Cargo (< 176 lbs)           | 176             |             | 47 010             |               |
| Baggage AFT (< 220 lbs)       |                             | 220             | 303.0       | 66 660             |               |
| Zero fuel weight (< 6032 lbs) |                             | 5 564           | 193.0       | 1 074 019          | 33.8          |
| Fuel (lbs)                    |                             | 1 850           | 189.8       | 351 130            |               |
| Ramp weight (< 7430 lbs)      |                             | 7 414           | 192.3       | 1 425 149          | 32.6          |
| Taxi fuel (lbs)               |                             | - 36            | 189.8       | - 6 833            |               |
| Takeoff weight (< 7394 lbs)   |                             | 7 378           | 192.3       | 1 418 316          | 32.6          |
| Trip fuel (lbs)               |                             | - 1 400         | 189.8       | - 265 720          |               |
| Landing weight (< 7024 lbs)   |                             | 5 978           | 192.9       | 1 152 596          | 33.6          |

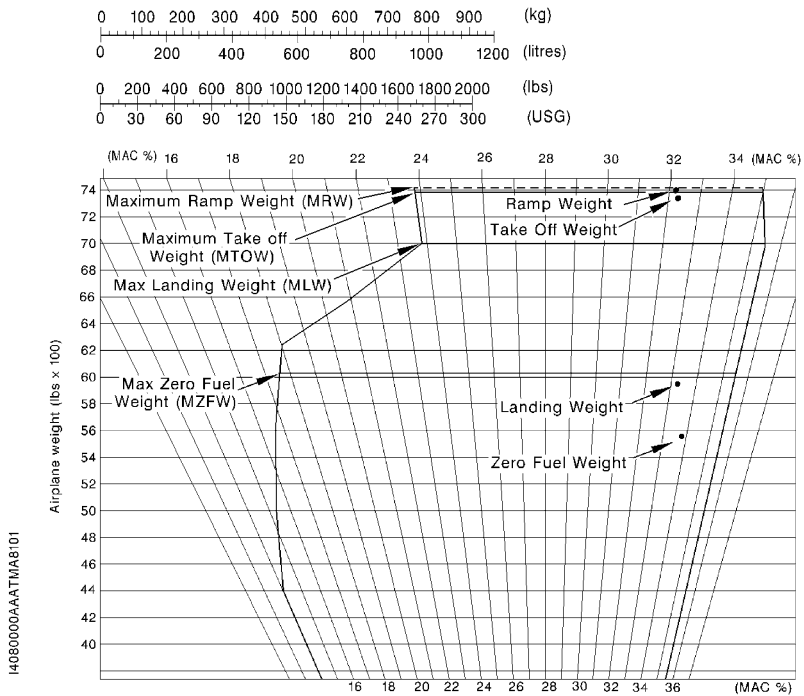


Figure 6.4.6 - Loading samples (in lbs and in)

## Determining empty airplane characteristics

Empty airplane characteristics (weight and balance) may vary with regard to those indicated on weighing form according to installed optional equipment and installed seats.

List of equipment (refer to chapter 6.5) contains the standard and optional equipment, as well as their characteristics (weight, arm), except those listed in this Chapter.

Use the chart below to compute new empty weight and corresponding balance if necessary.

| Date | Equipment or modification description | (+) (-) | Weight modification |         |                   | Basic empty weight |                       |        |
|------|---------------------------------------|---------|---------------------|---------|-------------------|--------------------|-----------------------|--------|
|      |                                       |         | Weight lb           | Arm in. | Moment lb.in/1000 | Weight W           | Arm "d <sub>o</sub> " | Moment |
|      | According to delivery                 |         |                     |         |                   |                    |                       |        |
|      |                                       |         |                     |         |                   |                    |                       |        |
|      |                                       |         |                     |         |                   |                    |                       |        |

Figure 6.4.7 - Sample weight and balance record

$$\text{CG m.a.c.\%} = \frac{(\text{d}_o - 172.93)}{59.45} \times 100$$

Use the above formula to express arm "d<sub>o</sub>" in % of mean aerodynamic chord.

• NOTE •

Arm expressed in inches with regard to reference.

•

|  |                     |
|--|---------------------|
| FWD baggage compartment :                  | 128.0 in. (3.250 m) |
| Baggage compartment in pressurized cabin : | 303.0 in. (7.695 m) |
| Fuel :                                     | 189.8 in. (4.820 m) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment  | Weight<br>per unit<br>lb<br>(kg)       | Arm<br>in.<br>(m) |
|---------------------|------------------------------|---|--|-------------------|
|                     |                              | <b>10 - Parking, mooring, storage and<br/>return to service</b> |  |                   |
|                     |                              | <b>Board kit</b>  |  |                   |
| S                   |                              | - Blanking caps bag   | 8.31<br>(3.77)                         | 128.00<br>(3.250) |
| S                   |                              | - Towing bar  | 8.77<br>(3.98)                         | 128.00<br>(3.250) |
| S                   |                              | - Control lock device   | 0.90<br>(0.41)                         | 133.86<br>(3.400) |
|                     |                              | <b>25 - Equipment and furnishings (partial)</b>                 |  |                   |
| S                   | 0500-25A                     | - Upholstery 2016 - Carbon finishing 2016                       | /                                      | /                 |
| S                   | 0503-25A                     | - Upholstery Version 2016, of which :                           | $\Delta + 2.73$<br>( $\Delta + 1.24$ ) | 228.30<br>(5.799) |
|                     |                              | · Carpets for 6-place configuration                             | 26.68<br>(12.100)                      | /                 |
|                     |                              | · Carpets for 4-place configuration                             | 20.59<br>(9.340)                       | /                 |
| A                   | 0503-25A                     | - Generation 2008 cabinets :                                    |  |                   |
|                     |                              | · L.H. low storage  | 9.48<br>(4.300)                        | 203.74<br>(5.175) |
|                     |                              | · R.H. low storage  | 9.48<br>(4.300)                        | 203.74<br>(5.175) |
|                     |                              | · L.H. low + high storage                                       | 17.20<br>(7.800)                       | 203.74<br>(5.175) |
|                     |                              | · R.H. low + high storage                                       | 17.20<br>(7.800)                       | 203.74<br>(5.175) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
| S                   | 0557-25A                     | · L.H. low + top pilot case support                            | 9.70<br>(4.400)                  | 203.74<br>(5.175) |
|                     |                              | · R.H. low + top pilot case support                            | 9.70<br>(4.400)                  | 203.74<br>(5.175) |
|                     |                              | - <b>Upholstery Version 2017, of which :</b>                   | △ Neglig.                        | /                 |
|                     |                              | · Carpets for 6-place configuration                            | 26.68<br>(12.100)                | /                 |
|                     |                              | · Carpets for 4-place configuration                            | 20.59<br>(9.340)                 | /                 |
| A                   | 0557-25A                     | - <b>Generation 2008 cabinets :</b>                            |                                  |                   |
|                     |                              | · L.H. low storage   | 9.48<br>(4.300)                  | 203.74<br>(5.175) |
|                     |                              | · R.H. low storage   | 9.48<br>(4.300)                  | 203.74<br>(5.175) |
|                     |                              | · L.H. low + high storage                                      | 17.20<br>(7.800)                 | 203.74<br>(5.175) |
|                     |                              | · R.H. low + high storage                                      | 17.20<br>(7.800)                 | 203.74<br>(5.175) |
|                     |                              | · L.H. low + top pilot case support                            | 9.70<br>(4.400)                  | 203.74<br>(5.175) |
|                     |                              | · R.H. low + top pilot case support                            | 9.70<br>(4.400)                  | 203.74<br>(5.175) |
|                     |                              |  |                                  |                   |
| S                   | 0207-00                      | Carpet   | 35.27<br>(16.000)                | 211.42<br>(5.370) |
|                     |                              | - Cabin furnishings  | 302.45<br>(137.19)               | 211.42<br>(5.370) |



| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>Leather seats</b>   |                                  |                   |
|                     |                              | >> <i>Up to S/N 1159</i>                                       |                                  |                   |
| S                   |                              | - L.H. intermediate seat (back to or in flight direction)      | 38.58<br>(17.50)                 | 224.80<br>(5.710) |
|                     |                              | >> <i>From S/N 1160 to S/N 1214</i>                            |                                  |                   |
| S                   | 0553-25A                     | - L.H. intermediate seat (back to or in flight direction)      | 33.07<br>(15.00)                 | 224.80<br>(5.710) |
|                     |                              | >> <i>Up to S/N 1159</i>                                       |                                  |                   |
| S                   |                              | - RH. intermediate seat (back to or in flight direction)       | 38.58<br>(17.50)                 | 224.80<br>(5.710) |
|                     |                              | >> <i>From S/N 1160 to S/N 1214</i>                            |                                  |                   |
| S                   | 0553-25A                     | - R.H. intermediate seat (back to or in flight direction)      | 33.07<br>(15.00)                 | 224.80<br>(5.710) |
|                     |                              | >> <i>Up to S/N 1159</i>                                       |                                  |                   |
| S                   |                              | - Double chair   |                                  |                   |
|                     |                              | . L.H. Seat  | 52.91<br>(24.00)                 | 278.19<br>(7.066) |
|                     |                              | . R.H. Seat  | 52.91<br>(24.00)                 | 278.19<br>(7.066) |
|                     |                              | >> <i>From S/N 1160 to S/N 1214</i>                            |                                  |                   |
| S                   | 0553-25                      | - Double chair   |                                  |                   |
|                     |                              | . L.H. Seat  | 45.20<br>(20.50)                 | 278.62<br>(7.077) |
|                     |                              | . R.H. Seat  | 45.20<br>(20.50)                 | 278.62<br>(7.077) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment   | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>Nets</b>  |                                  |                   |
| S                   | 0315-25                      | - Small cargo net<br>GP SOCT704CC-10   | 15.00<br>(7.00)                  | /                 |
| S                   | 0315-25                      | - Large cargo net<br>GP SOCT704CS-10   | 13.00<br>(6.00)                  | /                 |
| S                   | 25026B                       | - Partition net at Frame 14 (between the cabin and<br>the baggage compartment) T700B2590001, of<br>which | 3.638<br>(1.650)                 | 289.53<br>(7.354) |
| S                   |                              | - Partition net  | 1.698<br>(0.770)                 | 289.53<br>(7.354) |

## **6.5 - List of equipment**

■ The list of equipment is available in Report reference NAV No.34/90-RJ-App 4, located at the end of this POH.

A separate list of equipment of items installed at the factory in your specific airplane is provided in your airplane file.

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## Section 7

### Description

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## **7.1 - General**

This section provides description and operation of the airplane and its systems.

Some of the equipment described herein is optional and may not be installed in the airplane.

Complete description and operation of the GARMIN integrated flight deck are detailed in the GARMIN Integrated Flight Deck Pilot's Guide. References to this guide are often made all along this section to get more details about some systems.

Details of other optional systems and equipment are presented in section 9 Supplements of the POH.

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**7.2 - Airframe** - see figures 7.2.1, 7.2.1A and 7.2.1B

This airplane is a six-place, low wing airplane.

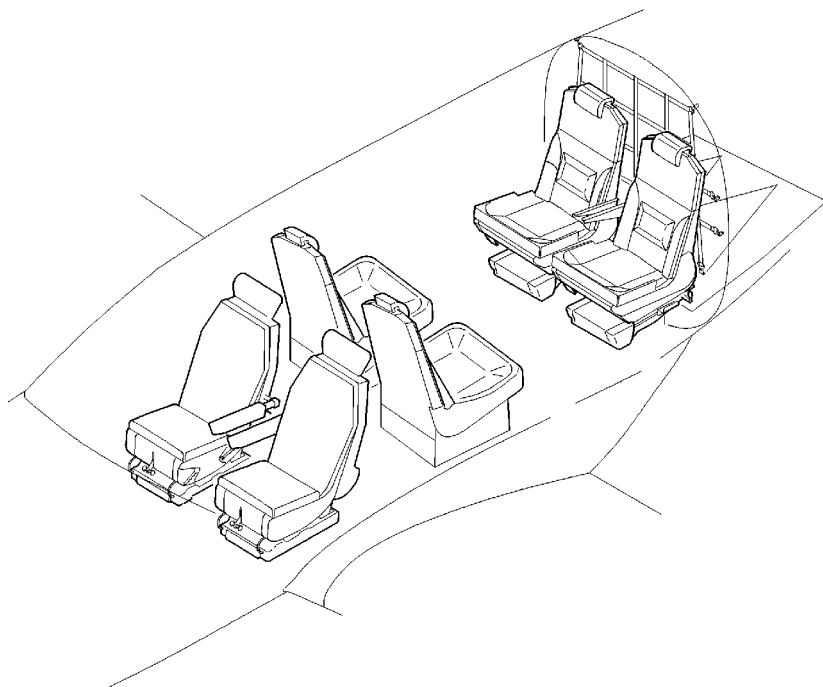
The airplane can be changed into 2, 3, 4 or 5-seat accommodation.

The structure is a semi-monocoque all-metal construction and is equipped with a retractable tricycle landing gear.

The pressurized cabin is equipped, on the left side of fuselage, with a one-piece access door and folding stairs comprising a hand rail allowing pilot and passengers boarding. The occupants have access to cockpit and to rear seats through a central aisle.

An optional pilot door located forward of the cabin on the left side allows access to the cockpit by means of folding stairs.

The aft cabin section is a baggage compartment.



14251201AAAKMA8000

Figure 7.2.1 - Cabin arrangement  
6-seat accommodation

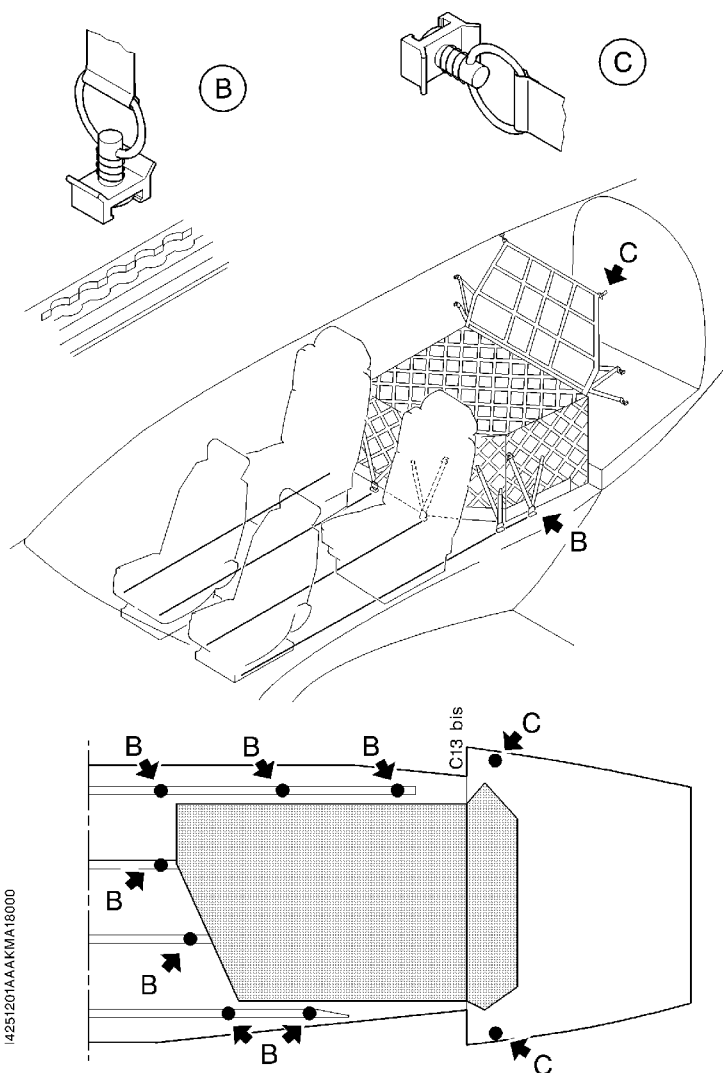


Figure 7.2.1A - Cabin arrangement  
4-seat accommodation with large securing net

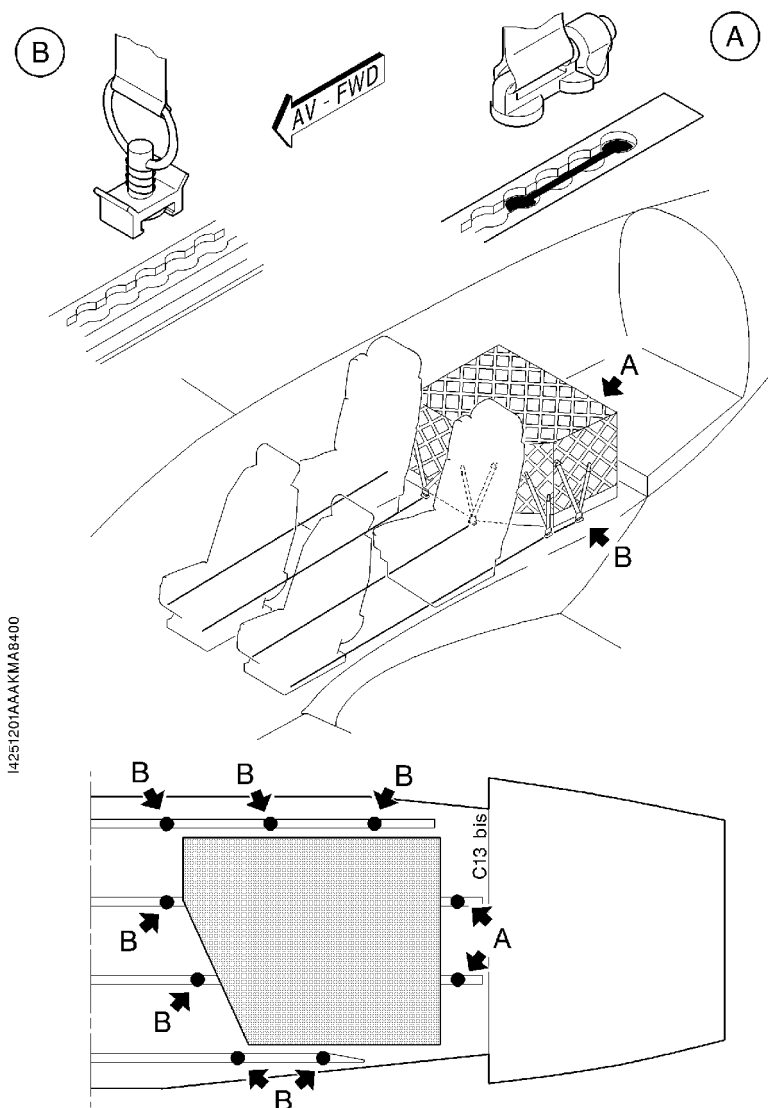


Figure 7.2.1B - Cabin arrangement  
4-seat accommodation with small securing net

## Wings

The wings are monocoque, bi-spar structures. Main spars of each wing are linked to the fuselage by two integral attach fittings. Each wing contains a main landing gear well and sealed casings forming the fuel tank. The wing leading edge is equipped with a deicing system.

Each wing extremity is equipped with a winglet.

## Ailerons, spoilers and pitch trim tab

The ailerons located on external trailing edge of each wing are hinged on two attach fittings fixed on the rear spar. They allow airplane lateral control and are controlled mechanically through control wheel rotation.

The spoilers located in front of flaps, on top skin side, are mechanically linked to the ailerons.

Trim tab attached on the trailing edge of L.H. aileron is electrically activated by a trim knob, through an actuator.

## Wing flaps - see figure 7.2.2

The wing flaps are large span slotted flaps with a single rotation point. They are activated by actuating rod-controlled screw jacks linked to an electric motor located under the floor, inside the fuselage.

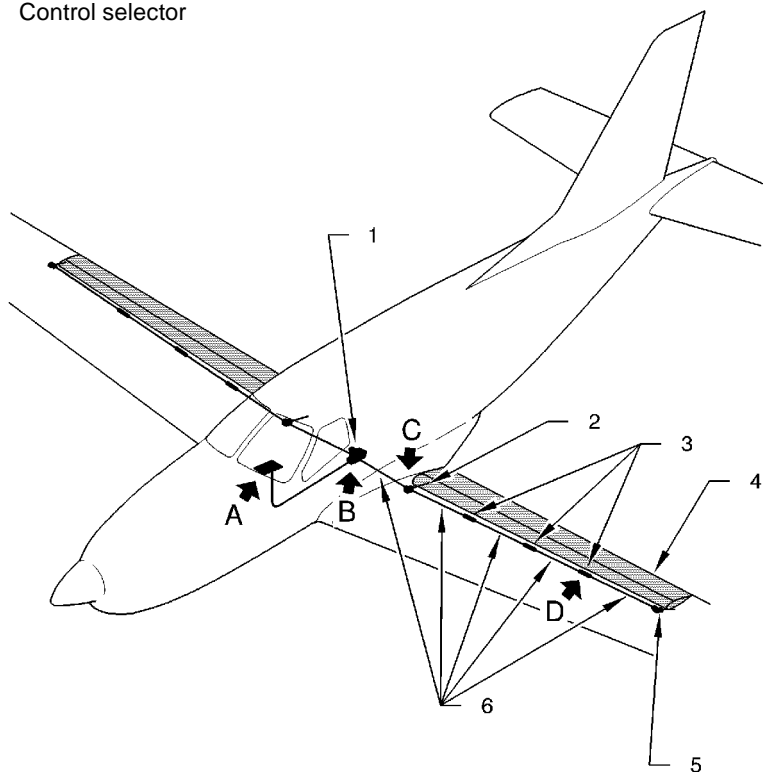
A preselection control located on the right side of pedestal console allows the pilot to select one of the three positions (UP - TO - LDG). For each control position, a deflection angle is defined (0°, 10°, 34°).

A monitoring device interrupts flaps movement as soon as a deflection dissymmetry is detected.

## Empennages

Empennages are composite structures. The horizontal empennage consists of a horizontal stabilizer (PHF), control surfaces and elevator trim tabs ; the vertical empennage consists of a vertical stabilizer, the rudder and the rudder trim tab. The empennage leading edge is equipped with a deicing system.

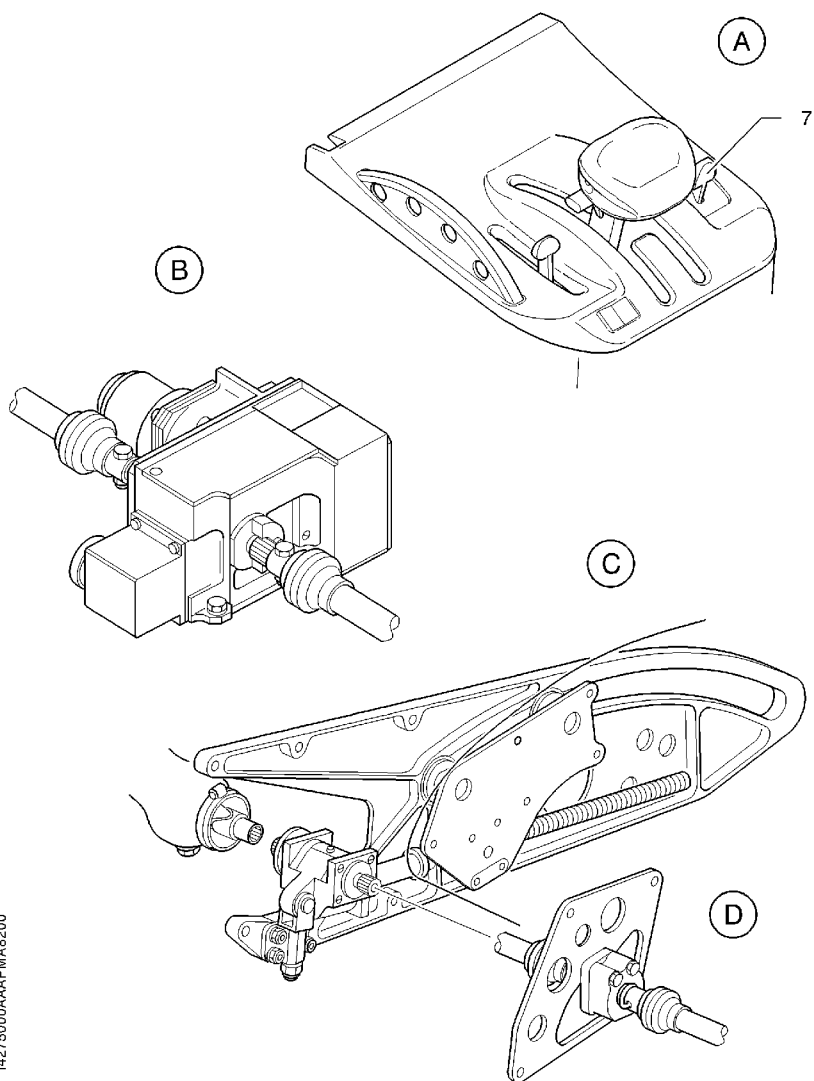
- 1) Geared motor
- 2) Internal actuator
- 3) Intermediate bearings
- 4) Wing flap
- 5) External actuator
- 6) Rods
- 7) Control selector



14275000/AAAAA8003

Figure 7.2.2 (1/2) - Wing flaps





14275000AAAPMA8200

Figure 7.2.2 (2/2) - Wing flaps

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## 7.3 - Accomodations

### Instrument panel

The instrument panel contains instruments and controls necessary for flight monitoring. The typical instrument panel consists of all standard equipment, as well as additional optional equipment.

**Upper panel** - see figure 7.3.2

The upper panel located at the top part of the windshield, contains electrical generation control panels, engine starting, ancillary electrical systems, AP/TRIMS switch, ELT remote control switch and the FUEL control panel.

Rearwards of upper panel, the central part of cockpit overhead panel provides loud-speakers and cockpit floodlights.

**Instrument panel** - see figure 7.3.1

The instrument panel consists of the integrated flight deck composed of three screens [two primary flight displays (PFD) and one multi-function display (MFD)] - refer to the GARMIN pilot's guide for detailed description. Apart from the GARMIN flight deck system, equipment listed below complete the instrument panel.

- Left area instrument panel includes - see figure 7.3.3 :
  - . on top : MD302, MASTER CAUTION and MASTER WARNING,
  - . on the left : DISPLAY BACKUP push-button, and A/C CONTROL panel,
  - . at bottom : deicing controls and indicators, MICRO/MASK switch, hourmeter (Post-MOD70-0533-31A), landing gear control panel, parking brake control and left station control wheel.
- Central area instrument panel includes - see figure 7.3.4 :
  - . on top : surmounted by the stand-by compass, AFCS control unit, BARO knobs and the LVL push-button
  - . at bottom : touchscreen controllers and PRESSURIZATION panel.
- Right area instrument panel includes - see figure 7.3.5 :
  - . on top : locations for optional equipment,
  - . on the right: DISPLAY BACKUP push-button,
  - . at bottom : alternate static source selector and the right station control wheel.

- Emergency air control is located under the right area instrument panel.

>> *Before digital hourmeter installation (Pre-MOD70-0533-31A)*

An hourmeter is located above circuit breakers panel.

>> *All*

An adjustable air outlet is located on both sides of instrument panel lower part.

Reception-micro jacks are located inside the recess under the arm-rest on both lateral sides of the cockpit, on R.H. side of intermediate R.H. passenger's seat and on the arm-rest of rear R.H. passenger's seat.

**Pedestal console** - see figure 7.3.6

The pedestal console, under the touchscreen controllers, comprises flaps controls, pitch trim tab control wheel, aileron trim switch, engine controls and fuel tank selector.

**Circuit breakers panel** - see figures 7.3.7 and 7.8.4

Circuit breakers for all electrical equipment supplied by bus bars are located on a separate panel installed on the right side of cockpit.

**General alarms warning lights and CAS messages**

**WARNING** and **CAUTION** messages appear on the MFD CAS window to alert crew about monitored systems discrepancies. As a message appears, a chime is heard. Refer to the GARMIN pilot's guide to know all possible CAS messages.

A **MASTER WARNING** red flashing indicator and a **MASTER CAUTION** amber indicator located on instrument panel - see figure 7.3.8, in front of the pilot, illuminate as soon as one or several messages of same color light on.

To cancel and reset a general alarm, press on the red or amber indicator. A pressure on the red indicator also stops red message associated chimes.

**Aural warnings** - see figure 7.3.2

The aural warnings are intended to alert the pilot during some configurations. The aural signals are heard through the loud-speakers installed in cockpit overhead panel and through the pilot's and R.H. station headsets.

The aural warnings consist of :

- the GARMIN flight deck system (GIA and GMA),
- the loud-speaker.

The system uses :

- the stall warning system,

- the airspeed indicator,
- the landing gear control unit,
- the flap geared motor,
- the idle position sensor.

**Aural warning alerts**

According to the airplane configuration, different aural warning alerts sound :

- gear up and idle → landing gear / landing gear
- gear up and extended flaps → landing gear / landing gear
- stall → stall / stall
- gear up, idle and stall → stall / landing gear
- gear up, extended flaps and stall → stall / landing gear
- IAS > 269 ± 3 KIAS → overspeed / overspeed

Refer to the GARMIN pilot's guide for description of the other aural warning alerts.

**Cockpit overhead panel** - see figure 7.3.2

This panel includes following elements :

- the loud-speaker of GMA,
- the TEST push-button,
- the emergency lighting rheostat.

It is attached to the cabin upper part between frames C6 and C7.

The emergency lighting rheostat is electrically supplied by BATT BUS bar and protected by PANEL EMER circuit breaker.

The TEST push-button allows to test :

- the autopilot control panel backlighting,
- the MASTER WARNING and MASTER CAUTION indicators,
- the deicing panel led,
- the stick shaker system,
- the fire detection system, if installed,
- the stall aural warning alert,
- the DISPLAY BACKUP push-buttons backlighting,
- the LVL push-button.

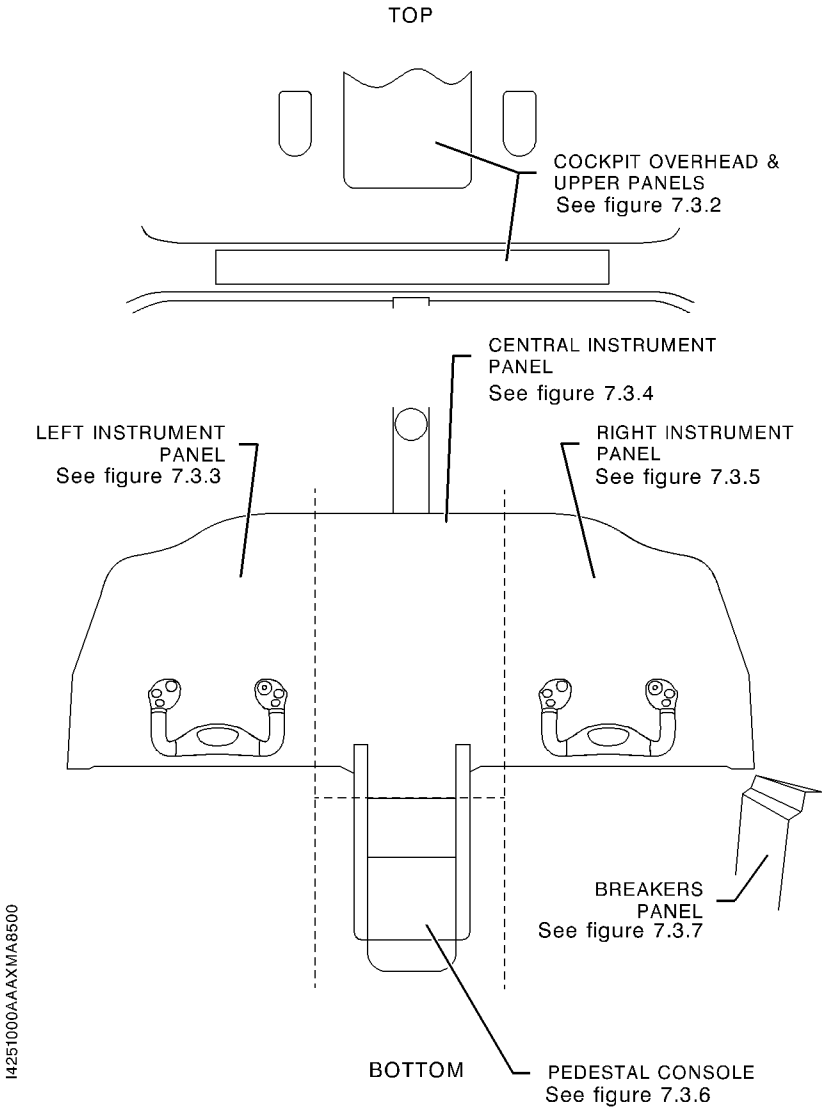


Figure 7.3.1 - Instrument panel assembly  
(Typical arrangement)

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- 1) L.H. instrument panel emergency lighting
- 2) Loud-speaker of GMA
- 3) R.H. instrument panel emergency lighting
- 4) Instrument panel emergency lighting switches (rheostats)
- 5) R.H. cockpit floodlight
- 6) ELT remote control switch
- 7) AP/TRIMS switch
- 8) FUEL control panel - see figure 7.7.3
- 9) ENGINE START switches - see figure 7.6.4
- 10) ELECTRIC POWER switches - see figure 7.8.5
- 11) INT LIGHTS internal lighting switches - see figure 7.8.7
- 12) EXT LIGHTS external lighting switches - see figure 7.8.6
- 13) L.H. cockpit floodlight
- 14) TEST push-button

Figure 7.3.2 (1/2) - Upper panel and cockpit overhead panel



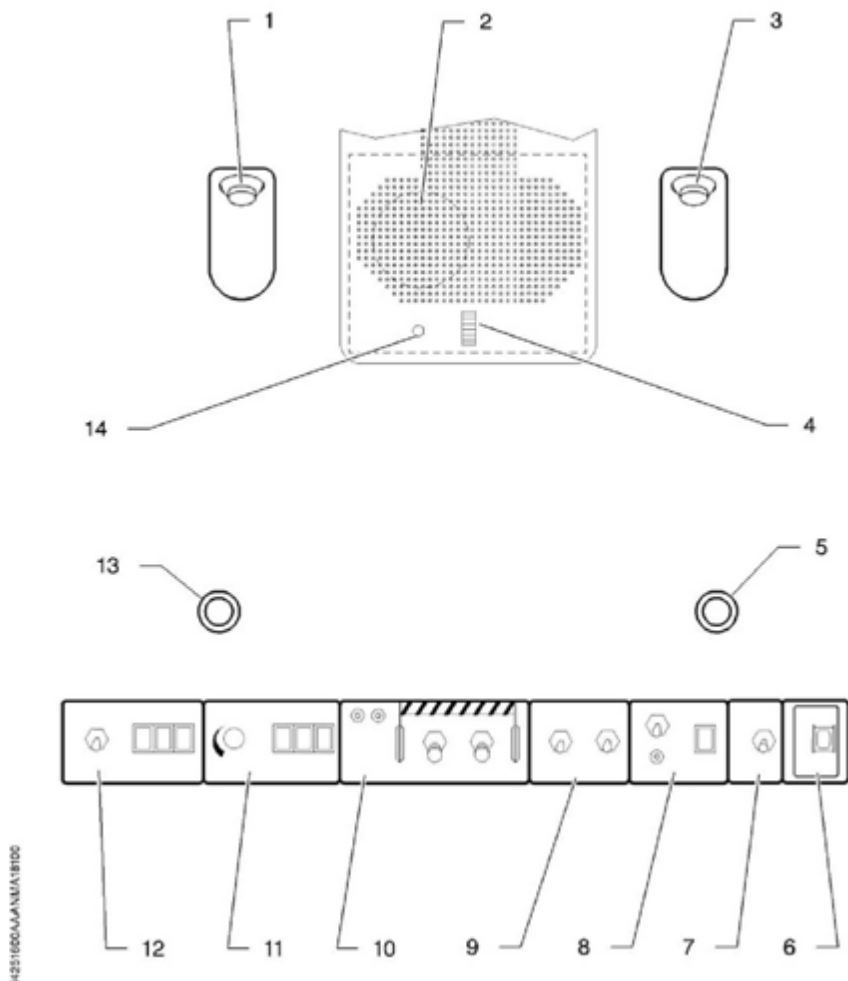


Figure 7.3.2 (2/2) - Upper panel and cockpit overhead panel

- 1) DISPLAY BACKUP and A/C CONTROL panels - see figure 7.9.2
- 2) General alarm red and amber indicators
- 3) PFD 1
- 4) MD302
- 5) Landing gear configuration and control panel - see figure 7.5.1
- 6) Parking brake control - see figure 7.5.6
- 7) Left station control wheel tube
- 8) Deicing control and check panel - see figure 7.13.1
- 9) L.H. station rudder pedals adjusting handle
- 10) Left station reception-micro jacks
- 11) Pitch & Yaw trim setting management
- 12) Push To Talk button (PTT)
- 13) AP / TRIM DISC push-button
- 14) CWS
- 15) Paper clip
- 16) Chronometer management
- 17) Transponder Ident sequence
- 18) Stormscope clear
- 19) COM 2 (Stand-by / active)
- 20) Flight conditions and instruction placard
- 21) Adjustable air outlet
- 22) Circuit breaker panel lighting switch
- 23) MICRO / MASK switch - see figure 7.10.1
- >> *After digital hourmeter installation (Post-MOD70-0533-31A)*
- 24) Hourmeter
- >> *After servicing plugs installation (Post-MOD70-0531-25)*
- 25) USB servicing plug

Figure 7.3.3 (1/2) - Left instrument panel



Figure 7.3.3 (2/2) - Left instrument panel  
(Typical arrangement)

- 1) Stand-by compass
- 2) BARO knobs
- 3) Registration
- 4) AFCS control unit
- 5) Touchscreen controllers
- 6) PRESSURIZATION panel - see figure 7.9.1
- 7) MFD
- 8) Micro LDR
- 9) LVL push-button

Figure 7.3.4 (1/2) - Central instrument panel



Figure 7.3.4 (2/2) - Central instrument panel  
(Typical arrangement)

>> Up to S/N 1159

- 1) PFD 2
- 2) DISPLAY BACKUP push-button
- 3) Right station control wheel tube
- 4) Crew music
- 5) Adjustable air outlet
- 6) Right station reception-micro jacks
- 7) Hourmeter
- 8) R. H. station rudder pedals adjusting handle
- 9) Circuit breakers panel postlight
- 10) Cigar lighter and two USB servicing plugs
- 11) Cabin emergency air control (EMERGENCY RAM AIR control knob)
- 12) Static source selector
- 13) COM 2 (Stand-by / active)
- 14) Stormscope clear
- 15) Transponder Ident sequence
- 16) Chronometer management
- 17) Paper clip
- 18) CWS
- 19) AP / TRIM DISC push-button
- 20) Push To Talk button (PTT)
- 21) Pitch & Yaw trim setting management

Figure 7.3.5 (1/2) - Right instrument panel



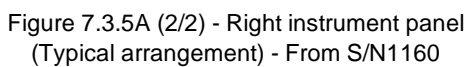
Figure 7.3.5 (2/2) - Right instrument panel  
(Typical arrangement) - Up to S/N1159

>> From S/N 1160

- 1) PFD 2
- 2) DISPLAY BACKUP push-button
- 3) Right station control wheel tube
- 4) Crew music
- 5) Adjustable air outlet
- 6) Right station reception-micro jacks
- 7) R. H. station rudder pedals adjusting handle
- 8) Circuit breakers panel postlight
- 9) USB servicing plugs
- 10) Cabin emergency air control (EMERGENCY RAM AIR control knob)
- 11) Static source selector
- 12) COM 2 (Stand-by / active)
- 13) Stormscope clear
- 14) Transponder Ident sequence
- 15) Chronometer management
- 16) Paper clip
- 17) CWS
- 18) AP / TRIM DISC push-button
- 19) Push To Talk button (PTT)
- 20) Pitch & Yaw trim setting management

Figure 7.3.5A (1/2) - Right instrument panel





- 1) THROTTLE
- 2) FLAPS lever
- 3) THROTTLE friction adjustment
- 4) Manual FUEL TANK SELECTOR - see figure 7.7.2
- 5) Roll trim tab control
- 6) MAN OVRD emergency fuel regulation lever
- 7) Pitch trim tab control
- 8) Lock for access door to landing gear emergency pump - see figure 7.5.2

Figure 7.3.6 (1/2) - Pedestal console

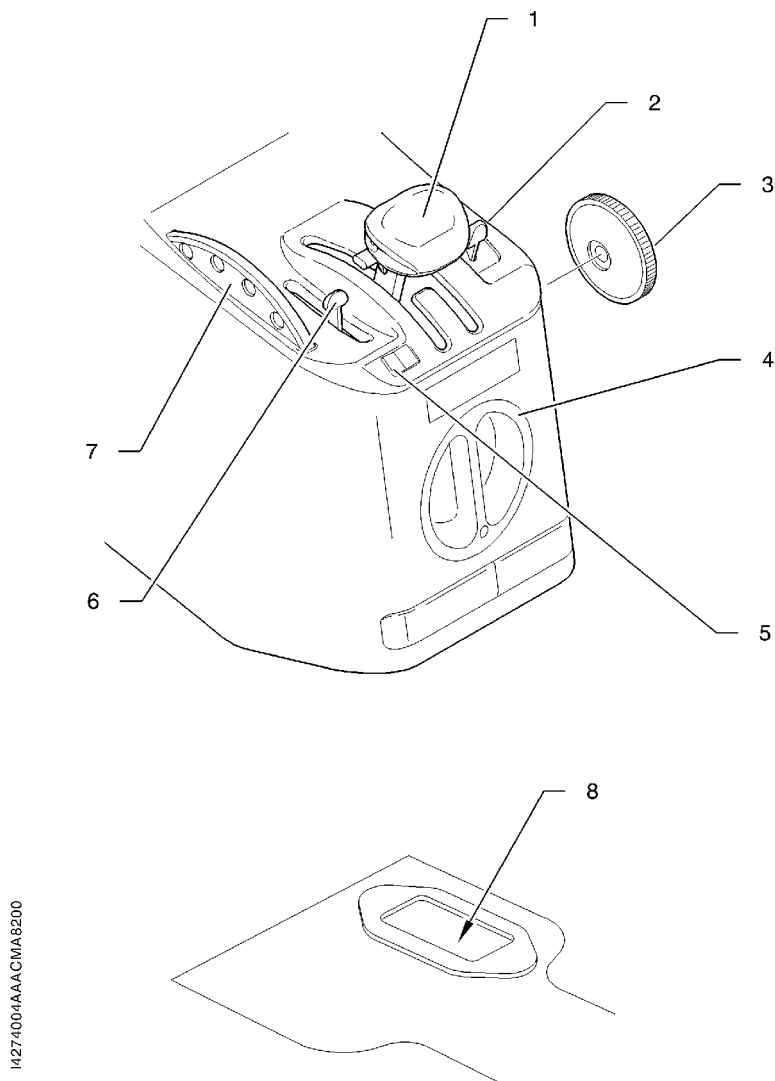
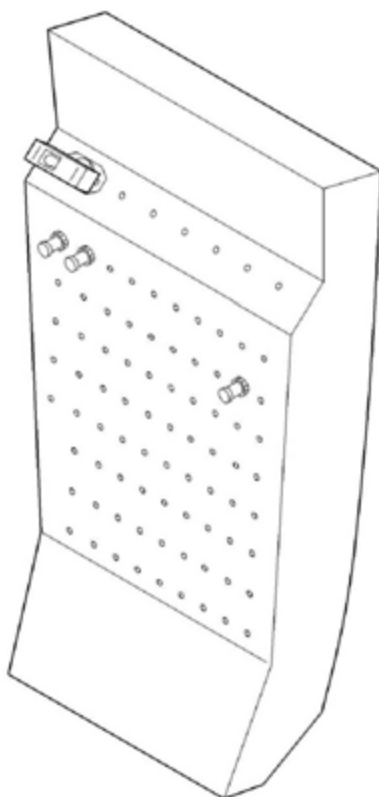
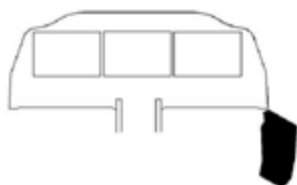


Figure 7.3.6 (2/2) - Pedestal console  
(Typical arrangement)



14255004AA/PMA 0200

Figure 7.3.7 - Circuit breakers panel



Figure 7.3.8 - General alarms warning lights

## Doors, windows and emergency exit

**Cabin access door** - see figure 7.3.9

The cabin one-piece access door, located on the left side of fuselage aft of the wings, opens outside. The retractable stairs and hand rail make boarding easier.

To open the door from outside the airplane (make sure the door is not locked), press on front end of the handle embedded in door (this pressure disengages the handle from its recess), then turn the handle upwards. Raise the door helping it to open. Two compensation actuators bring and maintain the door at its maximum opening position.

After door opening, tilt stairs downwards. Stairs down movement is damped by means of two gas struts and leads the hand rail to extend.

### ▲ CAUTION ▲

Retract stairs before closing access door and make sure door deflection area is clear.



To retract stairs, press on locking pin located on stairs front string board (see detail 1), raise retractable handle - see detail 2 and pull stairs inside cabin. While stairs are retracted, the hand rail folds up.

To close the door from inside the airplane, press on knob inside cabin forward of the door. The door driven by a geared motor tilts downwards up to a position near the complete closing. Pull the door until it aligns with fuselage and lock it by moving inside handle downwards. Check that all latch pins and hooks are correctly engaged (visible green marks).

**DOOR** lights on as long as cabin access door and pilot access door, if installed, are not correctly locked.

### ▲ CAUTION ▲

Before opening access door, make sure door deflection area is clear.



To open door from inside the cabin, unlock the handle by pressing on knob located on its left side, pull the handle toward inside and move it upwards. Open the door by pushing it upwards.

After door opening, tilt stairs downwards which leads the hand rail to extend.

**▲ CAUTION ▲**

Retract stairs before closing access door and make sure door deflection area is clear.

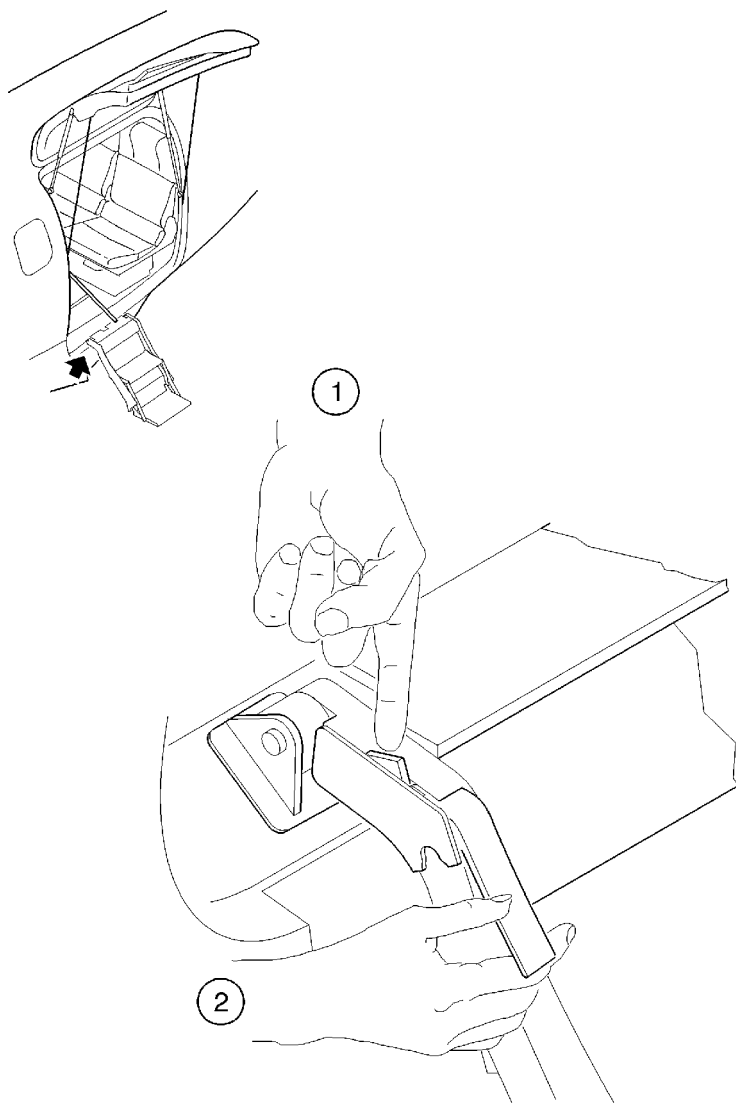


To retract stairs from outside the airplane, raise stairs by pushing them upwards from the lower part and fold them inside cabin. While stairs are retracted, the hand rail folds up.

To close the door from outside the airplane, press on knob on outside fuselage at the right side of the door. The door driven by a geared motor tilts downwards up to a position near the complete closing. Push the door until it aligns with fuselage and lock it by moving outside handle downwards, then fold handle in its recess.

Check that all latch pins and hooks are correctly engaged, with green marks visible.

In case of geared motor failure, the door can be manually tilted downwards by pulling sufficiently to override action of compensating struts.



14523000AAA FMA8001

Figure 7.3.9 - Cabin access door



**Cockpit access door** - see figure 7.3.9A

The cockpit access door, so-called pilot door, if installed located on the left side of fuselage forward of the wings, opens outside. Retractable footstep makes boarding easier.

**▲ WARNING ▲**

**As the pilot door is located in a dangerous area, wait for complete engine stop before operating this door.**



To open the door from outside the airplane (make sure the door is not locked), press on front end of the handle embedded in door (this pressure disengages the handle from its recess), then turn the handle downwards. Pull the door helping it to open until it reaches its maximum opening position.

After door opening, tilt and unfold footstep.

**▲ CAUTION ▲**

Retract footstep before closing access door.



Fold and tilt footstep upwards.

To close the door from inside the airplane, pull the door until it aligns with fuselage and lock it by moving inside handle downwards. Check that each latch is correctly engaged in its recess, with green marks visible.

**DOOR** lights on as long as cabin access door and pilot access door, if installed, are not correctly locked.

To open door from inside the cockpit, unlock the handle by pressing on knob located on its right side, pull the handle inwards and move it upwards. Open the door helping it to open until it reaches its maximum opening position.

After door opening, tilt and unfold footstep.

**▲ CAUTION ▲**

Retract footstep before closing access door.



Fold and tilt footstep upwards.

To close the door from outside the airplane, push the door until it aligns with fuselage and lock it by moving outside handle upwards, then fold handle in its recess.

### **FWD compartment door**

The FWD compartment door is located on the airplane left side between the firewall and the front pressure bulkhead. It is hinged at the top. It is maintained in the up position by a compensation rod. Two interlocking-type latches ensure its closing and it is equipped with a lock (same key as for the access door and the pilot door, if installed). When the door is closed, latches are flush with the fuselage profile.

**FRONT CARGO DOOR** lights on as long as FWD compartment door is not locked.

### **Windows**

Windows do not open. The windshield consists of two parts electrically deiced.

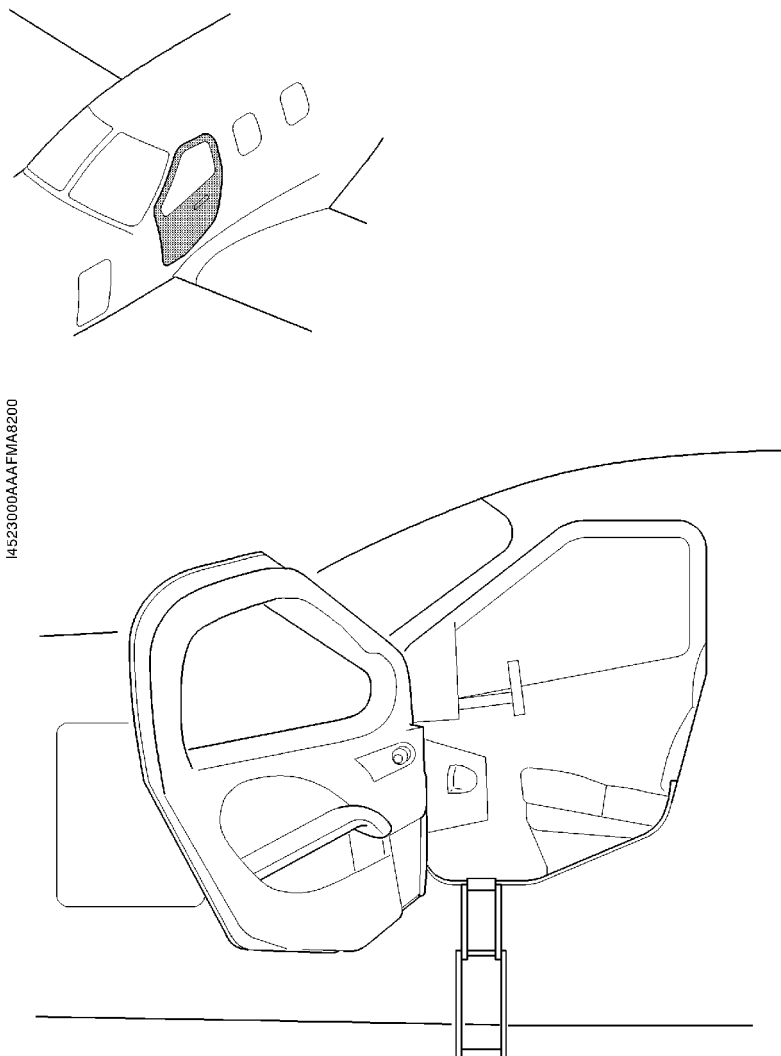


Figure 7.3.9A - Cockpit access door (pilot door)

**Emergency exit** - see figure 7.3.10

The emergency exit is installed on the right side of the fuselage and opens towards the inside. It is equipped with two handles, one inside and the other outside, each located on the upper frame.

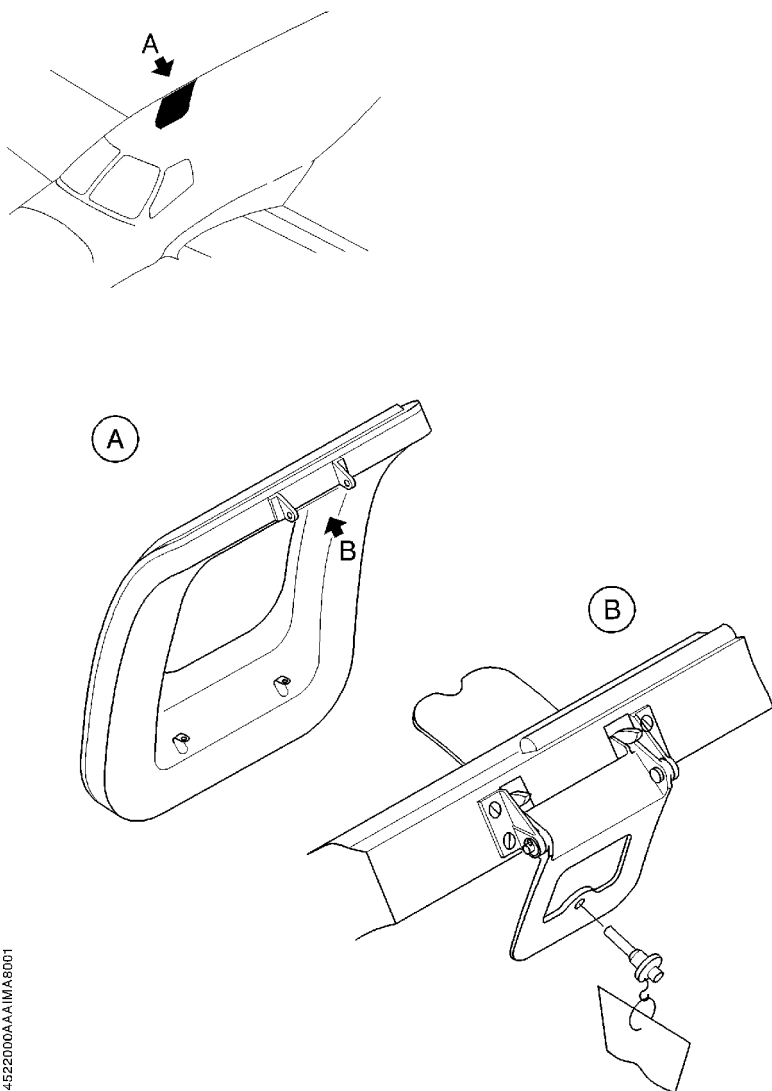
When the airplane is parked, the closing system may be locked by a safety pin provided with a flag marker. The handle is then inoperable.

**▲ WARNING ▲**

**Taxiing and flying with thief-proof safety pin installed is forbidden.**



To open the emergency exit, pull one of the two handles and tilt the emergency exit from top to bottom towards inside of airplane.



I-4522000AA-IMA8001

Figure 7.3.10 - Emergency exit

## Seats, belts and harnesses

**Cockpit seats** - see figure 7.3.11

L.H. and R.H. front seats are mounted on rails attached to the structure. Longitudinal position, height and back-rest tilting of each seat can be adjusted and the arm-rest is hinged.

Pull up the handle located forward for longitudinal setting.

The seat height is adjusted by pulling up side forward handle while relieving the seat from the body weight.

The seat back angle is adjusted by pulling up side rearward handle.

**Passenger seats** - see figures 7.3.11 and 7.3.11A

>> *With 6-seat accommodation*

The accommodation consists of :

- two individual seats, installed back to the flight direction, mounted on the same rails as the front seats.  
The seat back angle is adjusted by pulling up side handle.
- two rear seats arranged as a bench, mounted on the same rails as the front seats.  
The seat back-rests tilt forward by pulling up the handle located forward on L.H. side of each seat which may tilt forwards by pulling up a rear handle to ease baggage loading in baggage compartment.  
For longitudinal setting pull up the handle located forward, on R.H. side.

>> *With 4-seat accommodation*

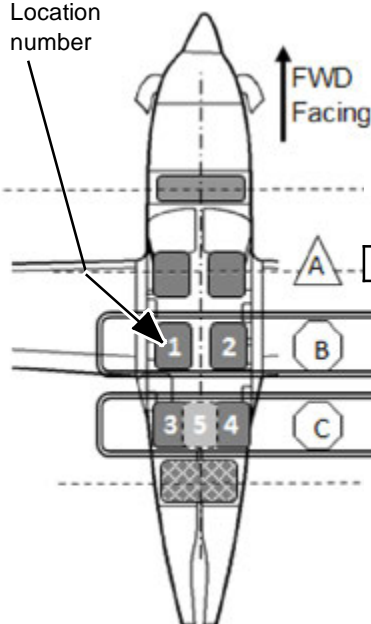
The accommodation consists of :

- two individual seats, installed facing flight direction, mounted on the same rails as the front seats.  
The seat back angle is adjusted by pulling up side handle.

Many accommodations are possible. They are described hereafter

ONLY zone **B** and zone **C** can be modified for seat configurations

Location  
number



For all configurations, verify that your luggages are stowed and attached in the appropriate areas

Fwd baggage zone

Pilot zone: No modification allowed

MID Seat Zone = Possibility of seat configuration

REAR Seat Zone = Possibility of seat configuration if no net installations

Cargo zone

If installed, cabinets can be removed or added by Service Center

### For the MID Seat zone **B**

ONLY the Middle Seats can be installed in MID Seat Zone.

This zone accepts Fwd and Aft Facing Mid Seat when rear seats are installed

The zone **B** accepts zero or 1 or 2 seats.

(The zone **B** is not a luggage area).

| Location number | FWD Facing | AFT Facing | Number of seat can be installed |
|-----------------|------------|------------|---------------------------------|
| 1               | YES        | YES        | 1 or 0                          |
| 2               | YES        | YES        | 1 or 0                          |

### For the REAR Seat zone **C**

ONLY the Rear Seat can be installed in Rear Seat Zone.

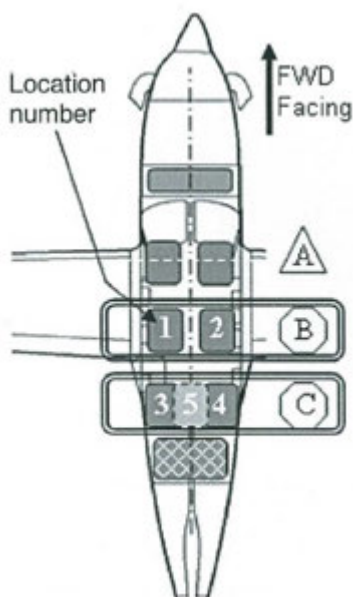
The Zone **C** accepts zero or 1 or 2 seats.

| Location number | FWD Facing | Number of seat can be installed |
|-----------------|------------|---------------------------------|
| 3               | YES        | 1 or 0                          |
| 4               | YES        | 1 or 0                          |
| 5 *(1)          | YES *(1)   | 1 or 0 *(1)                     |

\*(1) Centered on the fuselage axis

Here are all the configurations possibilities

| Configuration name | Location number |   |          |   |   |
|--------------------|-----------------|---|----------|---|---|
|                    | 1               | 2 | 3        | 4 | 5 |
| C1                 | X               | X | X        | X |   |
| C2                 | X               | X |          |   | X |
| C3                 | X               | X |          | X |   |
| C4 (1)             | X               | X |          |   |   |
| C5                 | X               | X | X        |   |   |
| C6                 | X               |   | X        | X |   |
| C7                 | X               |   | X        |   |   |
| C8                 | X               |   |          | X |   |
| C9                 | X               |   |          |   | X |
| C10 (1)            | X               |   |          |   |   |
| C11                |                 | X | X        | X |   |
| C12                |                 | X |          |   | X |
| C13                |                 | X | X        |   |   |
| C14                |                 | X |          | X |   |
| C15 (1)            |                 | X |          |   |   |
| C16                |                 |   | X        | X |   |
| C17                |                 |   | X        |   |   |
| C18                |                 |   |          | X |   |
| C19                |                 |   |          |   | X |
| C20 (1)            |                 |   |          |   |   |
|                    | Zone (B)        |   | Zone (C) |   |   |



(1) This configuration accepts small net or large net

Each cross indicates that you have a seat at the correspondent location number.



**Belts and harnesses** - see figure 7.3.12

**▲ WARNING ▲**

**Incorrect closure of the safety belt may introduce a risk. Make sure it is tightened when buckled. To be most efficient, the belt must not be twisted. Check that there is no constraint when operated. After a serious accident, replace all belts.**



Each cockpit seat is equipped with a four-point restraint system consisting of an adjustable lap belt and a dual-strap inertia reel-type shoulder harness with airbags, if installed.

Each passenger seat is equipped with a three-point restraint system consisting of an adjustable lap belt and an inertia reel-type shoulder harness.

Airbags, if installed, are inflated by two inflators located under the backrest fairing, which are activated by an accelerometer fixed under the floor panel in front of the seat.

## **Baggage compartments**

>> *With 6-seat accommodation*

There are two baggage compartments :

- An AFT compartment located in the pressurized cabin between rear passenger seats and rear pressure bulkhead.
- A FWD compartment (non-pressurized) located between firewall and fwd pressure bulkhead.

The AFT compartment is accessible through the cabin by tilting forward the L.H. rear seat and / or L.H. or R.H. rear seat back-rests. Rings fitted with lashing straps are provided for securing parcels and baggage on compartment floor.

The FWD compartment is accessible by opening the external door located on the left side of the airplane.

These locations are designed for the carrying of low density loads ; loading and unloading must be carried out with caution to avoid any damage to airplane.

The cabin is separated from the baggage compartment by a partition net intended to protect the passengers from injuries that could be caused by improper tie-down of a content.

The partition net is mounted at frame C14 - see figure 7.2.1, it is secured at the bottom to 4 points of the floor and on the sides to 6 points of the structure.

Maximum loads allowable in the baggage compartments depend on airplane equipment, refer to section 6 Weight and balance.

▲ **WARNING** ▲

**Any parcel or baggage must be stowed by straps.**

**It is the pilot's responsibility to check that all the parcels and baggage are properly secured in the cabin.**

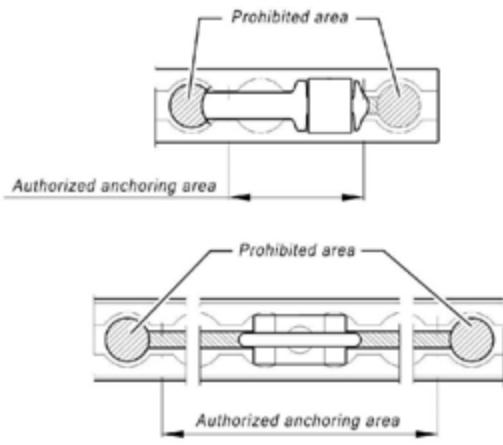
**In case of transport of dangerous materials, respect the law concerning transport of dangerous materials and any other applicable regulation.**



>> *With 4-seat accommodation*

Two cargo nets are available for the pilot to safely secure and transport baggage :

- the small cargo net is attached through nine anchoring points on seat rails, between frame C11 and frame C13bis - see figure 7.2.1B.

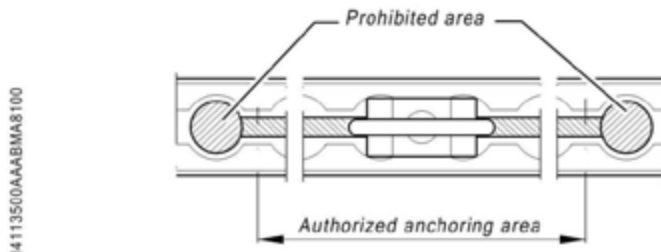


- the large cargo net is attached through seven anchoring points on seat rails, between frame C11 and frame C13bis and six anchoring points on fuselage sides, at frame C14 - see figure 7.2.1A.

● **NOTE** ●

Original partition net must be disconnected from side walls and placed on the floor.





Authorized anchoring points are identified with green self-adhesive labels affixed to the inside of the seat rail.

A placard indicates loading limits for each cargo net :

- for the small cargo net, it is affixed on frame C13bis,
- for the large cargo net, it is affixed on R.H. side upholstery panel, in the rear baggage compartment.

Maximum loads allowable in the baggage compartments depend on airplane equipment, refer to section 6 Weight and balance.

**▲ WARNING ▲**

**Any parcel or baggage in cabin must be stowed by cargo net and straps.**

**It is the pilot's responsibility to check that all the parcels and baggage are properly secured.**

**In case of transport of dangerous materials, respect the law concerning transport of dangerous materials and any other applicable regulation.**



## Use of cargo nets

### Net inspection

Before each use, visually inspect net for :

- webbing condition,
- seam condition of tensioning strap,
- metallic part condition.

### Installation instructions

Tensioning straps must be installed so that they make a V with a minimum angle of 40° between both strands attached on the net. The net must be properly tight.

### Damage acceptance criteria

If any damage is detected, such as :

- damage or absence of hook, buckle or stud on tensioning strap : strap must **mandatorily** be discarded and replaced,
- webbing frayed or cut on less than 30 % of its surface : reduce maximum load by 50 %,
- seam of vertical net tensioning straps damaged on less than 30 % of its length : reduce maximum load by 50 %,
- seam of tensioning straps attached on the rails damaged on less than 30 % of its length : reduce maximum load by 50 %,
- beyond 30% damage for above-mentioned cases, defective element must mandatorily be discarded and replaced,
- netting cut or torn on less than 3.9 in (100 mm) : still serviceable, no impact,
- netting cut or torn on more than 3.9 in (100 mm) : do not carry small objects which dimensions are smaller than 4.9 x 4.9 x 4.9 in (125 x 125 x 125 mm).

>> *With 6-seat accommodation*

- 1) Front passenger seat
- 2) L. H. pilot seat
- 3) R. H. intermediate passenger seat, back to flight direction
- 4) L. H. intermediate passenger seat, back to flight direction
- 5) R. H. rear passenger seat
- 6) L. H. rear passenger seat
- 7) Front seat(s) longitudinal shift control
- 8) Front seat(s) height control
- 9) Front seat(s) back-rest tilt control
- 10) Drawer for pilot's piddle pak, if installed  
(front side : new bags, rear side : used bags)
- 11) Intermediate seat(s) back-rest tilt control
- 12) Rear bench seat(s) back-rest tilt control
- 13) Rear bench L.H. seat tilt control
- 14) Rear bench seat(s) adjustment control handle

• NOTE •

To have access to the baggage compartment, pull forwards the back-rest of rear bench L.H. seat, then pull forwards control (Item 13) to tilt L.H. seat assembly forwards.

If necessary, pull forwards the back-rest of rear bench R.H. seat.

•

Figure 7.3.11 (1/2) - Seats

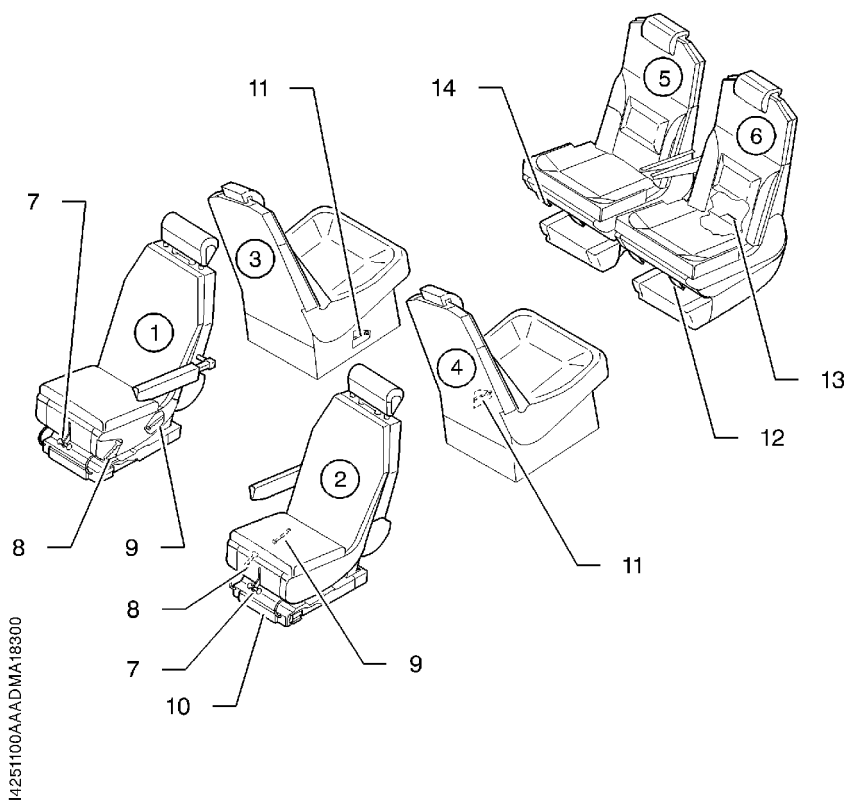
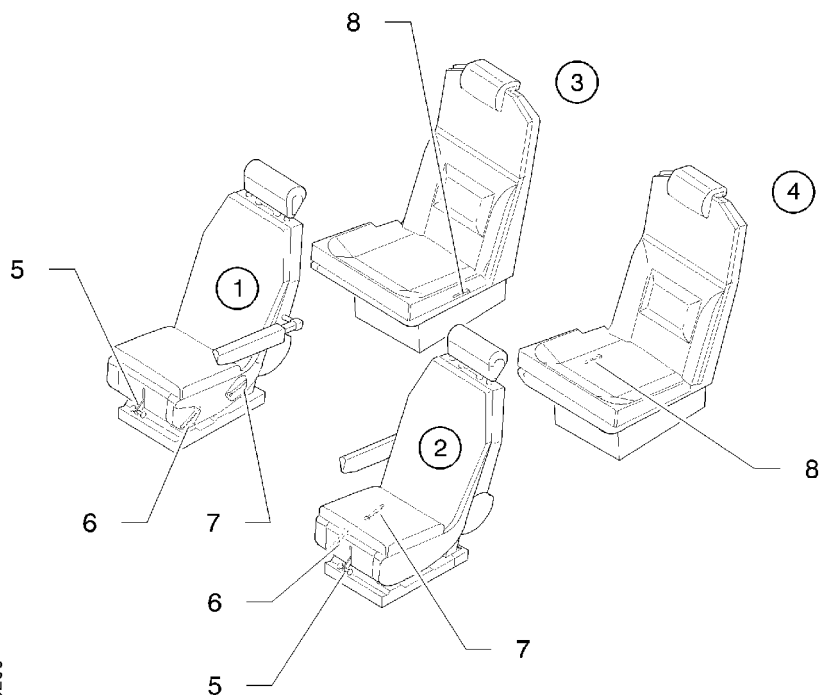


Figure 7.3.11 (2/2) - Seats

>> *With 4-seat accommodation*

- 1) Front passenger seat
- 2) L. H. pilot seat
- 3) R. H. intermediate passenger seat, facing flight direction
- 4) L. H. intermediate passenger seat, facing flight direction
- 5) Front seat(s) longitudinal shift control
- 6) Front seat(s) height control
- 7) Front seat(s) back-rest tilt control
- 8) Intermediate seat(s) back-rest tilt control

Figure 7.3.11A (1/2) - Seats



I4251100AAADMA18200

Figure 7.3.11A (2/2) - Seats



>> All

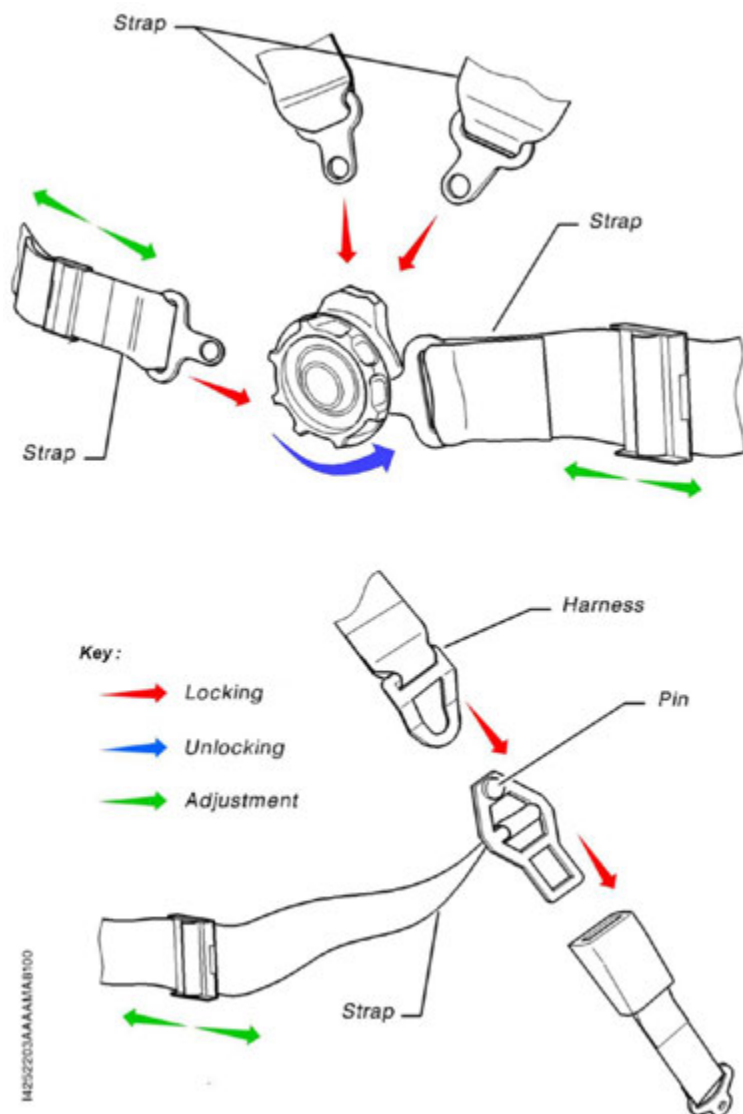


Figure 7.3.12 - Front and rear seat belts, with movable straps, and harnesses

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## 7.4 - Flight controls

Flight controls consist of roll, pitch and rudder controls, as well as roll trim tab, pitch trim tab and rudder trim tab controls.

• NOTE •

During airplane parking, it is recommended to lock flight controls -  
see figure 8.6.2

•

### **Roll** - see figure 7.4.1

The roll control is activated by an assembly of rods and cables which links control wheels with the ailerons and the spoilers.

Aileron displacement is combined with that of spoilers, located at upper surface of each wing forward of flaps.

The spoiler rises from wing upper surface profile, when the aileron is deflected upwards and remains in wing profile, when the aileron is deflected downwards.

Control wheel movement is transmitted through rods to fuselage roll lever located under the floor. The movement is then transmitted through cables to the spoiler mechanism and from the spoiler mechanism to wing roll lever which activates the aileron through a rod.

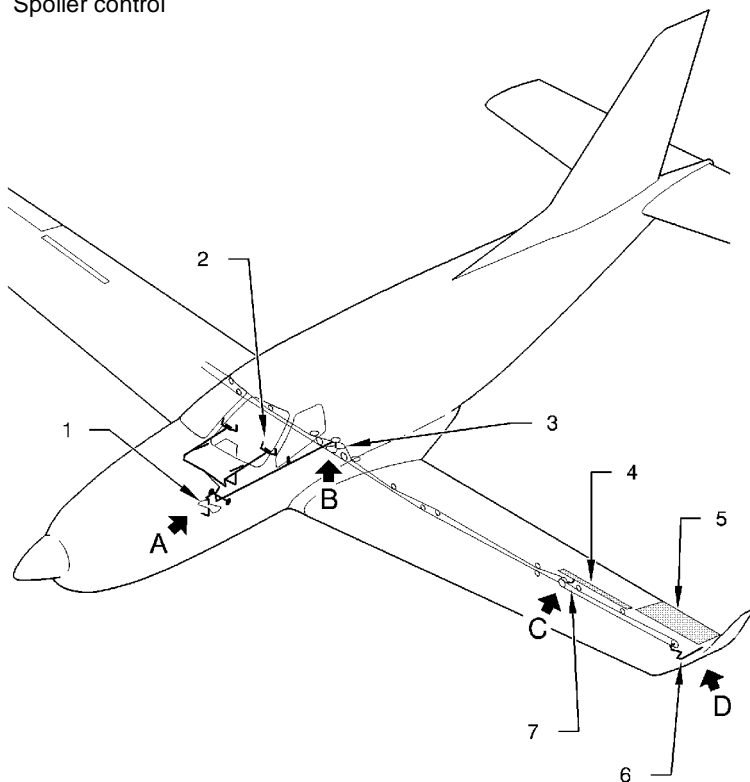
A rudder / roll combination spring-type system induces roll deflection at the time of pedals movement and vice versa.

### **Roll trim** - see figure 7.4.2

The roll trim is controlled by a trim tab attached at trailing edge of the L.H. aileron. The trim tab is connected through two links to an electric actuator located in the aileron. A trim switch located on pedestal controls the roll trim tab maneuver.

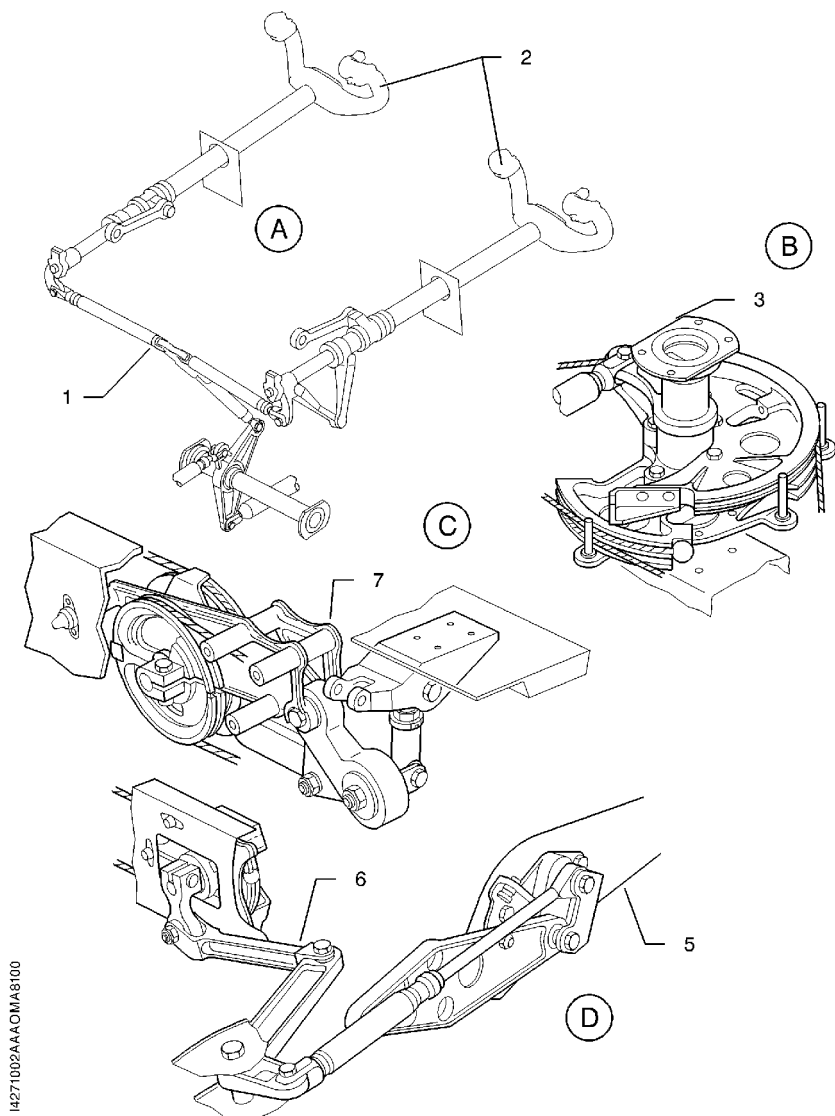
Roll trim tab electrical circuit is protected by the AIL TRIM breaker.

- 1) Pedestal assembly
- 2) Control wheels
- 3) Fuselage roll lever
- 4) Spoiler
- 5) Aileron
- 6) Aileron control in wing
- 7) Spoiler control



14271000AAAAMAB101

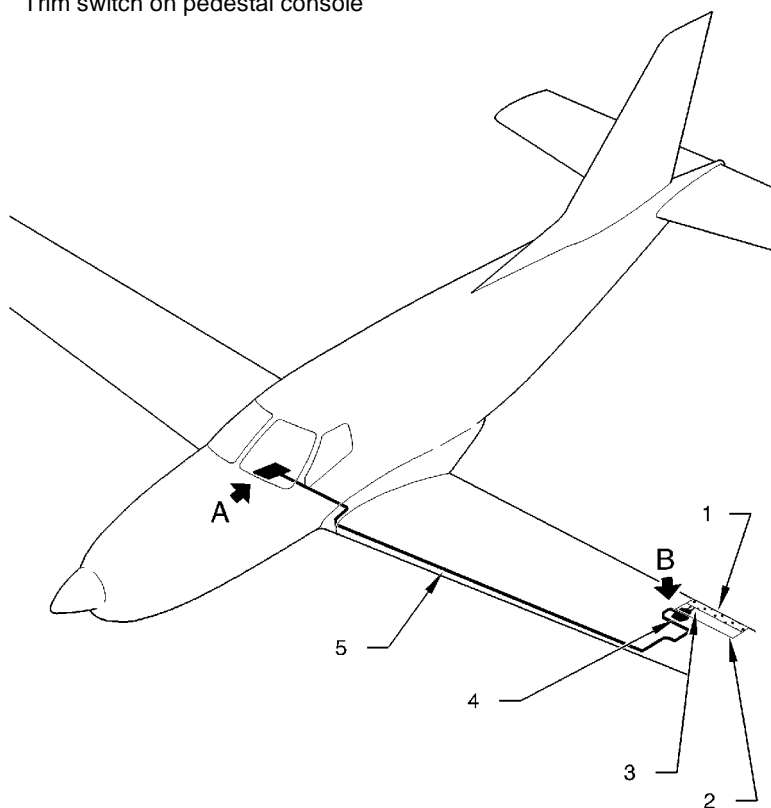
Figure 7.4.1 (1/2) - Roll



14271002AAACMA8100

Figure 7.4.1 (2/2) - Roll

- 1) Roll trim tab
- 2) Aileron
- 3) Adjustable rods
- 4) Actuator
- 5) Trim tab control wiring
- 6) Trim switch on pedestal console



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Figure 7.4.2 (1/2) - Lateral trim

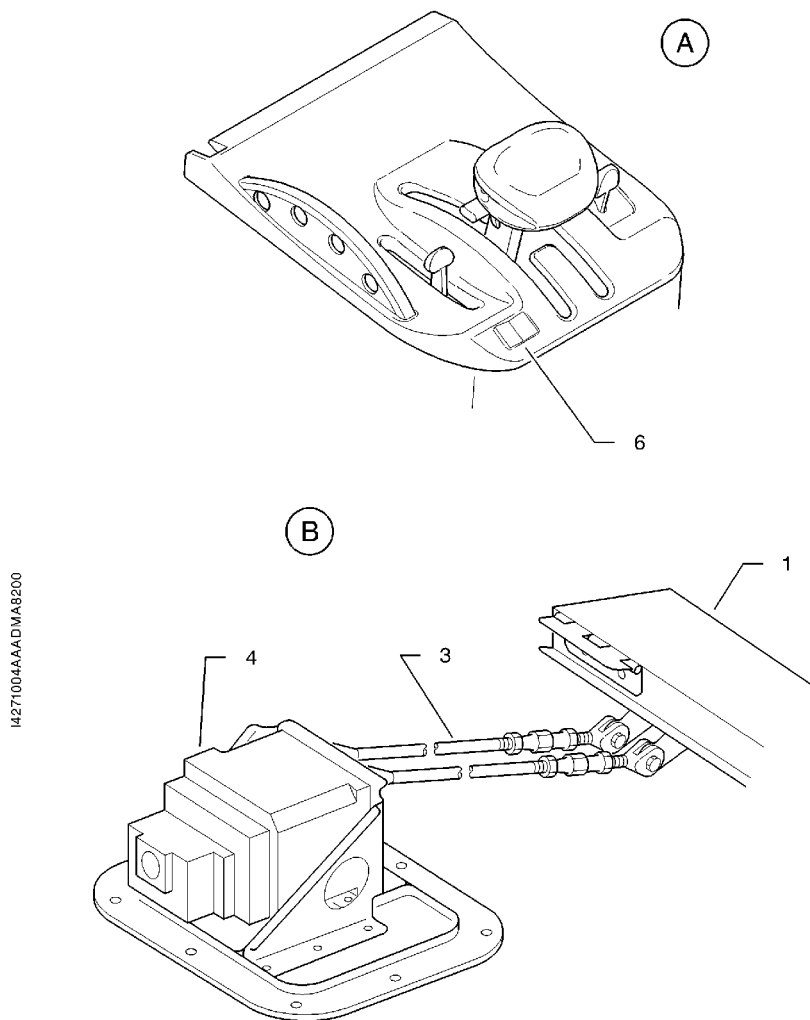


Figure 7.4.2 (2/2) - Lateral trim

**Elevator** - see figure 7.4.3

Both elevators are activated simultaneously by the same control. Each control surface is hinged at three points to the rear part of horizontal stabilizer.

The control wheel controls the two elevators through rods, bearings and bellcranks.

A stick shaker is fixed on the pitch lever linked to the pilot control column lever. This is a mechanical device to vibrate the control wheel to warn the pilot in case of an imminent stall. When the data received from the AoA (angle of attack) sensor indicates an imminent stall, the AoA computer actuates both the stick shaker and the stall warning.

A spring actuator creates a nose-down artificial force which allows a better static stability.

Each control surface is provided with an automatic anti-tab (automaticity about 0.3), which is also used as trim tab.

**Pitch trim** - see figure 7.4.4

The pitch trim is accomplished through the two anti-tabs located on left and right elevators.

The trim tab can be controlled electrically or manually. It is activated through cables and a chain on two screw actuators attached to the horizontal empennage.

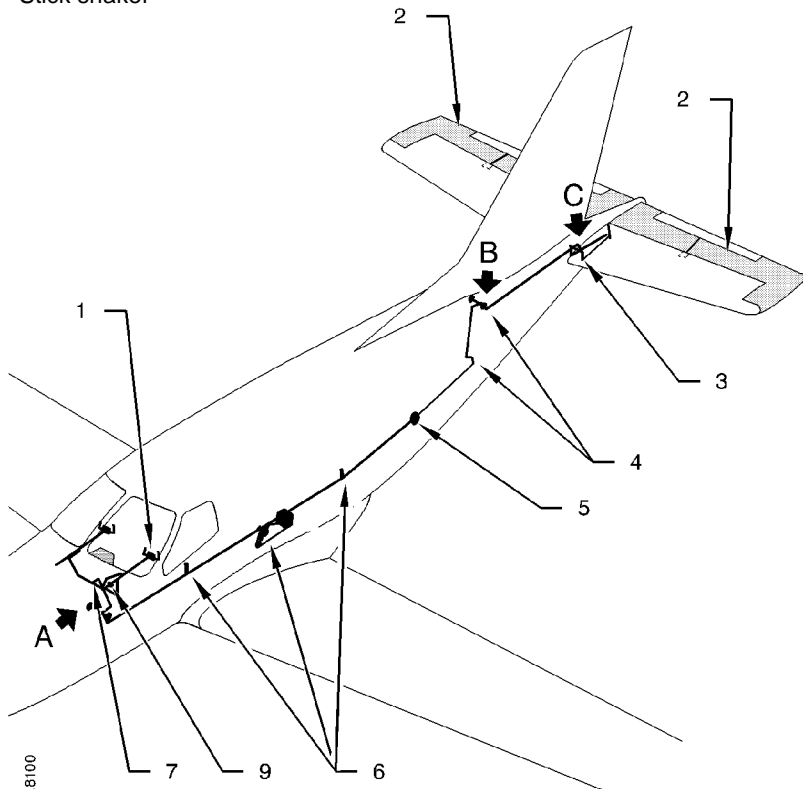
The electrical control consists of a switch (NOSE UP - NOSE DOWN) located on the pilot control wheel and a servo-motor attached under the pedestal.

The electrical circuit for pitch trims is protected by the AP SERVOS breaker.

Manual control wheel is installed vertically on left side of pedestal console.



- 1) Control wheel assembly
- 2) Elevators
- 3) Lever assembly, fuselage rear part
- 4) Elevator bellcrank
- 5) Rod with presseal connection
- 6) Lever assembly under floor
- 7) Pedestal assembly
- 8) Actuator
- 9) Stick shaker



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Figure 7.4.3 (1/2) - Elevator

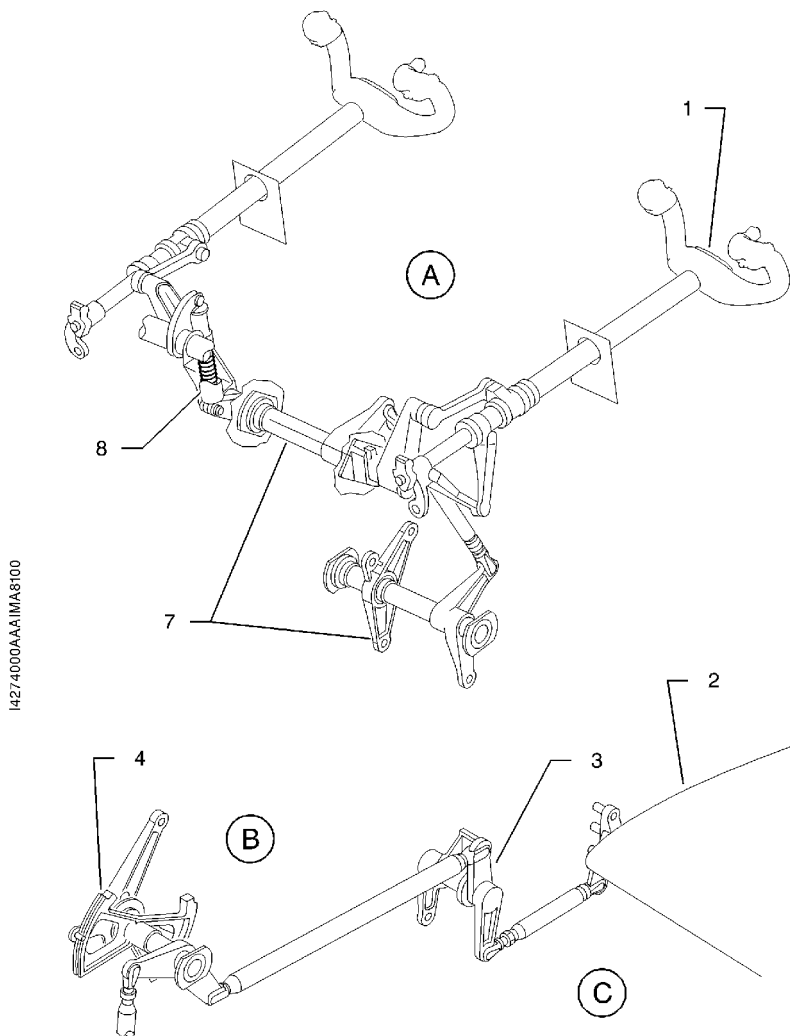
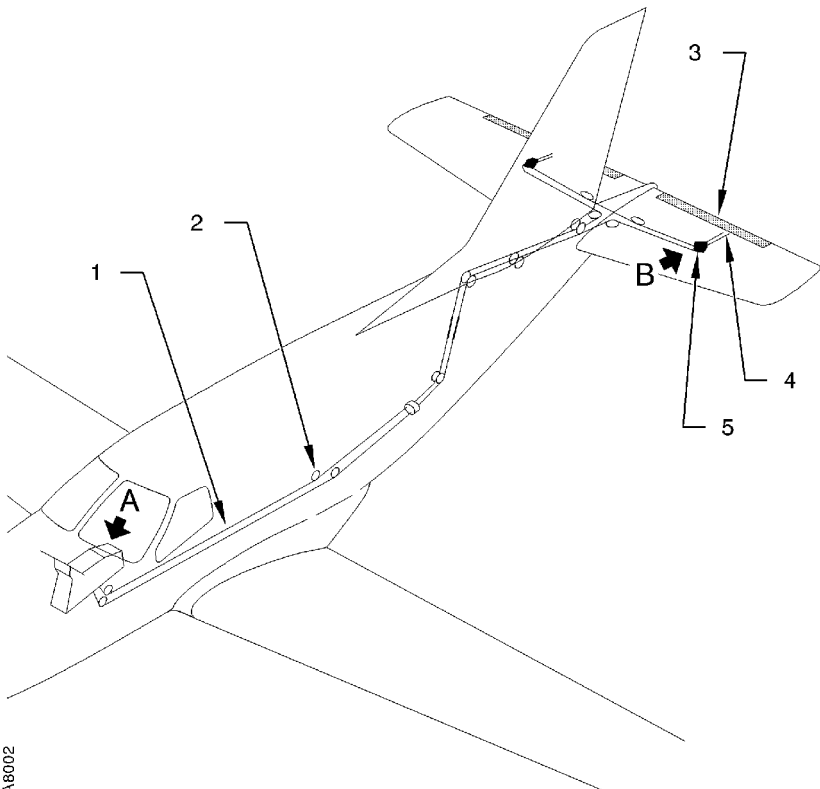


Figure 7.4.3 (2/2) - Elevator

- 1) Cables
- 2) Pulleys
- 3) Pitch trim tabs
- 4) Actuating rods
- 5) Actuator
- 6) Pitch trim manual control wheel
- 7) Electric pitch trim control



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Figure 7.4.4 (1/2) - Pitch trim

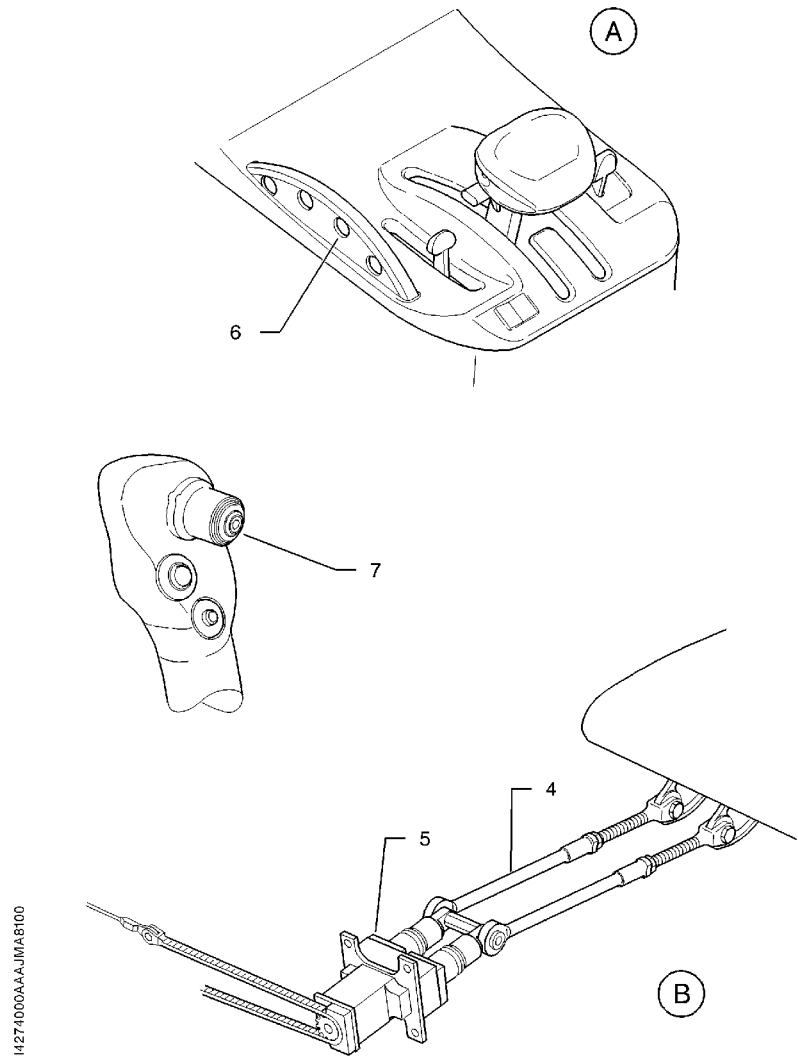


Figure 7.4.4 (2/2) - Pitch trim

**Rudder** - see figure 7.4.5

The rudder is hinged on three fittings attached to the vertical stabilizer rear spar.

The rudder pedals / rudder linkage is ensured through cables and a rod.

Pilot and R.H. station rudder pedal positions are adjustable at each station. The rudder pedal adjustment mechanism (for piloting comfort purposes) includes a manual control located against the external bulkhead beneath the instrument panel and a locking device on the rudder pedals. This ball locking device allows selecting six different positions.

When landing gear is down, rudder pedals are linked to nose gear steering system.

Spring system of rudder / roll combination induces aileron deflection at the time of pedal displacement and vice versa.

**Rudder trim** - see figure 7.4.6

A trim tab hinged at two points located at rudder trailing edge provides rudder trim.

Trim tab is linked by two rods to an electric actuator attached to rudder. It is controlled by rudder trim switch (Y L / Y R) located on pilot control wheel.

Electrical circuit of rudder trim tab is protected by RUD TRIM breaker.

- 1) Roll / rudder combination bellcrank installation
- 2) Rudder pedals assembly
- 3) Control cables
- 4) Pulleys
- 5) Rudder lever assembly
- 6) Rod
- 7) Rudder
- 8) Nose gear steering rod

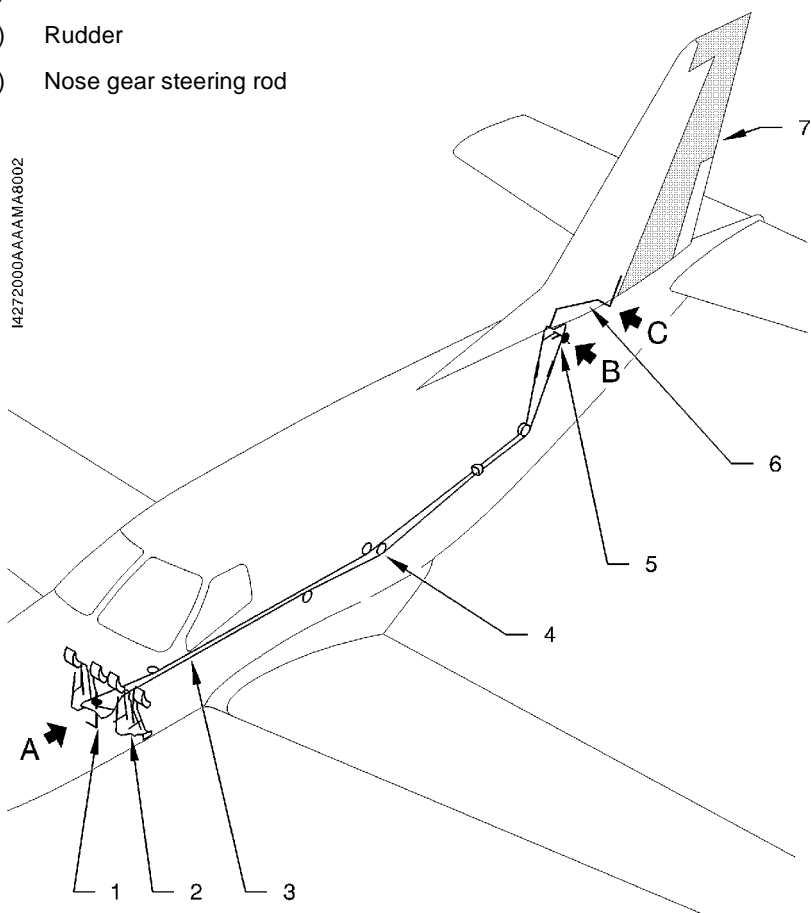


Figure 7.4.5 (1/2) - Rudder

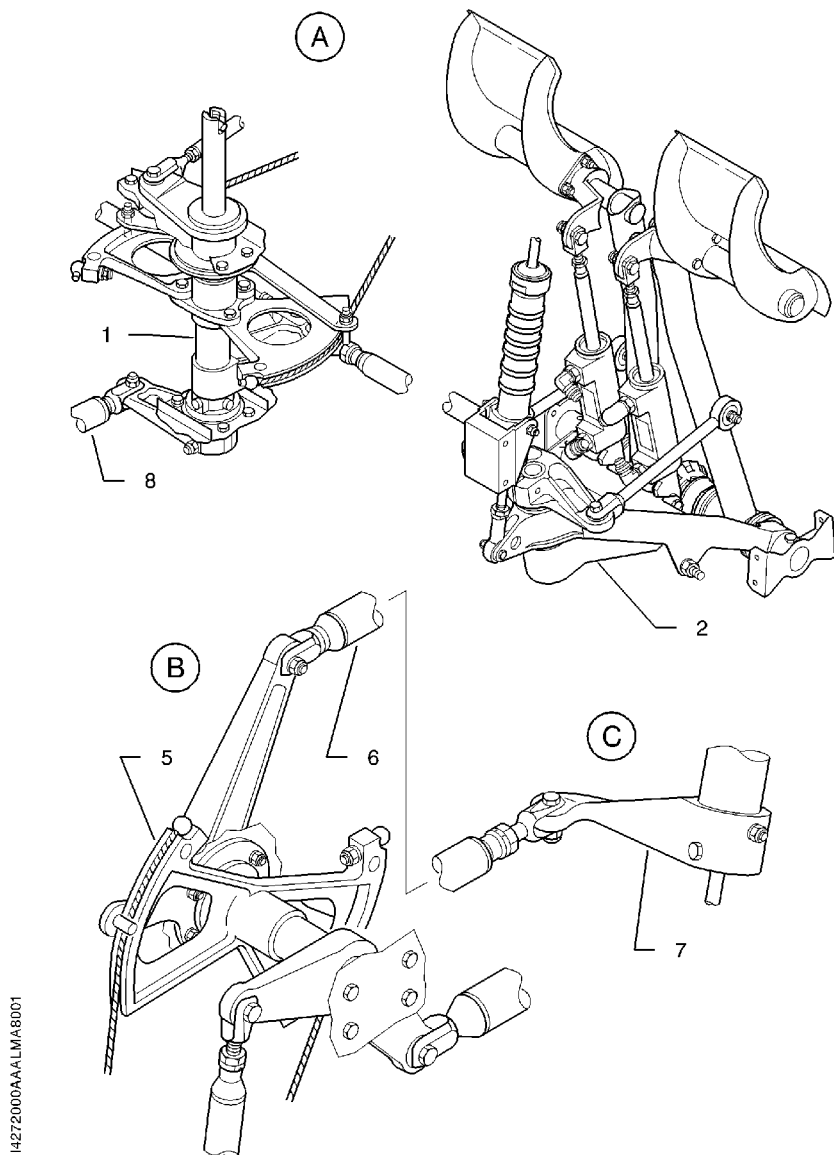
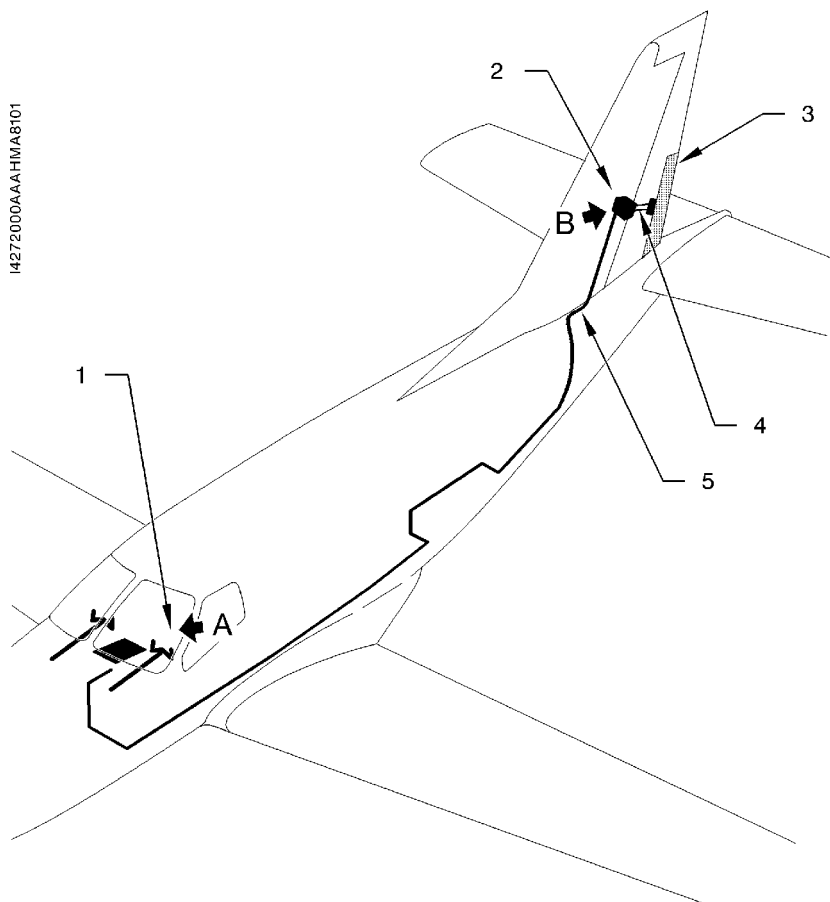
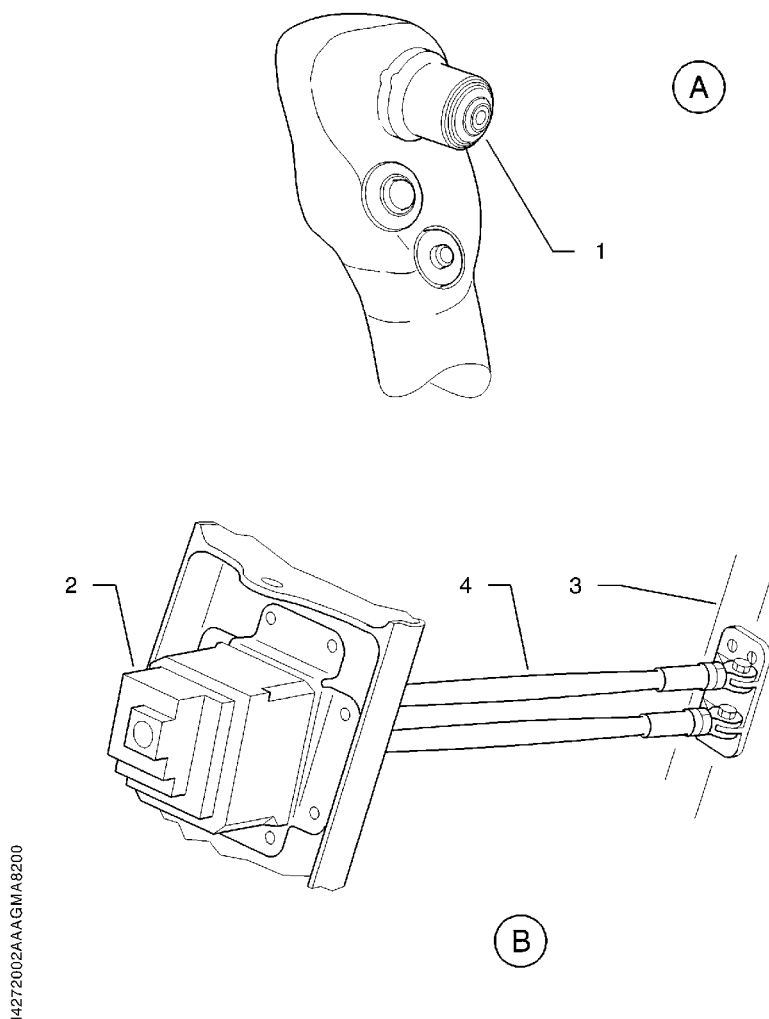


Figure 7.4.5 (2/2) - Rudder



Page 7.4.14





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Figure 7.4.6 (2/2) - Rudder trim

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## 7.5 - Landing gear

The airplane is equipped with electro-hydraulically actuated, fully retractable tricycle landing gear.

Each landing gear is equipped with one wheel and an oil-air shock absorber integrated in the strut.

**Main landing gears** swivel on two ball joints installed on wing spars. Each landing gear retracts toward airplane centerline. The operation is accomplished by a hydraulic actuating cylinder which also provides up and down locking.

**Nose gear** swivels on two ball joints installed on a tubular steel mount frame. Its operation is accomplished by a hydraulic actuating cylinder which also provides up and down locking. The nose wheel is steerable. It is connected to pedals through a spring rod and is provided with a shimmy damper. In UP position, nose wheel is automatically disconnected.

**Actuating cylinders** have a locking device integrated at both ends. This device maintains landing gear in up or down position.

**Landing gear doors**, two on the nose gear, two on each main landing gear, are driven and kept in UP position by the landing gear itself.

All doors are mechanically kept in down position.

### Hydraulic pressure

**Hydraulic pressure** required for landing gear operation is provided :

- during normal operation, by an electro-hydraulic generator with integrated reservoir,
- during emergency extension operation by a hand pump supplied with an auxiliary reservoir.

### Landing gear lever - see figure 7.5.1

LANDING GEAR lever, located on LANDING GEAR panel at the bottom of instrument panel left part, is accomplished by an electric selector actuated through a lever ending with a knob representing a wheel. Operation is carried out by pulling on lever and by putting it in the desired UP (retracted) or DN (extended) position. This selector controls hydraulic generator.

## **Landing gear position indicator** - see figure 7.5.1

Landing gear position indication is accomplished by 5 lights :

- On LANDING GEAR control panel
  - . 3 green indicator lights (one per landing gear),
  - . 1 red warning light GEAR UNSAFE
  - . 1 amber light in the LANDING GEAR lever.
- On MFD CAS window :
  - . **GEAR UNSAFE**

• NOTE •

The amber light flashes while the hydraulic pump is operating to extend or retract the landing gear.

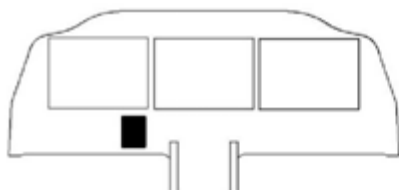
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When landing gear is correctly retracted, all lights are OFF.

Down-locked correct indication is when there are 3 green indicator lights ON, the GEAR UNSAFE red warning light is OFF, the **GEAR UNSAFE** is OFF and the amber caution light is OFF. All other cases mean the gear is not down-locked.

In case of doubt about landing gear down-locked position, an independent electrical circuit provides a countercheck capability of the indication system. Pressing the CHECK DOWN push-button, located on the landing gear panel, checks the down-lock of the gear making twinkle, at 16 hertz, the green indicator lights corresponding to the down-locked gear.

Pressing the LIGHT TEST push-button allows testing all landing gear panel lights making them flash at 1 hertz.



- 1) Green indicator light
- 2) Red warning light
- 3) LANDING GEAR lever
- 4) CHECK DOWN push-button
- 5) LIGHT TEST push-button
- 6) Amber light

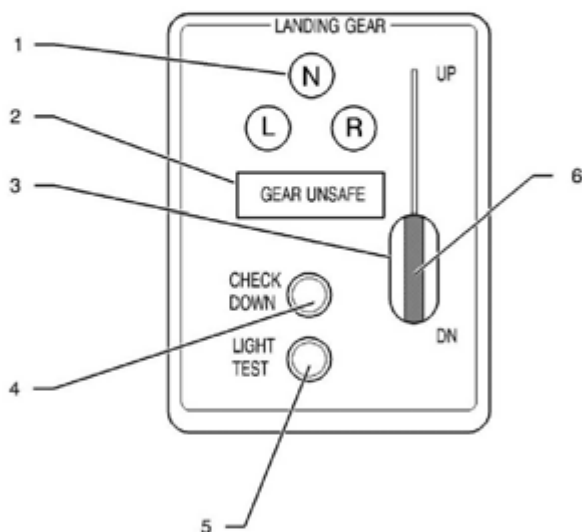


Figure 7.5.1 - Control panel and landing gear indicating

## **Safety**

### **Safety switch - landing gear retraction**

A safety switch installed on each main landing gear prevents, by detecting shock strut compression, landing gear accidental retraction when airplane is on ground.

### **Landing gear aural warning**

Landing gear / Landing gear aural warning alert sounds when :

- THROTTLE is on IDLE position and landing gear is not down-locked,
- flaps are beyond TO position (Takeoff) and landing gear is not down-locked.

• NOTE •

If one of above conditions exists and airplane is in stall configuration, the Stall/landing gear aural warning alert sounds and the control wheel vibrates.

•

### **Emergency landing gear extension control - see figure 7.5.2**

Emergency landing gear extension control consists of a hand pump and a by-pass selector.

This control is accessible by removing the floor panel located aft of the pedestal.

After bypass selector closing, hand pump operation sends hydraulic fluid directly into landing gear actuators ; landing gear full extension and locking requires up to 110 cycles.

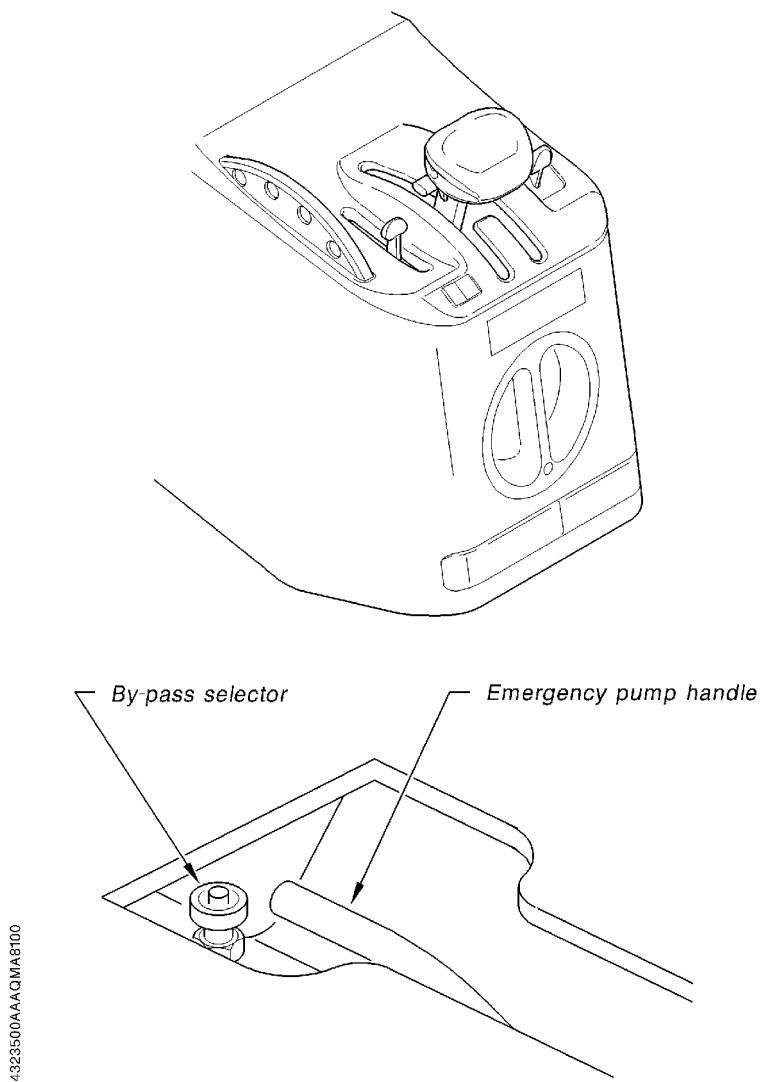


Figure 7.5.2 - Emergency landing gear extension control

## Ground maneuvers

**Nose gear steering control** - see figures 7.5.3 and 7.5.4

Nose gear steering control is combined with rudder pedals and is fitted with a shimmy damper. When one of rudder pedals is fully pushed, nose wheel swivels about 20°. Steering may be increased up to 28° by applying differential braking to each side.

Airplane may be towed by attaching a steering or towing bar on nose gear, refer to chapter 8.6 for operation. In that case nose wheel steering angle is limited to  $\pm 28^\circ$ .

### Minimum turn diameter

Minimum turn diameter, figure 7.5.4, is obtained by using nose gear steering and differential braking.

#### ▲ CAUTION ▲

Since tight turns lead to untimely tire wear, turns should be made using the largest possible turning radius.





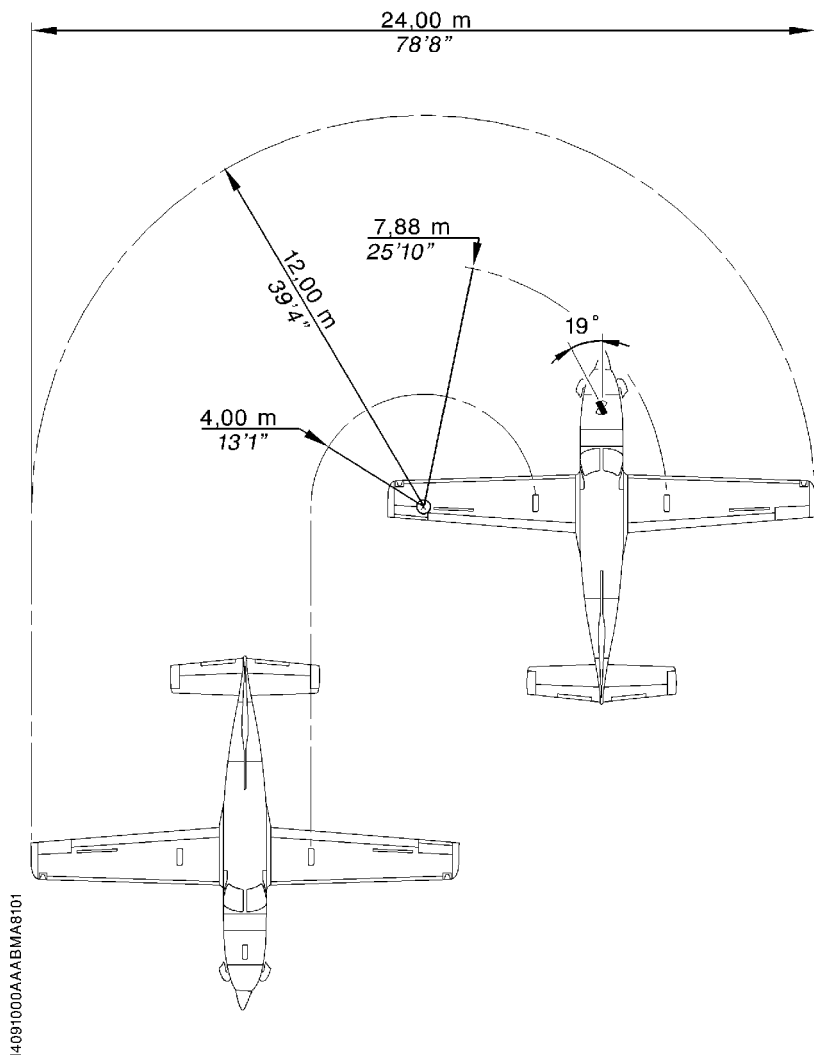


Figure 7.5.3 - Minimum turn diameter  
(Full rudder pedals travel without using differential braking)

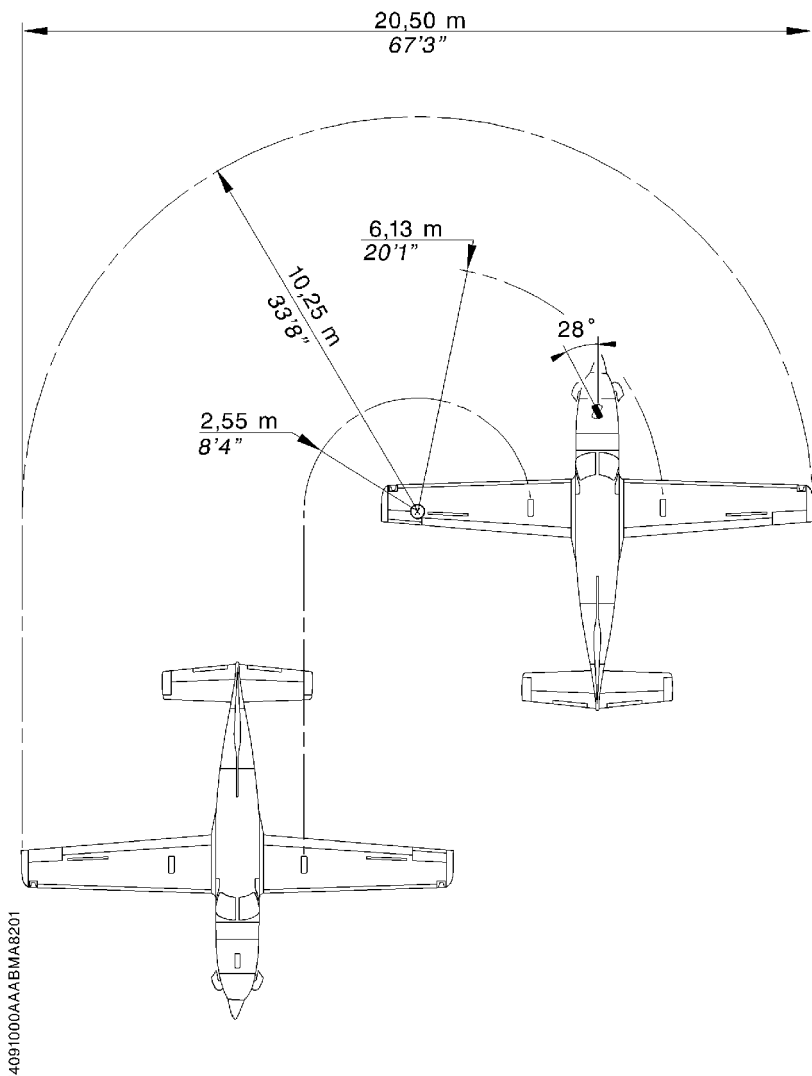


Figure 7.5.4 - Minimum turn diameter  
(Full rudder pedals travel by using differential braking)

**Brake system** - see figure 7.5.5

Airplane is equipped with a hydraulically actuated disc braking system installed on the main landing gear wheels.

Each toe brake at L.H. and R.H. stations is equipped with a master cylinder which sends hydraulic pressure to the corresponding disc brake : L.H. pedals L.H. brake ; R.H. pedals R.H. brake. This differential braking helps maneuvering during taxiing.

**Parking brake** - see figures 7.5.5 and 7.5.6

Parking brake control consists of a control knob located on pilot's side lower instrument panel and a valve which regulates brake pressure.

To apply the parking brake, press on toe brake of rudder pedals and set the control knob to ON.

**PARK BRAKE** lights on when the control knob is set to ON.

• NOTE •

Operating the parking brake knob without applying pressure on rudder pedals does not cause the wheels to be braked.

•

▲ CAUTION ▲

Failure to apply brake pressure while releasing the parking brake can damage the parking brake valve. This damage can cause the parking brake valve to not release the pressure.

▲

To release the parking brake, press on toe brake of rudder pedals and set the control knob to OFF. Check at the same time that **PARK BRAKE** disappears.

- 1) Reservoir
- 2) Vent
- 3) R.H. station master cylinders
- 4) PARK BRAKE control knob
- 5) PARK BRAKE valve
- 6) Drain
- 7) Pilot's station master cylinders
- 8) L.H. brake assembly
- 9) R.H. brake assembly

Figure 7.5.5 (1/2) - Brake system

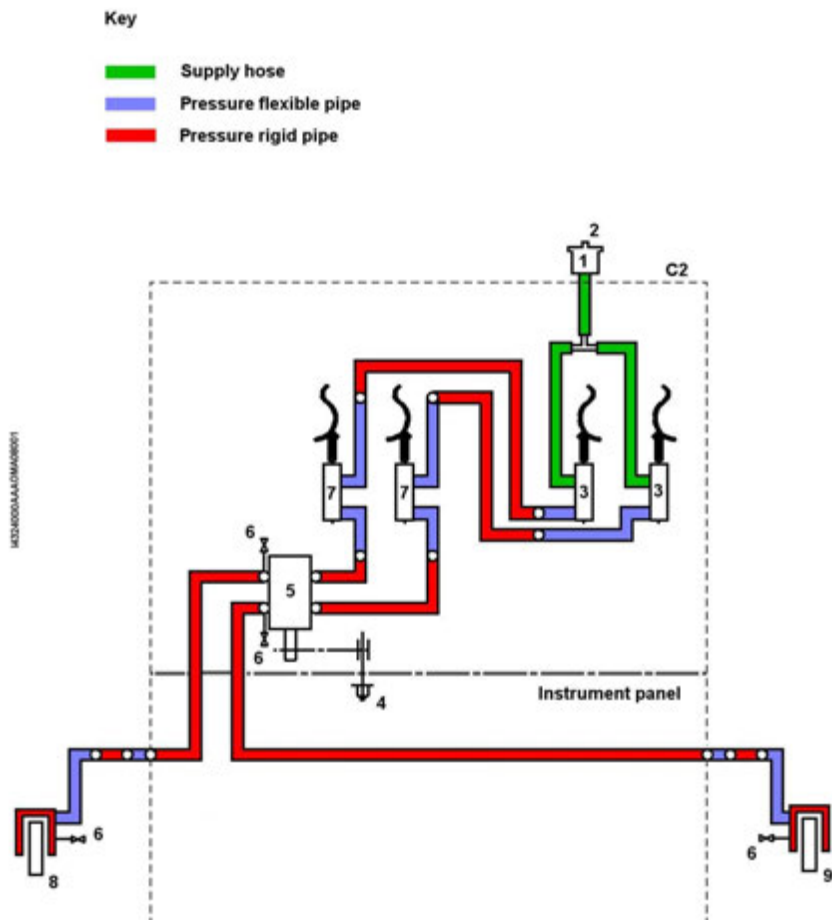
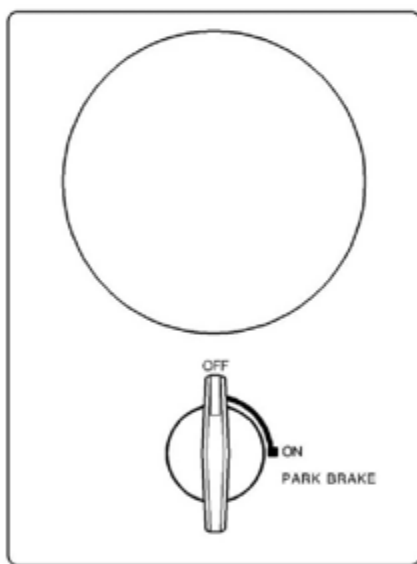
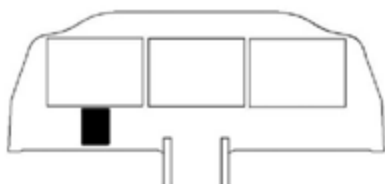


Figure 7.5.5 (2/2) - Brake system



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Figure 7.5.6 - Brake system

## **7.6 - Powerplant**

### **Turboprop engine operation - see figure 7.6.1**

The PRATT & WHITNEY CANADA turboprop engine (PT6A-66D model) is a free turbine engine rated at 850 SHP and developing a thermodynamic power of 1825 ESHP.

Intake air enters engine through an annular casing and is then ducted toward compressor. The latter consists of four axial stages and one single centrifugal stage assembly to form a whole assembly. Compressed air and fuel are mixed and sprayed into combustion chamber by fuel nozzles. The mixture is first ignited by two spark igniter plugs, then combustion continues as a result of air-fuel mixture flow. Gases resulting from combustion expand through a series of turbines. The first one (gas generator turbine) drives compressor assembly and accessories, the two other ones (power turbines), independant from the first one, drive propeller shaft through a reduction gear box. Hot gases are evacuated through two exhaust stubs located laterally on both sides forward of engine cowling.

All engine driven accessories, except power turbine tachometer, propeller governor and overspeed governor are installed on accessory gearbox located rearward of engine.

- 1) Propeller governor
- 2) Exhaust stub
- 3) Axial compressors
- 4) Accessory gearbox
- 5) FCU Fuel Control Unit
- 6) Oil to fuel heater
- 7) Input coupling shaft
- 8) Air intake
- 9) Centrifugal impeller
- 10) Combustion chamber
- 11) Compressor turbine
- 12) Power turbine 1st stage
- 13) Power turbine 2nd stage
- 14) Power turbine shaft

Figure 7.6.1 (1/2) - Powerplant



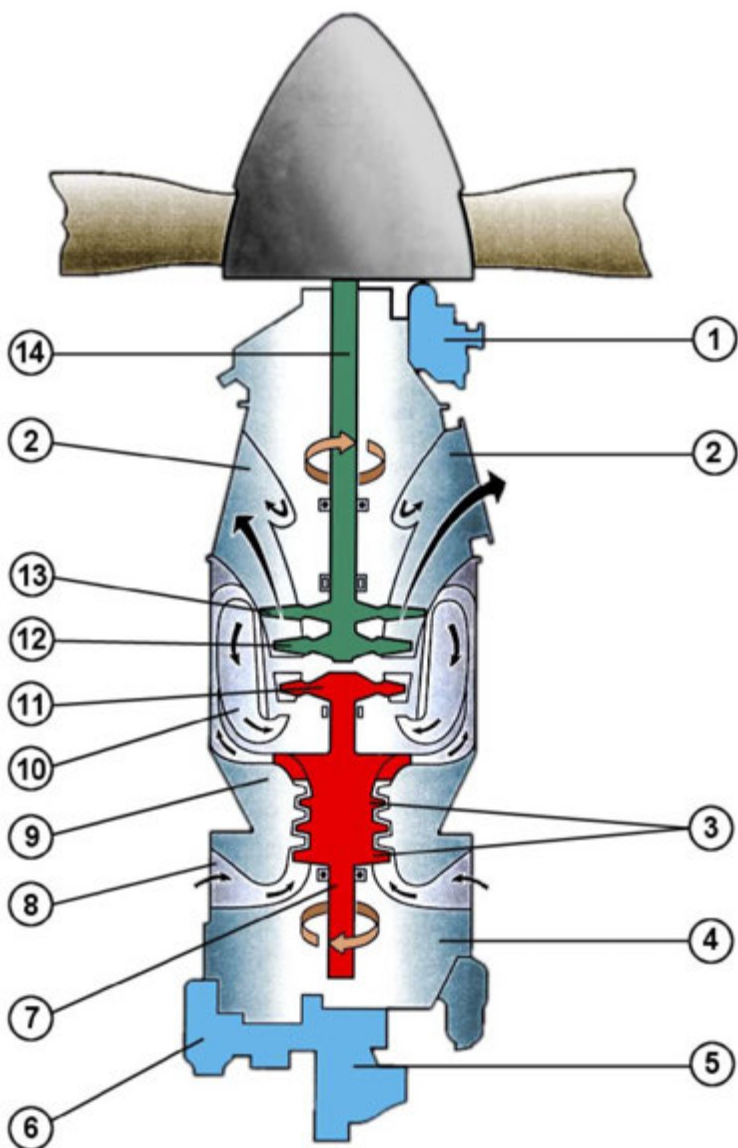


Figure 7.6.1 (2/2) - Powerplant

**Engine control levers** - see figure 7.6.2

Engine operation requires use of two levers located on pedestal console in cabin :

- THROTTLE (Item 1), and its detent for reverse (Item 4)
- MAN OVRD control for emergency fuel regulation (Item 3).

• NOTE •

Thumbwheel for lever friction (Item 2).

•

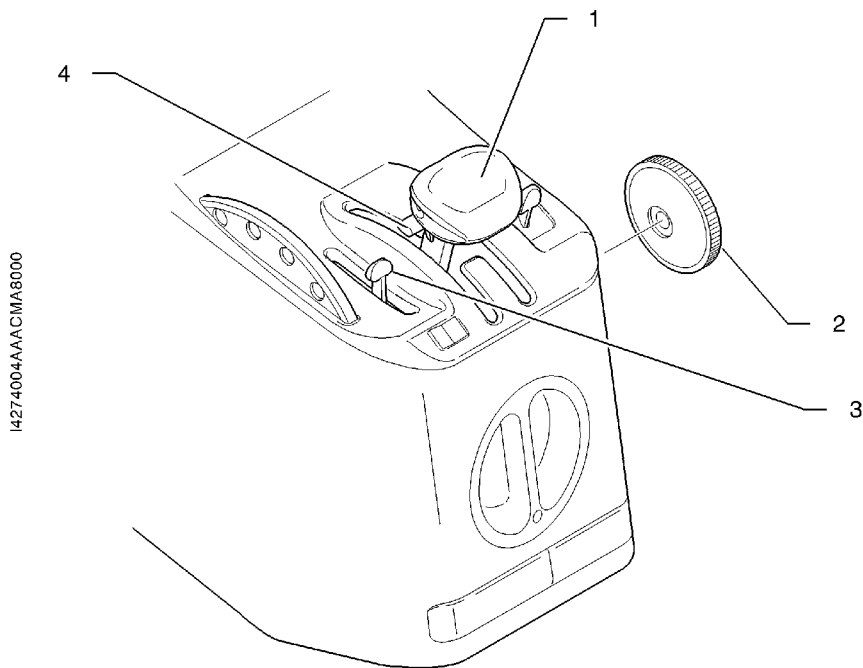


Figure 7.6.2 - Engine control levers

**THROTTLE** - see figure 7.6.3

The THROTTLE has two operating modes : thrust mode and condition mode.

- Thrust mode

The THROTTLE is in vertical position. It modulates engine power from full reverse to max power.

Engine running, the throttle rearward displacement, past the lock using the detent, allows to control :

- the engine power in the Beta range from idle to maximum reverse,
- the Beta valve to select the propeller pitch in reverse.

Return to idle position is accomplished by pushing the THROTTLE forward.

▲ **CAUTION** ▲

Do not move the cockpit THROTTLE into the propeller reverse position or damage to the linkage will result.

Reverse may only be selected with engine running and propeller turning.

Any rearward effort on the THROTTLE, past the idle stop, may damage or break the flexible control cable.



When engine is shutdown, there is no oil pressure in the propeller and the feathering spring locks the Beta ring and the propeller reversing interconnect linkage on the engine.

- Condition mode

The THROTTLE is moved to the condition side by lifting the knob.

As long as the THROTTLE is in condition mode, the propeller is in feather position. The THROTTLE can be positioned to CUT OFF, idle LO-IDLE or idle HI-IDLE.

Change from idle HI-IDLE to LO-IDLE position requires moving the THROTTLE rearwards.

Change from idle LO-IDLE to CUT OFF position is only possible after having overridden the idle gate. To override idle gate, raise the THROTTLE and move it rearwards.

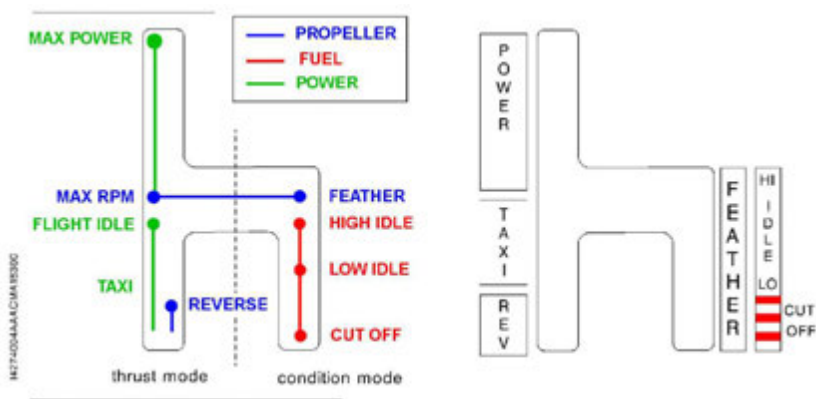


Figure 7.6.3 - THROTTLE

**MAN OVRD control** - see figure 7.6.2

MAN OVRD control (3) is normally notched in full backward position. In case of FCU or THROTTLE failure, it allows setting engine power manually.

To quit full backward position (notched), move the MAN OVRD control forward overriding the indexation.

• NOTE •

The power available if the THROTTLE fails will be limited by the position of the lever.

•

**Lever friction** - see figure 7.6.2

A thumbwheel (Item 2) located on right side of pedestal console increases friction to avoid control slip of the THROTTLE after setting.

## Engine instruments

Engine indicating consists of :

- engine torque expressed in percent (%), TRQ
- propeller speed in RPM, PROP RPM
- generator rotation speed expressed in percent (%), Ng
- ITT expressed in °C,
- oil pressure expressed in PSI.
- oil temperature expressed in °C.

• NOTE •

Engine monitoring is ensured by **ITT** and **OIL PRESS**.

Refer to the GARMIN pilot's Guide for further details.

•

## Engine lubrication

Engine oil is in a tank incorporated into the powerplant. It ensures lubrication and engine cooling. A cooler located on left side in engine compartment maintains oil temperature within limits. Oil flow into the cooler is metered by a thermostatic valve. Engine oil also supplies propeller governor and engine torquemeter.

A chip detection system enables the monitoring of engine oil system. The system includes one chip detector installed on propeller reduction gear box and a second chip detector installed on engine accessory gear box. In case of chip detection, **CHIP** will appear on integrated flight deck system screen.

Lubrication system content, cooler included, is 12.7 quarts (12 litres). A graduated dipstick allows checking oil quantity in system. A visual oil sight glass, located on engine left side, allows a rapid checking of oil level.

• NOTE •

For checking and oil filling-up, refer to section 8.

•

## **Engine starting** - see figure 7.6.4

### **Ignition function**

Ignition system consists of an ignition unit and two spark igniter plugs in powerplant, a three-position IGNITION switch OFF - AUTO - ON located on ENGINE START panel at upper panel.

Ignition unit supplies, from 28-volt source, high voltage current necessary to spark igniter plugs. When IGNITION switch is positioned to AUTO, ignition unit supply is ensured during the engine start.

**IGNITION** lights on as long as ignition unit is supplied.

### **Starter function**

Starting system consists of STARTER switch located on ENGINE START panel, starter generator and ignition circuit (Refer to paragraph Ignition function).

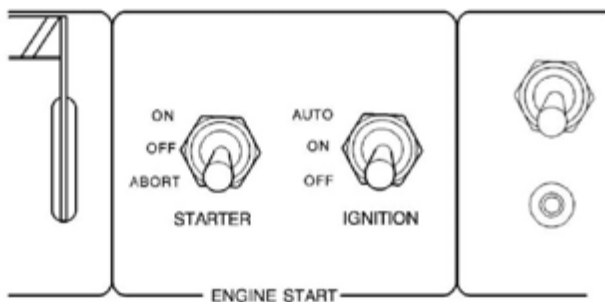
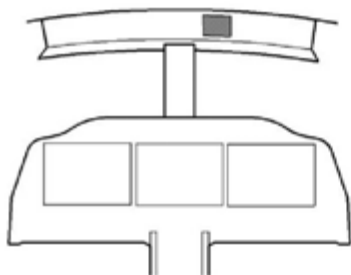
Starting procedure is semi-automatic. Setting STARTER switch to ON connects the starter generator which drives powerplant. **STARTER** lights on indicating that the starter generator is operating.

Starter operation is stopped automatically by the electrical power system once a sufficient starter-generator speed is reached or after 60 s. The pilot has the capability to interrupt the start process anytime by setting momentarily the STARTER switch to the ABORT position.

### **▲ WARNING ▲**

**Powerplant starting must be performed by qualified personnel and following procedures and parameters described in section 4 Normal procedures.**





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Figure 7.6.4 - Engine starting

## Engine air inlet

Engine air inlet is located at front lower section of engine cowling. Air inlet port is protected against icing by a hot air flux provided by engine. Air is driven throughout a duct in engine casing before entering engine through a protective screen. An inertial separator system inside the air duct protects the engine from ingesting dense particles (water, ice, fine gravels, sand).

Separator consists of two movable vanes. During normal operation, air is conducted directly towards engine air inlet. To separate particles suspended in the air, vanes are positioned to force engine induction air to execute a sharp turn : under the effect of centrifugal force denser particles separate from the air and are discharged overboard through two apertures located under engine cowling.

Operation of inertial separator vanes is electrically controlled by INERT SEP switch located on DE-ICE SYSTEM panel. When INERT SEP switch is set to ON, an electric actuator activates vanes ; **INERT SEP ON** lights on when vanes have reached their maximum deflection and remains visible as long as switch remains ON. Full deflection takes about 40 seconds. If the vanes do not reach the full deflection 50 seconds after activation or are not retracted 50 seconds after deactivation,

**INERT SEP FAIL** is displayed in CAS window.

## Exhaust system

Exhaust gases are evacuated through exhaust stubs located on sides of engine cowlings.

## Engine accessories

All engine driven accessories, except power turbine tacho-generator (Np), propeller governor and overspeed governor, are installed on accessory gearbox located rearwards of engine.

### Oil pump

Oil pump is a self-controlled gear pump located at the bottom of oil casing.

### Fuel high pressure pump (HP)

Fuel high pressure pump is installed on accessory gearbox. It supplies fuel nozzles, flow being controlled by fuel regulator (FCU). Fuel provided by engine driven main pump (mechanical) enters high pressure pump through a filter, then it is discharged under pressure into fuel regulator (FCU) through a second filter. In case of contamination of this second filter, a by-pass valve allows fuel to go directly from high pressure pump to the regulator.



**Compressor turbine tachogenerator (Ng)**

Compressor turbine tachogenerator (Ng) is attached on accessory gearbox. It supplies a voltage which is transmitted to the GARMIN system for display on the MFD, under normal display conditions.

**Power turbine tachogenerator (Np)**

Power turbine tachogenerator is attached on the right side of the reduction gearbox. It supplies a voltage which is transmitted to the GARMIN system for display on the MFD, under normal display conditions.

**Torque transmitter**

Torque transmitter is attached on the torque limiter, it measures torque produced by the power turbine by comparing oil pressures (reduction gear and power turbine) and converts pressure difference into a voltage. This voltage is transmitted to the GARMIN system for display on the MFD, under normal display conditions.

**Propeller overspeed limiter**

Propeller overspeed limiter is installed on left side of the reduction gear box. It prevents a propeller overspeed in case of main propeller governor failure.

Propeller overspeed limiter is equipped with a solenoid which makes feather the propeller when the THROTTLE is in condition mode.

**Torque limiter**

Torque limiter is located on right side of the reduction gear box. It is rated to limit engine torque to 109-110 % at sea level.

## Propeller

Airplane is equipped with a composite five-bladed, constant-speed and full-feathering propeller.

### Regulation

Propeller governor located on engine maintains rotation speed to the nominal value of 2000 RPM. Regulation is obtained through propeller blade pitch variation : counterweights drive propeller blades toward high pitch (low RPM) whereas oil pressure delivered by governor drives back blades toward low pitch (high RPM).

Propeller governor allows feathering either by voluntary pilot action via THROTTLE (Condition mode) or automatically in case of engine failure or shutdown.

Propeller reverse pitch allows reduced taxiing speed or landing roll. Change from idle to reverse position is performed with THROTTLE (Thrust mode) - refer to paragraph Engine controls.

**7.7 - Fuel system** - see figure 7.7.1

The fuel system comprises fuel tanks, fuel unit, selectors, manual and automatic, electric and mechanical boost pumps, engine fuel system, gaging installation, monitoring installation and drains.

**Fuel tanks**

Fuel tanks are formed by sealed casings in each wing. Each fuel tank comprises a filling port located at the end of wing upper surface, two drain valves located at the lower surface (one near main landing gear, at trailing edge side, the second one near wing root side, at leading edge), a vent valve located on the lower surface, a suction strainer and three level gages.

**Fuel unit**

The fuel unit combines shut-off valve, tank selector and filter functions. It is connected to the manual selector through a mechanical control. The fuel filter is located in a bowl at the lower part of the unit. It is fitted with a by-pass valve, a clogging indicator and a drain valve.

**Tank manual selector** - see figure 7.7.2

The FUEL TANK SELECTOR is located on the pedestal rear face. It allows selecting manually the tank (R or L) to be used and setting unit to OFF. To change from L position to OFF position, turn the selector clockwise (L → R → OFF) ; change from R position to OFF position requires a voluntary action from the pilot (pull and turn). The pull and turn maneuver prevents involuntary operation. When the unit is set to OFF,

**FUEL OFF** remains visible.

- |                                   |                        |
|-----------------------------------|------------------------|
| 1) Flow divider                   | 14) Fuel unit          |
| 2) Flowmeter                      | 15) Filter drain       |
| 3) Collector tank                 | 16) Fuel return pipe   |
| 4) Fuel regulator                 | 17) Filling port       |
| 5) High pressure pump (HP)        | 18) NACA scoop         |
| 6) Oil to fuel heater             | 19) Tank vent valve    |
| 7) Low pressure switch            | 20) Fuel level gages   |
| 8) Fuel jet                       | 21) Tank drain valve   |
| 9) Main mechanical boost pump     | 22) Check-valve        |
| 10) Electric boost pump           | 23) Low level detector |
| 11) Fuel filter                   | 24) Suction strainer   |
| 12) Filter clogging by-pass valve | 25) Fuel amplifier     |
| 13) Filter clogging indicator     | 26) Sequencer          |

Figure 7.7.1 (1/2) - Fuel system



Figure 7.7.1 (2/2) - Fuel system

## **Automatic tank selector** - see figures 7.7.2 and 7.7.3

Automatic tank selection allows, without pilot's intervention, feeding the engine from one tank or the other in predetermined sequences. These sequences depend on airplane configuration (ground, in-flight, fuel low level CAS messages appearance).

Automatic tank selection system comprises an electronic sequencer, an actuator attached on the fuel unit, FUEL SEL two-position selector (AUTO, MAN) and SHIFT push-knob located on FUEL panel.

To operate the automatic selector, set FUEL SEL switch to AUTO position and manual selector to R or L.

### **Selector operation**

When the system is operated, **AUTO SEL** disappears ; the sequencer chooses a tank (R or L) and through the actuator, positions the fuel unit selector on the selected tank. The sequencer controls the time during which the selected tank will operate. This time varies, depending on airplane conditions.

Airplane on ground : tank is changed every minute and 15 seconds.

Airplane in flight : tank is changed every five minutes, as long as **FUEL LOW L** or **FUEL LOW R** does not appear. When the first low level lights on, the sequencer immediately selects the other tank. The selected tank will operate until the second low level lights on. When **FUEL LOW L-R** is visible, the sequencer changes tanks every minute and 15 seconds.

#### • NOTE •

The manual selector is driven by the fuel unit and is positioned on R or L mark corresponding to the tank selected by the sequencer. Therefore, the pilot continuously knows the tank which is operating.

•

### **Test for system proper operation**

SHIFT push-button allows the pilot to test system proper operation anytime.

When the system operates, the fuel tank is changed when SHIFT push-button is pressed once.

If airplane is on ground or in flight, low level CAS messages not visible, the new selected tank remains operating and a new sequence is initiated.

• NOTE •

This procedure allows the pilot to preferably choose the tank from which he wants to take fuel.

•

In all cases, proper system operation is indicated by rotation of the manual selector.

Setting FUEL SEL switch to MAN position or setting FUEL TANK SELECTOR to OFF position leads to system de-activating and appearance of **AUTO SEL**.

**AUTO SEL** also lights on when order given by the sequencer has not been executed after 12 seconds.

### Electric boost pump (AUX BP)

Electric boost pump is an auxiliary pump located between fuel unit and main mechanical boost pump. It is controlled through AUX BP switch located on FUEL panel. This switch allows stopping or selecting the two pump operating modes :

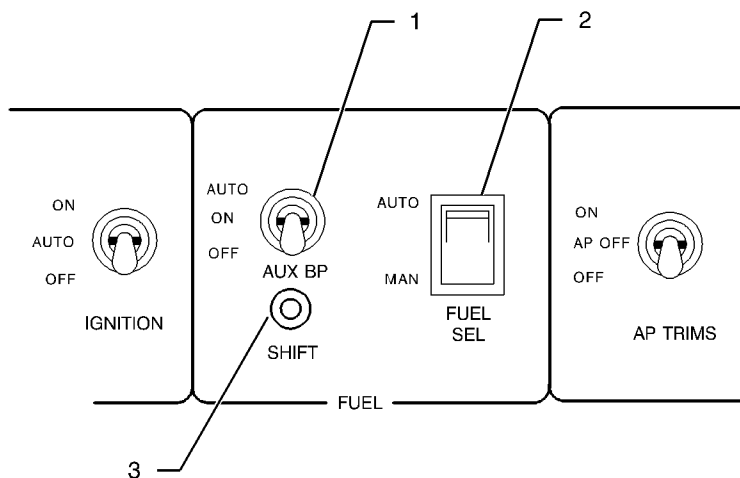
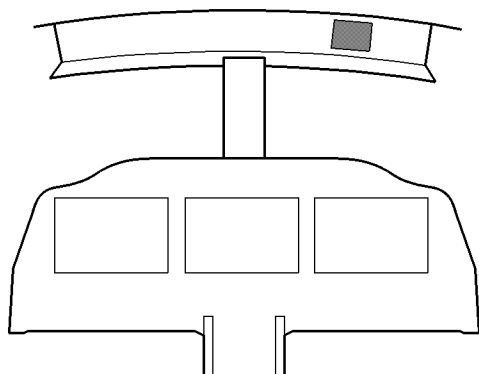
- when set to ON, electric boost pump operates permanently
- when set to AUTO, electric boost pump is automatically operated in case of fuel pressure drop at the mechanical boost pump outlet.



Figure 7.7.2 - Manual selector of fuel tanks



- 1) AUX BP switch
- 2) FUEL SEL switch
- 3) SHIFT push-button



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Figure 7.7.3 - Fuel control panel

## Main mechanical boost pump

The mechanical boost pump is attached to accessory gearbox and supplies fuel necessary for engine operation.

## Engine fuel system

The engine fuel system consists of a fuel regulator, pumps, filters, a fuel divider and fuel nozzles. The system provides the fuel flow necessary to satisfy the engine power and rating needs.

The fuel coming from airplane system goes through a heater which is automatically controlled by a thermostatic valve.

## Fuel gaging installation

Fuel gaging installation is a capacitive type. Fuel data are displayed in us gallons. Three fuel level gages are installed in each tank. The wing root side fuel level gage is equipped with a low level detector which leads to fuel low level CAS messages appearance, when usable fuel quantity remaining in the concerned fuel tank is under about 9 USG (34 Litres).

## Fuel system monitoring

Fuel system monitoring is ensured by CAS messages :

- **FUEL OFF** : Fuel tank selector set to OFF
- **FUEL PRESS** : Fuel pressure at mechanic pump outlet under 10 psi ( $\pm 2$  psi)
- **AUX BOOST PMP ON** : Electric fuel pump running (manual or automatic mode)
- **FUEL LOW L-R** \* : Fuel quantity less than or equal to 9 USG (34 Litres) of usable fuel in specified tank
- **AUTO SEL** : Sequencer inactive or operating defect
- **FUEL IMBALANCE** : Fuel tanks imbalanced by more than 15 USG (57 Litres) for more than 30 seconds

\* Only affected side (L, R or L-R) displayed in CAS message

**Fuel system draining and clogging indicator** - see figure 7.7.4

The fuel system comprises five drain points, a drain on the filter bowl, two drain valves on each tank, located on wing lower surface, one at wing root and the other past main landing gear well.

These drains allow draining water or sediments contained in fuel.

Fuel tank drain valves are provided with a slot which allows opening them with a screwdriver.

**▲ CAUTION ▲**

Fuel system draining shall be performed prior to the first flight of the day and after each tank refueling, using a sampler to pick off fuel at the two drain valves of each tank and at the filter vent valve.

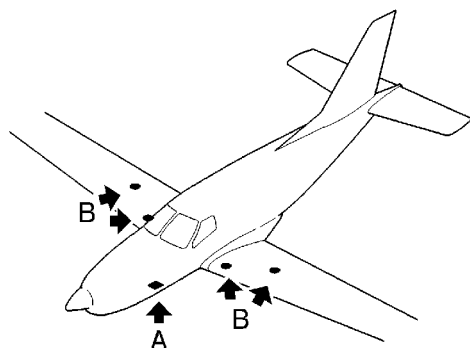


A red filter bypass flag on the fuel unit and visible from outside, when an inspection door located on L.H. side under front baggage compartment is open, indicates filter clogging. A push-button, adjacent to the inspection door, controls the illumination of a light provided to improve visibility of the clogging indicator. This indicator shall be observed during preflight inspection.

**● NOTE ●**

When filter gets clogged in flight, the filter is by-passed in order not to deprive powerplant from fuel. The powerplant is then supplied with non-filtered fuel.





- 1) Lighting switch
- 2) Mirror door
- 3) Clogging indicator
- 4) Central access door
- 5) Filter drain
- 6) Tank drain
- 7) Drain bowl

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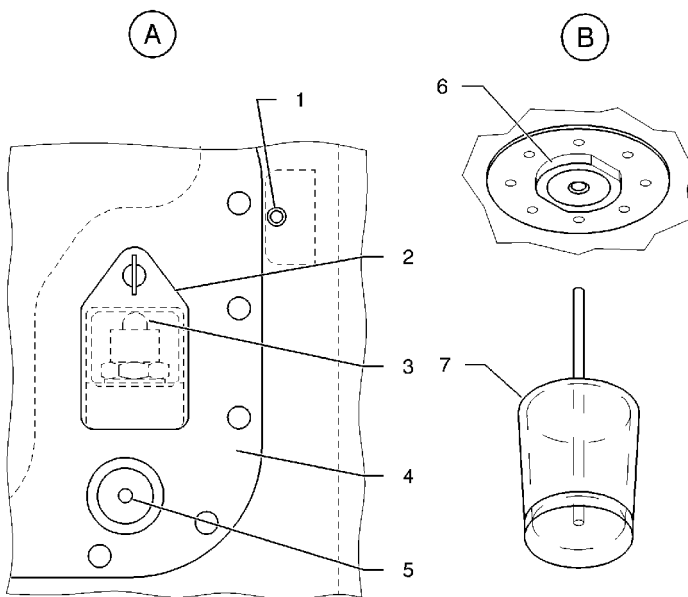


Figure 7.7.4 - Fuel system draining points and clogging indicator

## 7.8 - Electrical system - see figures 7.8.1, 7.8.2 and 7.8.5

The airplane is fitted with a 28-volt direct-current electrical system.

Electrical supply is obtained from various power supplies :

- a starter generator
- a stand-by generator
- a battery
- a ground power unit, via a plug, located on L.H. side.

Connection relays, main bus bar, generator regulation and protection systems and control logic systems are grouped in electrical power system box located in front baggage compartment upper section.

Electrical system indicating is displayed on the MFD and monitoring is ensured by CAS messages.

On ground, when the crash lever is positioned in the UP position (SOURCE selector in the OFF position), the battery supplies the electrical power system through the BATT BUS. A Power Up Built In Test (P-BIT) of the EPS internal functions is performed to verify the operating status. In case of failure detection, a white message EPS SERVICE REQUIRED appears in the message window on the PFD.

### Starter generator

The starter generator is the main electrical power source. It only performs its generator function when starting sequence is completed.

Generator connection with main bus bar is controlled through GENERATOR selector set to MAIN position. It will be effective when connection conditions are met.

Generator connection is indicated by **MAIN GEN** disappearance.

#### • NOTE •

Starter generator will not supply airplane if source switch is on GPU.

On ground, generator load should be maintained below 200 AMP.

•

## Stand-by generator

Stand-by generator supplies a 28-volt stand-by direct current which may be used in case of main generator failure.

Generator connection with main bus bar is controlled through GENERATOR selector set to ST-BY, it will be effective when connection conditions are met.

• NOTE •

Stand-by generator will not supply airplane if source switch is on GPU.

In order to prevent possible errors during flight, access to ST-BY position requires a double action from the pilot (pull to unlock). On ground, avoid using stand-by generator at full load.

•

## Battery

The battery provides the power required for starting when no ground power unit is available and is a power supply source when engine driven generators are stopped.

The battery is always connected to BATT BUS bus bar except when crash lever is pulled down.

Battery connection to main bus bar is controlled through SOURCE selector set to BATT position.

**BAT OFF** lights on when battery is isolated from the main bus and when main bus is supplied through another source.

## Ground power receptacle

The ground power receptacle allows connection to a ground power unit.

Ground power receptacle connection with main bus bar is controlled through SOURCE selector when set to GPU position, it will be effective when connection conditions are met.

When SOURCE selector is set to GPU position, the battery and ground power unit are connected simultaneously on main bus bar.

Ground power receptacle door opening is indicated by **GPU DOOR** appearance.

• NOTE •

Before connecting a GPU to the airplane, ensure that the voltage of the GPU is regulated between 27.5 volts and 28.5 volts.

The amperage output needs to be consistent with the airplane placard in front of compartment door : GPU shall provide a current limiting function, and current limit shall be set per placard.

Do not use batteries pack as GPU sources.



▲ CAUTION ▲

Use of a ground power source with voltage in excess of 28.5 volts or current exceeding current limit indicated on placard may damage the airplane electrical system.



## Distribution

Airplane electrical systems are connected to bus bars and protected by pull-off type breakers located on R.H. side panel - see figure 7.8.4. In case of overload of a system, the breaker triggers and switches the system off.

▲ CAUTION ▲

If a breaker corresponding to a non essential system trips, do not reset in flight.



If a breaker corresponding to an essential system trips:

- allow it to cool for about three minutes, then the breaker may be reengaged (pressed down)
- if the breaker trips again, do not reset.

BUS 1, BUS 2, BUS 3 and BUS 4 bus bars are directly connected to main bus bar and protected by fuses located in electrical power system.

The ESS BUS 1 and ESS BUS 2 essential bus bars are connected to main bus bar through ESS BUS TIE switch set to NORM position. ESS BUS TIE switch is attached to breaker panel ; NORM position is protected and locked by a cover. Common power supply to both essential bus bars is protected by a fuse, located in EPS box, and a breaker, located in the front cargo compartment on C2 frame right side, each bar being individually protected by a breaker.

BATT BUS bar is directly connected to the battery ; it is protected by a fuse, located in EPS box, and a breaker, located in the front cargo compartment on C2 frame left side.

• NOTE •

The electrical distribution of bus bars is described in figure 7.8.3.

•

## Emergency use

With both generators de-activated in flight, it is still possible to use battery power to supply all airplane systems maintaining SOURCE selector on BATT position.

In order to save battery power, it is possible to shed the charges which are not essential for flight safety, for that set :

- ESS BUS TIE switch to EMER position

In this configuration, only ESS BUS 1, ESS BUS 2 and BATT BUS bars are supplied.

• NOTE •

Supplying BUS 1, BUS 2, BUS 3 and BUS 4 bars is always possible, resetting temporarily ESS BUS TIE switch to NORM position.

•



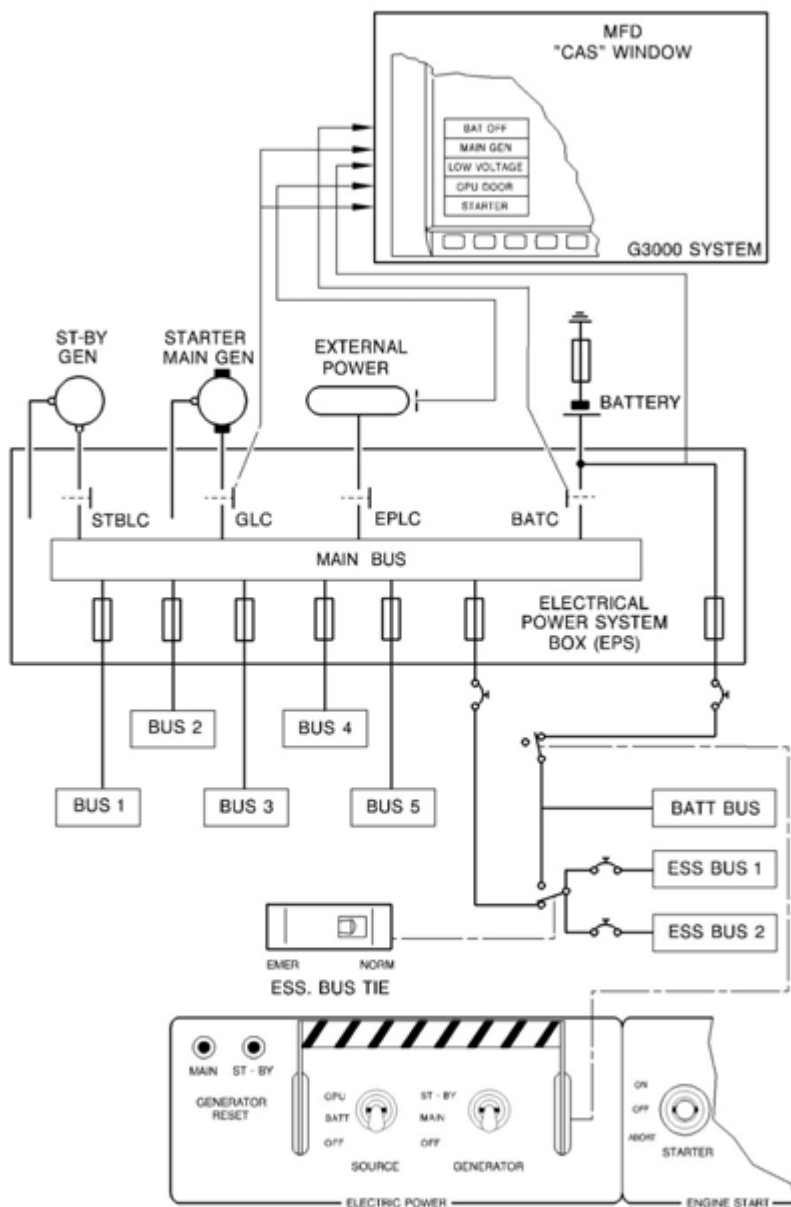
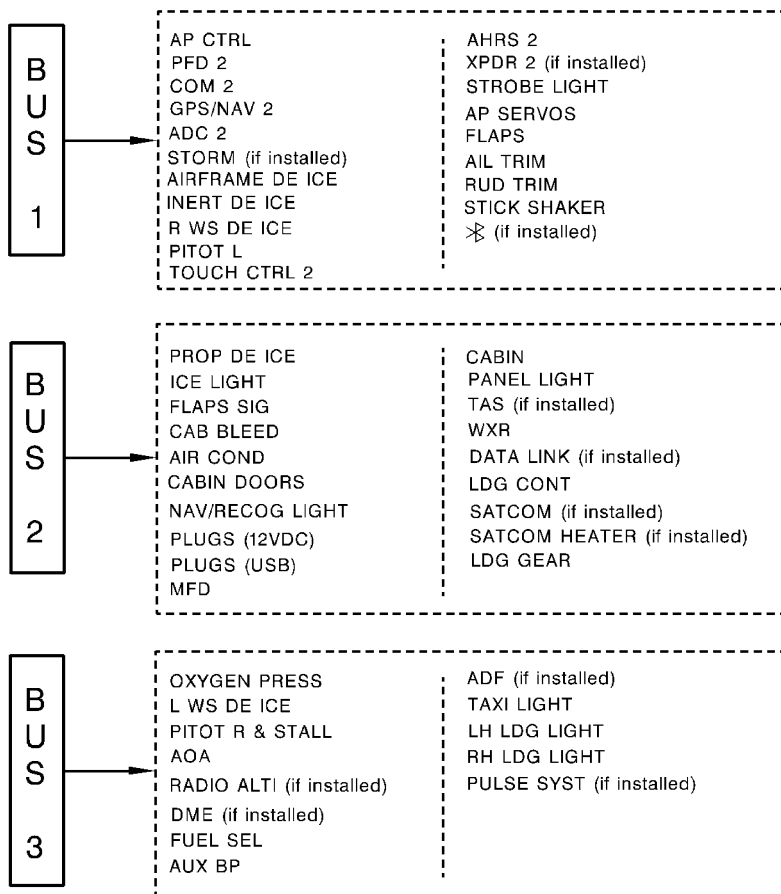


Figure 7.8.1 - Electrical diagram

| Switches    |        |           |             | Buses are powered by |                 |                 |                 |  |
|-------------|--------|-----------|-------------|----------------------|-----------------|-----------------|-----------------|--|
| Crash lever | Source | Generator | ESS BUS TIE | BATT BUS             | ESS BUS 1       | ESS BUS 2       | BUS 1 TO 5      |  |
| UP          | BATT   | OFF       | NORM        | Battery              | Battery         | Battery         | Battery         |  |
| UP          | BATT   | MAIN      | NORM        | Battery & MAIN       | Battery & MAIN  | Battery & MAIN  | Battery & MAIN  |  |
| UP          | BATT   | ST-BY     | NORM        | Battery & ST-BY      | Battery & ST-BY | Battery & ST-BY | Battery & ST-BY |  |
| UP          | OFF    | MAIN      | NORM        | MAIN                 | MAIN            | MAIN            | MAIN            |  |
| UP          | OFF    | ST-BY     | NORM        | ST-BY                | ST-BY           | ST-BY           | ST-BY           |  |
| UP          | BATT   | OFF       | EMER        | Battery              | Battery         | Battery         | None            |  |

(\*) In that case, power is done by MAIN or ST-BY and battery is used as a floated battery.

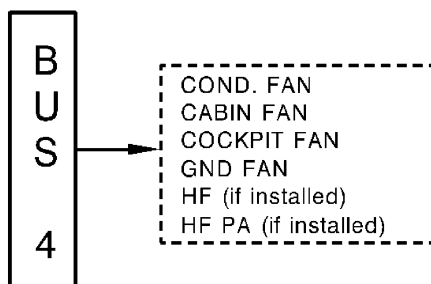
Figure 7.8.2 - Bus bars supply configurations



14246000AAANMA8400

Figure 7.8.3 - (1/3) - Electrical distribution of bus bars

I4246000AAN/A8101



NOTE: BREAKERS ON C13 BIS FRAME

I4246000AAG/MA8300

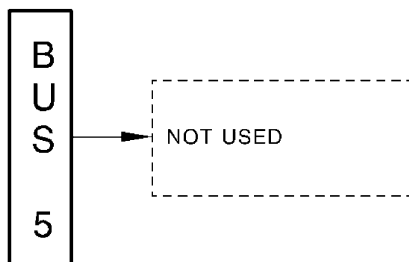
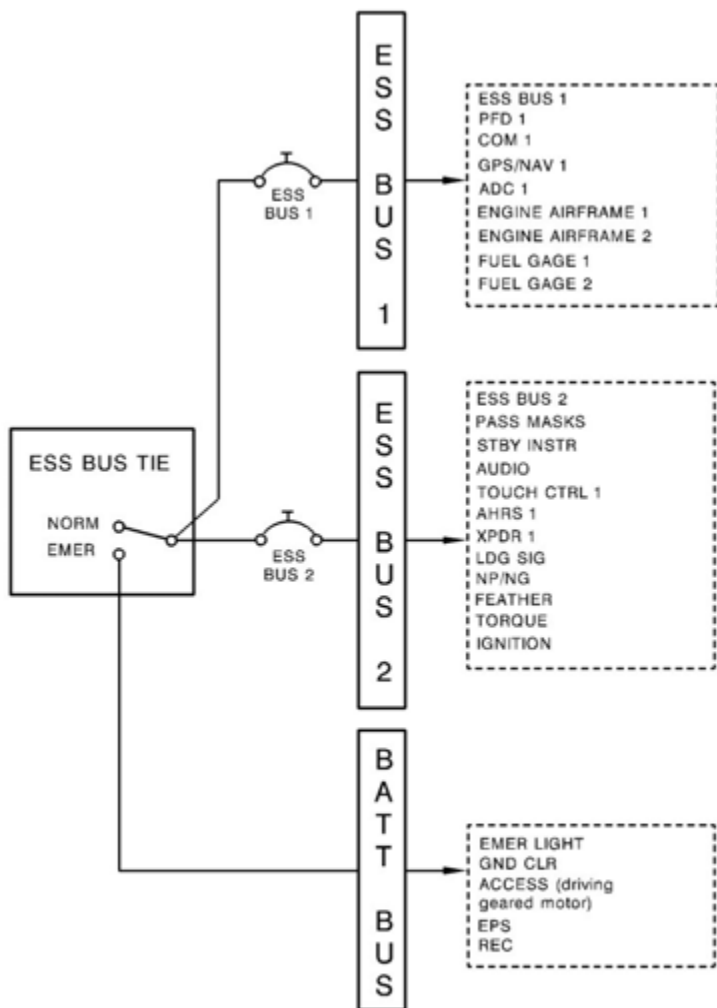


Figure 7.8.3 (2/3) - Electrical distribution of bus bars



14246000AAJGMA18200

Figure 7.8.3 (3/3) - Electrical distribution of bus bars

|                       |   |
|-----------------------|---|
| <b>ESS BUS TIE</b>    | Essential bus NORM & EMER switch                                |
| <b>BUS 1</b>          |   |
| AP SERVOS             | Autopilot servo protection                                      |
| FLAPS                 | Flaps protection  |
| AIL TRIM              | Aileron trim protection   |
| RUD TRIM              | Pitch trim protection   |
| <b>BUS 2</b>          |   |
| LDG GEAR              | Landing gear general supply protection                          |
| <b>ESS BUS 1</b>      |   |
| ESS BUS 1             | Essential bus 1 circuit protection                              |
| PFD 1                 | Primary Flight Display 1 protection                             |
| COM 1                 | VHF 1 protection  |
| GPS/NAV 1             | GPS NAV 1 protection  |
| ADC 1                 | Air Data Computer 1 protection                                  |
| ENGINE                | Powerplant cont. protec. : Oil temp. & pres., torque, propeller |
| AIRFRAME 1            |   |
| ENGINE                | Powerplant cont. protection : Ng, flowmeter & ITT               |
| AIRFRAME 2            |   |
| FUEL GAGE 1           | L.H. fuel gage protection                                       |
| FUEL GAGE 2           | R.H fuel gage protection  |
| <b>ESS BUS 2</b>      |   |
| ESS BUS 2             | Essential bus 2 circuit protection                              |
| PASS MASKS            | Passengers' oxygen masks protection                             |
| STBY INSTR            | Standby Attitude Module (MD302) protection                      |
| AUDIO                 | Audio control panel protection                                  |
| TOUCH CTRL 1          | Touchscreen controller 1 protection                             |
| AHRS 1                | Attitude and Heading Reference System 1 protection              |
| XPDR 1                | Transponder 1 protection  |
| LDG SIG               | Landing gear indicating system protection                       |
| NP/NG                 | Tachometer signal conditioner protection                        |
| FEATHER               | Propeller feather protection                                    |
| TORQUE                | Torque control protection                                       |
| IGNITION              | Powerplant ignition protection                                  |
| (Cont'd on next page) |   |

Figure 7.8.4 (1/5) - Breaker panel (Typical arrangement)

|                       |  |
|-----------------------|--|
| <b>BUS 1</b>          |  |
| AP CTRL               | Flight controller protection                                 |
| PFD 2                 | Primary Flight Display 2 protection                          |
| COM 2                 | VHF 2 & radio protection                                     |
| GPS/NAV 2             | GPS NAV 2 protection   |
| ADC 2                 | Air Data Computer 2 protection                               |
| STORM                 | Stormscope protection, if installed                          |
| AIRFRAME DE ICE       | Empennage and wing leading edges deicing                     |
| INERT DE ICE          | Inertial separator protection                                |
| R WS DE ICE           | R.H. windshield deicing protection                           |
| PITOT L               | Pitot L heating protection                                   |
| TOUCH CTRL 2          | Touchscreen controller 2 protection                          |
| AHRS 2                | Attitude and Heading Reference System 2 protection           |
| XPDR 2                | Transponder 2, if installed protection                       |
| STROBE LIGHT          | Strobe lights protection                                     |
| SHAKER                | Stick shaker protection                                      |
| BLUETOOTH             | Flight Stream (FS210) protection, if installed               |
| <b>BUS 2</b>          |  |
| PROP DE ICE           | Propeller deicing protection                                 |
| ICE LIGHT             | L.H. wing leading edge lighting and lighting test protection |
| FLAPS SIG             | Trim and flaps regulator protection                          |
| CAB BLEED             | Cabin pressurization protection                              |
| AIR COND              | Cabin ventilation and vapor cycle system protection          |
| CABIN DOORS           | Cabin doors opening protection                               |
| NAV/RECOG LIGHT       | Navigation and recognition lights protection                 |
| PLUGS                 | 12 VDC plugs protection                                      |
| PLUGS                 | USB plugs protection   |
| MFD                   | Multifunction display protection                             |
| CABIN                 | Passenger's reading lamps protection                         |
| PANEL LIGHT           | Instruments lighting protection                              |
| TAS                   | TAS, if installed protection                                 |
| WXR                   | Weather radar protection                                     |
| DATA LINK             | Data Link, if installed protection                           |
| LDG CONT              | Landing gear control protection                              |
| SATCOM                | SATCOM protection, if installed                              |
| SATCOM HEATER         | SATCOM heater protection, if installed                       |
| (Cont'd on next page) |  |

Figure 7.8.4 (2/5) - Breaker panel (Typical arrangement)

|                 |  |
|-----------------|--|
| <b>BUS 3</b>    |  |
| OXYGEN PRESS    | Oxygen/Pressure indication protection          |
| L WS DE ICE     | L.H. windshield deicing protection             |
| PITOT R & STALL | Pitot R and stall warning heating protection   |
| AOA             | Angle of Attack protection                     |
| RADIO ALTI      | RADIO ALTI, if installed protection            |
| DME             | DME protection, if installed                   |
| FUEL SEL        | Tank selector timer protection                 |
| AUX BP          | Electrical fuel pump protection                |
| ADF             | ADF protection, if installed                   |
| TAXI LIGHT      | Taxi light protection                          |
| LH LDG LIGHT    | L.H. landing light protection                  |
| RH LDG LIGHT    | R.H. landing light protection                  |
| PULSE SYST      | Pulse lite system protection, if installed     |
| <b>BATT BUS</b> |  |
| EMER LIGHT      | Instrument panel emergency lighting protection |
| GND CLR         | Ground clearance protection                    |
| ACCESS          | Cabin access lighting protection               |
| EPS             | Electrical power system protection             |
| REC             | Lightweight Data Recorder protection           |

Figure 7.8.4 (3/5) - Breaker panel (Typical arrangement)



&gt;&gt; Up to S/N 1159

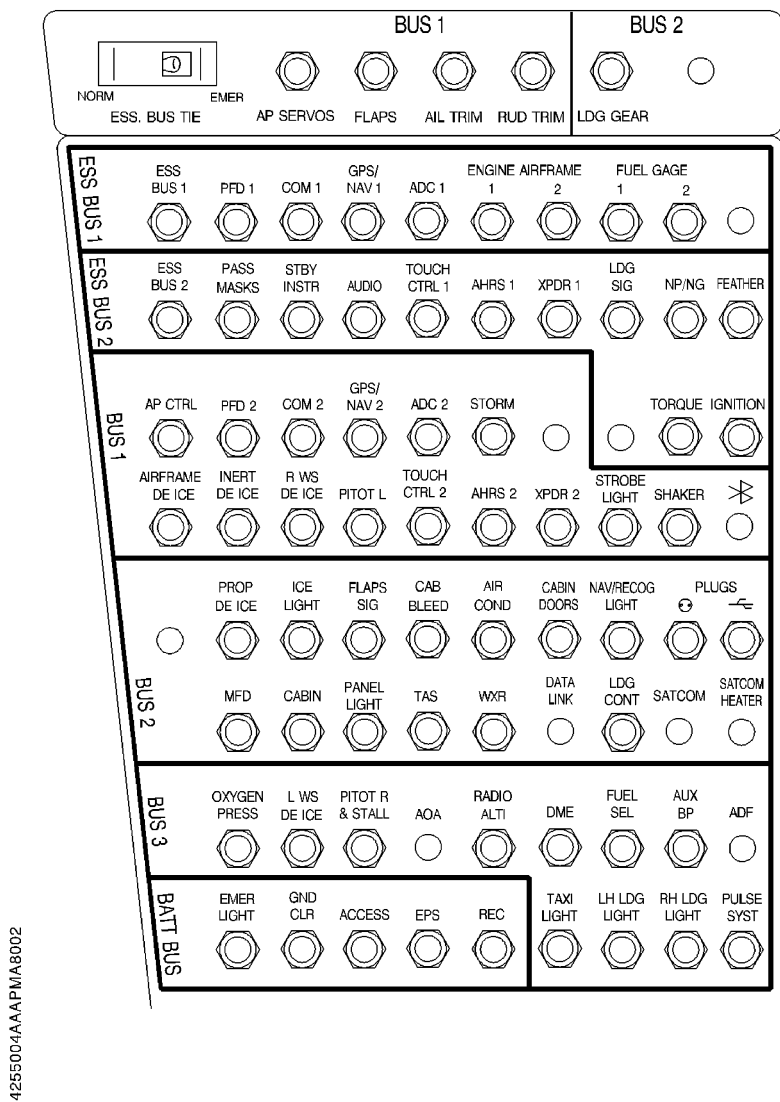


Figure 7.8.4 (4/5) - Breaker panel (Typical arrangement)

>> From S/N 1160

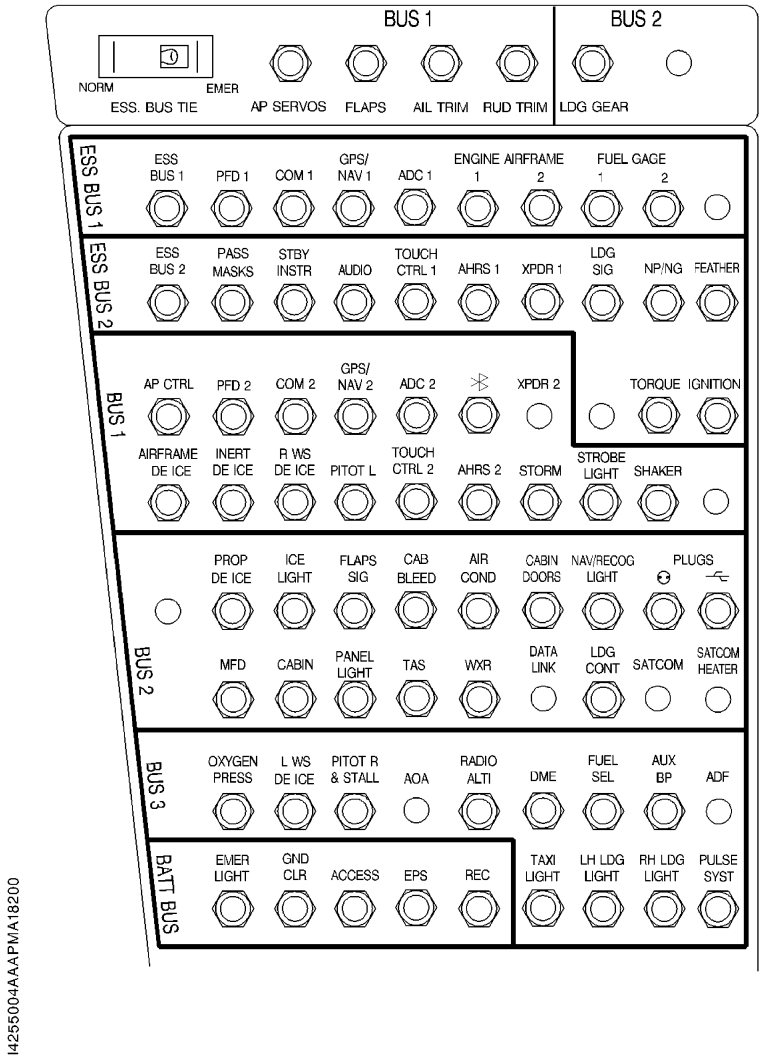


Figure 7.8.4 (5/5) - Breaker panel (Typical arrangement)

## Indicating

Electrical system indicating consists of voltage and ampere indicating - refer to GARMIN pilot's guide for further details.

Following CAS messages may appear on the MFD CAS window :

**BAT OFF**

: Battery is not connected to main bus bar

**MAIN GEN**

: Starter generator is not connected to main bus bar

**LOW VOLTAGE**

: Battery voltage is below the minimum value

**GPU DOOR**

: Ground power receptacle access door is not closed

## Protection - safety - see figures 7.8.2 and 7.8.5

The electrical power system provides systems protection in case of :

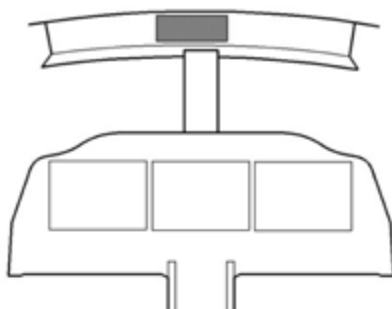
- overvoltage
- short-circuits

In case of disconnection of starter generator or stand-by generator following a failure, MAIN or ST-BY reset can be done by pressing corresponding GENERATOR RESET MAIN or ST-BY push-button.

A battery reset is done by setting the SOURCE selector to OFF and back to BATT.

In case of disconnection of ground power unit following a failure, it is possible to re-activate the system by turning the SOURCE selector to OFF and setting it again to GPU position to reset the protection.

A crash lever located on upper panel center part allows isolating simultaneously BATT BUS bar and setting to OFF the SOURCE and GENERATOR selectors when lowered. In this case all bus bars are isolated from generators.



- 1) MAIN reset knob
- 2) ST-BY reset knob
- 3) Crash lever
- 4) SOURCE selector
- 5) GENERATOR selector

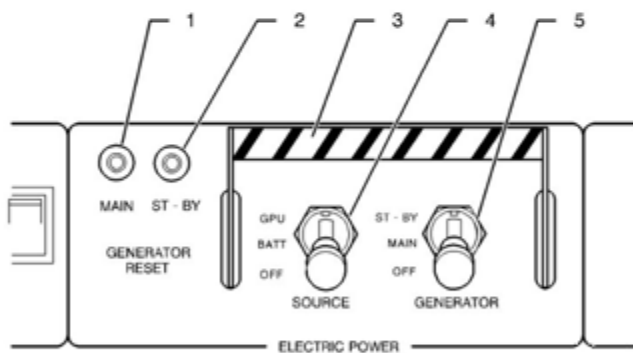


Figure 7.8.5 - Electrical control

**Exterior lighting** - see figure 7.8.6

■ The airplane is equipped with three strobe and navigation lights, two landing lights, two taxi lights, two recognition lights and a wing leading edge icing inspection light.

**Landing lights**

Landing lights are embedded in the winglets and located in leading edges. Lights illumination is controlled by setting to LDG, a switch located on upper panel.

The Pulse lite system, if installed, enables the pilot to control landing light flashing to be seen by the control tower or in heavy traffic areas.

**Taxi lights**

The taxi lights are embedded in the winglets and located in leading edges. They are controlled by setting to TAXI, a switch located on upper panel.

**Navigation lights and strobe lights**

■ Two strobe and navigation lights are installed in the winglets and one on the tail cone. They are controlled by NAV and STROBE switches located on upper panel.

● NOTE ●

By night, do not use anticollision lights in fog, clouds or mist as light beam reflexion may lead to dizziness and loss of sense of orientation.

●

**Recognition lights**

Recognition lights are embedded in the winglets.

They are automatically switched on when the airplane is on ground.

**Leading edge icing inspection light**

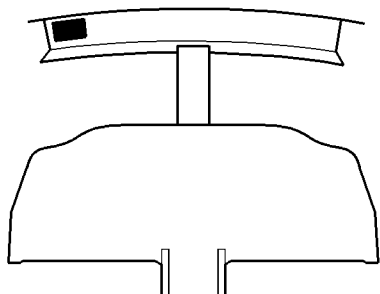
The leading edge icing inspection light is installed on fuselage L.H. side, its beam illuminates the wing leading edge. It is controlled by the ICE LIGHT switch installed on DE ICE SYSTEM panel - see figure 7.13.1.

### **FWD compartment light**

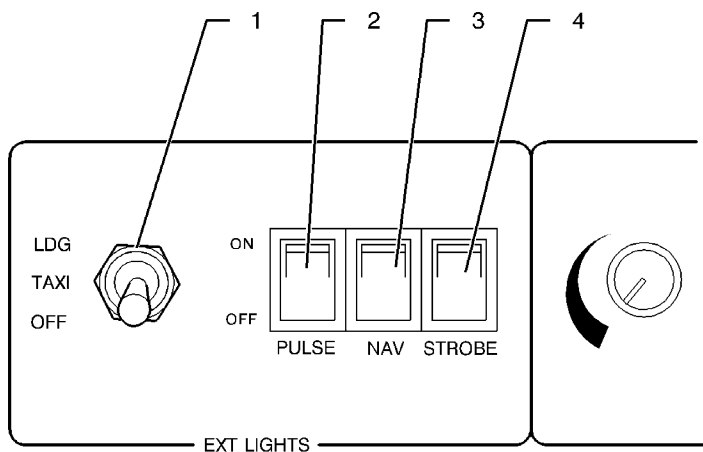
The dome light illumination of the FWD compartment is controlled by the switch located in the upper section of the door frame.

### **Fuel unit compartment light**

The lighting of the fuel unit compartment allows improving the visibility of the clogging indicator by pressing the push-button located besides the inspection door.



- 1) Taxi and landing light switch
- 2) Pulselite system switch
- 3) Navigation lights switch
- 4) Strobe lights switch



I4240000AAA VMA8400

Figure 7.8.6 - External lighting controls

## **Interior lighting** - see Figure 7.8.7

Interior lighting consists of access, cabin, instrument panel, instruments, baggage compartment and emergency lighting.

### **Access lighting**

Access lighting consists of two floodlights located on the ceiling upholstery (one at the level of the access door, the other at the level of the storage cabinet) and the L.H. dome light of baggage compartment. ACCESS push-button on INT LIGHTS panel and the push-button located on access door rear frame control these 3 lights via a delayed breaker.

If the crash lever is down, access lighting is automatically cut out after 3 minutes.

If the crash lever is up, there is no access lighting automatic cut out.

### **Cabin lighting**

■ Cabin lighting consists of two individual floodlights for front seats, six individual floodlights for rear passenger seats and the baggage compartment R.H. dome light.

■ Each floodlight is controlled by a push-button located near. The pilot can switch off the cabin floodlights and the baggage compartment dome light with the CABIN switch.

### **Instrument panel lighting**

Instrument panel lighting is controlled by the PANEL rheostat located on INT LIGHTS panel. This lighting consists of backlighted panels and a led lighting for the pedestal.

### **Breaker panel lighting**

Breaker panel lighting is controlled by a switch located on the instrument panel near the pilot's control wheel.

### **Emergency lighting**

Emergency lighting consists of two swiveling floodlights located on both sides of the cockpit overhead panel above front seats. It illuminates instrument panel assembly in case of visor lighting tubes and / or instrument integrated lighting failure.

A rheostat located on the cockpit overhead panel controls emergency lighting operation and intensity. Forward rotation of control knob allows changing from OFF position to minimum lighting then increasing lighting to maximum brightness.



- 1) Instrument panel lighting switch (rheostat)
- 2) DIMMER switch
- 3) Cabin lighting switch (rear seats reading light)
- 4) Access door, baggage compartment and FWD dome light (delayed breaker) push-button
- 5) Emergency lighting switch
- 6) Breaker panel lighting switch

Figure 7.8.7 (1/2) - Internal lighting controls

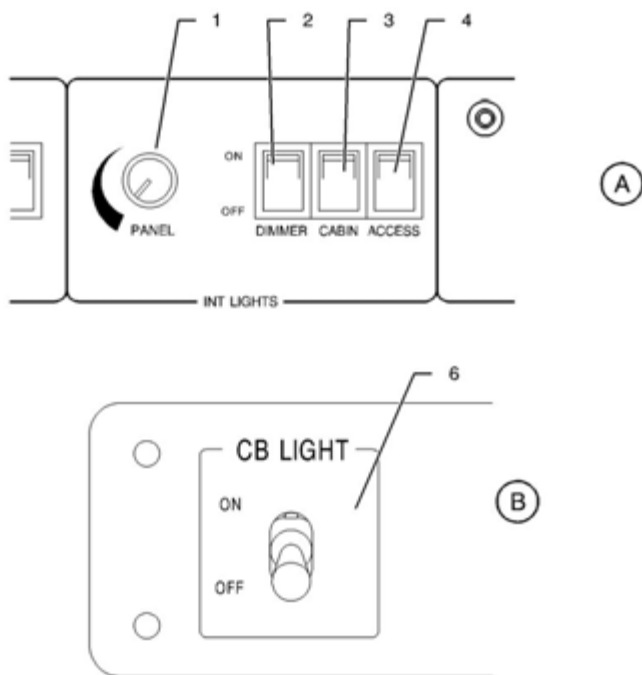
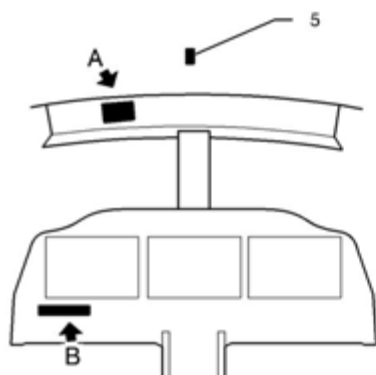


Figure 7.8.7 (2/2) - Internal lighting controls

## 7.9 - Air conditioning and pressurization

### • NOTE •

A list of abbreviations used in this chapter is given in figure 7.9.1A.

The airplane is equipped with a Global Air System (GAS), which ensures air conditioning and pressurization control - see figure 7.9.1A.

- Air conditioning corresponds to the cockpit / cabin air temperature management.
- Pressurization corresponds to the cabin altitude / rate of change management.

The GAS is composed of 3 sub-systems :

- Engine bleed air system,
- Cabin pressurization control system,
- Dual zones Environmental Control System, which includes heating and cooling functions.

These sub-systems are managed by a single digital controller, the GASC, which receives information from :

- the sensors within the sub-systems,
- the human interfaces set in the airplane.

The GASC elaborates the proper commands to the sub-systems actuators and indication or warning elements.

GAS controls are located on :

- the A/C panel on the left side of the left control wheel,
- the PRESSURIZATION panel on the right side of the left control wheel,
- a control panel above the arm rest of the L.H. side passenger's seat.

The pilot monitors the system through gauges and CAS messages appearing on the MFD. These indications are independent of the GASC controls and internal sensors.

## Engine bleed air system

The engine bleed air system is designed to ensure the following functions :

- to regulate the bleed air from the engine,
- to ensure a controlled airflow in the cabin,
- to adjust the bleed air temperature for cabin heating.

### To regulate the bleed air from the engine

The engine bleed air system operates from either P2.5 or P3 engine bleed ports.

The system normally operates on the P2.5 port as long as the pressure or temperature demands are met by this port.

If one of these conditions is not met, the system automatically switches to the P3 port.

When the pressure or temperature demand can be met by the P2.5 port, the system automatically returns to using the P2.5 port.

The Inlet Pressure Port Sensor (IPPS) measures the pressure at the P2.5 port and sends the value to the GASC which manages the ports switching on condition with the Shut Off Valve (SOV).

A Non Return Valve (NRV) prevents P3 air from entering the P2.5 port when the P3 port is opened.

### To ensure a controlled airflow in the cabin

The bleed air flow is controlled by the Flow Control and Shut Off Valve (FCSOV) driven by the GASC.

### To adjust the temperature of the bleed air

The bleed air outlet temperature control is ensured by the By-Pass Valve (BPV) in association with the Main Heat Exchanger (MHX).

Based on pilot's or passengers' TEMP selector position, the GASC computes the appropriate cabin air inlet temperature target and compares it to the actual measured inlet temperature in order to set the BPV position. The BPV derives a part of the bleed air through the MHX to cool it and mixes it to the remaining cabin air.

**System operation**

See figure 7.9.2.

The BLEED switch allows selection of the engine bleed air system provided that the engine is running.

The Ground Fan (GF) operates until takeoff, when BLEED switch is set to AUTO, and **MAIN GEN** is OFF.

The BLEED switch is fitted with a blocking device between AUTO, MAX DIFF and OFF positions. This prevents the operator from inadvertently setting the BLEED switch to OFF position.

To reset the system, set BLEED switch to OFF, then back to AUTO.

**System protection**

Power for the engine bleed air system is supplied by the BUS 2 bar and is protected by the CAB BLEED breaker.

**Cabin pressurization control system**

In flight, the GASC controls the modulation of the Outflow Valve (OFV) in order to reach the computed cabin altitude.

**System operation**

See figure 7.9.2.

The BLEED switch allows to activate the pressurization system.

Either one of two pressurization modes can be selected : AUTO or MAX DIFF

- When BLEED switch is set to AUTO, the GASC controls the cabin altitude rate of change in order to :
  - . optimize comfort,
  - . avoid reaching maximum or negative cabin differential pressure.
- MAX DIFF mode controls the cabin pressure to assist passengers that might require the lowest cabin altitude possible. When selecting this mode :
  - . flights below 13500 ft will result in cabin altitudes as low as 0 ft,
  - . for flights above 13500 ft, the cabin altitude is minimized throughout the flight while maintaining cabin differential pressure below 6.0 PSI.

The GASC controls the OFV through a torque motor on the valve.

## Cabin altitude management

In order to maximize comfort during all phases of flight, the cabin altitude is automatically computed by the GASC using flight parameters (such as aircraft altitude, altitude rate of change) sent by the avionics.

During descent, the GASC uses the Landing Field Elevation (LFE) to manage the optimal cabin altitude rate of change in order to land with a cabin altitude equal to LFE minus 200 ft.

The pilot selects LFE on the touchscreen controller :

- automatically, by setting a destination airport in the flight plan,
- manually, by pressing "MFD Home", "Aircraft Systems", "Landing Field Elev" and then "Manual".

## System monitoring

The pilot monitors information related to the pressurization system through gauges and information displayed on the MFD :

- landing field altitude,
- cabin altitude in ft,
- cabin climb speed in ft/min,
- cabin differential pressure ( $\Delta P$ ) in PSI.

These gauges are independent of the GASC controls and internal sensors.

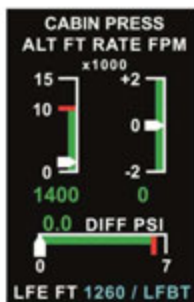


Figure 7.9.1 - Cabin altitude monitoring

**CAS messages** are displayed in the MFD CAS window :

- **BLEED TEMP** indicates that an overtemperature was detected by either the Bleed Temperature Switch (BTSW) or the Overheat Thermal Switch (OTSW).
- **BLEED OFF** indicates that either the BLEED switch is in OFF position or that a system malfunction was detected by the GASC. The Flow Control and Shut Off Valve (FCSOV) is automatically closed in flight if either a cabin inlet overtemperature, or a BDPS or FCSOV failure is detected.
- **BLEED OFF** is displayed 45 seconds after landing if a fault on the Overheat Thermal Switch was detected by the GASC during flight.
- **CABIN ALTITUDE** indicates that the cabin altitude is over 10000 ft.
- **CABIN DIFF PRESS** indicates that the cabin differential pressure is over 6.4 psi (441 mb). The DUMP switch could be used in case of necessity to depressurize the cabin.
- **CPCS BACKUP MODE** indicates that the GASC cannot compute optimal cabin altitude due to a system malfunction. Under this condition, the GASC will control the cabin altitude to 9800 ft default value.
- **MAX DIFF MODE** indicates that the BLEED switch is set to MAX DIFF.

### Protection - Safety

Cabin is automatically depressurized as soon as the airplane is on ground through landing gear switches (airplane on ground) or, if necessary, by actuating DUMP switch located on PRESSURIZATION panel (in normal operation, this switch is protected and locked by a cover).

Overpressure and negative relief safety are managed by both OFV and SFV. The safety functions are ensured by independent pneumatic modules fitted on both valves, which override the GASC control when necessary.

The DUMP switch allows the pilot to open the OFV in order to depressurize the cabin.

The OFV is fitted with a cabin altitude limitation device which overrides the DUMP function and forces the closure of the OFV if the cabin altitude reaches 14500 ft.

## **Dual zones Environmental Control System (ECS)**

The ECS ensures both Cockpit and Cabin heating and cooling functions.

The ECS consists of two independent air circuits :

- Heating circuit, controlled by Temperature Conditioning System (TCS)
- Cooling circuit, controlled by Vapor Cycle Cooling System (VCCS)

### **Heating circuit**

The TCS regulates hot air coming from the bleed air system (also used for pressurization) and mixes it with the recirculating cabin air at the Mixing Ejector (MIXEJ) in order to lower the delivered air temperature.

The resultant air flow enters the Hot Air Distributor (HAD) and is distributed in the cockpit / cabin zones depending on the demand.

The air is distributed :

- into the cockpit zone through :
  - . ports located on pedestal sides,
  - . ports under each seat or
  - . the demisting outlets.
- into the cabin zone through :
  - . ports located on the lower section of the L.H. side and R.H. side cabin upholstery.

### **Cooling circuit**

The VCCS is selected on only when the GASC receives a cooling request. It is composed of two independent circuits :

- one for the cockpit zone
- one for the cabin zone

For each circuit, air is sucked by means of a variable speed electrical fan and then blown through an evaporator and ducted to the different zones :

- cockpit circuit, by passing through :
  - . the upper panel equipped with swivelling and adjustable air outlets,
  - . air outlets located on arm rests of pilot and R.H. side front passenger stations and
  - . ports located under instrument panel.



- cabin circuit, by passing through :
  - the overhead duct equipped with swivelling and adjustable air outlets,
  - ports located on the floor between the cabinets and the intermediate passenger's seats.

**System operation**

See figure 7.9.2 and paragraph Air temperature management.

A/C control panel selection :

If the A/C switch is set to OFF :

>> *Before GASC software evolution (Pre-MOD70-0689-21)*

- Temperature is set by default by the GASC to 23°C,

>> *After GASC software evolution (Post-MOD70-0689-21)*

- The system maintains the cabin air inlet temperature previously selected. The pilot can modulate this temperature by using the TEMP selector on A/C panel,

>> *All*

- Cockpit / Cabin evaporator fans are OFF,
- VCCS is inhibited.

If the A/C switch is set to PILOT :

- Controls located in the cabin zone are inhibited.

If the A/C switch is set to PLT+PAX :

- Each zone is controlled by its own settings.

FAN speed selectors positions :

- OFF : prevents recirculation of cold air through the cold air circuit (VCCS).
- 1 - 4 : Cockpit / Cabin fan speeds are selectable.

TEMP selectors :

- Allow adjustment of the temperature for the cockpit and cabin zones.

>> *After GASC software evolution (Post-MOD70-0689-21)*

- If the pilot or passengers set the TEMP selector to the maximum heat position, the bleed air system automatically switches from the P2.5 to the P3 bleed port to increase the temperature and flow rate of the incoming bleed air. Except in the case of very cold environmental conditions, this switching is inhibited below 25000 ft.

>> *All*

HOT AIR FLOW distributor :

The HOT AIR FLOW distributor selects between windshield defog or cabin heating.

• NOTE •

For maximum efficiency, the HOT AIR FLOW distributor should be set either in defog position (fully turned to the left) or in cabin position (fully turned to the right).

•

>> *After GASC software evolution (Post-MOD70-0689-21)*

When the HOT AIR FLOW distributor is set in defog position (fully turned to the left), the bleed air system automatically switches from the P2.5 to the P3 bleed port to increase the temperature and flow rate of the incoming bleed air. Except in the case of very cold environmental conditions, this switching is inhibited below 25000 ft.

>> *All*

Emergency air system :

An emergency ventilation valve allows outside air to enter the cabin when the EMERGENCY RAM AIR control knob is pulled out. The EMERGENCY RAM AIR control knob is located under the right side of instrument panel near the right control wheel.

- In NORMAL position, the valve is closed and the control is locked.
- To open the emergency ventilation valve, press the locking button on the knob and pull out.

• NOTE •

Reduce the cabin differential pressure, if necessary by depressurizing the cabin, to be able to operate the EMERGENCY RAM AIR control knob.

•

**System protection**

Power for the ECS is supplied by the BUS 2 bar and is protected by the AIR COND breaker.

Four fans are supplied by BUS 4 bar and protected respectively by following breakers : COND FAN, CABIN FAN, COCKPIT FAN and GND FAN.

The system includes an automatic load shedding feature which :

- when **MAIN GEN** is ON :
  - . turns off the Ground Fan (GF),
  - . turns off the Condenser Fan (COND FAN),
  - . opens compressor clutch.
- during engine start :
  - . turns off all the Vapor Cycle Cooling System (VCCS)

**Air temperature management**





Cockpit and cabin heating / cooling

Cockpit and cabin air temperature management is operated by selecting :



















For optimal air temperature management, select :



| Air temperature expectations |      | Cockpit panel selection  |  | Cabin panel selection  |  |
|------------------------------|------|--|--|--|--|
| Cockpit area                 | Warm |  |  |  |  |
| Cabin area                   | Hot  |  |  |  |  |

|  |
|--|
| <u>Results</u>   |
| <u>Mixed Bleed Air Circuit</u> :<br>Warm air is distributed to both cockpit and cabin. More warm air is directed to cabin due to the greater temperature demand. |
| <u>Cold Air Circuit</u> :<br>Not active, as no TEMP selector is set in the blue arc.   |
| <u>FANS</u> select 0. Selecting a FAN speed will reduce cabin air temperature.   |

| Air temperature expectations   |      | Cockpit panel selection   |   | Cabin panel selection   |   |
|--|------|---|---|---|---|
| Cockpit area   | Warm |  |  |  |  |
| Cabin area   | Warm |   |   |   |   |
| <u>Results</u><br><u>Mixed Bleed Air Circuit</u> :<br>Warm air is distributed to both cockpit and cabin equally due to equal temperature demand.<br><u>Cold Air Circuit</u> :<br>Not active, as no TEMP selector is set in the blue arc.<br><u>FANS</u> select 0. Selecting a FAN speed will reduce cabin air temperature.   |      |   |   |   |   |
| Cockpit area   | Warm |  |  |  |  |
| Cabin area   | Cool |   |   |   |   |
| <u>Results</u><br><u>Mixed Bleed Air Circuit</u> :<br>Warm air is distributed to both cockpit and cabin. More warm air is directed to cockpit due to the greater temperature demand.<br><u>Cold Air Circuit</u> :<br>Cabin TEMP selector is set in the blue arc resulting in cabin air conditioning system supplying the cabin area.<br><u>Cockpit FAN</u> : select 0.<br><u>Cabin FAN</u> : select 1 or more to circulate the cooled air. |      |   |   |   |   |

| Air temperature expectations  |      | Cockpit panel selection   |   | Cabin panel selection   |   |
|---|------|---|---|---|---|
| Cockpit area  | Cool |  |  |  |  |
| Cabin area  | Cool |   |   |   |   |
| <u>Results</u><br><u>Mixed Bleed Air Circuit</u> :<br>With the cockpit and cabin TEMP selectors set in the blue arc, the Hot Air Distributor discharges the mixed bleed air below the floor towards the cold air circuit fans and evaporators to reduce the air to the desired temperature.<br><u>Cold Air Circuit</u> :<br>Both cockpit and cabin TEMP selectors are set in the blue arc resulting in both cockpit and cabin air conditioning systems supplying their respective area.<br><u>Cockpit and cabin FANS</u> :<br>Select 1 or more to circulate the cooled air. |      |   |   |   |   |
| Cockpit area  | Cool |  |  |  |  |
| Cabin area  | Warm |   |   |   |   |
| <u>Results</u><br><u>Mixed Bleed Air Circuit</u> :<br>Warm air is distributed to both cockpit and cabin. More warm air is directed to cabin due to the greater temperature demand.<br><u>Cold Air Circuit</u> :<br>Cockpit TEMP selector is set in the blue arc resulting in cockpit air conditioning system supplying the cockpit area.<br><u>Cockpit FAN</u> : select 1 or more to circulate the cooled air.<br><u>Cabin FAN</u> : select 0.  |      |   |   |   |   |

• NOTE •

FANS selected to zero ensures that no cool air is recirculated when trying to maximize the heating of the cockpit and / or cabin zones.





### Cabin override

Selecting the switch to PILOT position disables TEMP and FAN speed selectors in the cabin area.

The PILOT position also distributes warm air equally to the cockpit and cabin.

Only the cockpit TEMP and FAN selectors inputs are used by the GASC for air temperature management.



| Air temperature expectations |      | Cockpit panel selection   |   | Cabin panel selection   |   |
|------------------------------|------|---|---|---|---|
| Cockpit area                 | Warm |  |  |  |  |
| Cabin area                   | /    |   |   |   |   |

### Results





#### Mixed Bleed Air Circuit :

Warm air is distributed equally to both cockpit and cabin. The temperature is selected by the cockpit TEMP selector.

#### Cold Air Circuit :

Not active, as the cockpit TEMP selector is set in the red zone and cabin TEMP selector is inhibited.

Cockpit FAN : select 0. Selecting a FAN speed will reduce cabin air temperature.

| Air temperature expectations   |      | Cockpit panel selection   |   | Cabin panel selection   |   |
|--|------|---|---|---|---|
| Cockpit area   | Cool |  |  |  |  |
| Cabin area   | /    |   |   |   |   |
| <p><u>Results</u></p> <p><u>Mixed Bleed Air Circuit</u> :</p> <p>With the cockpit TEMP selector set in the blue arc, the Hot Air Distributor discharges the mixed bleed air below the floor towards the cold air circuit fans and evaporators to reduce the air to the desired temperature.</p> <p><u>Cold Air Circuit</u> :</p> <p>Cockpit TEMP selector is set in the blue arc resulting in both cockpit and cabin air conditioning systems to supply their respective area.</p> <p><u>Cockpit FAN</u> : select 1 or more to circulate the cooled air.</p> <p>Cockpit FAN speed selector determines cabin FAN speed.</p> |      |   |   |   |   |



### Windshield DEFOG

Windshield defog is operated by selecting :



or



| Air temperature expectations |   | Cockpit panel selection |  | Cabin panel selection |  |
|------------------------------|---|-------------------------|--|-----------------------|--|
| Cockpit area                 | / |                         |  |                       |  |
| Cabin area                   | / |                         |  |                       |  |

#### Results

##### Mixed Bleed Air Circuit :

Air is distributed to the windshields and cockpit side windows at a fixed temperature regardless of TEMP selector settings.

##### Cold Air Circuit :

Inhibited when DEFOG is selected.

FANS will continue to operate if selected to 1 or more for air circulation.

Intentionally left blank

- 1) Demisting outlets
- 2) Front vents
- 3) Cockpit ventilated temperature sensor (CKVTS)
- 4) Cabin ventilated temperature sensor (CBVTS)
- 5) Air ports
- 6) Cabin control panel
- 7) Global air system controller (GASC)
- 8) Out-flow valve (OFV)
- 9) Safety valve (SFV)
- 10) Condenser fan
- 11) Condenser
- 12) High pressure switch
- 13) Drier filter
- 14) Cabin fan
- 15) Cabin evaporator
- 16) Cabin blown temperature sensor (CBBTS)
- 17) Cabin thermostatic valve
- 18) Low pressure switch
- 19) A/C and PRESSURIZATION panel
- 20) Cockpit thermostatic valve
- 21) Cockpit fan
- 22) Cockpit evaporator
- 23) Cockpit blown temperature sensor (CKBTS)

Figure 7.9.1A (1/3) - GAS items list and abbreviations

- 24) Demisting microswitch
- 25) Hot air distributor (HAD)
- 26) Cabin inlet temperature sensor (ITS)
- 27) Cabin bleed temperature switch (BTWS)
- 28) Mixing ejector (MIXEJ)
- 29) Check valve
- 30) MFD unit
- 31) Ground safety microswitch
- 32) Differential pressure switch
- 33) By-pass valve (BPV)
- 34) Cabin altitude alarm switch
- 35) Emergency air supply system  
(EMERGENCY RAM AIR)
- 36) Main heat exchanger (MHX)
- 37) Ground fan (GF)
- 38) Flow control shut off valve (FCSOV)
- 39) Bleed differential pressure sensor (BDPS)
- 40) Compressor
- 41) Shut-off valve (SOV)
- 42) Overheat thermal switch (OTSW)
- 43) Non return valve (NRV)
- 44) Intermediate port pressure sensor (IPPS)
- 45) Cabin pressure sensor

Figure 7.9.1A (2/3) - GAS items list and abbreviations

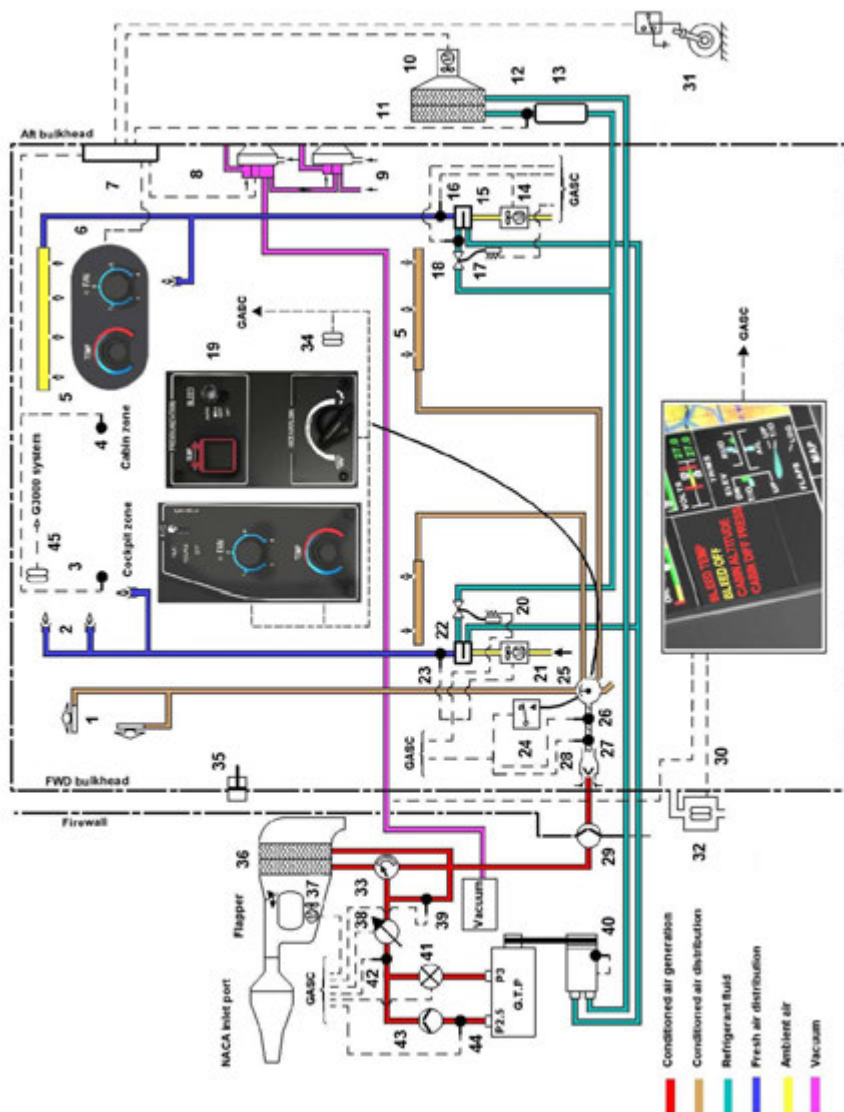


Figure 7.9.1A (3/3) - GAS

- 1) A/C switch
- 2) FAN speed selector (cockpit/cabin)
- 3) TEMP selector (cockpit/cabin)
- 4) DUMP switch
- 5) HOT AIR FLOW distributor
- 6) BLEED switch
- 7) FAN speed selector (cabin)
- 8) TEMP selector (cabin)

Figure 7.9.2 (1/2) - GAS controls

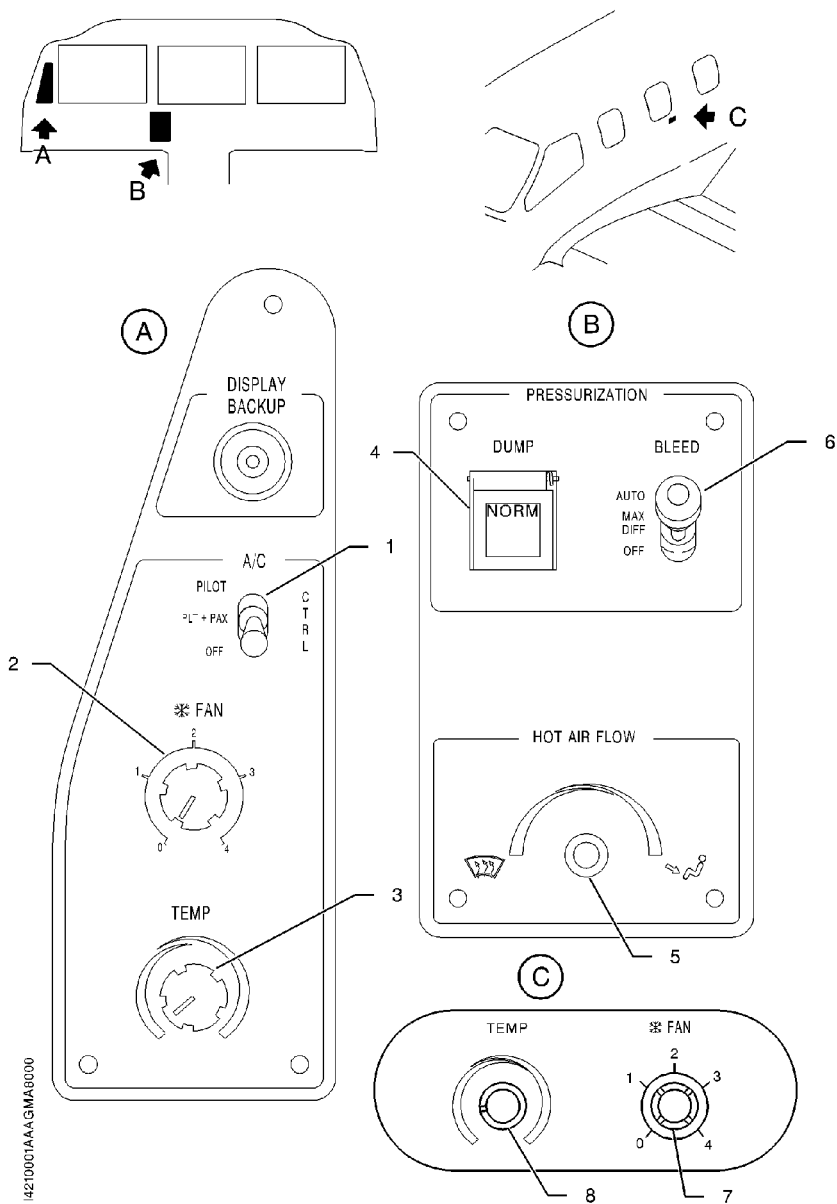


Figure 7.9.2 (2/2) - GAS controls

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**7.10 - Emergency oxygen system** - see figure 7.10.1

The gaseous oxygen system will be used by the crew and the passengers, when the cabin altitude is greater than 10000 ft following a loss of pressurization or in case of cabin air contamination.

**USE OXYGEN MASK** appears in the MFD CAS window (in normal conditions) and the USE OXYGEN MASK/USE OXYGEN MASK aural warning alert sounds when the cabin altitude is greater than 10000 ft.

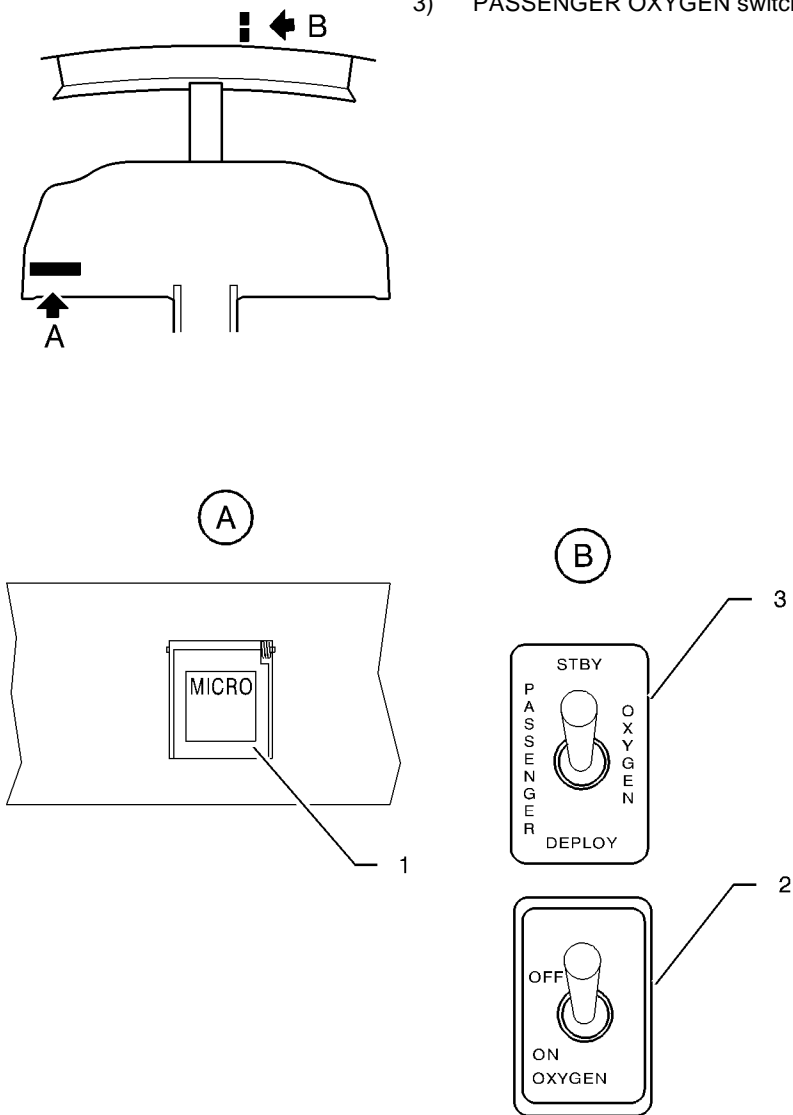
The oxygen reserve is contained in an oxygen cylinder made of composite material and located outside of the pressurized cabin into the R.H. karman. Its capacity is 50.3 cu.ft (1425 litres) STPD (Standard Temperature Pressure Dry) and use limit pressures are :

- maximum pressure 1850 PSIG (127 bars) at 70° F (21° C).  
Evolution of this pressure according to the outside temperature is given in section 8, figure 8.7.4, as well as on a placard on the inside of the cylinder service door,
- minimum pressure 217 PSIG (15 bars).

The oxygen cylinder head is equipped with :

- a hand-controlled isolation valve to permit cylinder installation and removal,
- a microswitch causing **OXYGEN** to light on. This message lights on, when the isolation valve is closed,
- a graduated pressure gage,
- a charging valve - refer to the replenishment procedure in section 8,
- an overpressure system consisting of a safety disc. This disc is designed to rupture between 2500 and 2775 PSIG (172 and 191 bars) discharging the cylinder contents outboard,
- a pressure reducing valve adjusting utilization pressure to a value comprised between 64 and 85 PSIG (4.4 and 5.9 bars),
- a low pressure safety valve calibrated to 116 PSIG (8 bars).

- 1) MICRO / MASK switch
- 2) OXYGEN switch
- 3) PASSENGER OXYGEN switch



14231204AAAFMA8002

Figure 7.10.1 - Emergency oxygen system

A control panel located in the cockpit overhead panel at the disposal of the pilot includes :

- a two-position valve ON/OFF (OXYGEN switch) to permit the supply of the front seats occupiers masks,
- a two-position valve DEPLOY/STBY (PASSENGER OXYGEN switch) with guard to permit the supply of the passengers four masks, when the first valve is open.

Oxygen pressure is displayed on the MFD.

An altimetric valve provides an automatic passengers masks actuation function at a cabin altitude between 13000 and 14000 ft when OXYGEN switch is set to ON.

Two pressure-demand type masks allowing quick donning with only one hand, covering the nose and the mouth, as well as two pairs of smoke goggles are at disposal of the pilot and of the R.H. front seat occupier. Masks are installed in cups on the cabin walls aft of the front seats. Permanently connected to the oxygen system, they are equipped with a micro controlled by the MICRO/MASK switch under cover located on the instrument panel near the pilot's control wheel. The cockpit masks are equipped with a microphone, a three-position selector NORMAL, 100 % and EMERGENCY and with a push-button PRESS TO TEST. The proper flow is signaled by a flow indicator (blinker) into the oxygen tubing.

The airplane is equipped with two smoke goggles.

Four passengers constant-flow type masks, covering the nose and the mouth and permanently connected, are installed in two containers on the cabin ceiling. The opening of these containers and the descent of the masks are controlled by the pilot, when OXYGEN switch is set to ON and PASSENGERS OXYGEN switch is set to DEPLOY, or automatically at a cabin altitude between 13000 and 14000 ft with the OXYGEN switch set to ON. The oxygen flow is obtained by pulling on the mask bounded by a lanyard cord to a pin. A proper flow is signaled by the filling of the green bag located on each passenger mask.

**▲ WARNING ▲**

**Do not smoke during oxygen system use.**

**Oil, grease, soap, make up, lipstick and any other greasy substances constitute a serious fire or burning hazard, when on contact with oxygen.**



**Flight above 15000 ft with possible emergency descent**

Minimum oxygen pressure (PSIG) for following conditions :

- 4 minutes from 31000 to 15000 ft. All equipment used from 31000 ft.
- Plus 30 minutes usage by each pilot and passenger at 15000 ft.
- Plus 86 minutes usage by each pilot at 10000 ft.

| Number of occupants |       | Outside temperature |                 |                 |                 |                 |                  |                   |
|---------------------|-------|---------------------|-----------------|-----------------|-----------------|-----------------|------------------|-------------------|
| Cockpit             | Cabin | 110° F/<br>43° C    | 90° F/<br>32° C | 70° F/<br>21° C | 50° F/<br>10° C | 30° F/<br>-1° C | 10° F/<br>-12° C | -10° F/<br>-23° C |
| 1                   | 0     | 631                 | 614             | <b>597</b>      | 580             | 563             | 546              | 529               |
| 1                   | 1     | 759                 | 736             | <b>713</b>      | 691             | 668             | 646              | 623               |
| 1                   | 2     | 885                 | 856             | <b>828</b>      | 799             | 771             | 743              | 715               |
| 1                   | 3     | 1010                | 976             | <b>941</b>      | 907             | 873             | 839              | 806               |
| 1                   | 4     | 1137                | 1096            | <b>1056</b>     | 1015            | 975             | 935              | 897               |
| 2                   | 0     | 1037                | 1001            | <b>965</b>      | 930             | 894             | 859              | 825               |
| 2                   | 1     | 1164                | 1122            | <b>1080</b>     | 1038            | 997             | 956              | 916               |
| 2                   | 2     | 1289                | 1241            | <b>1192</b>     | 1144            | 1097            | 1050             | 1004              |
| 2                   | 3     | 1416                | 1361            | <b>1306</b>     | 1252            | 1198            | 1145             | 1093              |
| 2                   | 4     | 1541                | 1480            | <b>1418</b>     | 1357            | 1297            | 1238             | 1180              |

Figure 7.10.2 - Minimum oxygen pressure (PSIG)  
[Flight above 15000 ft with possible emergency descent]

● NOTE ●

After a long parking time in the sunshine, increase pressures indicated in the table here above by 8 %.



**When required to remain above 15000 ft due to minimum enroute altitude**

Minimum oxygen pressure (PSIG) for following conditions :

- Flight above 15000 ft. All equipment used.
- 1 hour usage by each pilot and passenger.
- Plus 1 hour usage by each pilot under 15000 ft.

| Number of occupants |       | Outside temperature |                 |                 |                 |                 |                  |                   |
|---------------------|-------|---------------------|-----------------|-----------------|-----------------|-----------------|------------------|-------------------|
| Cockpit             | Cabin | 110° F/<br>43° C    | 90° F/<br>32° C | 70° F/<br>21° C | 50° F/<br>10° C | 30° F/<br>-1° C | 10° F/<br>-12° C | -10° F/<br>-23° C |
| 1                   | 0     | 618                 | 602             | <b>585</b>      | 569             | 552             | 536              | 520               |
| 1                   | 1     | 842                 | 816             | <b>789</b>      | 763             | 736             | 710              | 685               |
| 1                   | 2     | 1067                | 1029            | <b>992</b>      | 955             | 918             | 882              | 846               |
| 1                   | 3     | 1513                | 1240            | <b>1192</b>     | 1144            | 1097            | 1050             | 1004              |
| 1                   | 4     | 1513                | 1452            | <b>1392</b>     | 1333            | 1275            | 1217             | 1161              |
| 2                   | 0     | 992                 | 958             | <b>925</b>      | 891             | 858             | 825              | 793               |
| 2                   | 1     | 1215                | 1170            | <b>1125</b>     | 1081            | 1037            | 994              | 952               |
| 2                   | 2     | 1439                | 1382            | <b>1326</b>     | 1270            | 1215            | 1161             | 1108              |
| 2                   | 3     | 1662                | 1593            | <b>1525</b>     | 1457            | 1391            | 1326             | 1262              |
| 2                   | 4     | 1888                | 1807            | <b>1725</b>     | 1645            | 1567            | 1490             | 1415              |

Figure 7.10.3 - Minimum oxygen pressure (PSIG)

[When required to remain above 15000 ft due to minimum enroute altitude]

• NOTE •

After a long parking time in the sunshine, increase pressures indicated in the table here above by 8 %.

•

## Flight between 15000 ft and 10000 ft

Minimum oxygen pressure (PSIG) for following conditions :

- Flight under 15000 ft.
- 90 minutes usage by each pilot and **one** passenger.
- Plus 30 minutes usage by each pilot at 10000 ft.

| Number of occupants |       | Outside temperature |                 |                 |                 |                 |                  |                   |
|---------------------|-------|---------------------|-----------------|-----------------|-----------------|-----------------|------------------|-------------------|
| Cockpit             | Cabin | 110° F/<br>43° C    | 90° F/<br>32° C | 70° F/<br>21° C | 50° F/<br>10° C | 30° F/<br>-1° C | 10° F/<br>-12° C | -10° F/<br>-23° C |
| 1                   | 0     | 618                 | 602             | <b>585</b>      | 569             | 552             | 536              | 520               |
| 1                   | 1     | 961                 | 929             | <b>896</b>      | 864             | 833             | 801              | 770               |
| 1                   | 2     | 961                 | 929             | <b>896</b>      | 864             | 833             | 801              | 770               |
| 1                   | 3     | 961                 | 929             | <b>896</b>      | 864             | 833             | 801              | 770               |
| 1                   | 4     | 961                 | 929             | <b>896</b>      | 864             | 833             | 801              | 770               |
| 2                   | 0     | 992                 | 958             | <b>925</b>      | 891             | 858             | 825              | 793               |
| 2                   | 1     | 1333                | 1282            | <b>1231</b>     | 1181            | 1131            | 1083             | 1035              |
| 2                   | 2     | 1333                | 1282            | <b>1231</b>     | 1181            | 1131            | 1083             | 1035              |
| 2                   | 3     | 1333                | 1282            | <b>1231</b>     | 1181            | 1131            | 1083             | 1035              |
| 2                   | 4     | 1333                | 1282            | <b>1231</b>     | 1181            | 1131            | 1083             | 1035              |

Figure 7.10.4 - Minimum oxygen pressure (PSIG)  
[Flight between 15000 ft and 10000 ft]

• NOTE •

After a long parking time in the sunshine, increase pressures indicated in the table here above by 8 %.

•

## O<sub>2</sub> smart mike mask, if installed

The two cockpit masks, O<sub>2</sub> smart mike masks, are equipped with a system that detects when a communication is made operating one of the push-to-talk buttons. It activates a noise reduction system that attenuates the oxygen flow noise in the headsets.

Noise reduction function operates when the switch located on O<sub>2</sub> connecting line is set to ON.

14351100AAEMA8000

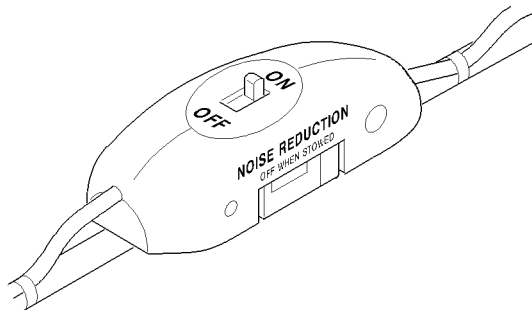


Figure 7.10.5 - Noise reduction switch

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**7.11 - Air data system and instruments** - see figure 7.11.1

Airplane air data system consists of :

- two separate static pressure systems supplying the standby attitude module and air data computers (ADC).

A part of system 1 is backed up by an alternate system which operation is controlled by a switching valve (normal / alternate) attached to instrument panel under R.H. control wheel. In case of obstruction or icing of ports, this selector isolates airplane normal static system. When selector is on alternate position (pulled rearwards), static pressure is picked from a port located in airplane rear fuselage.

- two separate dynamic pressure systems supplying the standby attitude module and air data computers.

**Static pressure systems****Primary systems**

Two dual static ports (one on either side of the fuselage tail part) supply a dual system routed towards the cockpit.

System 1 part, which is connected to the switching valve (normal / alternate), supplies the  $\Delta P$  cabin and the electronic standby indicator. The system remainder directly supplies one of the air data computers.

System 2 is directly connected to the second ADC.

Systems feature a drain valve located under the instrument panel on R.H. side.

**Alternate static source**

The alternate static port located in the rear fuselage supplies a system routed to the switching valve (normal / alternate) in order to replace static system 1.

The alternate line incorporates a drain plug located under the instrument panel on R.H. side.

- Static pressure from alternate line is only provided to standby instrument.

## Dynamic pressure system

One heated pitot probe is installed under the L.H. wing. The second one is installed under the R.H. wing. The first one supplies the standby attitude module and one ADC.

The second one supplies the other ADC.

Both lines incorporate a drain plug located in the root of L.H. and R.H. wings.

### Pitot heating

Pitot heating is controlled by PITOT L HTR and PITOT R & STALL HTR switches, installed on DE-ICE SYSTEM panel. Refer to chapter 7.13 for further details.

#### ▲ CAUTION ▲

Do not use heating during prolonged periods on ground to avoid pitot  
overheat.



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- 1) Pitot L
- 2) Dynamic system drain
- 3) Standby Attitude Module (MD302)
- 4) ADC
- 5) ADC
- 6) FWD pressure bulkhead
- 7) Static system drain
- 8) Static system drain
- 9) Static system drain
- 10) Emergency static system drain
- 11) Emergency static valve (Normal / alternate)
- 12) Instrument panel
- 13) Dynamic system drain
- 14) Pitot R
- 15) Rear pressure bulkhead
- 16) Static port
- 17) Emergency static port
- 18) Static port

Figure 7.11.1 (1/2) - Air data system

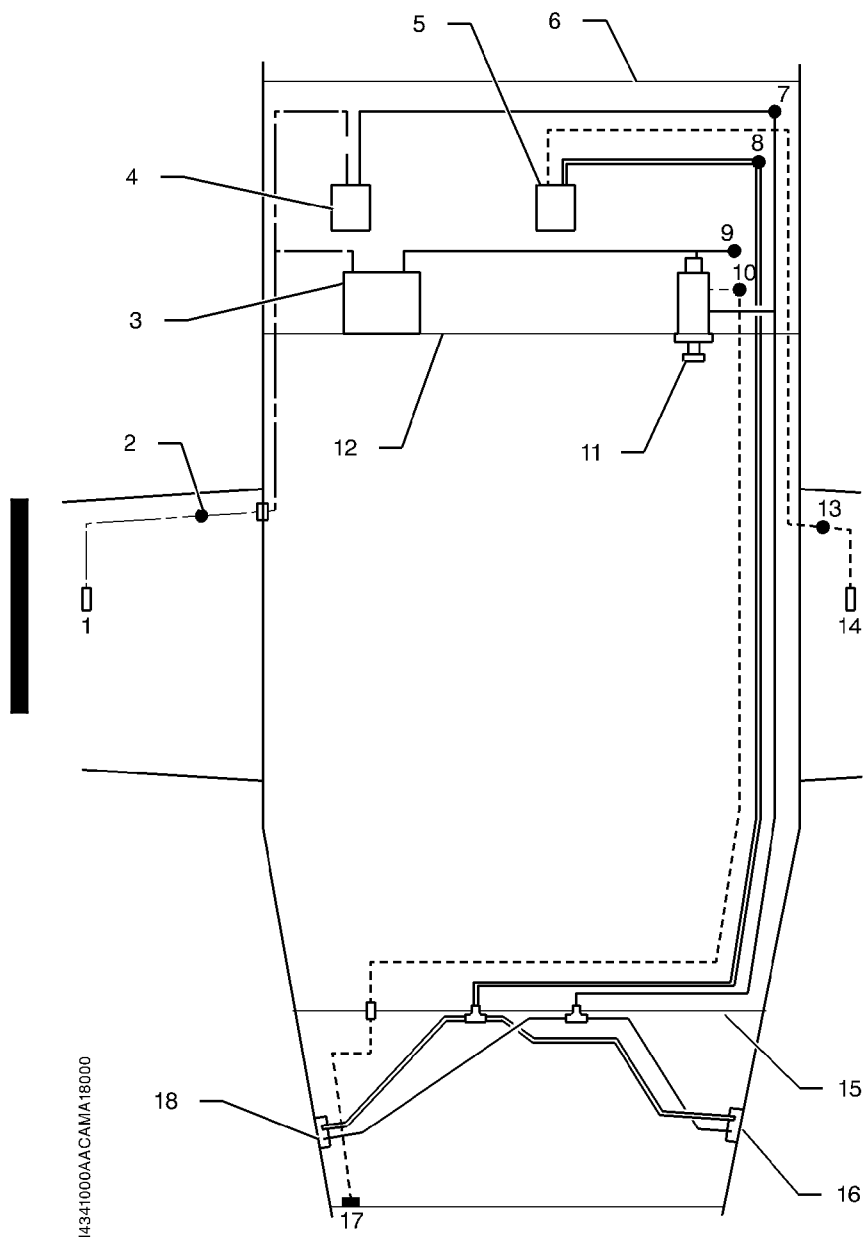


Figure 7.11.1 (2/2) - Air data system

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**7.12 - Vacuum system and instruments** - see figure 7.12.1

The airplane is fitted with a vacuum system providing the suction necessary to operate the cabin pressurization and the leading edge deicing.

Vacuum system includes :

- A pressure regulator
- An ejector
- A regulating and relief valve
- A pressure switch

Compressed air necessary for the ejector to create decompressed air is taken from the powerplant. The air flow is regulated before going into the ejector which creates necessary vacuum by venturi effect.

A relief valve fixed in cabin to frame C2, maintains the vacuum for pressurization system. In case of pressure drop, a pressure switch, installed in the system, indicates the failure by causing **VACUUM LOW** to light on.

**Standby attitude module (MD302)**

The Mid-Continent Instrument and Avionics MD302 Standby Attitude Module consists of two LCD screens. The first screen displays the airplane attitude (pitch and roll) and the second screen displays the airplane altitude and airspeed. The MD302 is powered from the ESSENTIAL BUS 2 or internal replaceable battery ensuring that the airplane can continue safe flight and landing in the event of a loss of primary attitude and air data displays. Pitot and static pressures are provided to the MD302 solid state electronic sensors using the airplane pitot probe and static sources.

The standby attitude module is located in the top left hand corner of the instrument panel.

- 1) Pressure regulator
- 2) Ejector
- 3) Valve
- 4) Regulating and relief valve
- 5) Pressure switch
- 6) Failure CAS message

Figure 7.12.1 (1/2) - Vacuum system



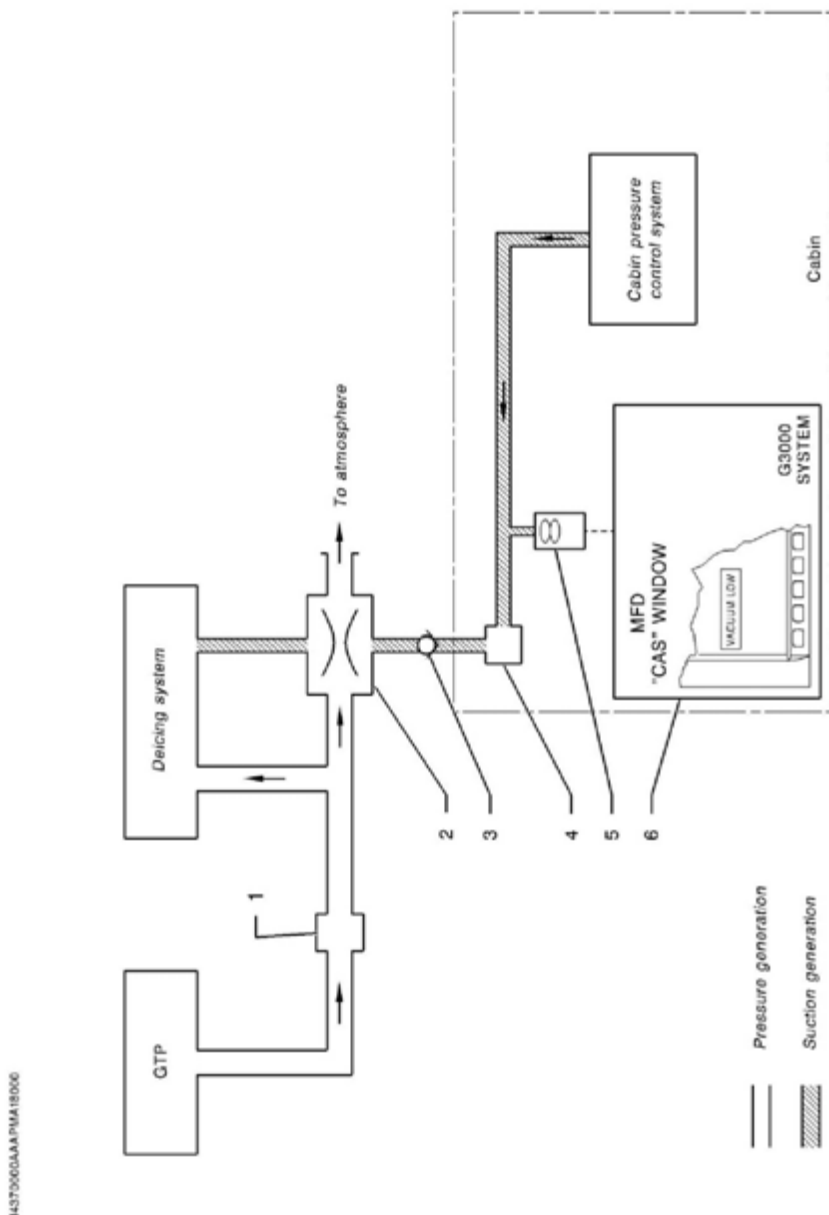


Figure 7.12.1 (2/2) - Vacuum system

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**7.13 - Ice protection equipment** - see figure 7.13.1

Ice protection equipment is as follows :

- Pneumatic deice system for inboard, central and outboard wing and for stabilizers : AIRFRAME DE ICE
- Propeller electrical deice system : PROP DE ICE
- Windshield electrical deice system : WINDSHIELD
- Electrical heating system for both pitots and for the stall warning sensor : PITOT L HTR and PITOT R & STALL HTR
- Turbine air inlet deice systems : INERT SEP

Deicing check and control panel is located on the lower L.H. side of the instrument panel.

**Wing and empennage deicing**

A pneumatic deice system assures protection of wing leading edges, horizontal stabilizer, elevator horns and vertical stabilizer. The system automatically cycles when AIRFRAME DE ICE switch is set to ON. The 67-second cycle breaks down in two inflation cycles :

- a first cycle induces inflation of leading edges deicer boots in wing central and outboard sections.
- the second cycle induces inflation of leading edges deicer boots in horizontal stabilizer, elevator horns, vertical stabilizer and wing inboard section,

During each inflation cycle, one of the two corresponding warning lights located above AIRFRAME DE ICE switch, remains illuminated.

Wing leading edge icing inspection light - see chapter 7.8 paragraph Exterior lighting.

## Propeller deicing

Propeller deicing is accomplished through electrical heating of blade roots. This system operates cyclically and alternately on the inboard and outboard zones of all blades. Each cycle is 180 seconds long. The system operation is correct when green warning light located above PROP DE ICE switch illuminates. The cycles continue as long as the switch remains set to ON.

**PROP DE ICE ON** illuminates if the engine is shut down with PROP DE ICE switch still ON.

### ▲ CAUTION ▲

When engine is shutdown, do not set the PROP DE ICE switch to ON for more than 10 seconds, damage to the propeller blades could result.



## Windshield deicing

The windshields are deiced electrically by integrated heating resistors. The system includes a controller and two heat probes embedded in each windshield. They are operated by the WINDSHIELD switch.

When the switch is positioned to ON, the controller supplies the heating resistors, the windshield temperature is monitored by probe 1. When the temperature reaches 45°C (113°F), the controller cuts the electrical supply to the heating resistors and resumes supply when the temperature falls below 30°C (86°F). The cycle continues as long as the switch remains set to ON.

In the event of failure of probe 1, the controller receives the temperature data from probe 2. The electrical supply to the heating resistors is cut when the windshield temperature reaches 56°C (133°F). In that case, the windshield is no longer heated, the pilot can reset the system by setting the switch to OFF, then to ON.

Two green lights located above the WINDSHIELD switch go on when the corresponding heating resistors are being supplied.

## Heating of pitots and stall warning sensor (PITOT L HTR and PITOT R & STALL HTR)

The two pitots, which supply ADCs, the airspeed indicator and the stall warning sensor are electrically heated. This deice equipment must be used even during flight into non-icing conditions.

The system condition messages **PITOT NO HT L** or **PITOT NO HT R**, **PITOT HT ON L** or **PITOT HT ON R**, **STALL HEAT ON** or **STALL NO HEAT** are displayed on the MFD CAS window. Refer to the GARMIN Pilot's guide for further details.

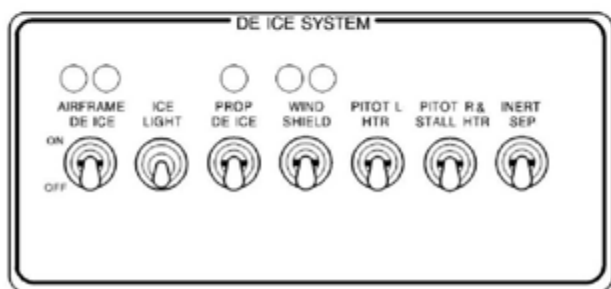
• NOTE •

Correct operation of the audible stall warning may be altered by severe or prolonged icing.

•

## Turbine air inlet protection

Operation and description are set forth in chapter 7.6 paragraph Engine air inlet.



14300001A/1G/M/8100

Figure 7.13.1 - Deicing control and check panel

## **7.14 - Miscellaneous equipment**

### **Stall warning system**

The airplane is equipped with an electrically deiced stall sensor in the leading edge of the right wing. This sensor fitted with a vane is electrically connected to an audible warning. The vane senses the change in airflow over the wing and operates the warning unit, which produces an aural warning alert. This warning alert begins no later than 5 knots above the stall in all configurations.

Simultaneously, the control wheel vibrates through the stick shaker.

The stall warning system should be checked during the preflight inspection by momentarily turning on the SOURCE selector and by manipulating the vane in the wing.

The stall warning system should also be checked during the preflight inspection by momentarily turning on the SOURCE selector and by depressing the TEST push-button on cockpit overhead panel.

The system is operational if a stall / stall aural warning alert is heard on the alarms speaker.

• NOTE •

The audible stall warning may be altered by severe or prolonged icing.

•

## **Static dischargers**

As an aid in flight, static dischargers are installed to improve radio communications during flight by reducing interference from dust or various forms of precipitations (rain, snow or ice crystals).

Under these conditions, the build-up and discharge of static electricity from the trailing edges of the wings (flaps and ailerons), rudder, stabilator, propeller tips and radio antennas can result in loss of usable radio signals on all communications and navigation radio equipment. Usually, the ADF is first and VHF communication equipment is the last to be affected.

Installation of static dischargers reduces interference from precipitation static, but it is possible to encounter severe precipitation static conditions which might cause the loss of radio signals, even with static dischargers installed. Whenever possible, avoid known severe precipitation areas to prevent loss of dependable radio signals. If avoidance is impractical, minimize airspeed and anticipate temporary loss of radio signals while in these areas.

## **Cabin fire extinguisher**

The fire extinguisher is located on R.H. front station side panel.

A pressure gage allows checking the fire extinguisher condition. Follow the recommendations indicated on the extinguisher.

## **Autopilot**

Autopilot control panel is located above the MFD. Refer to section 2 Limitations of this POH and to GARMIN Pilot's Guide for further details.

## **GPS**

GPS navigation is performed through the GARMIN system. Refer to section 2 Limitations and section 4 Normal procedures of this POH and to GARMIN Pilot's Guide for further details.



## **Weather radar GWX 70**

The weather information can be displayed on PFD1, PFD2 and MFD.

■ Refer to section 2 Limitations of this POH and to GARMIN Pilot's Guide for further details.

The controls for the weather radar are located on the touchscreen controllers.

- 1) MFD
- 2) Radar mode
- 3) Area of weather display
- 4) Antenna stabilization status
- 5) Touchscreen controllers
- 6) Scale for weather display

Figure 7.14.1 (1/2) - GWX 70 Weather radar display and controls

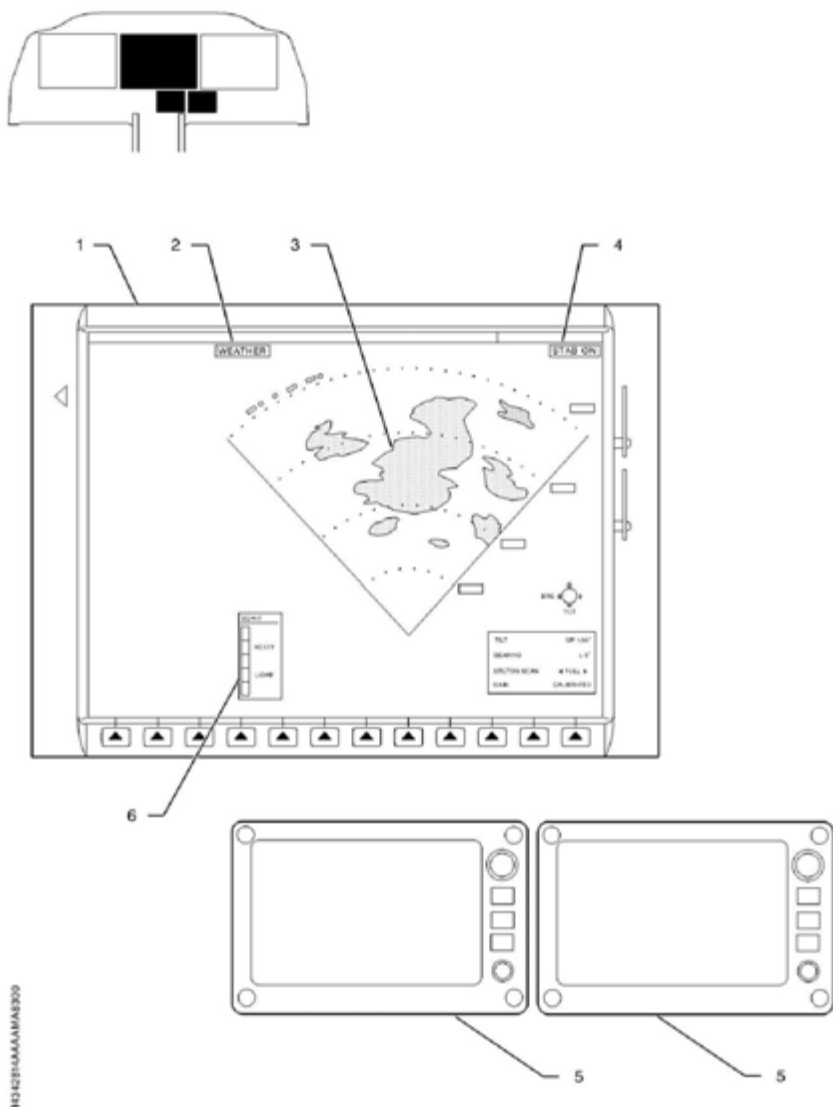


Figure 7.14.1 (2/2) - GWX 70 Weather radar display and controls

## Emergency locator transmitter

The airplane is equipped with an ELT ARTEX 1000 emergency locator transmitter which enables to locate it in case of distress. It is located in fuselage rear section with a service door on fuselage R.H. side.

The emergency locator transmitter assembly is constituted of a transmitter supplied by a battery, of an antenna attached on upper fuselage and of a remote control located on the upper panel.

● NOTE ●

For test sequences, refer to manufacturer manual.

●

Operation of the emergency locator transmitter is obtained as follows :

- from the instrument panel by setting ELT remote control switch to ON (locator transmitter ARM/OFF switch set to ARM/OFF),
- from the locator transmitter by setting its ARM/OFF control switch to ON,
- automatically in case of shock, when remote control switch is set to ARM/OFF and locator transmitter switch is set to ARM/OFF.

A red indicator light located on ELT remote control switch in the cockpit indicates to the pilot the emergency locator transmitter is transmitting.

A red indicator light located above locator transmitter switch and a buzzer located in the fuselage rear section indicate the emergency locator transmitter is transmitting.

▲ CAUTION ▲

Reset the ELT after an inadvertent activation.



● NOTE ●

The ELT cannot be reset if either the remote control switch or ELT switch is ON.

●

Reset procedure :

- 1) Set remote control switch or ELT switch to ON.
  - a) The ELT keeps on transmitting emergency signal.
  - b) On remote control box, red indicator light flashes.
  - c) On ELT, red indicator light flashes.
  - d) Near ELT, the buzzer sounds.
- 2) Wait approximately for 1 second.

- 3) Set remote control switch to ARM/OFF or ELT switch to ARM/OFF.
  - a) The ELT does not transmit emergency signal any longer.
  - b) On remote control box, red indicator light illuminates for about 1 second, then goes off.  
  
or
  - c) On ELT, red indicator light goes off.
  - d) Near ELT, the buzzer does no more sound.

Then ELT is reset.

*End of procedure.*

### **Lightweight data recorder (LDR 1000), if installed**

The airplane is equipped with a lightweight data recorder which is a crash-survivable system, recording both cockpit voices and flight data. These data are intended to be used after an accident or an incident.

The lightweight data recorder system includes a cockpit microphone located on instrument panel, between the standby instrument and the autopilot control panel.

The lightweight data recorder simultaneously records audio from the GMA audio control panel, audio from the cockpit microphone, data from the GASC, and data from the GIA integrated avionics unit 1 (GARMIN flight deck system).

The lightweight data recorder is powered from the BATT BUS and controlled by a printed circuit as follows :

- If the crash lever is set upward, the lightweight data recorder starts recording.
- If the crash lever is set downward, the lightweight data recorder goes on recording for 10 minutes (audio only) and then automatically stops recording.

## ADS-B OUT function

The ADS-B OUT function enables the airplane to broadcast data, such as position information, to ground stations and to other airplanes equipped with ADS-B IN system.

The loss of an interfaced input to the selected extended squitter transponder may cause the transponder to stop transmitting ADS-B OUT data. Depending on the nature of the fault or failure, the transponder may no longer be transmitting all of the required data in the ADS-B OUT messages.

*>> Airplane equipped with one extended squitter transponder*

ADS-B OUT data is only transmitted via transponder 1. Use of transponder 2 results in a loss of the ADS-B OUT data transmission.

If the transponder 1 detects any internal fault or failure with the ADS-B OUT functionality, the following advisory message XPDR1 ADS-B FAIL will be displayed on the PFDs.

After being informed of ADS-B OUT failure either by the advisory message XPDR1 ADS-B FAIL, or by Air Traffic Control, it is possible to disable ADS-B OUT function by selecting transponder 2, if installed.

*>> Airplane equipped with two extended squitter transponders*

ADS-B OUT data can be transmitted from any transponder upon pilot selection.

If the transponder 1 [2] detects any internal fault or failure with the ADS-B OUT functionality, the following advisory message XPDR1 ADS-B FAIL [XPDR2 ADS-B FAIL] will be displayed on the PFDs.

After being informed of ADS-B OUT failure either by the advisory message XPDR1 ADS-B FAIL [XPDR2 ADS-B FAIL] or by Air Traffic Control, it is possible to restore ADS-B OUT function by selecting transponder 2 [1].

## Flight deck information system (FS 210), if installed

The airplane is equipped with a flight deck information system allowing portable electronics devices to stream data to and from the GARMIN system.

For the system description and its utilization, refer to GARMIN Pilot's Guide.

## **GARMIN Integrated Flight Deck (GIFD) approaches**

The purpose of this section is to provide an overview of the GIFD capabilities and operation related to GIFD Approaches.

Detailed descriptions as well as operating instructions of these approaches are provided in the applicable Pilot's Guide and Cockpit Reference Guide.

### **RNP approaches operation**

The GIFD is capable of performing approaches with GNSS guidance also designated as RNP approaches.

#### **RNAV (GPS) or RNAV (GNSS) – LNAV, LNAV+V**

LNAV approaches provide lateral GPS-based guidance to legs defined by the navigation database.

Vertical deviations may be available if the necessary information to construct a vertical path is contained in the database.

Any vertical path information for LNAV approaches is strictly advisory.

There is no guarantee that stepdown fix altitudes will be honored and the crew must level off at the MDA if the runway is not visible.

LNAV approaches may be executed with or without SBAS, and advisory vertical guidance is dependent on sufficient GPS vertical error estimates rather than SBAS vertical integrity.

LNAV+V approaches do not downgrade in general because they do not require SBAS, although high GPS vertical error estimate anomalies could result in loss of advisory vertical guidance.

If GPS is lost the LNAV approach will be aborted.

#### **RNAV (GPS) or RNAV (GNSS) – L/VNAV**

LNAV/VNAV approaches add published vertical guidance in addition to LNAV guidance. They are different from LNAV+V in that the vertical deviations are not advisory, but rather published guidance.

The minimums of an LNAV/VNAV approach represent a DA rather than an MDA.

Execution of an LNAV/VNAV approach does not require SBAS integrity, as long as a system is configured to support barometric VNAV for approach.

If SBAS integrity is available, it will be used to provide vertical guidance.

During execution of a GPS approach with LNAV/VNAV service levels, while the aircraft is between the FAF and MAP, excessive deviation indicators appear as white vertical lines to indicate an area where the vertical deviation exceeds  $\pm 75$  feet.

If the glide-path indicator is within an area of excessive deviation, the glide-path indicator becomes yellow and the vertical lines also become yellow.

#### RNAV (GPS) or RNAV (GNSS) – LPV

LPV approaches provide both localizer precision lateral guidance and a vertical path definition.

SBAS integrity is required to execute the approach.

#### **Approach level downgrade**

Some automatic approach service downgrade may be performed automatically upon loss of SBAS or GPS approach alarm limits being exceeded, depending on the approach service level that has been loaded in the flight plan and activated.

This automatic downgrade is annunciated to the pilot through an APR DWNGRADE advisory message and a change in the annunciated service level in the HSI.

As the APR DWNGRADE advisory message may not be triggered under certain circumstances, the HSI annunciation shall be considered as the primary mean to annunciate any approach downgrade.

Under certain circumstances, when the GNSS integrity requirement are not met nor approach level is available the approach may be aborted.

This is annunciated through an ABORT APR advisory message and the service level annunciation being removed from the HSI.

If SBAS becomes unavailable on an RNAV LNAV/VNAV approach, **L/VNAV** is shown in yellow, the system will downgrade to LNAV service level (**LNAV** shown in magenta) and the APR DWNGRADE advisory message will be generated. The VDI will be flagged NO GP.

If SBAS becomes unavailable on an RNAV LPV approach, **LPV** will be shown in yellow, but the CDI and VDI will continue to be shown. At one minute to the FAF, an APR DWNGRADE advisory message will be generated.

The VDI will be flagged NO GP. Depending on the available lines of minima for the approach, the system will switch to either LNAV/VNAV or LNAV service level.



## Optional equipment

For optional equipment such as stormscope, SVS or TAWS, refer to section 9 Supplements.

Other optional equipment such as radio altimeter or chartview system or TAS are described in the GARMIN Pilot's Guide.

• NOTE •

Refer to section 2 Limitations for chartview system operating limitations.

•

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**Section 8****Handling, servicing and maintenance****Table of contents**

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## **8.1 - General**

This section contains the procedures recommended by the manufacturer for the proper ground handling and routine care and servicing of airplane. Also included in this section are the inspection and maintenance requirements which must be followed if your airplane is to retain its performance and dependability.

It is recommended that a planned schedule of lubrication and preventive maintenance be followed, and that this schedule be tailored to the climatic or flying conditions to which the airplane is subjected.

For this, see manufacturer maintenance manual.

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## **8.2 - Identification plate**

Any correspondence regarding your airplane should include its serial number. This number together with the model number, type certificate number and production certificate number are stamped on the identification plate attached to the left side of the fuselage beneath the horizontal stabilizer.

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### 8.3 - Publications

When the airplane is delivered from the factory, it is supplied with a POH, the GARMIN Integrated Flight Deck Pilot's Guide and supplemental data covering optional equipment installed in the airplane (refer to section 9 Supplements and pilot guides).

In addition, the owner may get access to the following publications online :

- Maintenance Manual
- Illustrated Parts Catalog
- Catalog of Service Bulletins, Service Letters

▲ **CAUTION** ▲

POH must always be in the airplane.



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## **8.4 - Inspection periods**

Refer to regulations in force in the certification country for information concerning preventive maintenance to be carried out.

A maintenance manual must be obtained prior to performing any preventive maintenance to make sure that proper procedures are followed. Maintenance must be accomplished by licensed personnel.

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## **8.5 - Alterations or repairs**

It is essential that the airworthiness authorities be contacted prior to any alterations or repairs on the airplane to make sure that airworthiness of the airplane is not violated. Alterations or repairs must be accomplished by licensed personnel.

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## 8.6 - Ground handling

### ▲ CAUTION ▲

Only move or tow the airplane with someone in the cockpit.



## Towing

### ▲ CAUTION ▲

Using the propeller for ground handling could result in serious damage, especially if pressure or pull is exerted on blade tips.



The airplane should be moved on the ground with a towing bar and a suitable vehicle in order not to damage the nose gear steering mechanism. Nose gear fork is equipped with an integrated towing fitting.

### ▲ CAUTION ▲

Do not tow the airplane when controls are secured.

When towing with a vehicle, do not exceed the nose gear turning angle, as this may result in damage to the gear and steering mechanism  
- see figure 8.6.1



## Parking

When parking the airplane, head it into the wind. Do not set the parking brake when brakes are overheated or during cold weather when accumulated moisture may freeze the brakes. Care should be taken when using the parking brake for an extended period of time during which an air temperature rise or drop could cause difficulty in releasing the parking brake or damage the brake system.

Make sure that the FUEL TANK SELECTOR is set to OFF.

### ● NOTE ●

Do not use solar screens or shields installed on the airplane inside, or leave sun visors down against windshield when airplane on ground. The reflected heat from these items causes a temperature increase which accelerates the crack growth or crazing and may cause the formation of bubbles in the inner layer of multilayer windshields.



Beyond 24 hours parking, use windshield protection screen provided with lateral and underside straps.

For long term parking, blanking covers (static ports, pitot, engine air inlet, NACAs, exhaust stubs), cockpit cover, tie-downs, wheel chocks, propeller lock and control lock are recommended.

In severe weather and high wind conditions, tie the airplane down as outlined in the following paragraph.

### **Tie-down**

Proper tie-down procedure is the best protection against damage to the airplane by gusty or strong winds. To tiedown the airplane securely, proceed as follows :

- Install control lock - see figure 8.6.2.
- Chock all wheels.
- Tie sufficiently strong ropes or chains to hold airplane down ; insert a rope in each tie-down hole located on flap hinge arm ; secure each rope to a ramp tie-down or to mooring rod.
- Check that doors are closed and locked.





Figure 8.6.1 - Turning angle limits

## **Jacking**

When it is necessary to jack the airplane off the ground, refer to maintenance manual for specific procedures and equipment required.

## **Leveling**

Level the airplane as described in maintenance manual.

## **Flyable storage (below 28 days)**

Airplanes placed in storage for a maximum of 28 days are considered in flyable storage.

Storage from 0 to 7 days :

- Engine : according to maintenance manual P & W C.

Airplane fueling :

- Keep fuel tanks full to minimize condensation in the tanks. Keep the battery fully charged to prevent the electrolyte from freezing in cold weather.  
Close oxygen cylinder isolation valve.

Storage from 8 to 28 days :

- Engine : according to maintenance manual P & W C.

Airplane fueling :

- Keep fuel tanks full to minimize condensation in the tanks. Keep the battery fully charged to prevent the electrolyte from freezing in cold weather.  
Close oxygen cylinder isolation valve.

Battery, remaining in the airplane or removed :

- Disconnect battery and check its charge level at regular intervals.

## **Long term storage without flying (over 28 days)**

Refer to maintenance manual for the procedures to follow.

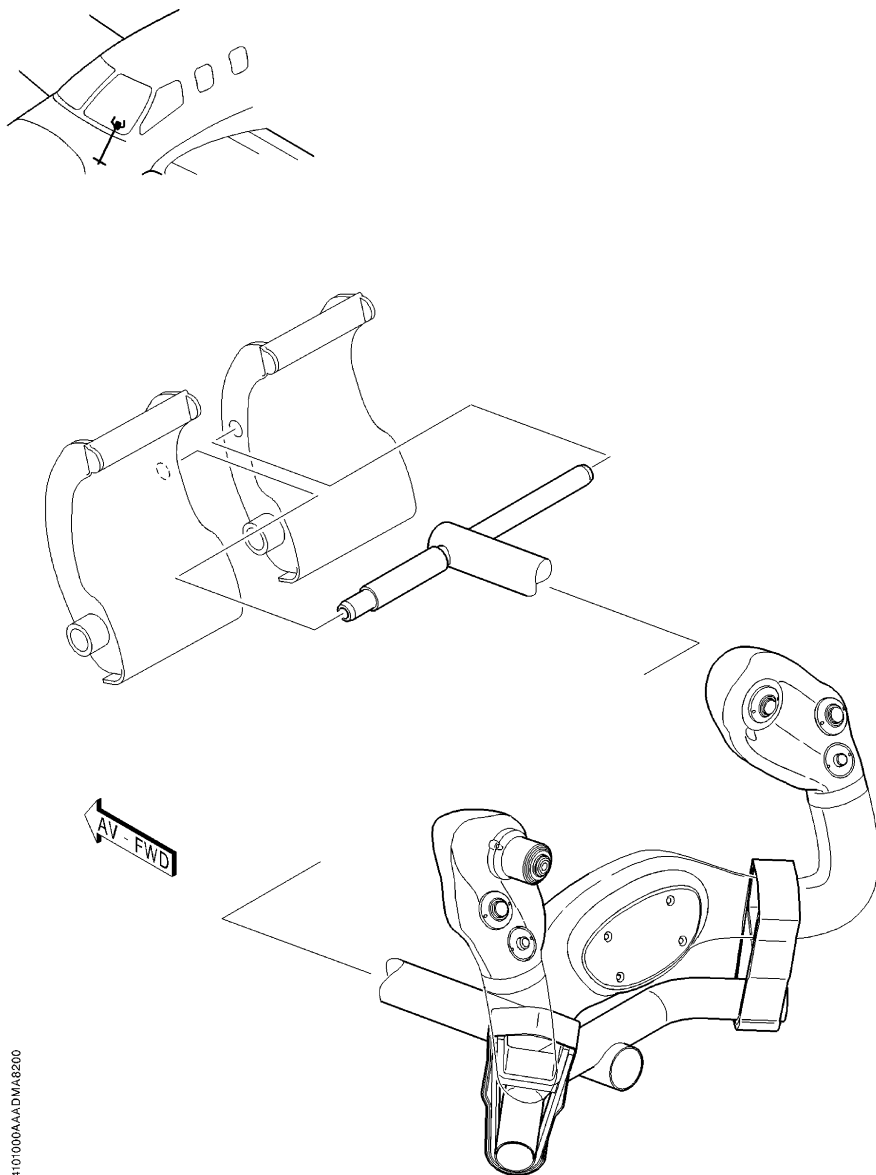


Figure 8.6.2 - Control lock device

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## 8.7 - Servicing

### Maintenance

In addition to the preflight inspection, refer to section 4, Normal procedures, servicing, inspection and test requirements for the airplane are detailed in the maintenance manual.

Maintenance manual outlines all items which require servicing, inspection, testing or overhaul.

### Engine oil

#### Type of oil

▲ **CAUTION** ▲

Do not mix different viscosities or specifications of oil as their different chemical structure can make them incompatible.



#### Specification

| Nominal Viscosity | Specification  | NATO Code                  |
|-------------------|----------------|----------------------------|
| 5cSt              | MIL-PRF-23699G | O-156 (STD)<br>O-154 (HTS) |

Figure 8.7.1 - Recommended engine oil types

Reference : Service Bulletin P & W C. No. 14001 at the latest revision

#### Oil capacity

System total capacity :

12.7 Quarts (12 Litres) (oil cooler included)

Usable capacity :

6 Quarts (5.7 Litres)

#### Servicing

The engine oil should be changed and the oil filter cleaned/replaced at intervals recommended in Pratt & Whitney Canada Engine Maintenance Manual (EMM) (Ref. chapter 72-00-00, table 601, periodic inspection).

### Oil level check

To avoid overfilling of oil tank, and high oil consumption, an oil level check is recommended within 30 minutes after engine shutdown. Ideal interval is 15 to 20 minutes. If more than 30 minutes have passed and the dipstick indicates that oil is needed, start the engine and run at LO-IDLE for five minutes, and recheck oil level.

Check oil level against marking on dipstick and top-up as required. Normal oil level is between MAX HOT and one US quart (0.83 Imp. Quart, 0.95 litres) below MAX HOT, with engine in horizontal attitude.

• NOTE •

Filling the oil to the maximum level may result in high consumption rate, with the oil exiting through the accessory gearbox breather.



▲ CAUTION ▲

When filler cap assembly is installed and locked, no movement is allowed.



### Fuel

**Total capacity each tank : 150.5 USG (570 l).**

• NOTE •

To minimize condensation, it is recommended that airplane be refueled after each flight, respecting weight and balance limits.



▲ CAUTION ▲

Never fly the airplane with contaminated (water, sand, rust, dust...) or unapproved fuel



Before each flight and after each fueling, use a sampler to bleed off some fuel through each tank and fuel filter drain to detect possible contamination and be sure that fuel used is the proper quality. If contamination is present, continue draining through all draining points until fuel is free of contamination. If quality of fuel used is not correct, defuel airplane completely and refuel with proper quality fuel.

**▲ WARNING ▲**

**During all fueling operations, fire fighting equipment must be available ; attach grounding wire to an unpainted metallic part of the airplane.**

**Do not operate any avionics or electrical equipment on the airplane during fueling. Do not allow open flame or smoking in the vicinity of the airplane while fueling.**



**▲ CAUTION ▲**

During fueling operations, take care not to damage pneumatic deicer boots located on wing leading edge.

The use of aviation gasoline (AVGAS) must be restricted to emergencies only. AVGAS will not be used for more than 150 cumulative hours during any period between engine overhaul.



**• NOTE •**

Use of AVGAS must be recorded in engine module logbook.



| US Specification<br>(US)                                  | French Specification<br>(FR) | English Specification<br>(UK) | NATO Code                          |
|---|------------------------------|-------------------------------|------------------------------------|
| ASTM-D1655 JET A<br>ASTM-D1655 JET A1<br>ASTM-D1655 JET B | AIR 3405C Grade F35          | DERD 2494 Issue 9             | F35 without additive               |
| MIL-DTL-5624<br>Grade JP-4                                | AIR 3407B                    | DERD 2454 Issue 4<br>Amdt 1   | F40 with additive                  |
| MIL-DTL-5624<br>Grade JP-5                                | AIR 3404C Grade F44          | DERD 2452 Issue 2<br>Amdt 1   | F44 with additive when utilization |
| MIL-DTL-83133<br>Grade JP-8                               | AIR 3405C Grade F34          | DERD 2453 Issue 4<br>Amdt 1   | F34 with additive S748             |
|   | AIR 3404C Grade F43          | DERD 2498 Issue 7             | F43 without additive               |

Figure 8.7.2 - Recommended fuel types  
Reference : Service Bulletin P & W C. No. 14004

### Fuel additives

Fuel used must contain an anti-ice additive conforming to MIL-I-27686 or MIL-I-85470 specification.

Strict adherence to recommended preflight draining instructions as called for in Section 4 will eliminate any free water accumulations from the tank sumps. While small amounts of water may still remain emulsified in the gasoline, it will normally be consumed and go unnoticed in the operation of the engine.

One exception to this can be encountered when operating under the combined effect of use of certain fuels, with high humidity conditions on the ground followed by flight at high altitude and low temperature. Under these unusual conditions, small amounts of water emulsified can precipitate from the fuel stream and freeze in sufficient quantities to induce partial icing of the engine fuel system.

While these conditions are quite rare and will not normally be a problem to owners and operators, they do exist in certain areas of the world and consequently must be dealt with, when encountered.

Therefore, to alleviate the possibility of fuel icing occurring under these unusual conditions, it is required to add an ethylene glycol monomethyl ether (EGME or DIEGME) compound to the fuel supply.

The introduction of an EGME or DIEGME compound into the fuel provides two distinct effects :

- it absorbs the dissolved water from the fuel
- alcohol has a freezing temperature depressant effect.

EGME or DIEGME must be carefully mixed with the fuel in concentration, it must be between a minimum of 0.06 % and a maximum of 0.15 % by volume. Figure 8.7.3 provides EGME or DIEGME / fuel mixing ratio information.

#### ▲ CAUTION ▲

Do not permit the concentrate of EGME or DIEGME to come in contact with the airplane finish or fuel tank

Mixing of the EGME or DIEGME with the fuel is extremely important. An excessive concentration (greater than 0.15 % by volume maximum) will result in detrimental effects to the fuel tanks by deterioration of protective primer, sealants and seals of system and engine components. Use only blending equipment recommended by the manufacturer to obtain proper proportioning.





Prolonged storage of the airplane will result in a water buildup in the fuel which leeches out the additive. An indication of this is when an excessive amount of water accumulates in the fuel tank sumps. The concentration can be checked using a differential refractometer. It is imperative that the technical manual for the differential refractometer be followed explicitly when checking the additive concentration.

**Fuel and fuel additives in Ukraine and CIS countries**

It is possible to use kerosene GOST 10227 RT with addition of anti-icing liquid :

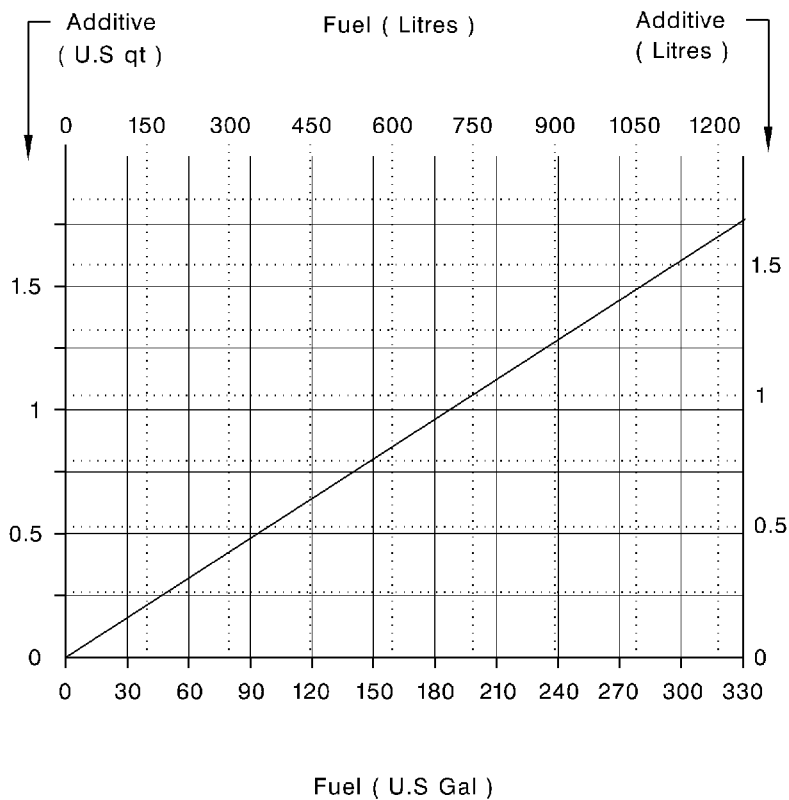
- liquid И - GOST 8313-88

Above-mentioned liquid is added in the quantity equal to 0.3 percent per volume.

▲ **CAUTION** ▲

Refer to Service Bulletin P & WC No. 14004 at its latest revision for appropriate quantities.





I4284000AAEMA8000

Figure 8.7.3 - Additive mixing ratio (EGME or DIEGME)

## **Landing gear**

### **Nose gear tire**

5.00-5 10 PR - Inflation pressure : 98 psi (6.7 bars) \*

### **Main gear tires**

18 5.5 10 PR - Inflating pressure : 135 psi (9.32 bars) \*

### **Nose gear shock absorber**

Fill with hydraulic fluid AIR 3520 B (MIL.H5606E) ; inflate with nitrogen to 87 psi (6 bars).

### **Main gear shock absorbers**

Fill with hydraulic fluid AIR 3520 B (MIL.H5606E) ; inflate with nitrogen to 160 psi (11 bars).

### **Hydraulic system**

Check every 100 hours and service with AIR 3520 B (MIL.H5606E) hydraulic fluid.

### **Brakes**

Service as required with AIR 3520 B (MIL.H5606E) hydraulic fluid.

• NOTE •

A higher inflation pressure has to be applied to tires and shock absorbers when in very cold conditions - refer to chapter 8.9.

•

- (\*) Tire inflation pressures are given for an airplane on ground at 21° C.  
An ambient temperature change of 3° C produces approximately 1 % pressure change.

Oxygen

The replenishment device of the oxygen cylinder is installed directly on the cylinder head. It consists of a charging valve and of a pressure gage graduated from 0 to 2000 PSIG. A chart - see figure 8.7.4, located on the inside of the cylinder service door, gives the cylinder charge maximum pressure according to the environment temperature.

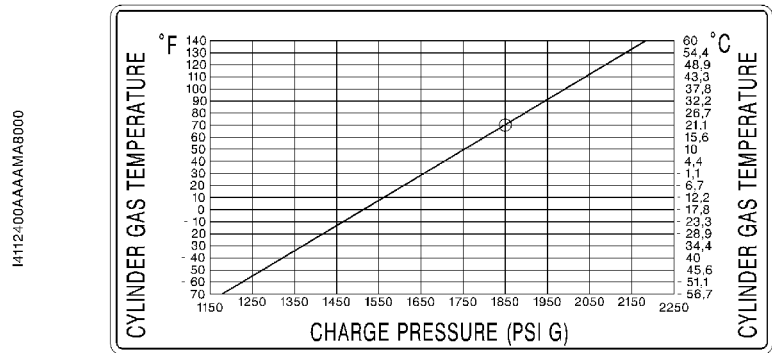


Figure 8.7.4 - Charge pressure chart

**Replenishment procedure****▲ WARNING ▲**

**Make sure that the airplane is fitted with a grounding cable and is properly grounded.**

**The oxygen cart must be electrically bonded to the airplane.**

**Do not operate the airplane electrical switches or connect/disconnect ground power during oxygen system replenishment.**

**Do not operate the oxygen system during refueling/defueling or perform any other servicing procedure that could cause ignition.**

**Introduction of petroleum based substances such as grease or oil to oxygen creates a serious fire hazard. Use no oil or grease with the oxygen replenishment equipment.**

**Always open shut-off valve slowly to avoid generating heat and replenish the system slowly at a rate not exceeding 200 PSIG (13.7 bars) per minute.**

**▲ CAUTION ▲**

Replenishment of the oxygen system should only be carried out by qualified personnel.

**● NOTE ●**

The cylinder full charge is assured for a pressure of 1850 PSIG (127 bars) at a temperature of 70° F (21° C). If the cylinder temperature differs from 70° F (21° C), refer to figure 8.7.4 which lists the required pressures according to the cylinder temperature.



- 1 - Open the oxygen service door on the R.H. rear karman.
- 2 - Measure the oxygen cylinder temperature.
- 3 - Make sure the thermometer indication is constant. Note the indication.
- 4 - Refer to the temperature/pressure chart for the correct oxygen cylinder pressure.

*If the pressure on the oxygen cylinder gage is low :*

Fill the oxygen cylinder

- 5 - Make sure the area around the oxygen cylinder charging valve is clean. Remove the cap from the charging valve.
- 6 - Make sure the oxygen supply hose is clean and connect it to the charging valve.
- 7 - Slowly pressurize the oxygen cylinder to the correct pressure.
- 8 - Close the oxygen supply and let the cylinder temperature become stable.
- 9 - Monitor the oxygen pressure on the gage and fill to the correct pressure if necessary.
- 10 - Release the pressure in the oxygen supply hose and disconnect from the charging valve.
- 11 - Install the cap on the charging valve.
- 12 - Make sure all the tools and materials are removed and the work area is clean and free from debris.
- 13 - Close the oxygen service door.

**Passenger masks repacking instructions****▲ CAUTION ▲**

Do not use oil or other petroleum based lubricants on passenger oxygen mask or deployment container. Oil based lubricants are a fire hazard in oxygen-rich environments.

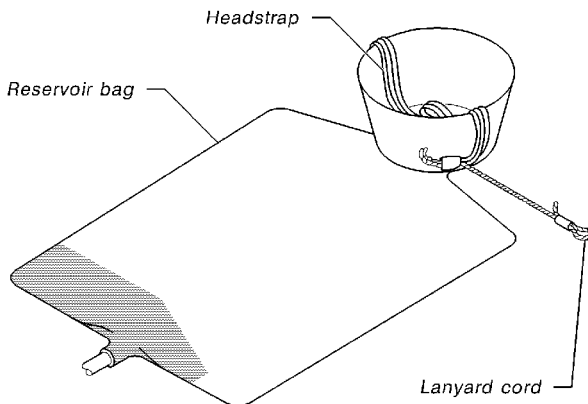
Repacking procedures shall be performed by personnel familiar with the instructions and warnings in this document. Improperly packed masks can damage the masks or result in failure of the masks to deploy.

**▲ WARNING ▲**

**Masks shall be repacked in an area free of oil, grease, flammable solvents or other contaminants.**

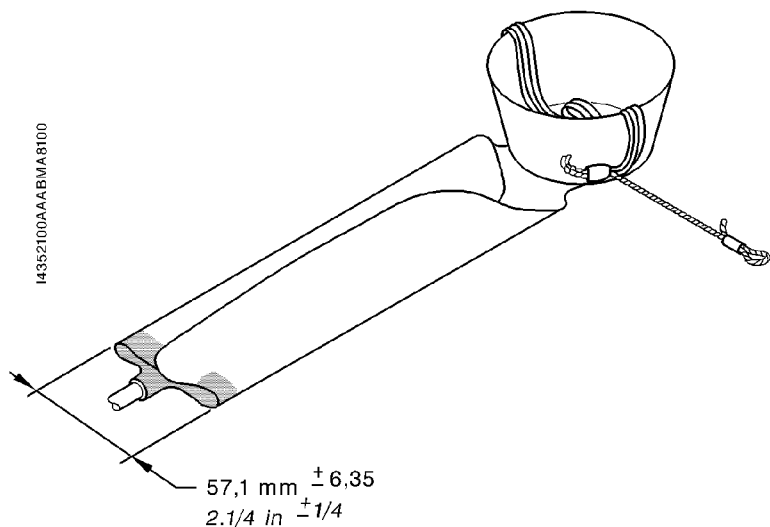


- 1 - Inspect and disinfect mask and deployment container with an aqueous solution of Zephiran Chloride (Scott Aviation P/N 00-2572) or with disinfection cleaners (EROS P/N SAN50). After disinfecting and thoroughly drying the mask, lightly dust the outside of the facepiece with Neo-Novacite powder (Scott Aviation P/N 00-736). Contamination can be removed with mild soap and water solution.
- 2 - Fold headstrap into facepiece. Pull lanyard cord out to side of facepiece so that it does not interfere with repacking.
- 3 - Lay reservoir bag on flat surface and smooth out wrinkles.



14352100AAABMA8300

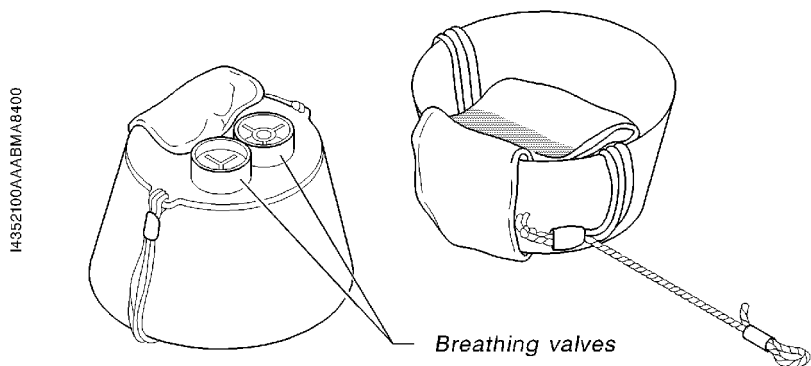
- 4 - Gently fold reservoir bag lengthwise into thirds (outside edges folded inward over center of bag). Do not crease bag.



- 5 - Fold reservoir bag away from breathing valves and into facepiece. Make sure bag does not cover breathing valves.

*Top view*

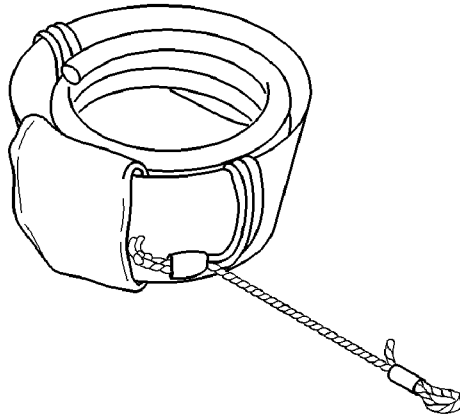
*Bottom view*





- 6 - Coil oxygen tubing inside facepiece over reservoir bag.

I4352100AAABMA18000



- 7 - Connect oxygen tubing to manifold oxygen fitting.

**▲ WARNING ▲**

**Make sure lanyard pin is inserted into correct check valve for mask being installed. Cross connected pins will result in passengers pulling lanyard cords only to initiate oxygen flow to another mask.**



- 8 - Insert lanyard pin into corresponding check valve.
- 9 - Place mask facepiece - first in deployment container. Make sure that oxygen tubing and lanyard cord are free to deploy and are not caught between the container and lid.
- 10 - Close and latch deployment container lid.

Intentionally left blank

## 8.8 - Airplane cleaning and care

### Windshield and windows

The windshield and windows should be cleaned with an airplane windshield cleaner.

• NOTE •

Refer to the maintenance manual for products and procedures to apply.



Apply the cleaner sparingly with soft cloths and rub with moderate pressure until all dirt, oil scum and bug stains are removed. Allow the cleaner to dry, then wipe it off with soft flannel cloth.

▲ CAUTION ▲

Do not use any of the following products on, or for cleaning windows :  
methanol, methylated alcohol, gasoline, benzene, xylene,  
methyl-ethyl-ketone, acetone, carbon tetrachloride, lacquer paint  
thinners, commercial or household window cleaning sprays. In case of  
doubt concerning a product, do not use it.

During cleaning operation, avoid wearing objects such as ring, watch,  
bracelet and exercise care to prevent buttons, buckles and any hard  
objects from touching the windshield and the windows.

Adhesive tapes other than Minnesota 3m type 670 shall not be used on  
acrylic surfaces.

Never use buffing machines as excessive forces or speeds might  
produce redhibitory defects.



Follow by carefully washing with a mild detergent and plenty of water. Rinse  
thoroughly, then dry with a clean moist chamois. Do not rub the plastic with a dry cloth  
since this builds up an electrostatic charge which attracts dust. Waxing will finish the  
cleaning operation. A thin, even coat of wax polished out by hand with clean soft  
flannel cloth will fill in minor scratches and help prevent further scratching.

Do not use a canvas cover on the windshield unless freezing rain or sleet is  
anticipated since the cover may scratch the plastic surface.

### Painted surfaces

Refer to maintenance manual for the products and procedures to apply.

## **Propeller care**

Preflight inspection of propeller blades for nicks and cleaning them occasionally with a cloth soaked with soapy water to clean off grass and bug stains will assure long blade life. Never use an alkaline cleaner on the blades ; remove grease and dirt. Refer to maintenance manual for the procedures to follow.

## **Engine care**

Refer to maintenance manual for the procedures to follow.

## **Interior care**

To remove dust and loose dirt from the upholstery and carpet, clean the interior regularly with a vacuum cleaner.

For additional information, refer to maintenance manual.

**■ 8.9 - Preparation of the airplane (equipment and furnishings)****▲ WARNING ▲**

**In any accommodation, make sure access to emergency exit is free.**

**▲ CAUTION ▲**

Removed equipment items must be stowed in a place which ensures their integrity.



Many accommodations are authorized by airplane manufacturer. They are enumerated in section 7.

This procedure specifies how to change your 6-seat accommodation into 4-seat accommodation, and conversely. However, it can be used partly to remove or install an equipment item.

However, the pilot must ensure that he gets all necessary authorizations from his regulatory authority.

**1 - Conversion of 6-seat accommodation into 4-seat accommodation - see figures 8.9.1, 8.9.2, 8.9.3 and 8.9.4**

**A - Tools and consumable materials**

- Seat protective covers

**B - Preparation**

- 1) Make sure the SOURCE selector is set to OFF and the crash lever is down.

**C - Removal of rear seats - see figure 8.9.1**

- 1) To remove rear seats, perform the following operations

**▲ CAUTION ▲**

In order to prevent cushion covering damage, protective covers should be put on seats.



- a) Install protective covers.

- b) Unlock backrest using backrest tilting handle (6) and fold it forward.

• NOTE •

For the R.H. rear seat, backrest tilting handle is located behind backrest.



- c) Unlock seat using seat tilting handle (1) and tilt it forward.
- d) Clear the carpet from under the seat to facilitate moving in rails.
- e) Hold the seat in tilted position and unscrew quick links (7) of strap (9) located under L.H. seatpan.

• NOTE •

This operation is specific to L.H. seat.



- f) Pull up and hold L.H. and R.H. rings (2), and turn knobs (8) by 90° in order to release and keep locks (3) in up position.
- g) Move the seat in the rails to line up pads (4) with rail (5) apertures.
- h) Remove the seat.

• NOTE •

Ensure proper storage of strap (9) with L.H. rear seat to avoid losing part.



D - Removal of intermediate seats - see figures 8.9.2 and 8.9.3

- 1) To remove intermediate seats, perform the following operations
  - a) Install protective covers.
  - b) Pull backrest bottom upholstery (25) to remove it.
  - c) Clear the carpet from under the seat to facilitate moving in rails.
  - d) Pull up locking handle (21) located under the pan, on the seat rear side, to unlock it.
  - e) Move the seat in the rails to line up pads (23) with rail (24) apertures.
  - f) Remove the seat.
  - g) Install backrest bottom upholstery (25).

**▲ CAUTION ▲**

In order to prevent deflector's damage, it is necessary to remove them.



- 2) Remove deflector (34) maintained with Velcro-type strap.
- 3) If necessary, remove the cabin central carpet.

**● NOTE ●**

If one of two cargo nets must be installed, it is necessary to use the carpet with appropriate cuttings.



E - Removal of a cabinet

**● NOTE ●**

This operation must be carried out by a service center.



F - Cabin comfort - see figure 8.9.3

- 1) Blank off the hot air outlet, located forward the large door, with blanking device assy (33) stored in storage bag - see figure 8.9.3 detail A.
- 2) Remove blanking plugs (32) located forward the large door and store them into storage bag - see figure 8.9.3 detail B.
- 3) Remove blanking plugs (31) located in line with R.H. front side window - see figure 8.9.3 detail C, and install them on holes located in line with card table - see figure 8.9.3 detail D.

G - Installation of intermediate seats - see figures 8.9.2, 8.9.3 and 8.9.4

- 1) Install deflector (34), ensuring that both red marks (36) are aligned with the deflector holes (35) - see figure 8.9.4.

**● NOTE ●**

Position deflectors (34) as indicated on label, according to future position of intermediate seat.



- 2) Install intermediate seats.

**● NOTE ●**

If seats are installed facing flight direction (frontwards), the L.H. seat must be installed on the right and the R.H. seat on the left in order to have the armrest on aisle side.



- a) Pull backrest bottom upholstery (25) to remove it.

- b) Clear the carpet from seat area to facilitate moving in rails.
- c) Position the seat and put lock (22) near the color mark (37) made on rail bottom on aisle side.

• NOTE •

The color mark (37) in the rail is aligned with red marks (36).



- d) Pull up locking handle (21), insert pads (23) into rail (24) apertures and then, move the seat so that lock (22) is in front of the color mark (37).
- e) Release locking handle (21) to lock the seat.

▲ **WARNING** ▲

**Verify that lock (22) and all pads (23) are engaged and locked into rails, trying to move seat forward and backward.**



- f) Install backrest bottom upholstery (25).

• NOTE •

Adjust it properly; make sure not to obstruct deflector (34) outlet.



- g) Slide properly the carpet under the seat.
- h) Remove protective covers.

H - Final operations

- 1) If removed, install cabin central carpet suited to the intended use.

• NOTE •

Slide properly the carpet under doorstep.



- 2) If necessary, remove the baggage compartment partition net and install the small or large cargo net - refer to section 7.
- 3) Make sure the work area is clean and free from debris.
- 4) Determine weight and balance - refer to section 6.



**2 - Conversion of 4-seat accommodation into 6-seat accommodation - see figures 8.9.1, 8.9.2, 8.9.3 and 8.9.4****A - Tools and consumable materials**

- Seat protective covers

**B - Preparation**

- 1) Make sure the SOURCE selector is set to OFF and the crash lever is down.
- 2) If installed, remove the cargo net.
- 3) Remove intermediate seats – refer to paragraph 1.D.
- 4) Remove the deflectors (34) maintained with Velcro-type strap.
- 5) If necessary, remove the cabin central carpet.

**C - Cabin comfort - see figure 8.9.3**

- 1) Remove blanking plugs (32) from their storage bag and install them on holes located forward the large door - see figure 8.9.3 detail B.
- 2) Remove blanking device assy (33) from the hot air outlet, located forward the large door, and store it into storage bag - see figure 8.9.3 detail A.
- 3) Remove blanking plugs (31) located in line with card table - see figure 8.9.3 detail D, and install them on holes located in line with R.H. front side window - see figure 8.9.3 detail C.

**D - Installation of cabinet****• NOTE •**

This operation must be carried out by a service center.

•

**E - Installation of intermediate seats**

- 1) Install intermediate seats – refer to paragraph 1 G.
- 2) If removed, install the baggage compartment partition net.
- 3) If removed, install cabin central carpet.

F - Installation of rear seats - see figure 8.9.1

- 1) Make sure the work area is clean and free from debris.
- 2) Clear the carpet from seat area to facilitate moving in rails.
- 3) Check that knobs (8) maintain locks (3) in up position.
- 4) Position the seat, fold it forward, refer to detail B, and insert pads (4) into rail (5) apertures.
- 5) Move the seat so that locks (3) are in front of the color mark made on rail bottom.
- 6) Pull up and hold L.H. and R.H. rings (2) and turn knobs (8) by 90° in order to insert locks (3) into rail (5) apertures.
- 7) Make sure the seat is correctly locked on rails (5).
- 8) Tilt seat forward, hold it and slip strap (9) around the locking control hinge pin. Screw quick links (7).
- 9) Tilt the seat rearward and lock it using seat tilting handle (1).
- 10) Fold up the backrest and lock it using backrest tilting handle (6).
- 11) Slide properly the carpet under the seat.
- 12) Remove protective covers.

G - Reconditioning

- 1) Make sure the work area is clean and free from debris.
- 2) Determine weight and balance - refer to section 6.

3 - **Additional configurations**

**▲ WARNING ▲**

**Removed seats can only be installed at their original location.  
Rear seat (L.H. or R.H.) is the only one which can be installed in  
cabin axis, on both central rails – refer to section 7.**



● NOTE ●

Many combinations of accommodations are authorized with seats (rear and intermediate) by pilot or service centers and cabinet(s) by service centers only. However, the pilot must ensure that he gets all necessary authorizations from his regulatory authority.



## ● NOTE ●

To remove or install these elements, use paragraph 1 or 2 – refer to table 1.

●

## ● NOTE ●

After these operations, determine weight and balance with the new C.G. - refer to section 6.

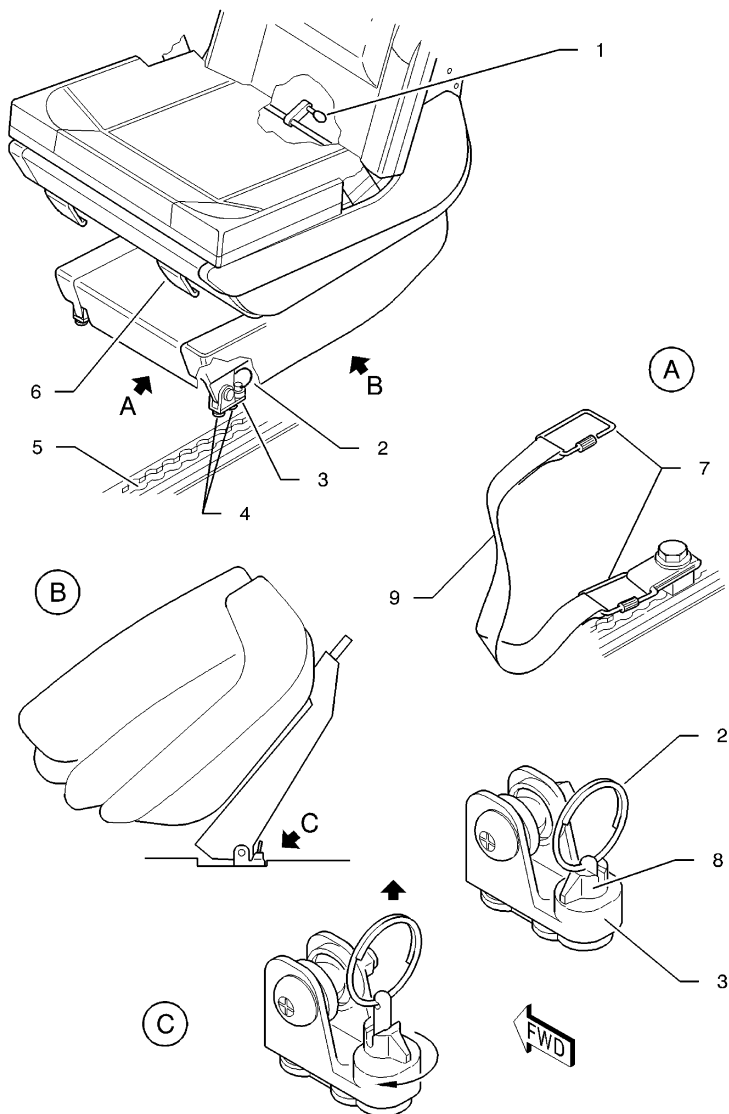
●

| Equipment         | Action       | Description operation |
|-------------------|--------------|-----------------------|
| Rear seat         | Removal      | Paragraph 1.C.        |
|                   | Installation | Paragraph 2. F.       |
| Intermediate seat | Removal      | Paragraph 1.D.        |
|                   | Installation | Paragraph 1.G.        |
| Cargo net         | Installation | Section 7             |

Table 1

- 1) Seat tilting handle
- 2) Ring
- 3) Lock
- 4) Pad
- 5) Rail
- 6) Backrest tilting handle
- 7) Quick link
- 8) Knob
- 9) Strap

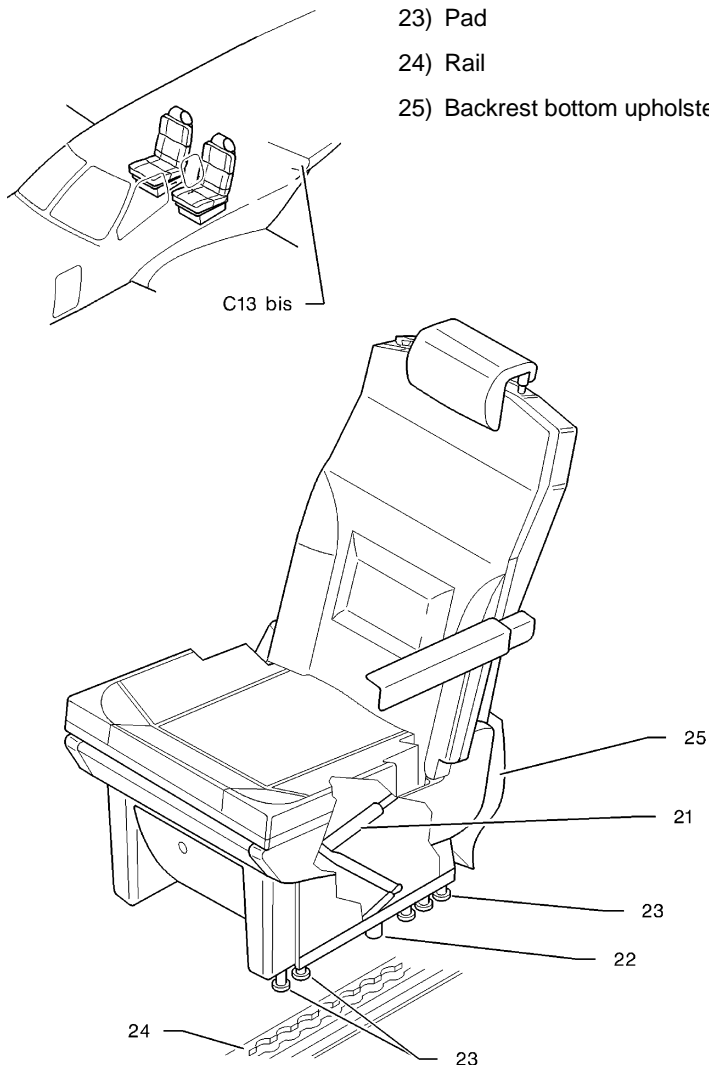
Figure 8.9.1 (1/2) - Removal / installation of rear seat



I42520202AAGMA8000

Figure 8.9.1 (2/2) - Removal / installation of rear seat

- 21) Locking handle
- 22) Lock
- 23) Pad
- 24) Rail
- 25) Backrest bottom upholstery



14252201AAA/ASMA8100

Figure 8.9.2 - Removal / installation of intermediate seat

- 31) Blanking plug
- 32) Blanking plug
- 33) Blanking device assy
- 34) Deflector

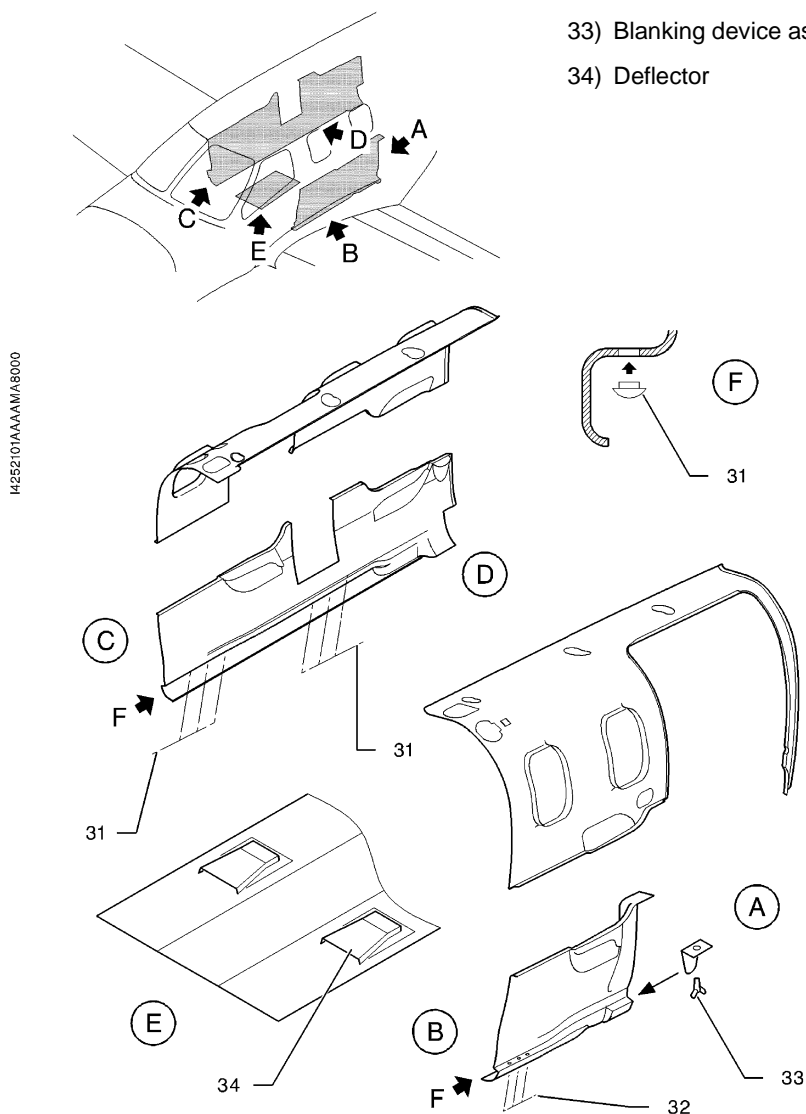


Figure 8.9.3 - Cabin comfort - Installation of blanking plugs and deflector

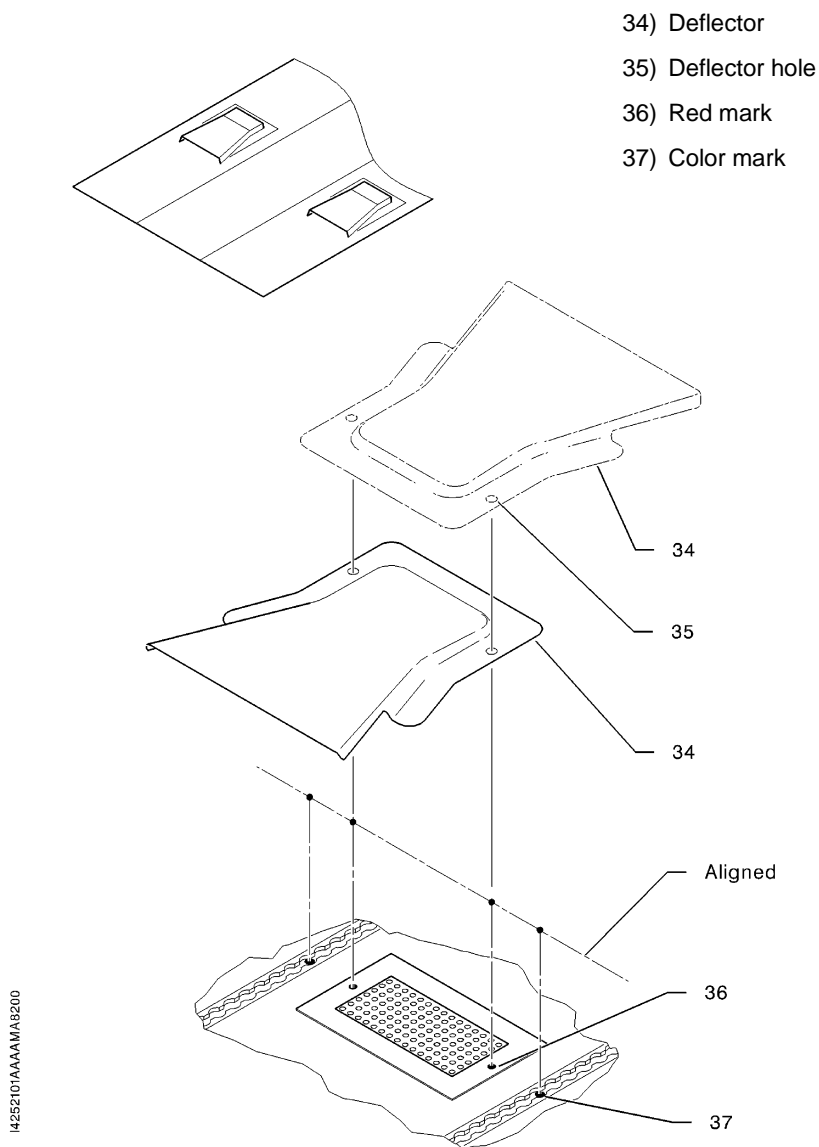


Figure 8.9.4 - Cabin comfort - Installation of deflector



**8.10 - Utilization by cold weather (- 0°C to - 25°C) or very cold weather (- 25°C to - 40°C)**

## ● NOTE ●

Check pressure values in a hangar heated at about 15°C with control equipment at room temperature.

●

If a landing is foreseen by cold or very cold weather or in case of airplane prolonged operation in such conditions, it is recommended to prepare the airplane as follows :

- 1 - Smear with silicone grease the door and engine cowlings seals, as well as the leading edge deicers.
- 2 - Apply engine oil on the engine cowlings latches.
- 3 - Inflate main landing gear shock absorbers to 247 psi (17 bars) at a room temperature of 15°C.
- 4 - Position a 0.59 in (15 mm) shim at the bottom of the piston tube and against forward landing gear half-fork to reduce shock absorber travel. Refill with hydraulic liquid. Remove the shim and inflate shock absorber to 138 psi (9.5 bars) at a room temperature of 15°C.
- 5 - Inflate main landing gear tires to 130 psi (8.96 bars) and nose tire to 102 psi (7 bars) at a room temperature of 15°C.

## ● NOTE ●

See table 1 hereafter to check pressure values and to inflate tires and shock absorbers.

●

Check pressure values and inflate, if necessary, according to following table 1 during operation in cold weather only :

|   |                                  | OAT (°C)      | - 40°         | - 30°         | - 20°         | - 10°         | + 15° |
|---|----------------------------------|---------------|---------------|---------------|---------------|---------------|-------|
| P<br>r<br>e<br>s<br>s<br>u<br>r<br>e<br><br>psi<br><br>(bars) | Main landing gear shock absorber | 189<br>(13)   | 196<br>(13.5) | 203<br>(14)   | 218<br>(15)   | 247<br>(17)   |       |
|   | Nose gear shock absorber         | 102<br>(7)    | 109<br>(7.5)  | 116<br>(8)    | 123<br>(8.5)  | 138<br>(9.5)  |       |
|   | Main landing gear tire           | 144<br>(9.96) | 144<br>(9.96) | 130<br>(8.96) | 130<br>(8.96) | 130<br>(8.96) |       |
|   | Nose gear tire                   | 94<br>(6.5)   | 94<br>(6.5)   | 102<br>(7)    | 102<br>(7)    | 102<br>(7)    |       |

Table 1



**TBM 930**

**With MOD70-0476-00**

# **List of equipment**

**Report reference NAV No. 34/90-RJ-App 4**

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**List of effective pages and validities****Edition 0 of December 31, 2015****Airplane equipped with MOD70-0476-00**

| <b>Page No.</b> | <b>Edit./Rev. No.</b> | <b>Rev. Date</b> | <b>Page No.</b> | <b>Edit./Rev. No.</b> | <b>Rev. Date</b> |
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| ■ Title         | 0-3                   | APR 19           | 18              | 0-2                   | APR 18           |
|                 |                       |                  | 19              | 0-2                   | APR 18           |
| ■ 0A            | 0-3                   | APR 19           | 20              | 0-2                   | APR 18           |
| 0B              | 0-2                   | APR 18           | 21              | 0-2                   | APR 18           |
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| ■ 0F            | 0-3                   | APR 19           | 25              | 0-2                   | APR 18           |
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| 4               | 0-2                   | APR 18           | 30              | 0-2                   | APR 18           |
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| 7               | 0-2                   | APR 18           | 33              | 0-2                   | APR 18           |
| 8               | 0-2                   | APR 18           | 34              | 0-2                   | APR 18           |
| 9               | 0-2                   | APR 18           | 35              | 0-2                   | APR 18           |
| 10              | 0-2                   | APR 18           | 36              | 0-2                   | APR 18           |
| 11              | 0-2                   | APR 18           | 37              | 0-2                   | APR 18           |
| 12              | 0-2                   | APR 18           | 38              | 0-2                   | APR 18           |
| 13              | 0-2                   | APR 18           | 39              | 0-2                   | APR 18           |
| 14              | 0-2                   | APR 18           | 40              | 0-2                   | APR 18           |
| 15              | 0-2                   | APR 18           | 41              | 0-2                   | APR 18           |
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| 17              | 0-2                   | APR 18           |                 |                       |                  |

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**List of amendments**

Revision 1 dated December 15, 2016

| Pages     | Description   |
|-----------|---|
| Title     | Modification of the date of the copyright                           |
| 0A        | List of effective pages   |
| 0C and 0D | List of amendments  |
| 1         | Modification of the table of contents                               |
| 4         | Modification of valves part number                                  |
| 8         | Addition of MOD/OPT70-0526-23A                                      |
| 9         | KHF-1050 HF Communication system option for G3000 (TR23)            |
| 12        | Modification and/or addition of servicing plugs unit                |
| 13        | Modification of weight value for upholstery version 2016            |
| 13        | Addition of MOD/OPT70-0515-25                                       |
| 13        | Addition of MOD/OPT70-0530-25A                                      |
| 13        | Addition of MOD/OPT70-05341-25A                                     |
| 14        | Addition of pilot's and front R.H. Seat for airplanes from S/N 1160 |
| 14        | Addition of Seat - Belts optional equipment                         |
| 16        | Modification of portable fire extinguisher unit versions            |
| 17        | Addition of control wheels part number                              |
| 17        | Addition of stick shaker (TR21)                                     |
| 19        | Addition of windshield deicing                                      |

## List of amendments

Revision 1 dated December 15, 2016 (Cont'd)

| Pages  | Description   |
|--|---|
| 21   | Addition of MOD/OPT70-0455-31D  |
| 21   | Addition of MOD/OPT70-0533-31A  |
| 24   | Addition of L.H. Wing inspection light and Leading edge ice detection LED light |
| 26   | Addition of version number of lift transducer and AoA computer installation     |
| 28   | Addition of MOD/OPT70-0541-23A  |
| 30, 31   | Modification and addition of MOD/OPT related to transponders                    |
| 40   | Modification of compressor and power turbine tachogenerator part numbers        |
| 10, 11, 15, 18, 20, 22, 23, 25, 27, 29, 32 thru 39, 41, 42 | Terminology, text moving and/or presentation                                    |

**List of amendments**

Revision 2 dated April 15, 2018

| Pages     | Description  |
|-----------|--|
| Title     | Copyright  |
| 0A        | List of effective pages                                |
| 0E        | List of amendments                                     |
| 3         | List of critical RVSM equipment Part Number correction |
| 21        | Addition of FASTBOX, and correction of LDR equipment   |
| All pages | Presentation and/or text moving                        |



**List of amendments**

Revision 3 dated April 8, 2019

| Pages | Description                            |
|-------|--|
| Title | Updating of the date on the copyright  |
| 0A    | List of effective pages and validities |
| 0F    | List of amendments Revision 3          |
| 3     | Terminology                            |

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| ATA 24 - Electrical power .....               | 11   |
| ATA 25 - Equipment and furnishings .....      | 12   |
| ATA 26 - Fire protection .....                | 16   |
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The following list contains standard equipment installed on each airplane and available optional equipment.

A separate list of equipment of items installed at the factory in your specific airplane is provided in your airplane file.

Columns showing weight (in pounds) and arm (in inches) provide the weight and center of gravity location for the equipment.

In the list of Required, Standard or Optional equipment (not restrictive), a letter "R", "S", "O" or "A" allows classifying the equipment :

- "R" : equipment items required for certification
- "S" : standard equipment items
- "A" : optional equipment items which are in addition to required or standard items
- "O" : optional equipment items replacing required or standard items

## List of critical RVSM equipment

Equipment listed hereafter, or later approved versions, is required for RVSM operation.

| Equipment                               | * | ** | P/N              |
|---|---|----|------------------|
| Barometric altimeter :                  |   |    |                  |
| - GDC74B (Air data computer)            | 2 | 2  | P/N 011-01110-01 |
| - GDU1200W (Display)                    | 3 | 2  | P/N 011-02333-00 |
| Autopilot Altitude Hold function :      |   |    |                  |
| - GMC710 (AFCS mode controller)         | 1 | 1  | P/N 011-01020-10 |
| - GIA63W (Integrated Avionics Computer) | 2 | 2  | P/N 011-00105-40 |
| - GRS77                                 | 2 | 2  | P/N 011-00868-10 |
| ATC transponder :                       |   |    |                  |
| - Altitude reporting transponder        | 1 | 1  | TSO C-74c        |

(\*) Quantity installed

(\*\*) Quantity required

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>01 - Specific optional equipment</b>                        |                                  |                   |
| S                   | 01026A                       | Flight ceiling at 31000 ft                                     | /                                | /                 |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>21 - Environmental system</b>                               |                                  |                   |
| S                   | 0454-21A                     | General Air System Controller (GASC)<br>82024A040701           | 1.98<br>(0.900)                  | 311.02<br>(7.900) |
|                     |                              | <b>21-20 - Distribution</b>                                    |                                  |                   |
| S                   |                              | Mixing unit 9723A010001  | 0.53<br>(0.240)                  | 151.57<br>(3.850) |
| S                   |                              | Hot Air Distributor 6044A010001                                | 4.06<br>(0.840)                  | 153.54<br>(3.900) |
| S                   |                              | Bleed temperature switch<br>92244B010002                       | 0.13<br>(0.060)                  | 153.54<br>(3.900) |
|                     |                              | <b>21-30 - Pressurization control</b>                          |                                  |                   |
| S                   |                              | Cabin altitude warn switch<br>214 C40.3.261                    | 0.077<br>(0.035)                 | 153.94<br>(3.910) |
| S                   |                              | Cabin differential pressure warn switch<br>17-600-01           | 0.143<br>(0.065)                 | 139.76<br>(3.550) |
| S                   | 0448-21                      | Outflow valve 81144A010101                                     | 2.976<br>(1.350)                 | 317.32<br>(8.060) |
| S                   | 0448-21                      | Safety valve 81145A010101                                      | 2.337<br>(1.060)                 | 317.32<br>(8.060) |
|                     |                              | <b>21-50 - Temperature conditioning system</b>                 |                                  |                   |
| S                   |                              | Flow control shut-off valve<br>784A010001                      | 4.74<br>(2.500)                  | 114.17<br>(2.900) |
| S                   |                              | Non-return valve 7085A010002                                   | 0.11<br>(0.050)                  | 102.36<br>(2.600) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
| S                   |                              | Shut-off valve 4589A010001                                     | 2.37<br>(1.075)                  | 114.17<br>(2.900) |
| S                   |                              | Intermediate pressure sensor<br>93557A010001                   | 0.33<br>(0.150)                  | 110.24<br>(2.800) |
| S                   |                              | Overheat thermal switch<br>A042010300-5                        | 0.18<br>(0.080)                  | 110.24<br>(2.800) |
| S                   |                              | Main heat exchanger 81249A010001                               | 7.72<br>(3.500)                  | 108.27<br>(2.750) |
| S                   |                              | Non-return valve 52704A010001                                  | 0.66<br>(0.300)                  | 118.11<br>(3.000) |
| S                   |                              | Ground Fan 8031A020  | 3.95<br>(1.790)                  | 90.55<br>(2.300)  |
|                     |                              | <b>21-55 - Vapor cycle cooling system</b>                      |                                  |                   |
| S                   |                              | Compressor 1377A010001   | 14.77<br>(6.700)                 | 98.43<br>(2.500)  |
| S                   |                              | Cockpit Evaporator Assembly<br>14720A010001                    | 9.06<br>(4.111)                  | 200.79<br>(5.100) |
| S                   |                              | Cabin Evaporator Assembly<br>14719A010001                      | 12.90<br>(5.850)                 | 311.02<br>(7.900) |
| S                   |                              | Condenser Assembly 81250A010001                                | 24.80<br>(11.250)                | 330.71<br>(8.400) |
|                     |                              | <b>21-60 - Temperature regulation</b>                          |                                  |                   |
| S                   |                              | By-pass valve 6043A010001                                      | 3.31<br>(1.500)                  | 106.30<br>(2.700) |
| S                   |                              | Bleed differential pressure sensor<br>93558A010001             | 0.44<br>(0.200)                  | 114.17<br>(2.900) |
| S                   |                              | Inlet temperature sensor<br>93276A010001                       | 0.11<br>(0.050)                  | 153.54<br>(3.900) |

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| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
| S                   |                              | Cockpit ventilated sensor<br>92279A010002                      | 0.18<br>(0.080)                  | 182.09<br>(4.625) |
| S                   |                              | Cabin ventilated sensor<br>92279A010002                        | 0.18<br>(0.080)                  | 250.00<br>(6.350) |



| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
| S                   | 0305-22                      | <b>22 - Auto flight</b>  |                                  |                   |
|                     |                              | Upgrading of AFCS GFC 700<br>composed of :                     |                                  |                   |
|                     |                              | - Pitch servo GSA 81<br>+ Servo mount GSM 86                   | 4.08<br>(1.85)                   | 247.40<br>(6.284) |
|                     |                              | - Roll servo GSA 81<br>+ Servo mount GSM 86                    | 4.08<br>(1.85)                   | 231.10<br>(5.870) |
|                     |                              | - Yaw servo GSA 81<br>+ Servo mount GSM 86                     | 4.08<br>(1.85)                   | 253.70<br>(6.444) |
|                     |                              | - Pitch trim servo GSA 81<br>+ Servo mount GSM 86              | 4.14<br>(1.88)                   | 157.87<br>(4.010) |
|                     |                              | - Trim adapter GTA 82  | 1.30<br>(0.59)                   | 240.87<br>(6.118) |
|                     |                              | - AFCS Control Unit GMC 710                                    | 0.91<br>(0.41)                   | 156.61<br>(3.978) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment            | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|---|----------------------------------|-------------------|
|                     |                              | <b>23 - Communications</b>  |                                  |                   |
| S                   |                              | Static dischargers DSC 740049 (Qty : 4)                                   | Neglig.                          | /                 |
| S                   |                              | Static dischargers 2-5 SCY (Qty : 2)                                      | Neglig.                          | /                 |
| S                   |                              | Static dischargers 2-9 SCY (Qty : 3)                                      | Neglig.                          | /                 |
| S                   | 0526-23A                     | Static dischargers on winglets  | Neglig.                          | /                 |
|                     |                              | - Static dischargers<br>DSC 740049 (Qty : 2)                              | Neglig.                          | /                 |
|                     |                              | or  |                                  |                   |
|                     |                              | - Static dischargers<br>2-5 SCY (Qty : 2)                                 | Neglig.                          | /                 |
| O                   | 0287-23A                     | Radio stereo-headset A20 with bluetooth                                   | Neglig.                          | /                 |
| O                   | 0487-23A                     | Radio stereo-headset A20  | Neglig.                          | /                 |
| O                   | 0331-23                      | Weather Data Link and Satellite Phone GSR 56                              |                                  |                   |
|                     |                              | Post-MOD70-0476-00 Version B with GDU SW 6.51                             |                                  |                   |
|                     |                              | - Version J : with antenna CI 490-490<br>(GSR unit support pre-installed) | 3.44<br>(1.561)                  | 58.00<br>(1.474)  |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment                                | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|---|----------------------------------|-------------------|
| A                   | 0410-23B                     | HF Communication System KHF1050,<br>of which  | 38.03<br>(17.250)                | 302.70<br>(7.689) |
|                     |                              | - Control Display unit  | 1.56<br>(0.707)                  | 155.43<br>(3.948) |
|                     |                              | - Receiver/Exciter  | 5.90<br>(2.676)                  | 123.07<br>(3.126) |
|                     |                              | - Antenna coupler   | 16.20<br>(7.348)                 | 342.28<br>(8.694) |
|                     |                              | - Power amplifier   | 8.40<br>(3.810)                  | 342.83<br>(8.708) |
|                     |                              | - HF Antenna kit  | 1.74<br>(0.790)                  | 324.80<br>(8.250) |
| A                   | 0458-23B                     | GDL 69A SXM - XM Generation 4<br>interfaced with G3000 system                                 | 1.41<br>(0.640)                  | 163.46<br>(4.152) |
| S                   | 0564-00A                     | G3000 COM # 1 system  |                                  |                   |
|                     |                              | - Transceiver<br>(integrated in GIA 64W Integrated Avionics<br>Unit # 1 - refer to ATA 34-28) | 7.03<br>(3.19)                   | 147.77<br>(3.753) |
|                     |                              | - VHF antenna (under fuselage)<br>16-21B-P3   | 0.86<br>(0.390)                  | 271.65<br>(6.900) |
| S                   | 0564-00A                     | G3000 COM # 2 system  |                                  |                   |
|                     |                              | - Transceiver<br>(integrated in GIA 64W Integrated Avionics<br>Unit # 2 - refer to ATA 34-28) | 7.03<br>(3.19)                   | 147.77<br>(3.753) |
|                     |                              | - VHF antenna (above fuselage)<br>16-21B-P3   | 0.86<br>(0.390)                  | 271.65<br>(6.900) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>24 - Electrical power</b>                                   |                                  |                   |
|                     |                              | <b>24-30 - DC generation</b>                                   |                                  |                   |
| R                   | 0234-24                      | Electric power system (EPS)<br>1408-1-1                        | 14.330<br>(6.500)                | 128.15<br>(3.255) |
| R                   |                              | Stand-by alternator<br>ES10024B-5                              | 13.000<br>(5.897)                | 104.84<br>(2.663) |
| R                   |                              | Starter generator<br>MG94K-1                                   | 31.989<br>(14.510)               | 118.83<br>(2.815) |
| S                   | 24002A                       | Lead-acid battery RG-380E/44                                   | 85.979<br>(39.000)               | 112.20<br>(2.850) |
| A                   | 0303-24                      | Charger/Maintainer for lead acid battery                       | 0.220<br>(0.100)                 | 114.17<br>(2.900) |
|                     |                              | <b>24-40 - External power supply</b>                           |                                  |                   |
| S                   |                              | Ground power receptacle<br>MS 3506-1                           | 0.794<br>(0.360)                 | 114.17<br>(2.900) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>25 - Equipment and furnishings</b>                          |                                  |                   |
| A                   | 25004D                       | Leather upholstery<br>- version D "Autolux"                    | 6.614<br>(3.000)                 | 212.60<br>(5.400) |
| A                   | 0386-25                      | Leather upholstery "Vulcain"                                   | 6.614<br>(3.000)                 | 212.60<br>(5.400) |
| S                   |                              | Smoke goggles MXP 210  | 0.855<br>(0.388)                 | 200.00<br>(5.080) |
| A                   | 25032                        | Front seats ease covers  | 2.756<br>(1.250)                 | 183.78<br>(4.668) |
| A                   | 25035                        | JetFly type cabin arrangement                                  | /                                | /                 |
| A                   | 25036                        | Cabin furnishings - "Loupe d'Orme" wood                        | /                                | /                 |
| A                   | 0151-25                      | CD reader PCD 7100   | 2.20<br>(1.00)                   | 205.04<br>(5.208) |
| A                   | 0304-25                      | Cabin fitting out<br>("Autolux" leather upholstery variants)   |                                  |                   |
|                     |                              | - Version A : Heather-leather light blue-coloured<br>seats     | /                                | /                 |
|                     |                              | - Version B : Blue jeans-coloured carpets                      | /                                | /                 |
|                     |                              | - Version C : Sateen Chocolate-coloured seats<br>and cabinets  | /                                | /                 |
|                     |                              | - Version D : Carbon-coloured Finishing                        | /                                | /                 |
|                     |                              | - Version E : Grey-coloured seats and cabinets                 | /                                | /                 |

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| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment   | Weight<br>per unit<br>lb<br>(kg)   | Arm<br>in.<br>(m)  |
|---------------------|------------------------------|--|--|--|
| S                   | 0374-25B                     | Servicing plugs unit, of which :<br><br>- 12 VDC servicing plugs unit (Qty : 2 - one in the cockpit, one in the cabin), of which :<br><br>. 28-12 VDC Converter<br>AK-551-18S<br><br>- 5 VDC servicing plugs unit (USB type)<br>(Qty : 4 - two in the cockpit, two in the cabin) with integrated charger                               | 3.75<br>(1.700)<br><br>3.31<br>(1.500)<br><br>2.98<br>(1.350)<br><br>0.44<br>(0.200) | /<br><br>195.28<br>(4.960)<br><br>195.28<br>(4.960)<br><br>187.99<br>(4.775)                 |
| S                   | 0374-25C                     | Additional servicing plugs unit<br><br>- 5 VDC servicing plugs unit (USB type)<br>(Qty : 2 - on L.H. intermediate seat) with integrated charger  | /<br><br>0.20<br>(0.100)   | /<br><br>219.29<br>(5.570)   |
| O                   | 0374-25D                     | Servicing plugs unit, of which :<br><br>- 12 VDC servicing plugs unit (Qty : 2 - one in the cockpit, one in the cabin), of which :<br><br>. 28-12 VDC Converter<br>6430120-1<br><br>- 5 VDC servicing plugs unit (USB type)<br>(Qty : 4 - two in the cockpit, two in the cabin)<br>6430102-1<br><br>. 28-12 VDC Converter<br>6430120-1 | 3.31<br>(0.800)<br><br>0.60<br>(0.270)<br><br>0.44<br>(0.200)<br><br>0.60<br>(0.270) | 195.28<br>(4.960)<br><br>195.28<br>(4.960)<br><br>187.99<br>(4.775)<br><br>195.28<br>(4.960) |
| O                   | 0531-25A                     | 13.8 VDC servicing plug unit, of which :<br><br>- 28-12 VDC Converter<br><br>- 13.8 VDC servicing plug   | 0.74<br>(0.334)<br><br>0.60<br>(0.270)<br><br>0.14<br>(0.064)                        | 193.00<br>(4.900)<br><br>193.00<br>(4.900)<br><br>288.00<br>(7.310)                          |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment   | Weight<br>per unit<br>lb<br>(kg)    | Arm<br>in.<br>(m)                                |
|---------------------|------------------------------|--|-------------------------------------|--|
| A                   | 0417-25                      | Paper clips (one on each control wheel)  | /                                   | /  |
| A                   | 0503-25A                     | Upholstery Version 2016  | $\Delta+2.73$<br>( $\Delta+1.240$ ) | 228.30<br>(5.799)                                |
| O                   | 0515-25                      | One piece cabin floor protection   | $\Delta 0$                          | /  |
| S                   | 0530-25A                     | Hi-power USB servicing plugs, of which :   | 0.33<br>(0.15)                      | 188.00<br>(4.775)                                |
|                     |                              | - one 5 VDC servicing single plug (USB type)<br>6430202-9 - on instrument panel, pilot side                          | /                                   | /  |
|                     |                              | - one 5 VDC servicing double plug unit (USB type)<br>6430202-5 - on instrument panel, front R.H.seat<br>side         | /                                   | /  |
|                     |                              | - two 5 VDC servicing double plug units (USB type)<br>6430202-9 - in the cabin (R.H. and L.H.<br>Intermediate seats) | /                                   | /  |
| S                   | 0531-25A                     | 13.8 VDC servicing plug unit, of which :   | 0.74<br>(0.334)                     | 193.00<br>(4.900)                                |
|                     |                              | - 28-12 VDC converter<br>6430120-1   | 0.360<br>(0.270)                    | 193.00<br>(4.900)                                |
|                     |                              | - 13-8 VDC servicing plug 77 00 808 844 - in the<br>cabin (R.H. Rear seat)   | 0.14<br>(0.064)                     | 288.00<br>(7.310)                                |
|                     |                              | <b>Seats - Belts</b>   |                                     |  |
|                     |                              | <b>Belts</b>   |                                     |  |
| S                   |                              | Reels  | 1.79<br>(0.810)                     | 192.91<br>or<br>287.40<br>(4.900<br>or<br>7.300) |

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| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment                              | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|---|----------------------------------|-------------------|
| O                   | 0516-25A                     | Airbag seat belts   | 15.08<br>(6.840)                 | /                 |
| O                   | 0516-25A                     | Pilot's and passengers' seat belts  | 10.24<br>(4.640)                 | /                 |
|                     |                              | <b>Seats</b>  |                                  |                   |
| S                   |                              | Pilot's seat T700C2500002   | 55.12<br>(25.00)                 | 183.90<br>(4.671) |
| S                   |                              | Pilot's seat T700J2500005<br>(From S/N 1160)  | 48.90<br>(22.20)                 | 185.86<br>(4.671) |
| S                   |                              | Front R.H. seat T700C2500002  | 55.12<br>(25.00)                 | 183.90<br>(4.671) |
| S                   |                              | Front R.H. seat T700J2500005<br>(From S/N 1160)   | 48.90<br>(22.20)                 | 185.86<br>(4.721) |
|                     |                              | <b>25-61 - Emergency locator transmitter</b>  |                                  |                   |
| A                   | 0437-25B                     | Emergency locator transmitter ELT 1000 (airplanes<br>equipped with reinforcement), of which | 2.385<br>(1.082)                 | 340.91<br>(8.659) |
|                     |                              | - ELT 1000 with base  | 1.764<br>(0.800)                 | 354.72<br>(9.010) |
|                     |                              | - Antenna 110-338   | 0.449<br>(0.204)                 | 318.70<br>(8.095) |



| S/<br>R/<br>A/<br>O         | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment                               | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|-----------------------------|------------------------------|--|----------------------------------|-------------------|
| <b>26 - Fire protection</b> |                              |  |                                  |                   |
| A                           | 0391-26                      | Portable fire extinguisher unit 74-00<br><br>- Version A (on R.H. Station R.H. Upholstering) | 4.89<br>(2.220)                  | 170.11<br>(4.321) |
| A                           | 0496-26A                     | Engine fire detection system   | 1.464<br>(0.664)                 | 96.06<br>(2.440)  |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m)  |
|---------------------|------------------------------|--|----------------------------------|--------------------|
|                     |                              | <b>27 - Flight controls</b>                                    |                                  |                    |
|                     |                              | <b>27-10 - Roll control</b>                                    |                                  |                    |
| R                   |                              | Roll trim actuator 145700.02                                   | 1.543<br>(0.700)                 | 212.60<br>(5.400)  |
|                     |                              | <b>27-20 - Yaw control</b>                                     |                                  |                    |
| R                   |                              | Rudder trim actuator 145700.02                                 | 1.543<br>(0.700)                 | 395.27<br>(10.040) |
| S                   | 0348-27                      | New control wheels   |                                  |                    |
| S                   |                              | - L.H. equipped control wheel<br>83912112                      | 2.535<br>(1.150)                 | 157.48<br>(4.000)  |
| S                   |                              | - R.H. equipped control wheel<br>83912113                      | 2.535<br>(1.150)                 | 157.48<br>(4.000)  |
|                     |                              | <b>27-30 - Pitch control</b>                                   |                                  |                    |
| S                   |                              | Pitch trim actuator 145400-02                                  | 1.213<br>(0.550)                 | 425.20<br>(10.800) |
| A                   | 0510-27A                     | Stick shaker<br>C-101702-1                                     | 1.053<br>(0.477)                 | 144.00<br>(3.658)  |
|                     |                              | <b>27-50 - Wing flaps (control)</b>                            |                                  |                    |
| R                   |                              | Flap control including :                                       | 15.520<br>(7.040)                | 218.50<br>(5.550)  |
|                     |                              | - Flap motor 6157-1  | 2.866<br>(1.300)                 | 216.54<br>(5.500)  |
|                     |                              | - Flap actuator 1-5297 / 2-5297                                | 1.830<br>(0.830)                 | 220.47<br>(5.600)  |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>28 - Fuel system</b>  |                                  |                   |
|                     |                              | <b>28-20 - Fuel supply</b>                                     |                                  |                   |
| R                   |                              | Electric boost pump 1B9-5                                      | 4.409<br>(2.000)                 | 129.92<br>(3.300) |
| R                   |                              | Engine driven fuel pump 1127-02                                | 1.543<br>(0.700)                 | 110.24<br>(2.800) |
| R                   |                              | Fuel unit L88A15-651   | 4.586<br>(2.080)                 | 133.07<br>(3.380) |
| R                   |                              | A35 fuel sequencer unit  | 1.102<br>(0.500)                 | 125.98<br>(3.200) |
|                     |                              | <b>28-40 - Fuel indication</b>                                 |                                  |                   |
| R                   | 0158-28C                     | Fuel gage amplifier (in us gal)<br>738574-1-0                  | 1.08<br>(0.49)                   | 278.74<br>(7.080) |
| R                   |                              | Inboard L.H. Gage<br>762 438.1.0                               | 0.331<br>(0.150)                 | 183.07<br>(4.650) |
| R                   |                              | Inboard R.H. Gage<br>762 439.1.0                               | 0.331<br>(0.150)                 | 183.07<br>(4.650) |
| R                   |                              | Intermediate gage<br>762 440.1.0                               | 0.220<br>(0.100)                 | 190.94<br>(4.850) |
| R                   |                              | Outboard gage<br>762 441.1.0                                   | 0.220<br>(0.100)                 | 190.94<br>(4.850) |
| R                   | 0427-28A                     | Low level sensor<br>747-971-1-0                                | 0.143<br>(0.065)                 | 185.28<br>(4.706) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment  | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m)  |
|---------------------|------------------------------|---|----------------------------------|--------------------|
|                     |                              | <b>30 - Ice and rain protection</b>                             |                                  |                    |
| S                   |                              | Deicer T700A3013003000,<br>L.H. horizontal stabilizer           | 4.189<br>(1.900)                 | 398.42<br>(10.120) |
| S                   |                              | Deicer T700A3013003001,<br>R.H. horizontal stabilizer           | 4.189<br>(1.900)                 | 398.42<br>(10.120) |
| S                   |                              | Deicer T700A3014003000,<br>vertical stabilizer                  | 3.968<br>(1.800)                 | 374.02<br>(9.500)  |
| S                   |                              | Deicer T700A3010001002,<br>inboard L.H. wing                    | 5.732<br>(2.600)                 | 173.23<br>(4.400)  |
| S                   |                              | Deicer T700A3010001003,<br>inboard R.H. wing                    | 5.732<br>(2.600)                 | 173.23<br>(4.400)  |
| S                   |                              | Deicer T700A3010001004,<br>middle L.H. wing                     | 3.748<br>(1.700)                 | 173.23<br>(4.400)  |
| S                   |                              | Deicer T700A3010001005,<br>middle R.H. wing                     | 3.748<br>(1.700)                 | 173.23<br>(4.400)  |
| S                   |                              | Deicer T700A3010012000,<br>outboard L.H. wing                   | 2.65<br>(1.200)                  | 173.23<br>(4.400)  |
| S                   |                              | Deicer T700A3010001007,<br>outboard R.H. wing                   | 3.307<br>(1.500)                 | 173.23<br>(4.400)  |
| S                   |                              | Dual port distribution valve 1532-10C                           | 2.425<br>(1.100)                 | 125.98<br>(3.200)  |
| S                   |                              | Timer 42E25-2A  | 0.772<br>(0.350)                 | 177.17<br>(4.500)  |
| S                   |                              | Water separator and filter 44E21-2A                             | 1.102<br>(0.500)                 | 125.98<br>(3.200)  |
|                     |                              | <b>30-40 - Windshield deicing</b>                               |                                  |                    |
| S                   |                              | Windshield heater controllers (Qty 2 : L.H. + R.H.<br>TWH 93-01 | 1.984<br>(0.900)                 | 149.61<br>(3.800)  |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
| S                   |                              | <b>30-60 - Propeller deicing</b><br><br>Timer 3E2311-4         | 0.44<br>(0.200)                  | 200.79<br>(5.100) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment         | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>31 - Indicating/recording systems</b>                               |                                  |                   |
|                     |                              | <b>31-20 - Independent instruments</b>                                 |                                  |                   |
| O                   | 31002A                       | Hourmeter 56457-3<br>(engine running time)                             | 0.551<br>(0.250)                 | 156.30<br>(3.970) |
| S                   |                              | Hourmeter 56457-3 (flying time)  | 0.551<br>(0.250)                 | 156.30<br>(3.970) |
| S                   | 0455-31                      | Light weight Flight Data Recorder (ADRS - CARS),<br>of which :         | 5.659<br>(2.567)                 | 256.50<br>(6.515) |
|                     |                              | - Light Data Recorder  | 4.982<br>(2.260)                 | 260.63<br>(6.620) |
|                     |                              | - MK170 microphone   | 0.198<br>(0.090)                 | 153.54<br>(3.900) |
|                     |                              | - Support (pre-installed)  | 0.478<br>(0.217)                 | 260.63<br>(6.620) |
| S                   | 0533-31A                     | Digital hourmeter (flying time)<br>(From S/N 1160)                     | 0.412<br>(0.187)                 | 148.62<br>(3.775) |
| A                   | 0578-31B                     | Data collection and transmission system<br>(FASTBOX A171bFAST-A-042-1) | 3.62<br>(1.64)                   | 125.2<br>(3.180)  |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>32 - Landing gears</b>                                      |                                  |                   |
|                     |                              | <b>32-10 - Main landing gear</b>                               |                                  |                   |
| R                   | 0190-32                      | L.H. main landing gear<br>D23767001                            | 53.79<br>(24.400)                | 200.39<br>(5.090) |
| R                   | 0190-32                      | R.H. main landing gear<br>D23768001                            | 53.79<br>(24.400)                | 200.39<br>(5.090) |
|                     |                              | <b>32-20 - Nose landing gear</b>                               |                                  |                   |
| R                   | 0134-32                      | Nose gear D23766000  | 53.57<br>(24.300)                | 93.70<br>(2.380)  |
|                     |                              | <b>32-30 - Extension and retraction</b>                        |                                  |                   |
| O                   | 0334-32                      | Main locking actuator VSTS 083560                              | 13.228<br>(6.000)                | 208.07<br>(5.285) |
| O                   | 0334-32                      | Nose locking actuator VSTS 083560                              | 13.228<br>(6.000)                | 110.24<br>(2.800) |
| R                   |                              | Hand pump 914-8D27   | 2.326<br>(1.055)                 | 181.10<br>(4.600) |
|                     |                              | <b>32-35 - Hydraulic generation</b>                            |                                  |                   |
| R                   | 060-32                       | Hydraulic power pack 1118-04                                   | 10.362<br>(4.700)                | 84.65<br>(2.150)  |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>32-40 - Wheels and brakes</b>                               |                                  |                   |
| R                   |                              | Brake assembly 030-19100                                       | 14.991<br>(6.800)                | 204.33<br>(5.190) |
| R                   |                              | Main tire 18x5.5-10PR  | 13.50<br>(6.123)                 | 204.33<br>(5.190) |
| R                   | 0409-32                      | Main tire 18x5.5-10PR  | 14.396<br>(6.530)                | 204.33<br>(5.190) |
| R                   |                              | Master cylinder 010-07802                                      | 0.882<br>(0.400)                 | 145.67<br>(3.700) |
|                     |                              | >> <i>With MICHELIN tire</i>                                   |                                  |                   |
| R                   |                              | Nose tire 5.00-5-10PR TL                                       | 5.600<br>(2.540)                 | 89.57<br>(2.275)  |
|                     |                              | >> <i>With GOOD YEAR tire</i>                                  |                                  |                   |
|                     |                              | Nose tire 5.00-5-10PR TL                                       | 6.300<br>(2.858)                 | 89.57<br>(2.275)  |
| R                   | 0408-32                      | Nose tire 5.00-5-10PR  | 6.834<br>(3.100)                 | 89.57<br>(2.275)  |
| R                   |                              | Nose wheel 40-262A   | 2.976<br>(1.350)                 | 89.57<br>(2.275)  |
| R                   |                              | Main wheel (Model 40-434)                                      | 11.28<br>(5.120)                 | 204.33<br>(5.190) |
| R                   |                              | Parking brake valve T700A3240010<br>or T700B3240001            | 0.331<br>(0.150)                 | 157.48<br>(4.000) |



| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m)  |
|---------------------|------------------------------|--|----------------------------------|--------------------|
|                     |                              | <b>33 - Lights</b>   |                                  |                    |
|                     |                              | <b>33-10 - Instrument panel lighting</b>                       |                                  |                    |
| S                   |                              | Instruments emergency lighting 2240-3                          | 0.110<br>(0.050)                 | 181.10<br>(4.600)  |
| S                   | 0322-00                      | PULSELITE unit   | Neglig.                          | /                  |
|                     |                              | <b>33-40 - External lighting</b>                               |                                  |                    |
| S                   |                              | L.H. wing inspection light (icing detection)<br>T700G3340020   | 0.20<br>(0.090)                  | 151.57<br>(3.850)  |
| S                   | 0509-33                      | Leading edge ice detection LED light<br>01-0771904-00          | 0.25<br>(0.113)                  | 172.00<br>(4.371)  |
| S                   | 0322-00                      | LED L.H. taxi and landing lights<br>01-0771674-01              | 1.400<br>(0.635)                 | 181.10<br>(4.600)  |
| S                   | 0322-00                      | LED R.H. taxi and landing lights<br>01-0771674-01              | 1.400<br>(0.635)                 | 181.10<br>(4.600)  |
| S                   | 0322-00                      | NAV/Anticollision system (LED lights) :                        |                                  |                    |
| S                   |                              | Central units :  |                                  |                    |
| S                   |                              | - L.H. strobe light power supply<br>01-0771234-07              | 0.609<br>(0.277)                 | 191.38<br>(4.861)  |
| S                   |                              | - R.H. strobe light power supply<br>01-0771234-07              | 0.609<br>(0.277)                 | 191.38<br>(4.861)  |
| S                   |                              | - Rear strobe light power supply                               | 0.609<br>(0.277)                 | 397.87<br>(10.106) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m)  |
|---------------------|------------------------------|--|----------------------------------|--------------------|
| S                   |                              | Lights :   |                                  |                    |
| S                   |                              | - L.H. navigation/strobe/recognition lights<br>01-0771170-02   | 0.499<br>(0.227)                 | 184.29<br>(4.681)  |
| S                   |                              | - R.H. navigation/strobe/recognition lights<br>01-0771170-01   | 0.499<br>(0.227)                 | 184.29<br>(4.681)  |
| S                   |                              | - Rear tail navigation/strobe lights<br>01-0790667-00          | 0.499<br>(0.227)                 | 444.21<br>(11.283) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>34 - Navigation</b>   |                                  |                   |
|                     |                              | <b>34-11 - Air data systems</b>                                |                                  |                   |
| S                   |                              | Pitot L heated probe<br>AN 5812-1                              | 0.750<br>(0.340)                 | 200.79<br>(5.100) |
| S                   |                              | Pitot R heated probe<br>AN 5812-1                              | 0.750<br>(0.340)                 | 200.79<br>(5.100) |
| R                   |                              | Static reference plug T700A3415017A                            | Neglig.                          | /                 |
| S                   |                              | Static reference selector TB30 77010000                        | 0.220<br>(0.100)                 | 157.48<br>(4.000) |
| S                   | 0160-34A                     | Authorization to operate in RVSM area                          | /                                | /                 |
| S                   | 0423-34A                     | Lift transducer and AoA computer installation, of<br>which :   | 1.66<br>(0.752)                  | 242.01<br>(6.147) |
| R                   |                              | - Lift transducer<br>P/N C-101-707-1 or<br>P/N C-101-707-2     | 0.50<br>(0.226)                  | 173.23<br>(4.400) |
| S                   |                              | - AoA computer<br>P/N C-101-706-1                              | 0.74<br>(0.336)                  | 273.62<br>(6.950) |
| S                   |                              | - K59 and K590 relays  | 0.25<br>(0.115)                  | 265.55<br>(6.745) |
| S                   | 0544-00A                     | Air Data Computer # 1 GDC 74B                                  | 1.83<br>(0.83)                   | 148.11<br>(3.762) |
| S                   | 0544-00A                     | Air Data Computer # 2 GDC 74B                                  | 1.83<br>(0.83)                   | 148.11<br>(3.762) |
| S                   | 0462-34                      | Standby Attitude Module MD302                                  | 1.61<br>(0.730)                  | 154.53<br>(3.925) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>34-21 - Heading reference system</b>                        |                                  |                   |
| S                   | 0543-00A                     | Attitude and Heading Reference System<br># 1 GRS 77            | 3.46<br>(1.57)                   | 171.77<br>(4.363) |
| S                   | 0543-00A                     | Attitude and Heading Reference System<br># 2 GRS 77            | 3.46<br>(1.57)                   | 171.77<br>(4.363) |
| S                   | 0476-00A                     | Magnetometer # 1 GMU 44  | 0.48<br>(0.22)                   | 180.98<br>(4.597) |
| S                   | 0476-00A                     | Magnetometer # 2 GMU 44  | 0.48<br>(0.22)                   | 180.98<br>(4.597) |
|                     |                              | <b>34-23 - Magnetic compass</b>                                |                                  |                   |
| R                   |                              | Stand-by compass C2350 L4.M23                                  | 0.551<br>(0.250)                 | 163.39<br>(4.150) |
|                     |                              | <b>34-28 - Electronic flight instrumentation<br/>system</b>    |                                  |                   |
| A                   | 0226-00C                     | Synthetic Vision System  | /                                | /                 |
| S                   | 0549-00A                     | Integrated Flight Deck System G3000 composed of :              |                                  |                   |
|                     |                              | - PFD1 GDU 1200W   | 8.29<br>(3.76)                   | 154.88<br>(3.934) |
|                     |                              | - PFD2 GDU 1200W   | 8.29<br>(3.76)                   | 154.88<br>(3.934) |
|                     |                              | - MFD GDU 1200W  | 8.29<br>(3.76)                   | 154.88<br>(3.934) |
|                     | 0565-00A                     | - Engine/Airframe Interface<br>Unit # 1 GEA 71                 | 2.58<br>(1.17)                   | 148.27<br>(3.766) |
|                     | 0565-00A                     | - Engine/Airframe Interface<br>Unit # 2 GEA 71                 | 2.58<br>(1.17)                   | 148.27<br>(3.766) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
| S                   | 0564-00A                     | - Integrated Avionics Unit # 1 GIA 63W                         | 7.19<br>(3.26)                   | 147.87<br>(3.756) |
|                     | 0564-00A                     | - Integrated Avionics Unit # 2 GIA 63W                         | 7.19<br>(3.26)                   | 147.87<br>(3.756) |
|                     | 0476-00A                     | - Touch Controller Unit # 1 GTC 580                            | 3.99<br>(1.81)                   | 156.69<br>(3.980) |
|                     | 0476-00A                     | - Touch Controller Unit # 2 GTC 580                            | 3.99<br>(1.81)                   | 156.69<br>(3.980) |
|                     | 0563-00A                     | - Audio box GMA 36   | 3.00<br>(1.36)                   | 149.02<br>(3.785) |
|                     |                              | <b>34-31 - Marker</b>  |                                  |                   |
|                     |                              | MARKER antenna DM N27-3  | 0.750<br>(0.340)                 | 129.92<br>(3.300) |
|                     | O 0541-23A                   | MARKER antenna 6216-82-00                                      | 0.990<br>(0.450)                 | 121.00<br>(3.065) |
|                     |                              | <b>34-41 - Stormscope</b>                                      |                                  |                   |
|                     | A 34056D                     | Stormscope WX 500, G3000 coupled :                             | 4.94<br>(2.24)                   | 232.28<br>(5.900) |
| S                   |                              | - Antenna NY163  | 0.84<br>(0.38)                   | 311.02<br>(7.900) |
|                     |                              | - Processor WX500  | 2.27<br>(1.03)                   | 255.91<br>(6.500) |
|                     |                              | <b>34-42 - Weather radar</b>                                   |                                  |                   |
|                     | 0394-34C                     | Weather radar GWX 70, of which                                 | 10.35<br>(4.47)                  | 169.10<br>(4.295) |
|                     |                              | - SAINT-GOBAIN radome 4906-100-V2                              | 2.82<br>(1.280)                  | 169.10<br>(4.295) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
| S                   | 0430-34A                     | New SAINT-GOBAIN radome 4906-100-V2                            | 2.82<br>(1.280)                  | 169.10<br>(4.295) |
|                     |                              | <b>34-43 - Radioaltimeter</b>                                  |                                  |                   |
| A                   | 0451-34B                     | GRA 55 radar altimeter, of which :                             | 4.127<br>(1.872)                 | 220.47<br>(5.600) |
|                     |                              | - Transceiver RA4500   | 3.527<br>(1.600)                 | 228.82<br>(5.812) |
|                     |                              | - Transmitting antenna S67-2002                                | 0.300<br>(0.136)                 | 182.09<br>(4.625) |
|                     |                              | and  |                                  |                   |
|                     |                              | - Receiving antenna S67-2002                                   | 0.300<br>(0.136)                 | 205.83<br>(5.228) |
|                     |                              | <b>34-44 - Traffic advisory system</b>                         |                                  |                   |
| A                   | 0176-00F                     | TAWS system  | /                                | /                 |
| A                   | 0258-00D                     | TAS system GTS 820, G3000 coupled,<br>of which :               | 22.53<br>(10.220)                | 177.68<br>(4.513) |
|                     |                              | - Processor GTS 820  | 9.92<br>(4.500)                  | 143.11<br>(3.635) |
|                     |                              | - Power amplifier/low noise amplifier GPA 65                   | 1.90<br>(0.860)                  | 221.42<br>(5.624) |
|                     |                              | - Antenna GA 58 (above fuselage)                               | 0.79<br>(0.360)                  | 230.71<br>(5.860) |
|                     |                              | - Antenna GA 58 (under fuselage)                               | 0.79<br>(0.360)                  | 260.63<br>(6.620) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment  | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m)  |
|---------------------|------------------------------|---|----------------------------------|--------------------|
|                     |                              | <b>34-51 - NAV 1 installation</b>   |                                  |                    |
| S                   |                              | VHF GS-NAV antenna DM N4-17N  | 3.307<br>(1.500)                 | 401.57<br>(10.200) |
| S                   |                              | Receiver<br>(integrated in the MOD70-0476-00A GIA 63W<br>Integrated Avionics Unit # 1 : refer to ATA 34-28) | /                                | /                  |
|                     |                              | <b>34-52 - NAV 2 installation</b>   |                                  |                    |
| S                   |                              | Receiver<br>(integrated in the MOD70-0476-00A GIA 63W<br>Integrated Avionics Unit # 2 : refer to ATA 34-28) | /                                | /                  |
|                     |                              | <b>34-53 - Transponder</b>  |                                  |                    |
| A                   | 0475-34A                     | Transponder # 1 GTX 33D - Mode S with antenna<br>diversity and extended squitter                            | 3.39<br>(1.54)                   | 147.68<br>(3.751)  |
|                     |                              | + Antenna KA 61 (under fuselage)  | 0.40<br>(0.18)                   | 150.08<br>(3.812)  |
|                     |                              | + Antenna KA 61 (above fuselage)  | 0.40<br>(0.18)                   | 193.22<br>(4.908)  |
| S                   | 0475-34B                     | Transponder # 1 GTX 33 - Mode S without antenna<br>diversity and with extended squitter                     | 3.11<br>(1.41)                   | 147.68<br>(3.751)  |
|                     |                              | + Antenna KA 61   | 0.40<br>(0.18)                   | 193.22<br>(4.908)  |
| A                   | 0475-34C                     | Transponder # 2 GTX 33 - Mode S with antenna<br>diversity and extended squitter                             | 3.11<br>(1.41)                   | 147.68<br>(3.751)  |
|                     |                              | + Antenna KA 61 (under fuselage)  | 0.40<br>(0.18)                   | 193.22<br>(4.908)  |
|                     |                              | + Antenna KA 61 (above fuselage)  | 0.40<br>(0.18)                   | 193.22<br>(4.908)  |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment   | Weight<br>per unit<br>lb<br>(kg)   | Arm<br>in.<br>(m)  |
|---------------------|------------------------------|--|--|--|
| S                   | 0542-34A                     | Transponder # 1 GTX 345R - full Mode S enhanced surveillance capability, without antenna diversity, with extended squitter and ADS-B IN acquisition (with ethernet connection)<br><br>+ Antenna KA 61  | 2.40<br>(1.09)<br><br>0.40<br>(0.18)   | 147.20<br>(3.738)<br><br>150.08<br>(3.812)   |
| O                   | 0542-34B                     | Transponder # 2 GTX 345R - full Mode S enhanced surveillance capability, without antenna diversity, with extended squitter and ADS-B IN acquisition (without ethernet connection)<br><br>+ Antenna KA 61   | 2.40<br>(1.09)<br><br>0.40<br>(0.18)   | 147.20<br>(3.738)<br><br>150.08<br>(3.812)   |
| O                   | 0542-34C                     | Transponder # 2 GTX 345R - full Mode S enhanced surveillance capability, without antenna diversity, with extended squitter and ADS-B IN acquisition (with ethernet connection)<br><br>+ Antenna KA 61<br><br><b>34-54 - Automatic Direction Finder (ADF)</b> | 2.40<br>(1.09)<br><br>0.40<br>(0.18)   | 147.20<br>(3.738)<br><br>150.08<br>(3.812)   |
| A                   | 0507-34A                     | ADF RA 3500 system (European countries only), of which :<br><br>- Receiver RA3502 P/N 0505.757-912<br><br>- Antenna AN3500 P/N 0832.601-912<br><br>- RMI converter AC3504 P/N 0856.010-912   | 7.61<br>(3.45)<br><br>2.205<br>(1.000)<br><br>3.754<br>(1.700)<br><br>1.653<br>(0.750) | 216.48<br>(5.498)<br><br>259.84<br>(6.000)<br><br>171.85<br>(4.365)<br><br>259.84<br>(6.000) |



| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>34-55 - DME installation</b>                                |                                  |                   |
| A                   | 34014G                       | DME KN63, G3000 coupled  | 2.80<br>(1.27)                   | 232.28<br>(5.900) |
|                     |                              | + Antenna KA 61  | 0.40<br>(0.18)                   | 238.82<br>(6.066) |
|                     |                              | <b>34-57 - Global Positioning System<br/>(GPS)</b>             |                                  |                   |
| S                   | 0476-00A                     | GPS/WAAS Antenna GA 36   | 0.48<br>(0.22)                   | 204.84<br>(5.203) |
| S                   | 0476-00A                     | GPS/WAAS and Datalink Antenna GA 37                            | 0.55<br>(0.25)                   | 204.84<br>(5.203) |
|                     |                              | <b>34-62 - Multifunction display</b>                           |                                  |                   |
| A                   | 0176-00G                     | Chartview function   | /                                | /                 |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>35 - Oxygen</b>   |                                  |                   |
| S                   | 0207-00                      | Gaseous oxygen system<br>With EROS oxygen system               | 22.73<br>(10.310)                | 226.77<br>(5.760) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>37 - Vacuum</b>   |                                  |                   |
| S                   |                              | Air ejector valve 19E17-5A                                     | 0.661<br>(0.300)                 | 116.14<br>(2.950) |
| S                   |                              | Regulator and relief valve 38E-96-2D                           | 1.323<br>(0.600)                 | 116.14<br>(2.950) |
| S                   |                              | Vacuum relief valve 691-21A                                    | 0.331<br>(0.150)                 | 139.76<br>(3.550) |
| S                   |                              | Valve 557-18 E   | 0.353<br>(0.160)                 | 118.11<br>(3.000) |

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| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>52 - Doors</b>  |                                  |                   |
| O                   | 0320-52B                     | New "Pilot" door   | 45.607<br>(20.687)               | 173.23<br>(4.400) |
| S                   | 0342-52                      | Additional landing gear doors                                  | 6.613<br>(3.000)                 | 204.33<br>(5.190) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg)    | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|-------------------------------------|-------------------|
| <b>56 - Windows</b> |                              |  |                                     |                   |
| S                   | 56001A                       | Deiced R.H. windshield   | $\Delta$ 1.764<br>( $\Delta$ 0.800) | 158.27<br>(4.020) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg)        | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|---|-------------------|
| <b>57 - Wings</b>   |                              |  |   |                   |
| S                   | 57001A                       | Utilization on runways covered<br>with melting snow            | $\Delta$ - 7.716<br>( $\Delta$ - 3.500) | 200.00<br>(5.080) |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>61 - Propeller</b>  |                                  |                   |
|                     |                              | <b>61-10 - Propeller assembly</b>                              |                                  |                   |
| S                   | 0345-61                      | Propeller (5-blade) HC-E5N-3C / NC 8834 K<br>+ spinner 104552P | 171.08<br>(77.60)                | 43.11<br>(1.095)  |
|                     |                              | <b>61-20 - Controls</b>  |                                  |                   |
| S                   |                              | Propeller governor 8210.007                                    | 2.646<br>(1.200)                 | 59.06<br>(1.500)  |
| R                   | 0445-72                      | Overspeed governor 1439292                                     | 2.535<br>(1.200)                 | 59.06<br>(1.330)  |

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>71 - Power plant</b>  |                                  |                   |
| R                   |                              | Turboprop engine PT6 A-66D                                     | 497.30<br>(226.00)               | 79.72<br>(2.025)  |
| S                   |                              | Top silentblocks 95007-16 (Qty 2)                              | 2.647<br>(1.201)                 | 79.72<br>(2.025)  |
| S                   |                              | Bottom silentblocks 95007-19 (Qty 2)                           | 2.654<br>(1.204)                 | 79.72<br>(2.025)  |
|                     |                              | <b>71-60 - Air inlet</b>                                       |                                  |                   |
| R                   | 0359-71                      | Inertial separator actuator<br>JA23372-1000-1                  | 2.156<br>(0.978)                 | 62.99<br>(1.600)  |



| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment           | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>77 - Engine indicating</b>  |                                  |                   |
| R                   |                              | Compressor turbine tachogenerator (Ng)<br>MIL-G-26611 GEU-7/A / 32005-00 | 0.981<br>(0.445)                 | 108.27<br>(2.750) |
| R                   |                              | Power turbine tachogenerator (Np)<br>MIL-G-26611 GEU-7/A / 32005-007     | 0.981<br>(0.445)                 | 55.12<br>(1.400)  |
| R                   | 0328-77                      | Torque transducer APTE-438-1000-75D                                      | 0.473<br>(0.215)                 | 54.84<br>(1.393)  |
|                     |                              | <b>77-12 - Fuel management</b>   |                                  |                   |
| S                   |                              | Fuel flow transmitter 660 526AS  | 0.683<br>(0.310)                 | 110.20<br>(2.799) |

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| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>79 - Lubrication</b>  |                                  |                   |
|                     |                              | <b>79-20 - Distribution</b>                                    |                                  |                   |
| R                   |                              | Oil cooler L8538233  | 10.472<br>(4.750)                | 90.55<br>(2.300)  |
|                     |                              | <b>79-30 - Indicating</b>                                      |                                  |                   |
| R                   | 0327-79A                     | Oil pressure transmitter<br>APT-369A-1000-150G (5 VDC)         | 0.337<br>(0.153)                 | 105.35<br>(2.676) |
| S                   | 0512-79                      | Chip detector monitoring                                       | Neglig.                          | /                 |

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**List of supplements and validities**

| Sup.<br>No.   | Edit.<br>No. | Description   | Airplane type * |           |           |           |           |           |           |
|---|--------------|---|-----------------|-----------|-----------|-----------|-----------|-----------|-----------|
|   |              |   | 900<br>E0       | 900<br>E1 | 910<br>E0 | 910<br>E1 | 930<br>E0 | 930<br>E1 | 940<br>E0 |
| A   | 5            | List of supplements<br>From S/N 1000, plus S/N 687  | X               | X         | X         | X         | X         | X         | X         |
| 06  | 3            | WX-500 stormscope<br>OPT70-34-056<br>From S/N 1000, plus S/N 687  | X               | X         | X         | X         | X         | X         | X         |
| 18  | 3            | Engine fire detection system<br>OPT70-26-002G<br>OPT70-26-002H<br>MOD70-0496-26A<br>From S/N 1000, plus S/N 687 | X               | X         | X         | X         | X         | X         | X         |
| 45  | 2            | Mexico specifics<br>MOD70-0212-11<br>From S/N 1000 up to S/N 1269, plus<br>S/N 687                              | X               | X         | X         | X         | X         | X         | X         |
| 45  | 3            | Mexico specifics<br>MOD70-0619-11 Version C<br>From S/N 1270  |                 |           |           | X         |           |           | X         |
| 47  | 1            | GARMIN GWX70 color weather radar<br>MOD70-0394-34<br>From S/N 1000 to S/N 1049, plus<br>S/N 687                 | X               |           |           |           |           |           |           |
| 49  | 2            | GARMIN TAWS System<br>MOD70-0176-00 Version F<br>From S/N 1000, plus S/N 687                                    | X               | X         | X         | X         | X         | X         | X         |
| 50  | 2            | GARMIN Synthetic Vision System<br>MOD70-0226-00<br>From S/N 1000, plus S/N 687                                  | X               | X         | X         | X         | X         | X         | X         |
| * 900 E0 : From S/N 1000 to S/N 1049, plus S/N 687      |              |   |                 |           |           |           |           |           |           |
| 900 E1 : From S/N 1050                                  |              |   |                 |           |           |           |           |           |           |
| 910 E0 : From S/N 1170 to 1269                          |              |   |                 |           |           |           |           |           |           |
| 910 E1 : Airplane equipped with MOD70-0539-00 Version H |              |   |                 |           |           |           |           |           |           |
| 930 E0 : Airplane equipped with MOD70-476-00            |              |   |                 |           |           |           |           |           |           |
| 930 E1 : Airplane equipped with MOD70-549-00            |              |   |                 |           |           |           |           |           |           |
| 940 E0 : Airplane equipped with MOD70-0649-00 Version A |              |   |                 |           |           |           |           |           |           |

| Sup. No.  | Edit. No. | Description   | Airplane type * |        |        |        |        |        |        |
|---|-----------|---|-----------------|--------|--------|--------|--------|--------|--------|
|   |           |   | 900 E0          | 900 E1 | 910 E0 | 910 E1 | 930 E0 | 930 E1 | 940 E0 |
| 56  | 3         | GARMIN GSR56 weather datalink and satellite phone<br>MOD70-0331-23<br>From S/N 1000, plus S/N 687                                 | X               | X      | X      | X      | X      | X      | X      |
| 57  | 1         | Public transportation for French-registered airplanes<br>MOD70-0352-11<br>From S/N 1000, plus S/N 687 with MOD70-0176-00          | X               | X      |        |        |        |        |        |
| 58  | 0         | Five-bladed propeller<br>MOD70-0345-61<br>From S/N 1000 up to S/N 1049, plus S/N 687  | X               |        |        |        |        |        |        |
| 59  | 1         | Brazil specifics<br>OPT70-01004<br>From S/N 1000 up to S/N 1269, plus S/N 687   | X               | X      | X      |        | X      | X      |        |
| 59  | 3         | Brazil specifics<br>MOD70-0619-11 Version B<br>From S/N 1270  |                 |        |        | X      |        |        | X      |
| 60  | 1         | ADS-B OUT function<br>MOD70-0264-34<br>MOD70-0542-34<br>From S/N 1000 up to S/N 1159, plus S/N 687 with MOD70-0176-00             | X               | X      |        |        |        |        |        |
| 62  | 0         | Flight envelope protection<br>MOD70-0423-34 and<br>MOD70-0488-27<br>From S/N 1000 up to S/N 1169, plus S/N 687 with MOD70-0176-00 | X               | X      |        |        |        |        |        |
| * 900 E0 : From S/N 1000 to S/N 1049, plus S/N 687      |           |   |                 |        |        |        |        |        |        |
| 900 E1 : From S/N 1050                                  |           |   |                 |        |        |        |        |        |        |
| 910 E0 : From S/N 1170 to 1269                          |           |   |                 |        |        |        |        |        |        |
| 910 E1 : Airplane equipped with MOD70-0539-00 Version H |           |   |                 |        |        |        |        |        |        |
| 930 E0 : Airplane equipped with MOD70-476-00            |           |   |                 |        |        |        |        |        |        |
| 930 E1 : Airplane equipped with MOD70-549-00            |           |   |                 |        |        |        |        |        |        |
| 940 E0 : Airplane equipped with MOD70-0649-00 Version A |           |   |                 |        |        |        |        |        |        |

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| Sup.<br>No.  | Edit.<br>No. | Description  | Airplane type * |           |           |           |           |           |           |
|--|--------------|--|-----------------|-----------|-----------|-----------|-----------|-----------|-----------|
|  |              |  | 900<br>E0       | 900<br>E1 | 910<br>E0 | 910<br>E1 | 930<br>E0 | 930<br>E1 | 940<br>E0 |
| 63   | 1            | Lavatory compartment<br>MOD70-0505-25<br>From S/N 1000, plus S/N 687   | X               | X         | X         | X         | X         | X         | X         |
| 64   | 1            | Stick Shaker<br>MOD70-0510-27 Version C<br>From S/N 1000 to S/N 1049, plus<br>S/N 687  | X               |           |           |           |           |           |           |
| 66   | 1            | GARMIN G1000 NXi retrofit<br>MOD70-0539-00<br>From S/N 1000 to S/N 1169  | X               | X         |           |           |           |           |           |
| 67   | 0            | Data collection and transmission<br>system (FAST BOX)<br>MOD70-0578-31 Version A<br>From S/N 1000, plus S/N 687  | X               | X         | X         |           | X         |           |           |
| 68   | 0            | TBM930 2018<br>From S/N 1216   |                 |           |           |           | X         |           |           |
| 69   | 0            | Baro-VNAV approaches<br>MOD70-0636-34<br>From S/N 1000, plus S/N 687,<br>equipped with G1000 system<br>(MOD70-0176-00) or G1000 NXi<br>system retrofit (MOD70-0539-00<br>Versions D & E) | X               | X         | X         |           |           |           |           |
| 71   | 0            | JCAB Supplement for TBM940<br>airplane<br>MOD70-0619-11 Version D<br>From S/N 1272   |                 |           |           |           |           |           | X         |
| 72   | 0            | France specifics<br>MOD70-0619-11 Version E<br>From S/N 1270   |                 |           |           | X         |           |           | X         |
| * 900 E0 : From S/N 1000 to S/N 1049, plus S/N 687<br>900 E1 : From S/N 1050<br>910 E0 : From S/N 1170 to 1269<br>910 E1 : Airplane equipped with MOD70-0539-00 Version H<br>930 E0 : Airplane equipped with MOD70-476-00<br>930 E1 : Airplane equipped with MOD70-549-00<br>940 E0 : Airplane equipped with MOD70-0649-00 Version A |              |  |                 |           |           |           |           |           |           |

| Sup.<br>No.  | Edit.<br>No. | Description  | Airplane type * |           |           |           |           |           |           |
|--|--------------|--|-----------------|-----------|-----------|-----------|-----------|-----------|-----------|
|  |              |  | 900<br>E0       | 900<br>E1 | 910<br>E0 | 910<br>E1 | 930<br>E0 | 930<br>E1 | 940<br>E0 |
| 73   | 0            | Extended large storage cabinet<br>MOD70-0684-25<br>From S/N 1270 |                 |           |           | X         |           |           | X         |
| 75   | 0            | GDR66 COM3 VHF communication<br>MOD70-0722-23<br>From S/N 1272   |                 |           |           |           |           |           | X         |
| * 900 E0 : From S/N 1000 to S/N 1049, plus S/N 687<br>900 E1 : From S/N 1050<br>910 E0 : From S/N 1170 to 1269<br>910 E1 : Airplane equipped with MOD70-0539-00 Version H<br>930 E0 : Airplane equipped with MOD70-476-00<br>930 E1 : Airplane equipped with MOD70-549-00<br>940 E0 : Airplane equipped with MOD70-0649-00 Version A |              |  |                 |           |           |           |           |           |           |

## **SUPPLEMENT**

### **WX-500 stormscope**

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| 4 - Normal procedures .....                   | 9.6.3 |
| 5 - Performance .....                         | 9.6.3 |
| 6 - Weight and balance .....                  | 9.6.4 |
| 7 - Description .....                         | 9.6.4 |
| 8 - Handling, Servicing and maintenance ..... | 9.6.4 |



## SECTION 1

### General

This supplement is intended to inform the pilot about the equipment limitations, description and operations necessary for operation when the TBM airplane is equipped with the option WX-500 stormscope.

Whenever this supplement refers to the WX-500 Pilot's Guide, it states the one described in section 2.

## SECTION 2

### Limitations

The limitations hereafter supplement or replace those of the standard airplane described in section 2 Limitations of the basic POH when the airplane is equipped with the option WX-500 stormscope.

The WX-500 stormscope systems signal displays are not intended for the purpose of penetrating thunderstorm areas or areas of severe turbulence ; such intentional use is prohibited.

#### • NOTE •

Range selection determines receiver sensitivity and therefore relative range. Displayed range is based on signal strength and is not to be used for accurate determination of thunderstorm location.

•

#### ▲ CAUTION ▲

The stormscope must not be used for thunderstorm penetration.

▲

The WX-500 Pilot's guide, Series II, No. 009-11501-001 and the GARMIN Integrated flight deck pilot's guide, as applicable, at their latest revision shall be readily available to the pilot, whenever the operation of the WX-500 stormscope is predicted.

## **SECTION 3**

### **Emergency procedures**

Installation and operation of WX-500 stormscope do not change the basic emergency procedures of the airplane described in section 3 Emergency procedures of the basic POH.

## **SECTION 4**

### **Normal procedures**

Normal operating procedures of the WX-500 stormscope are outlined in the WX-500 Pilot's Guide.

## **SECTION 5**

### **Performance**

Installation and operation of WX-500 stormscope do not change the basic performance of the airplane described in section 5 Performance of the basic POH.

## SECTION 6

### Weight and balance

The weight and balance hereafter supplement or replace those of the standard airplane described in section 6 Weight and balance of the basic POH when the airplane is equipped with the option WX-500 stormscope.

| S/<br>R/<br>A/<br>O    | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|------------------------|------------------------------|--|----------------------------------|-------------------|
| <b>34 - NAVIGATION</b> |                              |  |                                  |                   |
| A                      | 34056                        | Stormscope<br>- shared with the integrated flight deck system  | 4.94<br>(2.240)                  | 232.28<br>(5.900) |

## SECTION 7

### Description

Information hereafter supplement or replace those of the standard airplane described in section 7 Description of the basic POH when the airplane is equipped with the option WX-500 stormscope.

The WX-500 (series II) stormscope, weather mapping system provides a visual screen readout of the electrical discharges associated with thunderstorms. This information with proper interpretation, will allow the pilot to detect severe thunderstorm activity. A series of green dots or of strike points will be displayed on the screen to indicate the electrical discharge areas.

The WX-500 (series II) stormscope, weather mapping system interfaces with the integrated flight deck system.

## SECTION 8

### Handling, servicing and maintenance

Installation and operation of WX-500 stormscope do not change the handling, servicing and maintenance procedures of the airplane described in section 8 Handling, servicing and maintenance of the basic POH.

## SUPPLEMENT

### Engine fire detection system

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| 4 | - Normal procedures .....                   | 9.18.5 |
| 5 | - Performance .....                         | 9.18.5 |
| 6 | - Weight and balance .....                  | 9.18.6 |
| 7 | - Description .....                         | 9.18.7 |
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## **SECTION 1**

### **General**

This supplement is intended to inform the pilot about the equipment limitations, description and operations necessary for operation when the airplane is equipped with the option Engine fire detection system.

The general hereafter supplement or replace those of the standard airplane described in section 1 General of the basic POH when the airplane is equipped with the option Engine fire detection system.

The fire detection system allows engine fire monitoring and indicating.

## **SECTION 2**

### **Limitations**

Installation and operation of Engine fire detection system do not change the basic limitations of the airplane described in section 2 Limitations of the basic POH.

## SECTION 3

### Emergency procedures

The emergency procedures hereafter supplement or replace those of the standard airplane described in section 3 Emergency procedures of the basic Pilot's Operating Handbook when the airplane is equipped with the option Engine fire detection system.

#### Engine fire on ground

Symptoms : ITT increasing, **ITT** , **FIRE** , smoke, ...

1 - THROTTLE ..... CUT OFF

>> Airplane with G1000 or G1000 NXi Flight deck (MOD70-0176-00 or MOD70-0539-00)

2 - BLEED switch ..... OFF / RST

>> Airplane with G3000 Flight deck (MOD70-0476-00)

3 - BLEED switch ..... OFF

>> All

4 - A/C switch ..... OFF

5 - Brakes ..... As required

6 - FUEL TANK SLECTOR ..... OFF

7 - Warn ground assistance, if necessary

8 - Crash lever ..... Pull down

► Evacuate as soon as possible ◄

## Engine fire in flight

Symptoms : **FIRE**

Try to confirm the fire warning by looking for other indications such as ITT increase, **ITT** , smoke from engine cowls or air conditioning system.

▲ **CAUTION** ▲

No air start attempt after an engine fire.



► Fly the airplane ◀

*If the fire warning is not confirmed :*

- 1 - Monitor the engine parameters, ITT in particular
- 2 - Look for smoke coming from engine cowls or from air conditioning system

► Land as soon as possible ◀

*If the fire warning is confirmed :*

- |     |                          |         |
|-----|--------------------------|---------|
| 1 - | THROTTLE .....           | CUT OFF |
| 2 - | AUX BP switch .....      | OFF     |
| 3 - | FUEL TANK SELECTOR ..... | OFF     |
| 4 - | Oxygen mask .....        | Use     |

>> *Airplane with G1000 or G1000 NXi Flight deck (MOD70-0176-00 or MOD70-0539-00)*

- 5 - BLEED switch ..... OFF / RST

>> *Airplane with G3000 Flight deck (MOD70-0476-00)*

- 6 - BLEED switch ..... OFF

>> *All*

- 7 - A/C switch ..... OFF

- 8 - If necessary, ..... Emergency descent

- 9 - Perform ..... Forced landing

## SECTION 4

### Normal procedures

The normal procedures hereafter supplement or replace those of the standard airplane described in section 4 Normal procedures of the basic Pilot's Operating Handbook when the airplane is equipped with the option Engine fire detection system.

- Before starting the engine

■ >> *Up to S/N 1105, plus S/N 687, on left side of left instrument panel*

FIRE TEST push-button ..... Press

■ >> *From S/N 1106, on upper panel*

TEST push-button ..... Press

■ >> *All*

**FIRE** lights on and causes the illumination of the MASTER WARNING light.

## SECTION 5

### Performance

Installation and operation of Engine fire detection system do not change the basic performance of the airplane described in section 5 Performance of the basic Pilot's Operating Handbook.



## SECTION 6

### Weight and balance

The weight and balance hereafter supplement or replace those of the standard airplane described in section 6 Weight and balance of the basic Pilot's Operating Handbook when the airplane is equipped with the option Engine fire detection system.

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment                    | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|---|----------------------------------|-------------------|
|                     |                              | <b>26 - Fire protection</b>   |                                  |                   |
| A                   | 26002G<br>or 26002 H         | Engine fire detection system L'HOTELLIER<br>(From S/N 1000 to 1105, plus S/N 687) | 1.455<br>(0.660)                 | 96.06<br>(2.440)  |
| A                   | 0496-26A                     | Engine fire detection system L'HOTELLIER<br>(From S/N 1106)                       | 1.464<br>(0.66)                  | 96.06<br>(2.440)  |

## SECTION 7

### Description

Information hereafter supplement or replace those of the standard airplane described in section 7 Description of the basic Pilot's Operating Handbook when the airplane is equipped with the option Engine fire detection system.

The engine fire detection system enables the monitoring and indication of a fire in the engine area.

The system includes :

- 7 detectors
- the test push-button
- the integrated flight deck system.

#### Detectors

The 7 detectors are secured on supports positioned in the most sensitive engine areas. They consist of thermal switches detecting a temperature greater than 200°C.

#### Push-button

The push-button enables the pilot to test the detection system by opening the grounding circuit. It is connected in series with the 7 detectors.

>> *Up to S/N 1105, plus S/N 687*

The FIRE TEST push-button is located on left side of left instrument panel.

>> *From S/N 1106*

The TEST push-button is located on upper panel.

#### Display

Refer to the GARMIN Integrated Flight Deck Pilot's Guide, as applicable, at its latest revision.

## SECTION 8

### Handling, servicing and maintenance

Installation and operation of Engine fire detection system do not change the basic handling, servicing and maintenance procedures of the airplane described in section 8 Handling, Servicing and Maintenance of the basic Pilot's Operating Handbook.

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# **SUPPLEMENT**

## **Mexico specifics**

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| 7 - Description .....                         | 9.45.19 |
| 8 - Handling, servicing and maintenance ..... | 9.45.19 |

## **SECTION 1**

### **General**

This supplement is intended to inform the pilot about the airplane specifics, among others those required by the relevant Certification Authorities (limitations, description and operations necessary to the operation of the TBM airplane).

## **SECTION 2**

### **Limitations**

The limitations hereafter supplement or replace those of the standard airplane described in section 2 Limitations of the basic POH.

### **2.9 - Placards**

#### **Internal placards**

- 1 - Rear pressurized baggage compartment (in cabin)

On partition wall

**MÁXIMO 100 kg - (220 lbs)**

**ES RESPONSABILIDAD DEL PILOTO  
COMPROBAR QUE TODO EL EQUIPAJE ESTÁ  
ASEGURADO CORRECTAMENTE.  
PARA INSTRUCCIONES DE CARGA REFIERASE A  
LOS "DATOS DE PESO Y BALANCE"  
DEL MANUAL DE OPERACIÓN DEL PILOTO.**

Pilot's Operating Handbook

For the small cargo net, on frame C13bis



For the large cargo net, on R.H. Side upholstery panel, in the rear baggage compartment



2 - Non pressurized FWD baggage compartment

On baggage compartment door frame

**MÁXIMO 50 kg - (110 lbs)**

**PARA INSTRUCCIONES DE CARGA REFIERASE  
A LOS "DATOS DE PESO Y BALANCE"  
DEL MANUAL DE OPERACIÓN DEL PILOTO.**

- 3 - On R.H. side at front seat level and on the first rear passengers masks container (R.H. side on the ceiling)

14112003AAA/FMA18101



- 4 - On rear passengers masks containers (on R.H. side on the ceiling and left side)

14112003AAA/FMA18201



- 5 - On rear passenger's table casing

LA MESA DEBE ESTAR GUARDADA DURANTE EL DESPEGUE Y ATERRIZAJE.

## Pilot's Operating Handbook

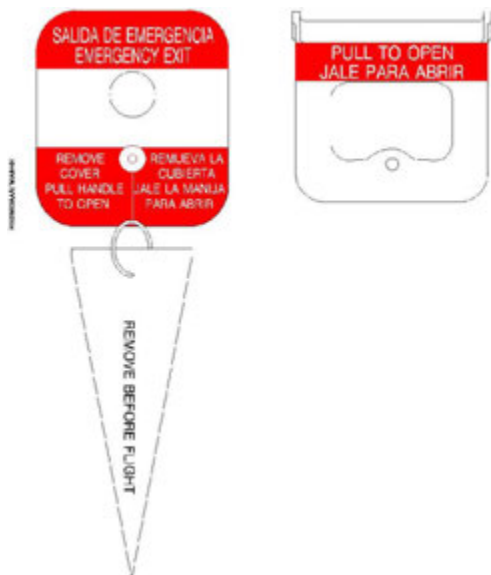
- 6 - Door internal side  
On access door



On pilot door, if installed



- 7 - On emergency exit handle





- 8 - On landing gear emergency control access door

14112003AAAMA18400

**LDG GEAR EMERGENCY  
ACCESS PULL  
TREN DE ATERRIZAJE  
DE EMERGENCIA  
JALE AQUI**

- 9 - At the upper corner of the window on each side of the cockpit

14112003AAAHMA8301

**HANDLE  
HERE  
MANIJA  
AQUI**

- 10 - On cabinet drawer (optional)

14112003AAALMA18000

**GAFAS PARA HUMO**

>> *Airplane equipped with coat hanger (Post-MOD70-0557-25)*

- 11 - On the upper edge of the L.H. Passenger access door panel

14113200AAAKMA18200

**SOLO PRENDAS DE VESTIR**

## Pilot's Operating Handbook

>> *Airplane equipped with lavatory compartment (Post-MOD70-0505-25)*

12 - On fixed panel, cabin side

I4113200AAKMA8300

**EL DIVISOR DEBE ESTAR ALMACENADO DURANTE EL DESPEGUE Y EL ATERRIZAJE**

13 - On fixed panel, toilet side

I4113200AAKMA8000

**EL INODORO NO DEBE  
ESTAR OCUPADO DURANTE EL  
DEPEGUE Y EL ATERRIZAJE**

**CIERRE Y ASEGURE LA TAPA  
DEL INODORO CUANDO NO ESTÉ EN USO**

**NO CUELQUE O GUARDE OBJETOS  
EN EL BAÑO O DIVISOR**

**EL DIVISOR DEBE ESTAR ALMACENADO  
DURANTE EL DESPEGUE Y EL ATERRIZAJE**

**USE LOS AURICULARES CUANDO EL  
DIVISOR ESTÉ DESPLEGADO**

- 14 - On access door, cabin side and toilet side

14113200AAAKMA8200



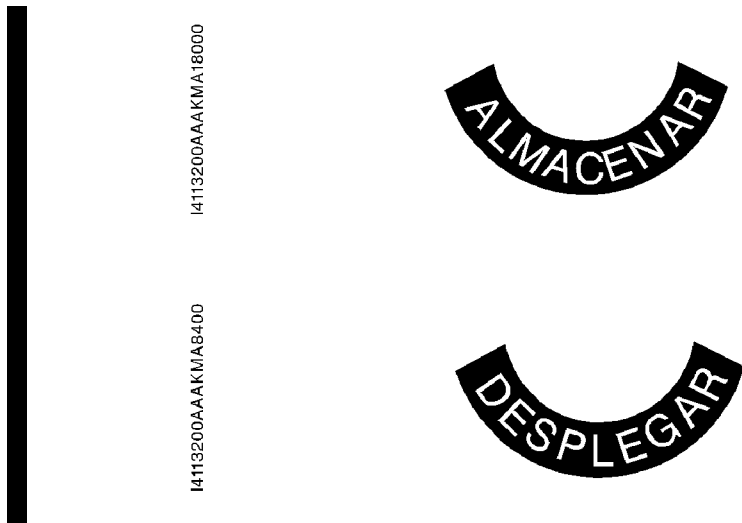
- 15 - Behind access door, cabin side and toilet side

14113200AAAKMA8100



Pilot's Operating Handbook

16 - Front face of lavatory compartment, near opening / closing switches



17 - On the magazine rack

I4113200AAAKMA18100

1,5 KG (3.3 LBS)

>> All

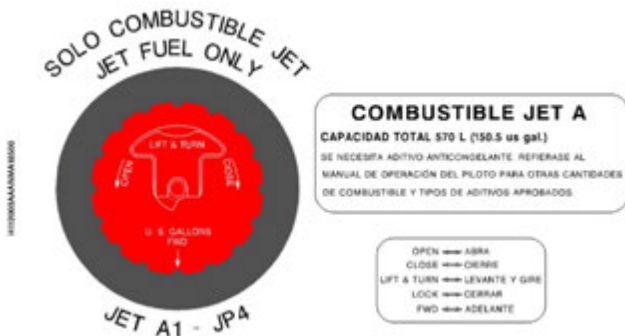
## External placards

18 - Under engine cowling and under each wing

I4112003AAA-HMA18300



19 - Near fuel tank caps



I4112003AAA-HMA18300

I4112003AAA-HMA8201



## Pilot's Operating Handbook

- 20 - Above brakes hydraulic fluid reservoir against firewall

I4112003AAA HMA18101

FRENOS  
MIL - H - 5606  
AIR 3520  
FLUIDO HIDRÁULICO

- 21 - On landing gear hydraulic fluid reservoir

I4112003AAA HMA18001

GEARS  
TRENES  
  
MIL - H - 5606  
AIR 3520  
  
HYDRAULIC FLUID  
FLUIDO HIDRÁULICO

- 22 - On fuse box in engine cowling

I4112003AAA HMA18009

CAJA DE FUSIBLES  
Y FOCOS

23 - On internal face of L.H. engine cowling

I4112003AAAEMA5300



I4112003AAA-HMA8101

| OILS - ACEITES           |                                      |
|--------------------------|--------------------------------------|
| <input type="checkbox"/> | AEROSHELL 560                        |
| <input type="checkbox"/> | EXXON 2380 OR ESSO 2380 OR BPTO 2380 |
| <input type="checkbox"/> | MOBIL JET OIL II                     |
| <input type="checkbox"/> | MOBIL JET OIL 254                    |
| <input type="checkbox"/> | AERO SHELL TURBINE OIL 500           |
| <input type="checkbox"/> | ROYCO TURBINE OIL 500                |
| <input type="checkbox"/> | CASTROL 5000                         |
| <input type="checkbox"/> | TURBONYCOIL 525-2A                   |

24 - On front lower portion of firewall L.H. side

M112003AAA-HMA8401



Pilot's Operating Handbook

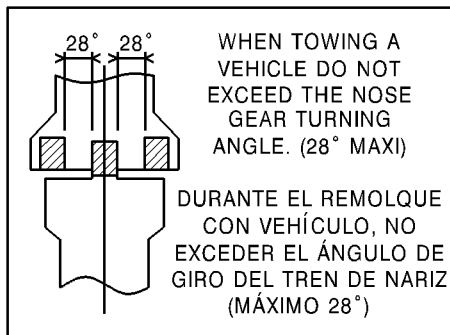
25 - On engine cowling, in front of compartment door

I4112003AAAGMA18500

**ALIMENTACIÓN EXTERNA:  
28 VOLTS C.D. NOMINAL.  
CAPACIDAD MÍNIMA DE ARRANQUE:  
800 AMPS  
NO EXCEDER 1000 AMPS**

26 - On nose gear door

I4112003AAAEIMA18101



27 - On nose gear leg

I4112003AAAIMA8200

**TREN DE ATERRIZAJE  
DE NARIZ**

**PRESIÓN DE LLANTA: 6,5 bar  
94 psi**



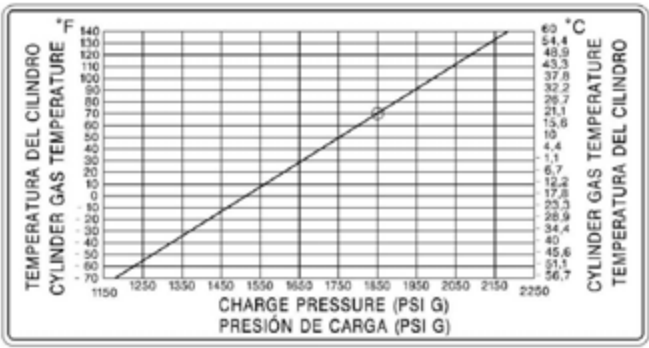
28 - On main gear leg

I4112003AAA IMA 8300

TREN DE ATERRIZAJE  
PRINCIPAL  
  
PRESIÓN DE LLANTA: 8,96 bar  
130 psi

29 - On internal face of the oxygen cylinder service door

I4112003AAA IMA 18301



30 - On the oxygen service door

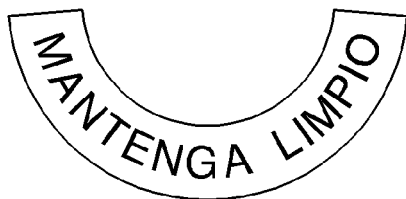
I4112003AAA IMA 18101

PUNTO DE SERVICIO PARA  
OXÍGENO. NO USAR LUBRICANTES

Pilot's Operating Handbook

- 31 - Near air data system port

I4112003AAA HMA8001



- 32 - On external side of emergency locator transmitter inspection door

I4112003AAA HMA1B400



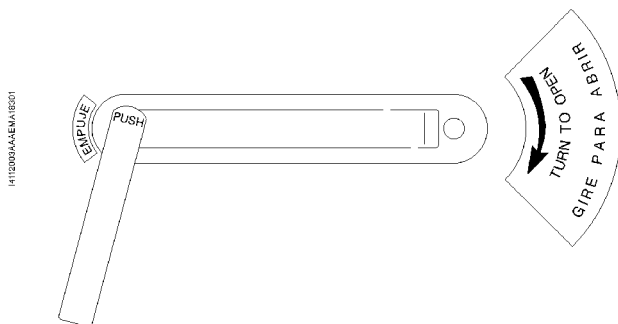
- 33 - On emergency exit external side

I4112003AAAFMAB500

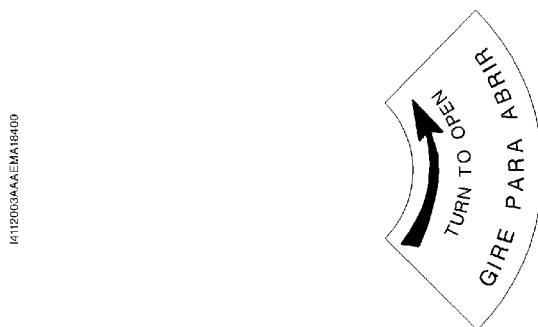


34 - Door external side

On pilot door



On access door



On outer fuselage skin aft of access door and in the cabin forward of access door



## Pilot's Operating Handbook

35 - On last step of stairs

**CARGA MÁXIMA SOBRE LA ESCALERA : UNA PERSONA**

36 - On R.H. access door jamb

14112003MANK/MA118001

**NO USAR EL  
PASAMANO  
PARA RETRAER  
O GUARDAR LA  
ESCALERA**

### SECTION 3

#### Emergency procedures

No specifics

### SECTION 4

#### Normal procedures

No specifics

### SECTION 5

#### Performance

No specifics

### SECTION 6

#### Weight and balance

The weight and balance hereafter supplement or replace those of the standard airplane described in section 6 Weight and balance of the basic POH.

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>01 - Specific optional equipment</b>                        |                                  |                   |
| S                   | 0212-11                      | Mexico certification markings                      SOCATA      | /                                | /                 |

## **SECTION 7**

### **Description**

No specifics

## **SECTION 8**

### **Handling, servicing and maintenance**

No specifics

Intentionally left blank

## **SUPPLEMENT**

### **GARMIN TAWS system**

#### **Table of contents**

|   | Page    |
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| 1 - General .....                             | 9.49.2  |
| 2 - Limitations .....                         | 9.49.2  |
| 3 - Emergency procedures .....                | 9.49.3  |
| 4 - Normal procedures .....                   | 9.49.4  |
| 5 - Performance .....                         | 9.49.6  |
| 6 - Weight and balance .....                  | 9.49.6  |
| 7 - Description .....                         | 9.49.7  |
| 8 - Handling, servicing and maintenance ..... | 9.49.14 |



## SECTION 1

### General

This supplement is intended to inform the pilot about the equipment limitations, description and operations necessary for operation when the airplane is equipped with the option GARMIN TAWS system.

The TAWS function enables to detect if the airplane path is in compliance with the overflowed terrain relief.

## SECTION 2

### Limitations

The limitations hereafter supplement or replace those of the standard airplane described in section 2 Limitations of the basic POH when the airplane is equipped with the option GARMIN TAWS system.

The TAWS function provides terrain proximity alerting and detection to the pilot. It must not be used for airplane vertical and horizontal navigation.

AC 2318 recommendation : in order to avoid unwillingly warnings, TAWS function must be inhibited for any landing on a terrain which is not mentioned in the data base.

The use of the terrain awareness warning and terrain display functions is prohibited during QFE (atmospheric pressure at airport elevation) operations.

■ >> *Airplane equipped with GARMIN flight deck as standard*

The GARMIN Integrated Flight Deck Pilot's Guide mentioned in section 2 Limitations of the basic POH, as applicable, or any further edition applicable to the latter, shall be readily available to the pilot, whenever the operation of TAWS system is predicted.

■ >> *Airplane retrofitted with GARMIN G1000 NXi flight deck (MOD70-0539-00)*

The GARMIN G1000 NXi Integrated Flight Deck Pilot's Guide for the TBM850/900 P/N 190-02348-00 or any further edition applicable to the latter, shall be readily available to the pilot, whenever the operation of TAWS system is predicted.

## SECTION 3

### Emergency procedures

The emergency procedures hereafter supplement or replace those of the standard airplane described in section 3 Emergency procedures of the basic POH when the airplane is equipped with the option GARMIN TAWS system.

#### TAWS FAIL annunciation

The TAWS function is not operational.

**SECTION 4**

**Normal procedures**

The normal procedures hereafter supplement or replace those of the standard airplane described in section 4 Normal Procedures of the basic POH when the TBM airplane is equipped with the option GARMIN TAWS system.

**Before takeoff**

- "TAWS System Test OK" voice message ..... Heard  
*End of procedure.*

**4.1 - Warnings of the TAWS function**

**"PULL UP" voice alert**

**PULL UP** PFD/MFD alert annunciation and **PULL UP** MFD pop-up alert light ON.

- 1 - Level the wings.
- 2 - TRQ ..... Maximum
- 3 - Choose the optimum rate of climb adapted to airplane configuration and speed, until the warning disappears.

*End of procedure.*

**"Terrain Terrain, Pull up Pull up",  
"Obstacle Obstacle, Pull up Pull up", voice alerts**

**PULL UP** PFD/MFD alert annunciation and **TERRAIN/OBSTACLE - PULL UP** MFD pop-up alert light ON.

- 1 - Adjust airplane path in order to make the warning disappear.  
*End of procedure.*

## 4.2 - Cautions of the TAWS function

"Caution terrain", "Caution obstacle",  
"Too low terrain" voice alerts

**TERRAIN** PFD/MFD alert annunciation and **CAUTION TERRAIN/OBSTACLE**  
or **TOO LOW TERRAIN** MFD pop-up alerts light ON.

- 1 - Adjust airplane path in order to make the warning disappear.

*End of procedure.*

"Don't sink" voice alert

**TERRAIN** PFD/MFD alert annunciation and **DON'T SINK** MFD pop-up alert  
light ON.

- 1 - Re-establish a positive rate of climb.

*End of procedure.*

"Sink rate" voice alert

**TERRAIN** PFD/MFD alert annunciation and **SINK RATE** MFD pop-up alert light  
ON.

- 1 - Reduce rate of descent.

*End of procedure.*

## SECTION 5

### Performance

Installation and operation of GARMIN TAWS system do not change the basic performance of the airplane described in section 5 Performance of the basic POH.

## SECTION 6

### Weight and balance

The weight and balance hereafter supplement or replace those of the standard airplane described in section 6 Weight and balance of the basic POH when the airplane is equipped with the option GARMIN TAWS system.

| S/<br>R/<br>A/<br>O    | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|------------------------|------------------------------|--|----------------------------------|-------------------|
| <b>34 - Navigation</b> |                              |  |                                  |                   |
| A                      | 0176-00<br>Version F         | TAWS system<br><br>GARMIN                                      | /                                | /                 |

## SECTION 7

### Description

Information hereafter supplement or replace those of the standard airplane described in section 7 Description of the basic POH when the TBM airplane is equipped with the option GARMIN TAWS system.

>> *Airplane with G3000 Flight deck (MOD70-0476-00)*

#### **TAWS-B terrain and obstacle alerts**

- Alerts include visual annunciations and voice alerts.
- Alerts are accompanied by visual annunciation on PFD's and pop-up alerts on either Touchscreens Controllers
- Pilot acknowledges the Alert on the Touchscreen Controller

#### **Voice alerts inhibiting**

- TAWS Alerts can be inhibited by the pilot selecting Inhibit TAWS on Touchscreens Controllers
- Discretion should be used when inhibiting alerts and the system should be enabled when appropriate.

>> *All*

The TAWS function has 7 modes.

#### **1. Forward Looking Terrain Avoidance alert**

The Forward Looking Terrain Avoidance (FLTA) alert is used by TAWS and is composed of :

- **Reduced Required Terrain Clearance and Reduced Required Obstacle Clearance**

Reduced Required Terrain Clearance (RTC) and Reduced Required Obstacle Clearance (ROC) alerts are issued when the airplane flight path is above terrain, yet is projected to come within the minimum clearance values in table 9.49.1. When an RTC or ROC alert is issued, a potential impact point is displayed on the TAWS Page.

- **Imminent Terrain Impact and Imminent Obstacle Impact**

Imminent Terrain Impact (ITI) and Imminent Obstacle Impact (IOI) alerts are issued when the airplane is below the elevation of a terrain or obstacle cell in the airplane's projected path. ITI and IOI alerts are accompanied by a potential impact point displayed on the TAWS Page. The alert is annunciated when the projected vertical flight path is calculated to come within minimum clearance altitudes in table 9.49.1.

| Phase of flight | Minimum Clearance<br>Altitude Level Flight (ft) | Minimum Clearance<br>Altitude Descending (ft) |
|-----------------|---|---|
| Enroute         | 700   | 500   |
| Terminal        | 350   | 300   |
| Approach        | 150   | 100   |
| Departure       | 100   | 100   |

Table 9.49.1 - Minimum Terrain and Obstacle Clearance values for FLTA alerts

During the final approach phase of flight, FLTA alerts are automatically inhibited when the airplane is below 200 feet AGL while within 0.5 Nm of the approach runway or below 125 feet AGL while within 1.0 Nm of the runway threshold.

## Pilot's Operating Handbook

The aural/displayed messages associated with the FLTA function are described in the table 9.49.2.

| Alert Type  | PFD/MFD<br>TAWS Page<br>Annunciation | MFD Map Page<br>Pop-Up Alert    | Aural Message  |
|---|--------------------------------------|---------------------------------|--|
| Reduced Required<br>Terrain Clearance<br>Warning (RTC)<br>(Red)     | <b>PULL UP</b>                       | <b>TERRAIN - PULL UP</b>        | "Terrain, Terrain ;<br>Pull up, Pull up"                     |
| Imminent Terrain<br>Impact Warning<br>(ITI) (Red)                   | <b>PULL UP</b>                       | <b>TERRAIN AHEAD - PULL UP</b>  | "Terrain Ahead,<br>Pull up ;<br>Terrain Ahead,<br>Pull up"   |
| Reduced Required<br>Obstacle<br>Clearance<br>Warning (ROC)<br>(Red) | <b>PULL UP</b>                       | <b>OBSTACLE - PULL UP</b>       | "Obstacle, Obstacle ;<br>Pull up, Pull up"                   |
| Imminent Obstacle<br>Impact Warning<br>(IOI) (Red)                  | <b>PULL UP</b>                       | <b>OBSTACLE AHEAD - PULL UP</b> | "Obstacle Ahead,<br>Pull up ;<br>Obstacle Ahead,<br>Pull up" |
| Reduced Required<br>Terrain Clearance<br>Caution (RTC)<br>(Amber)   | <b>TERRAIN</b>                       | <b>CAUTION - TERRAIN</b>        | "Caution, Terrain ;<br>Caution, Terrain"                     |
| Imminent Terrain<br>Impact Caution<br>(ITI) (Amber)                 | <b>TERRAIN</b>                       | <b>TERRAIN AHEAD</b>            | "Terrain Ahead ;<br>Terrain Ahead"                           |
| Reduced Required<br>Obstacle<br>Clearance Caution<br>(ROC) (Amber)  | <b>TERRAIN</b>                       | <b>CAUTION - OBSTACLE</b>       | "Caution, Obstacle ;<br>Caution, Obstacle"                   |
| Imminent Obstacle<br>Impact Caution<br>(IOI) (Amber)                | <b>TERRAIN</b>                       | <b>OBSTACLE AHEAD</b>           | "Obstacle Ahead ;<br>Obstacle Ahead"                         |

Table 9.49.2 - FLTA alerts



2. Premature descent alerting

A Premature Descent Alert (PDA) is issued when the system detects that the airplane is significantly below the normal approach path to a runway (Figure 9.49.1).

PDA alerting begins when the airplane is within 15 Nm of the destination airport. PDA alerting ends when the airplane is either :

- 0.5 Nm from the runway threshold
- or
- at an altitude of 125 feet AGL while within 1.0 Nm of the threshold.

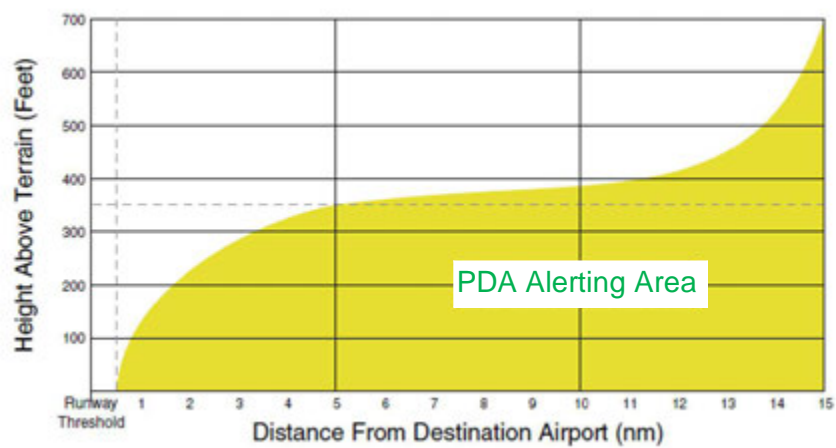


Figure 9.49.1 - PDA alerting threshold

The aural/displayed messages associated with the PDA function are described in the table 9.49.3.

| Alert Type  | PFD/MFD<br>TAWS Page<br>Annunciation | MFD Map Page<br>Pop-Up Alert | Aural Message      |
|---|--------------------------------------|------------------------------|--------------------|
| Premature Descent<br>Alert Caution<br>(PDA) (Amber) | <b>TERRAIN</b>                       | <b>TOO LOW - TERRAIN</b>     | "Too low, Terrain" |

Table 9.49.3 - PDA alerts

# Pilot's Operating Handbook

## 3. Excessive descent rate alert

The purpose of the Excessive Descent Rate (EDR) alert is to provide suitable notification when the airplane is determined to be closing (descending) upon terrain at an excessive speed. Figure 9.49.2 shows the parameters for the alert as defined by TSO-C151b.

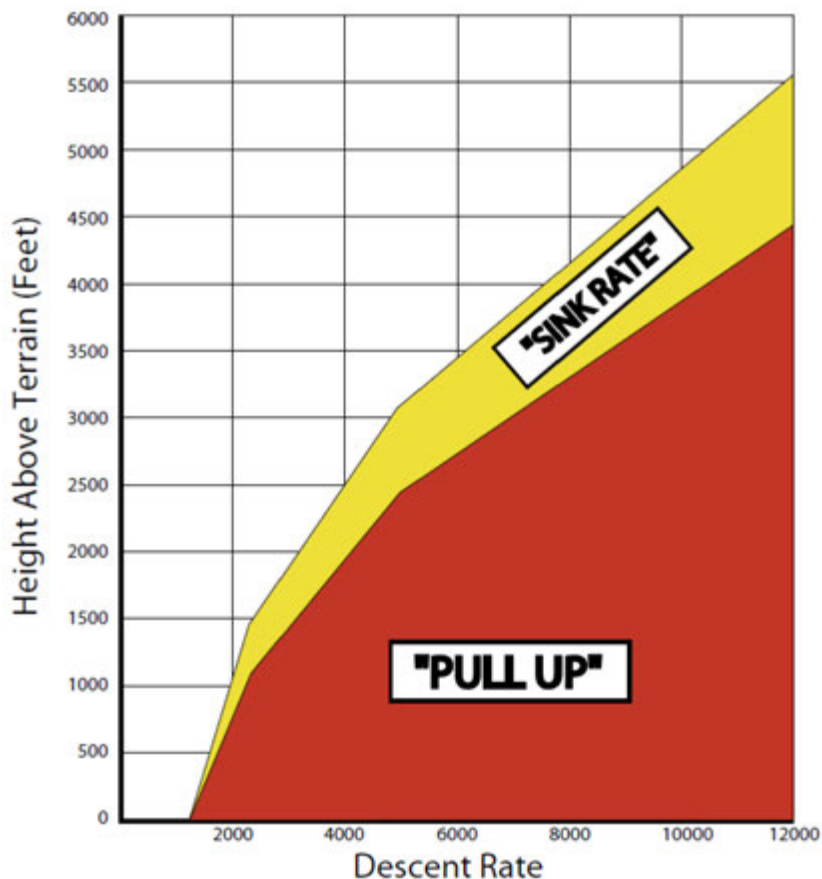


Figure 9.49.2 - Excessive Descent Rate Alert Criteria

The aural/displayed messages associated with the EDR function are described in the table 9.49.4.

| Alert Type   | PFD/MFD<br>TAWS Page<br>Annunciation | MFD Map Page<br>Pop-Up Alert | Aural Message |
|--|--------------------------------------|------------------------------|---------------|
| Excessive Descent<br>Rate Warning<br>(EDR) (Red)   | <b>PULL UP</b>                       | <b>PULL UP</b>               | "Pull up"     |
| Excessive Descent<br>Rate Caution<br>(EDR) (Amber) | <b>TERRAIN</b>                       | <b>SINK RATE</b>             | "Sink rate"   |

Table 9.49.4 - EDR alerts

#### 4. Negative climb rate after takeoff alert (NCR)

The purpose of the Negative Climb Rate (NCR) After Takeoff alert (also referred to as Altitude Loss After Takeoff) is to provide suitable alerts to the pilot when the system determines that the airplane is losing altitude (closing upon terrain) after takeoff. The aural message "Don't sink" is given for NCR alerts, accompanied by an annunciation and a pop-up terrain alert on the PFD's and Touchscreen Controllers. NCR alerting is only active when departing from an airport and when the following conditions are met :

- The height above the terrain is less than 700 feet.
- The distance from the departure airport is 5 Nm or less.
- The heading change from the heading at the time of departure is less than 110 degrees.

# Pilot's Operating Handbook

Figure 9.49.3 shows two figures which illustrate the NCR alerting parameters as defined by TSO-C151b.

The NCR alert is issued when the altitude loss and height are within the range in the first figure, or when the sink rate (negative vertical speed) and height are within the range in the second figure.

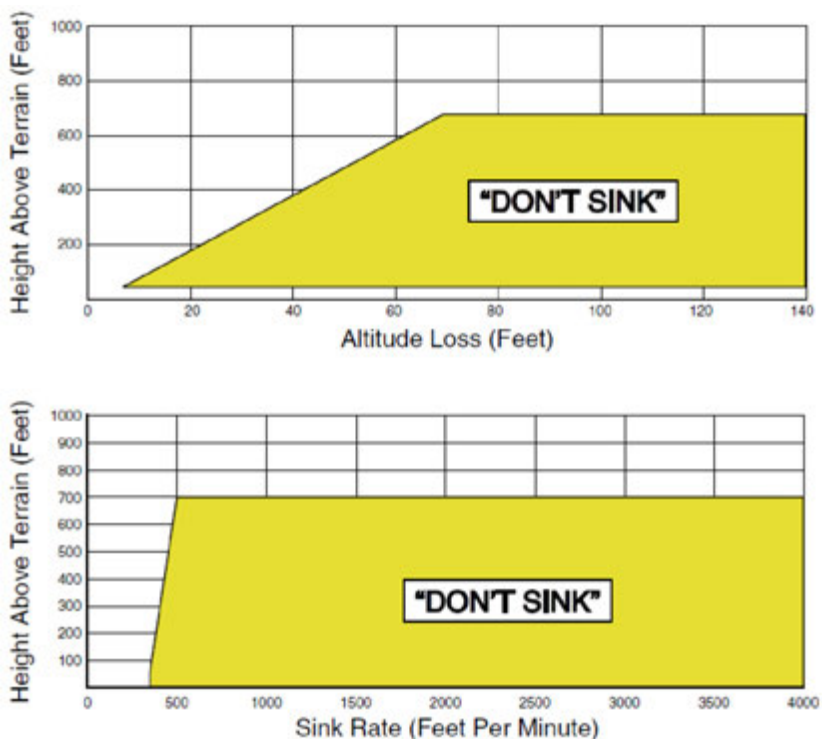


Figure 9.49.3 - Negative Climb Rate (NCR) Alert Criteria

The aural/displayed messages associated with the NCR function are described in the table 9.49.5.

| Alert Type                                      | PFD/MFD<br>TAWS Page<br>Annunciation | MFD Map Page<br>Pop-Up Alert | Aural Message |
|---|--------------------------------------|------------------------------|---------------|
| Negative Climb<br>Rate Caution<br>(NCR) (Amber) | <b>TERRAIN</b>                       | <b>DONT' SINK</b>            | "Don't sink"  |

Table 9.49.5 - NCR alerts

**5. "FIVE-HUNDRED" aural alert, altitude voice callout (VCO)**

The purpose of the aural alert message "Five-Hundred" is to provide an advisory alert to the pilot that the airplane is 500 feet above terrain. When the airplane descends within 500 feet of terrain, the aural message "Five-Hundred" is generated. There are no display annunciations or pop-up alerts that accompany the aural message.

**6. TAWS not available alert**

TAWS requires a 3-D GPS navigation solution along with specific vertical accuracy minimums. Should the navigation solution become degraded or if the airplane is out of the database coverage area, the annunciation TAWS N/A is generated in the annunciation window and on the TAWS Page. The aural message "TAWS Not Available" is generated. When the GPS signal is re-established and the airplane is within the database coverage area, the aural message "TAWS Available" is generated.

**7. TAWS inhibit**

TAWS also has an inhibit mode that deactivates the PDA/FLTA aural and visual alerts. Pilots should use discretion when inhibiting TAWS and always remember to enable the system when appropriate. Only the PDA and FLTA alerts are disabled in the inhibit mode.

**SECTION 8**

**Handling, servicing and maintenance**

Installation and operation of GARMIN TAWS system do not change the basic handling, servicing and maintenance procedures of the airplane described in section 8 Handling, Servicing and Maintenance of the basic POH.

# **SUPPLEMENT**

## **GARMIN Synthetic Vision System**

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| 4 - Normal procedures .....                   | 9.50.4 |
| 5 - Performance .....                         | 9.50.5 |
| 6 - Weight and balance .....                  | 9.50.5 |
| 7 - Description .....                         | 9.50.5 |
| 8 - Handling, servicing and maintenance ..... | 9.50.6 |

## SECTION 1

### General

This supplement is intended to inform the pilot about the equipment limitations, description and operations necessary for operation when the TBM airplane is equipped with the option GARMIN Synthetic Vision System (SVS).

The SVS does not replace and is not intended to be used independently of the TAS and/or TAWS system(s).

The SVS does not replace and is not intended to be used independently of the horizontal and vertical primary flight instruments.

The SVS does not replace and is not intended to be used independently of the Course Deviation Indicator and the Vertical Deviation Indicator.

## SECTION 2

### Limitations

The limitations hereafter supplement or replace those of the standard airplane described in section 2 Limitations of the basic POH when the TBM airplane is equipped with the option GARMIN Synthetic Vision System.

The following document, or any further edition applicable to the latter, shall be readily available to the pilot, whenever operation of the SVS is predicted :

>> *Airplane equipped with G1000 Flight deck (MOD70-0176-00)*

- GARMIN Integrated Flight Deck Pilot's Guide, No. 190-00709-05 or its latest revision.

>> *Airplane equipped with G1000 NXi Flight deck (MOD70-0539-00)*

- GARMIN Integrated Flight Deck Pilot's Guide, No. 190-02218-XX at its latest revision.

>> *Airplane retrofited with GARMIN G1000 NXi Flight deck (MOD70-0539-00)*

- GARMIN G1000 NXi Integrated Flight Deck Cockpit Pilot's Guide for the TBM850/900 P/N 190-02348-00 or any later revision as applicable.

>> *Airplane equipped with G3000 Flight deck (MOD70-0476-00)*

- GARMIN Integrated Flight Deck Pilot's Guide, No. 190-02046-XX at its latest revision.

The use of the Synthetic Vision System display elements alone for airplane control without reference to the GARMIN system primary flight instruments is prohibited.

## Pilot's Operating Handbook

The use of the Synthetic Vision System alone for vertical and/or horizontal navigation, or obstacle or terrain avoidance is prohibited.

Pathway boxes must be selected OFF when flying an instrument approach. Turn Pathways OFF when ACTIVATE VECTORS-TO-FINAL, ACTIVATE APPROACH is selected, or the airplane is established on any segment of the approach.

The use of the Synthetic Vision System traffic display alone to avoid other airplane is prohibited.

The Terrain Database has an area of coverage from North 75° latitude to South 60° latitude in all longitudes.

### SECTION 3

#### Emergency procedures

The emergency procedures hereafter supplement or replace those of the standard airplane described in Section 3 Emergency Procedures of the basic Pilot's Operating Handbook when the TBM airplane is equipped with the option GARMIN Synthetic Vision System.

#### Inconsistent display between SVS and GARMIN system primary flight instruments

>> *Airplane with G1000 Flight deck (MOD70-0176-00)*

From PFD display unit

- PFD softkey ..... Press
- SYN VIS softkey ..... Press
- SYN TERR softkey ..... Press to disable
- SVS is removed from the PFD ..... Verify

>> *Airplane with G1000 Nxi Flight deck (MOD70-0539-00)*

From PFD display unit

- PFD OPT softkey ..... Press
- SVT softkey ..... Press
- Terrain softkey ..... Press to disable
- SVS is removed from the PFD ..... Verify



■ >> *Airplane with G3000 Flight deck (MOD70-0476-00)*

From PFD display unit

- PFD Settings softkey ..... Press
- Attitude Overlays softkey ..... Press
- Synthetic Terrain softkey ..... Press to disable
- SVS is removed from the PFD ..... Verify

>> *All*

Use GARMIN system primary displays for navigation and airplane control.

**SECTION 4**

**Normal procedures**

The normal procedures hereafter supplement or replace those of the standard airplane described in section 4 Normal procedures of the basic POH when the TBM airplane is equipped with the option GARMIN Synthetic Vision System.

**▲ CAUTION ▲**

SVS information is not a substitute for standard course and altitude deviation information provided by the CDI, VSI, VDI and the primary flight instruments, as well as for the Traffic Advisory System (TAS) or the Terrain Awareness Warning System (TAWS).



**SVS activation**

Refer to GARMIN Integrated Flight Deck Pilot's Guide, as applicable, listed in section 2 Limitations of this supplement for further information.

## SECTION 5

### Performance

Installation and operation of GARMIN Synthetic Vision System do not change the basic performance of the airplane described in Section 5 Performance of the basic POH.

## SECTION 6

### Weight and balance

The weight and balance hereafter supplement or replace those of the standard airplane described in Section 6 Weight and balance of the basic POH when the airplane is equipped with the option GARMIN Synthetic Vision System.

| S/<br>R/<br>A/<br>O    | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|------------------------|------------------------------|--|----------------------------------|-------------------|
| <b>34 - Navigation</b> |                              |  |                                  |                   |
| A                      | 0226-00                      | Synthetic Vision System<br>GARMIN                              | /                                | /                 |

## SECTION 7

### Description

Information hereafter supplement or replace those of the standard airplane described in section 7 Description of the basic POH when the airplane is equipped with the option GARMIN Synthetic Vision System.

SVS provides additional features on the primary flight display (PFD) - refer to GARMIN Integrated Flight Deck Pilot's Guide, as applicable, listed in section 2 Limitations of this supplement for further information.

## **SECTION 8**

### **Handling, servicing and maintenance**

Installation and operation of GARMIN Synthetic Vision System do not change the basic handling, servicing and maintenance procedures of the airplane described in section 8 Handling, servicing and maintenance of the basic POH.

**SUPPLEMENT****GARMIN GSR 56 weather datalink and satellite phone****Table of contents**

|   | Page   |
|---|--------|
| 1 - General .....                             | 9.56.2 |
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| 5 - Performance .....                         | 9.56.7 |
| 6 - Weight and balance .....                  | 9.56.7 |
| 7 - Description .....                         | 9.56.8 |
| 8 - Handling, servicing and maintenance ..... | 9.56.9 |

## SECTION 1

### General

This supplement is intended to inform the pilot about the equipment limitations, description and operations necessary for operation when the TBM airplane is equipped with the option GARMIN GSR 56 weather datalink and satellite phone.

Unless otherwise mentioned, whenever a G1000 system is called in this supplement it concerns either a G1000 system or a G1000 NXi system.

## SECTION 2

### Limitations

The limitations hereafter supplement or replace those of the standard airplane described in section 2 Limitations of the basic POH when the TBM airplane is equipped with the option GARMIN GSR 56 weather datalink and satellite phone.

The GARMIN Integrated Flight Deck Pilot's Guide mentioned in section 2 Limitations of the basic POH (G1000 or G3000 as standard) or of the POH supplement (G1000 NXi retrofit), as applicable, or any further applicable edition, shall be readily available to the pilot, whenever the operation of GARMIN GSR 56 weather datalink and satellite phone is predicted.

#### Satellite phone functions

### ▲ WARNING ▲

#### Use of phone by PIC prohibited during all airplane operations



##### >> Airplane with G1000 Flight deck

- It is forbidden to activate Pilot In Command on-side GMA TEL button as long as the airplane is in the air or moving on the ground.
- Only the Pilot In Command cross side GMA TEL input can be activated at all time of flight for the front passenger and passengers to have the GSR 56 telephone audio functions.

##### >> Airplane with G3000 Flight deck

- It is forbidden to activate TEL button on Pilot Tab (located in NAV COM/Audio & Radios page) on GTC Touchscreen Controllers as long as the airplane is in the air or moving on the ground.

- Only the TEL button, on Copilot and Pass Tabs (located in NAV COM/Audio & Radios page) on GTC Touchscreen Controllers can be activated at all time of flight for the front passenger and passengers to have the GSR 56 telephone audio functions.

>> All

### Weather datalink functions

- The GSR 56 weather datalink is only an advisory weather source, it does not relieve the pilot to comply with the applicable operational regulation in terms of flight preparation especially with regard to the use of an approved weather and NOTAM sources during flight planning.

### International telecommunication regulation

The GSR 56 is a telecommunication device approved under FCC ID Q639522B and registered by the ITU (International Telecommunication Union) for international use according to the GMPCS-MoU.

The receiver transmitter RF module embedded in the GSR 56 is a 9522 B manufactured by Iridium Satellite LLC.

Terms of use are subject to changes and are available from the ITU website.

## 2.1 - Placards

Under L.H. front side window, under instruction plate

I4113207AAA/MA4200

USE OF PHONE BY PIC PROHIBITED DURING ALL AIRCRAFT OPERATIONS

## **SECTION 3**

### **Emergency procedures**

Installation and operation of GARMIN GSR 56 weather datalink and satellite phone do not change the basic emergency procedures of the airplane described in section 3 Emergency procedures of the basic POH.

## **SECTION 4**

### **Normal procedures**

The normal procedures hereafter supplement or replace those of the standard airplane described in section 4 Normal Procedures of the basic POH when the TBM airplane is equipped with the option GARMIN GSR 56 weather datalink and satellite phone.

Normal operating procedures of the GARMIN GSR 56 weather datalink and satellite phone system are outlined in the GARMIN Integrated Flight Deck Pilot's Guide.

>> Airplane with G1000 Flight deck

### Before starting engine

On L.H. GMA audio panel

1 - TEL button ..... OFF

*End of procedure.*

### Before starting a phone call in flight

On L.H. GMA audio panel

1 - TEL button ..... OFF

*If rear passengers intend to take part in a phone call :*

2 - CABIN button or PASS ICS button ..... OFF

2 types of wording may exist for the same button

*If front passenger intends to take part in a phone call :*

3 - INTRCOM button or CREW ICS button ..... OFF

2 types of wording may exist for the same button

On R.H. GMA audio panel

4 - TEL button ..... ON

*If rear passengers intend to take part in a phone call :*

5 - CABIN button or PASS ICS button ..... ON

2 types of wording may exist for the same button

*End of procedure.*



>> Airplane with G3000 Flight deck

Before starting engine

- In one of the GTC's NAV COM / Audio & Radios / pilot Tab
  - 1 - TEL button ..... OFF
- End of procedure.

Before starting a phone call in flight

- In one of the GTC's NAV COM / Audio & Radios / pilot Tab
  - 1 - TEL button ..... OFF
- If passengers intend to take part into a phone call :
- In one of the GTC's NAV COM / Intercom Page
  - 2 - Pilot/Passenger Link Arrow ..... OFF
- If front passenger intends to take part into a phone call :
- In one of the GTC's NAV COM / Intercom Page
  - 3 - Pilot/Copilot Link Arrow ..... OFF
- In one of the GTC's NAV COM / Audio & Radios / copilot Tab
  - 4 - TEL button ..... ON
- if necessary, switch off Sync to Pilot function
- If passengers intend to take part into a phone call :
- In one of the GTC's NAV COM / Audio & Radios / Pass Tab
  - 5 - TEL button ..... ON
- if necessary, switch off Sync to Pilot function
- End of procedure.

## SECTION 5

### Performance

Installation and operation of GARMIN GSR 56 weather datalink and satellite phone. do not change the basic performance of the airplane described in section 5 Performance of the basic POH.

## SECTION 6

### Weight and balance

The weight and balance hereafter supplement or replace those of the standard airplane described in section 6 Weight and balance of the basic POH when the airplane is equipped with the option GARMIN GSR 56 weather datalink and satellite phone.

| S/<br>R/<br>A/<br>O       | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------------|------------------------------|--|----------------------------------|-------------------|
| <b>23 - Communication</b> |                              |  |                                  |                   |
| A                         | 0331-23                      | Weather datalink and satellite<br>phone system GSR 56          | GARMIN<br>3.82<br>(1.736)        | 58.03<br>(1.474)  |

## SECTION 7

### Description

Information hereafter supplement or replace those of the standard airplane described in section 7 Description of the basic POH when the airplane is equipped with the option GARMIN GSR 56 weather datalink and satellite phone.

GARMIN GSR 56 weather datalink and satellite phone system provides airborne low speed datalink and voice communication capability to Integrated Flight Deck system excluding any voice mail function. GSR 56 weather datalink and satellite phone system contains a transceiver that operates on the Iridium Satellite network.

The weather information are displayed on the MFD maps and on the PFD inset map.

#### *>> Airplane with G1000 Flight deck*

The satellite phone interface is embedded in the MFD : Phone communication and SMS can be received and sent through the dedicated pages on the MFD.

The controls for the MFD are located on both the MFD bezel and the MFD control unit.

The telephone audio including the incoming call ringing is controlled by the TEL button on the GMA audio panels and can be played in the pilot, front passenger and passengers headphones.

#### *>> Airplane with G3000 Flight deck*

The satellite phone interface is embedded in the Touchscreen Controllers : Phone communication and SMS can be received and sent through the dedicated pages on the Touchscreen Controllers.

The telephone audio including the incoming call ringing is controlled by the Touchscreen controllers & GMA audio processor and can be played in the pilot, front passenger and passengers headphones.

#### *>> All*

Although it is possible to leave a message when calling the airplane, as voice mail communication is not supported by the GSR 56 :

- it is not possible to access the GSR 56 voice mail from the airplane
- there is no indication on the Integrated Flight Deck system when a new message has been left on the GSR 56 voice mail.

## **SECTION 8**

### **Handling, servicing and maintenance**

Installation and operation of GARMIN GSR 56 weather datalink and satellite phone. do not change the basic handling, servicing and maintenance procedures of the airplane described in section 8 Handling, Servicing and Maintenance of the basic POH.

Intentionally left blank

## **SUPPLEMENT**

### **Brazil specifics**

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| 7 - Description .....                         | 9.59.14 |
| 8 - Handling, servicing and maintenance ..... | 9.59.14 |

## SECTION 1

### General

This supplement is intended to inform the pilot about the airplane specifics, among others those required by the relevant Certification Authorities (limitations, description and operations necessary to the operation of the TBM airplane).

## SECTION 2

### Limitations

The limitations hereafter supplement or replace those of the standard airplane described in section 2 Limitations of the basic POH.

### 2.5 - Weight and C.G. limits

#### Weight limits

>> *With 4-seat accommodation*

- in rear part of pressurized cabin : 396 lbs (180 kg), with small or large net (see sketch below)

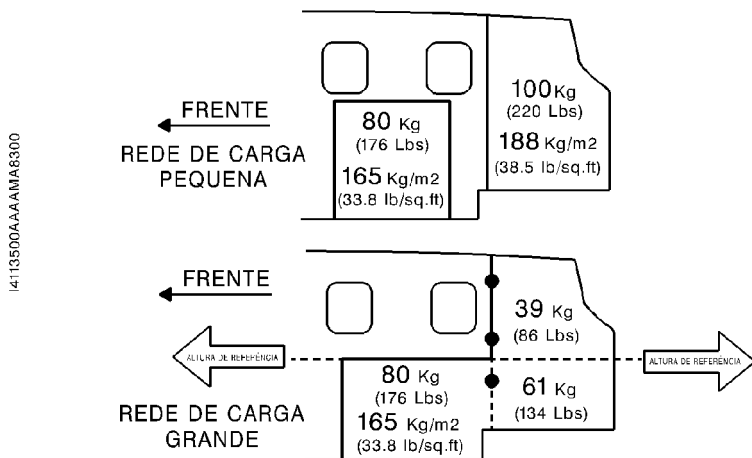


Figure 2.5.1 - Baggage limits

## 2.6 - Operation limits

When operating the VHF-COMM system in Brazilian air space, the selection of 8.33 kHz in the channels spacing can cause the loss of communication with the Air Traffic Control (ATC).

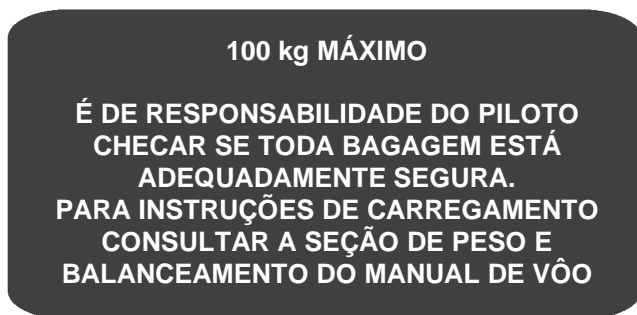
### **GNSS (GPS/SBAS) navigation system limitations**

In accordance with Brazilian IS 21-013A, use of GNSS/GPS is prohibited under IFR unless other means of navigation, suitable and approved for the intended route, are installed and operational. It must be possible - at any point along the route - to navigate to the destination or alternate, using such means.

The SBAS functionality is not available in Brazil, therefore operations that require such functionality, such as GNSS vertical navigation modes, are prohibited in Brazilian airspace.

## 2.9 - Placards

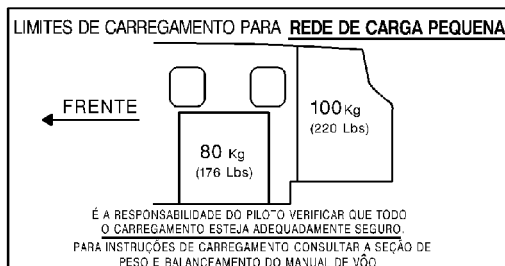
On pressurized baggage compartment partition wall





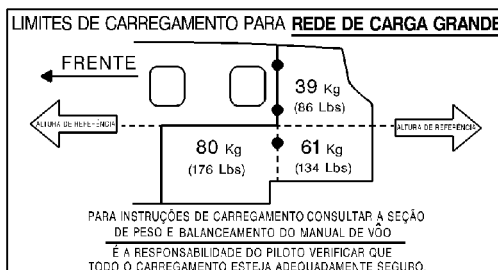
For the small cargo net, on frame C13bis

14113500AAAAB400



For the large cargo net, on R.H. side upholstery panel, in the rear baggage compartment

14113500AAAAB400



On FWD baggage compartment door frame, non pressurized

**50 kg MÁXIMO**

**PARA INSTRUÇÕES DE CARREGAMENTO  
CONSULTAR A SEÇÃO DE PESO E  
BALANCEAMENTO DO MANUAL DE VÔO**

# Pilot's Operating Handbook

Near fuel tank caps



On internal face of L.H. engine cowl

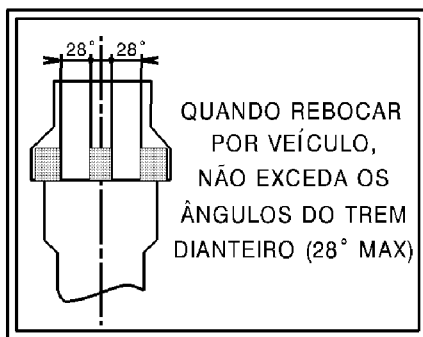


On rear passenger's table casing

**A MESA DEVERÁ ESTAR RECOLHIDA PARA DECOLAGEM E POUSO**

On nose gear door

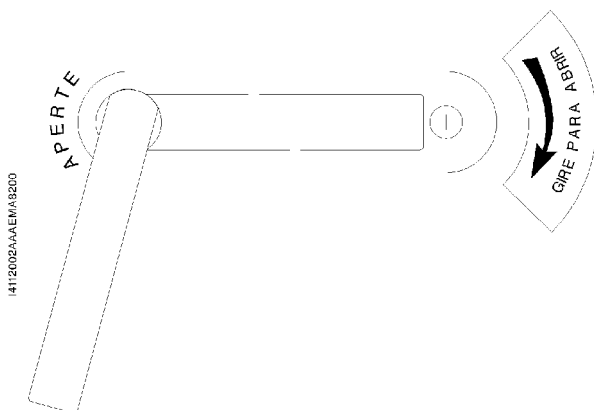
I4112001AAACMA8100



On engine cowling, in front of compartment door

**- TOMADA EXTERNA  
- 28 VOLTS D.C. NOMINAL  
- 800 AMPS  
CAPACIDADE MÍNIMA PARA PARTIDA  
- NÃO EXCEDA 1000 AMPS**

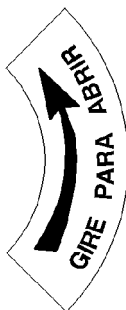
On pilot door - External side, if installed



# Pilot's Operating Handbook

On access door - External side

14112002AAAE1A6300



On outer fuselage skin aft of access door and in the cabin forward of access door

14112002AAAD1A6400



On access door - Internal side



**ATENÇÃO: DESTRAVAR ANTES  
DE OPERAR A MANOPLA**

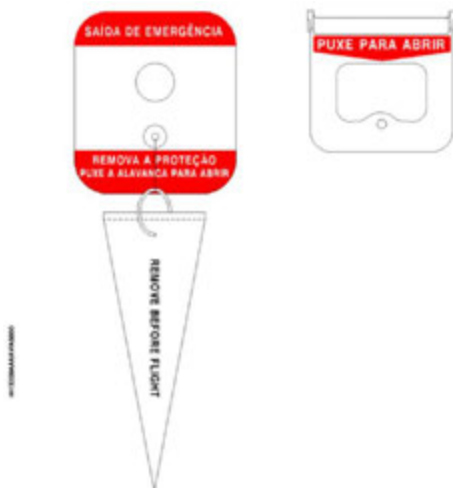
**GIRE A MANOPLA  
PARA ABRIR** ↑



On emergency exit handle

### Marking on cover

### Marking on handle



On last step of stairs

**MAX. UMA PESSOA NA ESCADA**

## Pilot's Operating Handbook

On R.H. access door jamb

I4113400AAA/BMA8200

**NÃO USE O  
CORRIMÃO  
PARA RECOLHER  
E PARA ALOJAR  
A ESCADA**

On R.H. side at front seat level and on the first rear passengers masks container (R.H. side on the ceiling)

I4113400AAA/BMA8300

**CUIDADO**  
PROIBIDO FUMAR DURANTE O USO DE OXIGÊNIO.  
GRAXAS E SUBSTÂNCIAS OLEOSAS ESTÃO  
SUJEITAS À COMBUSTÃO ESPONTÂNEA QUANDO  
EM CONTATO COM OXIGÊNIO

On rear passengers masks containers

I4113400AAA/BMA8400

**MÁSCARAS DE OXIGÊNIO**  
**PUXE A MÁSCARA PARA**  
**ATIVAR O OXIGÊNIO**



On the oxygen service door

I4112400AAAAM/A8200

**ABASTECIMENTO DE OXIGÊNIO.  
NÃO USE LUBRIFICANTES**

>> *Airplanes equipped with Lavatory compartment (Post-MOD70-0505-25)*

On fixed panel, cabin side

I4113200AAAAM/A8300

**A DIVISÓRIA DEVE ESTAR RECOLHIDA DURANTE A DECOLAGEM E O POUSO**

On fixed panel, toilet side

I4113200AAAAM/A8000

**O ASSENTO DO SANITÁRIO NÃO DEVE ESTAR  
OCUPADO DURANTE A DECOLAGEM E O POUSO**

**FECHE E TRAVE A TAMPA DO SANITÁRIO  
QUANDO NÃO ESTIVER EM USO**

**NÃO PENDURE OU MANTENHA OBJETOS  
SOBRE O SANITÁRIO OU NA DIVISÓRIA**

**A DIVISÓRIA DEVE ESTAR RECOLHIDA  
DURANTE A DECOLAGEM E O POUSO**

**OS FONES DE OUVIDO DEVEM SER  
UTILIZADOS QUANDO A  
DIVISÓRIA ESTIVER ESTENDIDA**

## Pilot's Operating Handbook

On access door, cabin side and toilet side

I4113200AAAMMA8200



Behind access door, cabin side and toilet side

I4113200AAAMMA18200





Front face of lavatory compartment, near opening / closing switches

I4113200AAAMMA18000

RECOLHER

I4113200AAAMMA8400

ESTENDER

On the magazine rack and on side wall of storage volume

I4113200AAAMMA18100

1,5 KG (3.3 LB)

>> *Airplanes equipped with Coat hanger (Post-MOD70-0557-25)*

On the upper edge of the L.H. Passenger access door panel

I4113200AAALMA8200

SOMENTE VESTUÁRIO

## SECTION 3

### Emergency procedures

No specifics

## SECTION 4

### Normal procedures

No specifics

## SECTION 5

### Performance

No specifics

## SECTION 6

### Weight and balance

The weight and balance hereafter supplement or replace those of the standard airplane described in section 6 Weight and balance of the basic POH.

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>01 - Specific optional equipment</b>                        |                                  |                   |
| S                   | 01004                        | Brazil certification markings                                  | /                                | /                 |

## **SECTION 7**

### **Description**

No specifics

## **SECTION 8**

### **Handling, servicing and maintenance**

No specifics

## **SUPPLEMENT**

### **Lavatory compartment**

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| 6 - Weight and balance .....                  | 9.63.8  |
| 7 - Description .....                         | 9.63.19 |
| 8 - Handling, servicing and maintenance ..... | 9.63.23 |

## SECTION 1

### General

This supplement is intended to inform the pilot about the equipment limitations, description and operations necessary for operation when the TBM airplane is equipped with the option lavatory compartment.

## SECTION 2

### Limitations

The limitations hereafter supplement or replace those of the standard airplane described in section 2 Limitations of the basic POH when the TBM airplane is equipped with the option lavatory compartment.

- toilet seat must not be occupied during take-off and landing
- divider must be stowed during take-off and landing
- headset shall be worn at all time when seat is occupied

■ >> *From S/N 1000 to S/N 1269*

### 2.9 - Placards

On fixed panel, cabin side

DIVIDER MUST BE STOWED DURING TAKE-OFF AND LANDING

14113200AAAHMA8300

## Pilot's Operating Handbook

On fixed panel, toilet side

14113200AAAHUA18000

TOILET SEAT MUST NOT  
BE OCCUPIED DURING  
TAKE OFF AND LANDING  
CLOSE AND LOCK TOILET COVER  
WHEN NOT IN USE  
DO NOT HANG OR STORE ITEMS  
ON TOILET OR DIVIDER  
DIVIDER MUST BE STOWED DURING  
TAKE OFF AND LANDING  
HEADSET MUST BE USED WHEN  
DIVIDER IS DEPLOYED

On access door, cabin side and toilet side

14113200AAAHUA18000

EMERGENCY  
STOWAGE  
REMOVE  
COVER

Behind access door, cabin side and toilet side

14113200AAAHUA18100

PUSH TO STOW

Inner face of toilet cover

I4113200AAA/HUA18200



Front face of lavatory compartment, near opening/closing switches

I4113200AAA/IMA8100



I4113200AAA/IMA8200



On the magazine rack

I4113200AAA/IMA8000

3.3 LBS (1,5 KG)

## Pilot's Operating Handbook

&gt;&gt; From S/N 1270

**2.9 - Placards**

On fixed panel, cabin side

I4113200AAANMA8200

DIVIDER MUST BE STOWED DURING TAKE-OFF AND LANDING

On fixed panel, toilet side

I4113200AAANMA8300

TOILET SEAT MUST NOT TO BE  
OCCUPIED DURING TAKE-OFF  
AND LANDINGCLOSE AND LOCK TOILET COVER  
WHEN NOT IN USEDO NOT HANG OR STORE ITEMS  
ON TOILET OR DIVIDERDIVIDER MUST BE STOWED DURING  
TAKE-OFF AND LANDINGHEADSET MUST BE USED WHEN  
DIVIDER IS DEPLOYED

On access door, cabin side and toilet side

I4113200AAANMA8100

**EMERGENCY  
STOWAGE  
  
REMOVE  
COVER**



Behind access door, cabin side and toilet side

I4113200AAA1MA8000



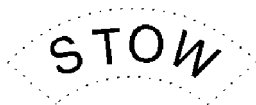
Inner face of toilet cover

I4113300AAA1MA8400



Front face of lavatory compartment, near opening/closing switches

I4113200AAA1MA8400



I4113200AAA1MA18000



On the magazine rack

I4113200AAA/IMA8300

1,5 kg - 3.3 lbs

## SECTION 3

### Emergency procedures

The emergency procedures hereafter supplement those of the standard airplane described in section 3 Emergency procedures of the basic POH.

#### 3.10 - Pressurization and air conditioning

>> *Without v15 GARMIN software (Pre-MOD70-0407-00)*

**CABIN ALTITUDE**

Inform passengers to use emergency stowing of the divider and oxygen mask.

>> *With v15 GARMIN software (Post-MOD70-0407-00) or airplane with G3000 Flight deck (MOD70-0476-00)*

**CABIN ALTITUDE** and **USE OXYGEN MASK**

or

**CABIN ALTITUDE** and **USE OXYGEN MASK** and **EDM**

Inform passengers to use emergency stowing of the divider and oxygen mask.

Other procedures in the basic POH are unchanged.

## SECTION 4

### Normal procedures

The normal procedures hereafter supplement or replace those of the standard airplane described in section 4 Normal procedures of the basic POH when the TBM airplane is equipped with the option lavatory compartment.

#### **BRIEFING to passengers to be performed before entering the airplane**

Normal and Emergency stowing operations of the divider.

In case of depressurization : emergency stowing of the divider, use oxygen mask, and remain seated unless otherwise instructed by the crew.

The headset must be used when the divider is deployed to allow communication with the crew in case of emergency.

## SECTION 5

### Performance

Installation and operation of Lavatory compartment do not change the basic performance of the airplane described in section 5 Performance of the basic POH.

## SECTION 6

### Weight and balance

The weight and balance hereafter supplement or replace those of the standard airplane described in section 6 Weight and balance of the basic POH when the airplane is equipped with the option lavatory compartment.

#### **6.1 - General**

This paragraph is intended to provide the pilot with a simple and rapid means of determining weight and balance of the airplane when equipped with the lavatory compartment option.

#### **▲ WARNING ▲**

**It is the pilot's responsibility to ensure that the airplane is properly loaded and the weight and balance limits are adhered to.**



## **6.4 - Determining the new airplane empty weight and balance after the application of the lavatory compartment option**

• NOTE •

The new empty weight determination after lavatory compartment installation shall be performed from the 6-seat configuration airplane characteristics

•

- 1) Record the basic empty weight (1a) and moment (1b) and CG (MAC %) (1c) from the last Weight and Balance Report in 6-seat configuration (see samples Figures 6.4.1 and 6.4.2 of the basic Pilot's Operating Handbook).
- 2) Compute the new empty weight (2a) and moment (2b) as sum of all above weights (1a) [removed equipment + installed equipment] and associated moments (1b) [removed equipment + installed equipment]
- 3) Compute the new empty weight arm (3) and CG (MAC %) (3c) using given formulas.
- 4) Report the new empty weight arm (3) and CG (MAC %) (3c) into the WEIGHT AND BALANCE FORM AND DIAGRAM of the airplane loading form in order to perform the weight and balance determination with the lavatory compartment installed.

>> From S/N 1000 to S/N 1159

$$\text{Moment} = \text{Weight} \times \text{Arm} \qquad \text{CG (MAC \%)} = \frac{(\text{Arm (m)} - 4.392)}{1.51} \times 100$$

| Item   | Weight<br>(kg) | Arm<br>(m) | Moment<br>(m.kg) | CG<br>(MAC %) |
|--|----------------|------------|------------------|---------------|
| Empty Weight (kg)  | (1a)           |            | (1b)             | (1c)          |
| Weight and moment variation after lavatory compartment option installation | + 12           |            | + 86             |               |
|  |                |            |                  |               |
| New empty weight<br>(ready for cargo preparation)                          | (2a)           | (3)        | (2b)             | (3c)          |

$$\text{Moment} = \text{Weight} \times \text{Arm} \qquad \text{CG (MAC \%)} = \frac{(\text{Arm (in)} - 172.93)}{59.45} \times 100$$

| Item   | Weight<br>(lbs) | Arm<br>(in) | Moment<br>(in.lbs) | CG<br>(MAC %) |
|--|-----------------|-------------|--------------------|---------------|
| Empty Weight (lbs)   | (1a)            |             | (1b)               | (1c)          |
| Weight and moment variation after lavatory compartment option installation | + 27            |             | + 7530             |               |
|  |                 |             |                    |               |
| New empty weight<br>(ready for cargo preparation)                          | (2a)            | (3)         | (2b)               | (3c)          |

# Pilot's Operating Handbook

>> From S/N 1160

$$\text{Moment} = \text{Weight} \times \text{Arm} \qquad CG (MAC \%) = \frac{(\text{Arm} (m) - 4.392)}{1.51} \times 100$$

| Item  | Weight<br>(kg)   | Arm<br>(m) | Moment<br>(m.kg)   | CG<br>(MAC %) |
|---|------------------|------------|--------------------|---------------|
| Empty Weight (kg)   | (1a)             |            | (1b)               | (1c)          |
| Weight and moment variation after lavatory compartment option installation without seat heating system<br>or<br>Weight and moment variation after lavatory compartment option installation with seat heating system | + 19<br><br>+ 18 |            | + 134<br><br>+ 127 |               |
|   |                  |            |                    |               |
| New empty weight<br>(ready for cargo preparation)   | (2a)             | (3)        | (2b)               | (3c)          |

$$\text{Moment} = \text{Weight} \times \text{Arm} \qquad CG (MAC \%) = \frac{(\text{Arm} (in) - 172.93)}{59.45} \times 100$$

| Item  | Weight<br>(lbs)  | Arm<br>(in) | Moment<br>(in.lbs)     | CG<br>(MAC %) |
|---|------------------|-------------|------------------------|---------------|
| Empty Weight (lbs)  | (1a)             |             | (1b)                   | (1c)          |
| Weight and moment variation after lavatory compartment option installation without seat heating system<br>or<br>Weight and moment variation after lavatory compartment option installation with seat heating system | + 42<br><br>+ 40 |             | + 11643<br><br>+ 11056 |               |
|   |                  |             |                        |               |
| New empty weight<br>(ready for cargo preparation)   | (2a)             | (3)         | (2b)                   | (3c)          |

## Using the weight and balance form

### ▲ CAUTION ▲

Empty weight, arm and CG % position to be considered are the ones from the last weight and balance report issued after the lavatory compartment option installation.



Refer to POH section 6.4 using the weight and balance form procedure to determine the weight and balance of the airplane equipped with the lavatory compartment option together with the use of the loading form hereafter.

| S/<br>R/<br>A/<br>O | Item<br>OPT70<br>or<br>MOD70 | Required (R) or Standard (S) or Optional (A or O)<br>equipment | Weight<br>per unit<br>lb<br>(kg) | Arm<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
| O                   | 0505-25C                     | <b>25 - Equipment and furnishings</b>                          |                                  |                   |
|                     |                              | Lavatory compartment   | 138.9<br>(63)                    | 267.7<br>(6.799)  |
|                     |                              | Lavatory compartment carpet                                    | 28.7<br>(13)                     | 211.4<br>(5.370)  |

## Weight and balance form and diagram (m, kg) - only applicable if lavatory compartment is installed

>> From S/N 1000 to S/N 1159

Moment = Weight x Arm

$$CG (MAC \%) = \frac{(Arm (m) - 4.392)}{1.51} \times 100$$

| Item                          |                  | Weight<br>(kg) | Arm<br>(m) | Moment<br>(m.kg) | CG<br>(MAC %) |
|-------------------------------|------------------|----------------|------------|------------------|---------------|
| Empty Weight (kg)             |                  |                |            |                  |               |
| Baggage FWD (< 50 kg)         |                  |                | 3.250      |                  |               |
| Front Seats (kg)              |                  |                | 4.534      |                  |               |
| Inter. Seats                  | -17 kg           |                | 5.710      |                  |               |
|                               | per seat removed |                |            |                  |               |
|                               | Pax              |                |            |                  |               |
| Baggage AFT (< 100 kg)        |                  |                | 7.695      |                  |               |
| Zero Fuel Weight (< 2 736 kg) |                  |                |            |                  |               |
| Fuel (kg)                     |                  |                | 4.820      |                  |               |
| Ramp Weight (< 3 370 kg)      |                  |                |            |                  |               |
| Taxi Fuel (kg)                |                  |                | 4.820      |                  |               |
| Takeoff Weight (< 3 354 kg)   |                  |                |            |                  |               |
| Trip Fuel (kg)                |                  |                | 4.820      |                  |               |
| Landing Weight (< 3 186 kg)   |                  |                |            |                  |               |



>> From S/N 1160

Moment = Weight x Arm

$$CG (MAC \%) = \frac{(Arm (m) - 4.392)}{1.51} \times 100$$

| Item                          |                              | Weight<br>(kg) | Arm<br>(m) | Moment<br>(m.kg) | CG<br>(MAC %) |
|-------------------------------|------------------------------|----------------|------------|------------------|---------------|
| Empty Weight (kg)             |                              |                |            |                  |               |
| Baggage FWD (< 50 kg)         |                              |                | 3.250      |                  |               |
| Front Seats (kg)              |                              |                | 4.534      |                  |               |
| Inter. Seats                  | -15 kg<br>per seat removed * |                | 5.710      |                  |               |
|                               | Pax                          |                |            |                  |               |
| Baggage AFT (< 100 kg)        |                              |                | 7.695      |                  |               |
| Zero Fuel Weight (< 2 736 kg) |                              |                |            |                  |               |
| Fuel (kg)                     |                              |                | 4.820      |                  |               |
| Ramp Weight (< 3 370 kg)      |                              |                |            |                  |               |
| Taxi Fuel (kg)                |                              |                | 4.820      |                  |               |
| Takeoff Weight (< 3 354 kg)   |                              |                |            |                  |               |
| Trip Fuel (kg)                |                              |                | 4.820      |                  |               |
| Landing Weight (< 3 186 kg)   |                              |                |            |                  |               |

\* Seats weights do not include seat heating system weight

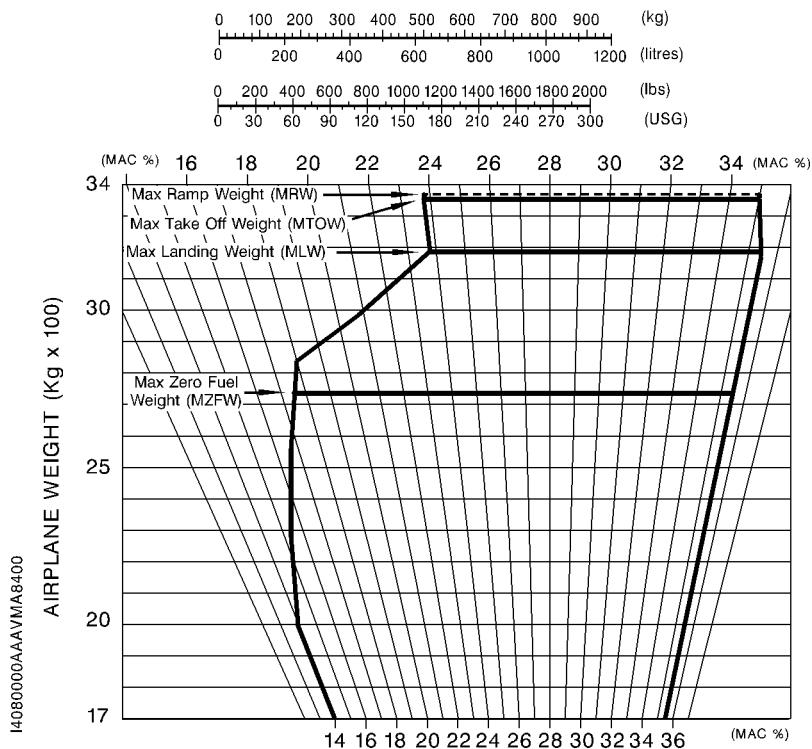


Figure 9.63.1 – Weight and Balance diagram

**Weight and balance form and diagram (in, lbs) - only applicable if lavatory compartment is installed**

>> From S/N 1000 to S/N 1159

$$\text{Moment} = \text{Weight} \times \text{Arm} \qquad CG \text{ (MAC \%)} = \frac{(\text{Arm (in)} - 172.93)}{59.45} \times 100$$

| Item                           |                               | Weight<br>(lbs) | Arm<br>(in) | Moment<br>(in.lbs) | CG<br>(MAC %) |
|--------------------------------|-------------------------------|-----------------|-------------|--------------------|---------------|
| Empty Weight (lbs)             |                               |                 |             |                    |               |
| Baggage FWD (< 110 lbs)        |                               |                 | 128.0       |                    |               |
| Front Seats (lbs)              |                               |                 | 178.5       |                    |               |
| Inter. Seats                   | -37.5 lbs<br>per seat removed |                 | 224.8       |                    |               |
|                                | Pax                           |                 |             |                    |               |
| Baggage AFT (< 220 lbs)        |                               |                 | 303.0       |                    |               |
| Zero Fuel Weight (< 6 032 lbs) |                               |                 |             |                    |               |
| Fuel (lbs)                     |                               |                 | 189.8       |                    |               |
| Ramp Weight (< 7 430 lbs)      |                               |                 |             |                    |               |
| Taxi Fuel (lbs)                |                               |                 | 189.8       |                    |               |
| Takeoff Weight (< 7 394 lbs)   |                               |                 |             |                    |               |
| Trip Fuel (kg)                 |                               |                 | 189.8       |                    |               |
| Landing Weight (< 7 024 lbs)   |                               |                 |             |                    |               |

Pilot's Operating Handbook

>> From S/N 1160

$$\text{Moment} = \text{Weight} \times \text{Arm} \qquad \text{CG (MAC \%)} = \frac{(\text{Arm (in)} - 172.93)}{59.45} \times 100$$

| Item                           |                                 | Weight<br>(lbs) | Arm<br>(in) | Moment<br>(in.lbs) | CG<br>(MAC %) |
|--------------------------------|---------------------------------|-----------------|-------------|--------------------|---------------|
| Empty Weight (lbs)             |                                 |                 |             |                    |               |
| Baggage FWD (< 110 lbs)        |                                 |                 | 128.0       |                    |               |
| Front Seats (lbs)              |                                 |                 | 178.5       |                    |               |
| Inter. Seats                   | -33.1 lbs<br>per seat removed * |                 | 224.8       |                    |               |
|                                | Pax                             |                 |             |                    |               |
| Baggage AFT (< 220 lbs)        |                                 |                 | 303.0       |                    |               |
| Zero Fuel Weight (< 6 032 lbs) |                                 |                 |             |                    |               |
| Fuel (lbs)                     |                                 |                 | 189.8       |                    |               |
| Ramp Weight (< 7 430 lbs)      |                                 |                 |             |                    |               |
| Taxi Fuel (lbs)                |                                 |                 | 189.8       |                    |               |
| Takeoff Weight (< 7 394 lbs)   |                                 |                 |             |                    |               |
| Trip Fuel (kg)                 |                                 |                 | 189.8       |                    |               |
| Landing Weight (< 7 024 lbs)   |                                 |                 |             |                    |               |

\* Seats weights do not include seat heating system weight

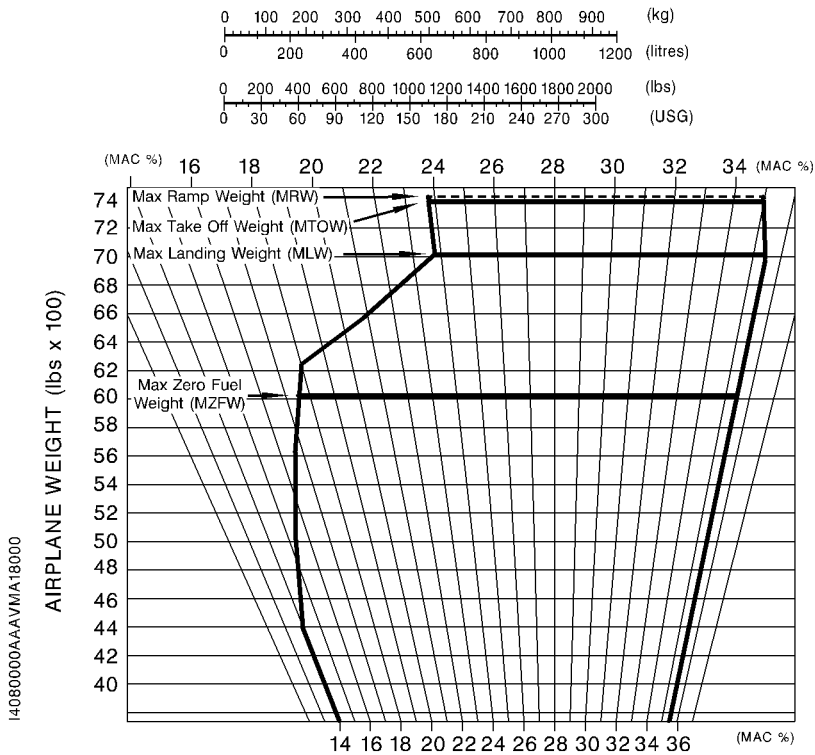


Figure 9.63.2 – Weight and Balance diagram

## SECTION 7

### Description

Information hereafter supplement or replace those of the standard airplane described in section 7 Description of the basic POH when the airplane is equipped with the option Lavatory compartment.

For operation, refer to equipment User's Guide.

The lavatory compartment is installed against right interior upholstery panel, facing large door. The lavatory compartment is installed at the place of the rear seats, removed to allow this installation. It is attached to the fuselage structure on the cabin floor, using the seats tracks with four pads and screws.

The lavatory compartment structure is made of composite panels.

The lavatory compartment assembly is composed of :

- A chemical toilet,
- Electrically deployable separating panels (divider),
- Two (2) actuating switches (DEPLOY, STOW),
- Two (2) emergency stowing buttons (PUSH TO STOW), accessible from inside or outside the lavatory compartment,
- One (1) mirror,
- One (1) electric power plug,
- One (1) headset allowing communication between the passenger and the crew.

When the lavatory compartment is not occupied, the divider is stored unfolded in the lavatory compartment structure.

A breaker integrated into the lavatory compartment structure protects its electrical system. The circuit breaker is only accessible when the lavatory compartment is removed.

Two (2) switches, located on the seat front face, left side, hidden when latching strap snap fastener is locked, control the deployment/stowing of the moveable parts of the divider.

Two (2) access doors (Velcro tape attached), located on each side of the fixed part of the divider, give access to the emergency stowing push button, allowing the emergency (manual) retraction of the divider, using the application of a vertical force (by hand) on the upper edge of the divider.

Electric connection of the system is performed via a power plug :

>> Without optional 12V power plugs (Pre-MOD70-0174-25)

- 28 volts

>> With optional 12V power plugs (Post-MOD70-0174-25)

- 12 volts

>> All

The power plug is located on the right hand side upholstery panel. Connection is only accessible when the lavatory compartment structure is unscrewed from the floor and moved slightly aside to access the plug.

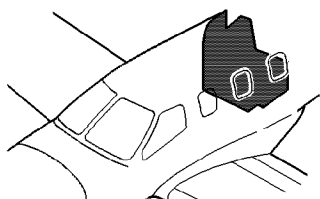
Mirror is automatically illuminated during the deployment of the divider.

A safety anti pinching sensor stops the deployment of the divider in case an interference is detected.

To remove the chemical toilet system from the lavatory compartment structure, it is necessary to unlatch the toilet cover, remove the top frame, if installed then lift upward the forward face of the structure and pull out the toilet from the structure.

A storage volume on the left side of the toilet is accessible when toilet cover is up.

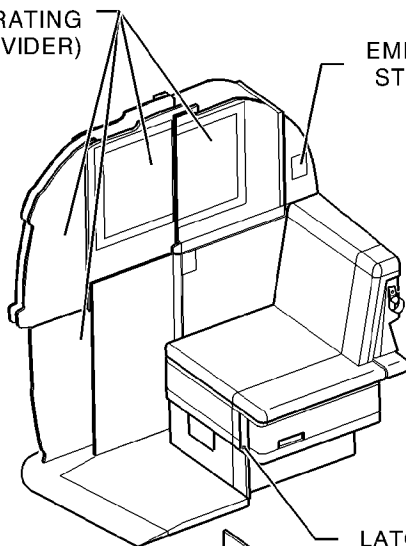
A magazine rack is located on the forward side of the fixed part of the divider.



DEPLOYABLE SEPARATING  
PANELS (DIVIDER)

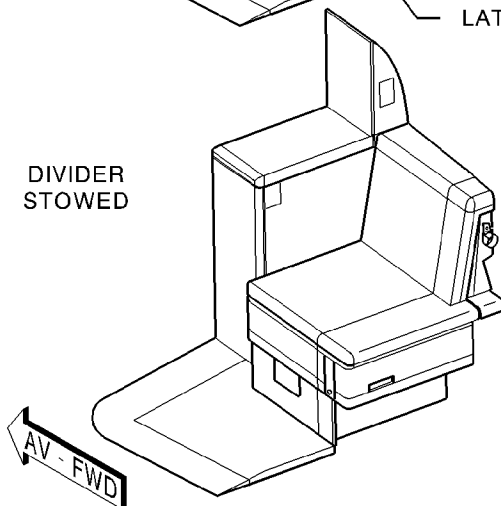
EMERGENCY  
STOWING BUTTON

DIVIDER  
DEPLOYED



LATCHING STRAP

DIVIDER  
STOWED



14252108AA/BMA8100

Figure 9.63.3 (1/2) - Lavatory compartment



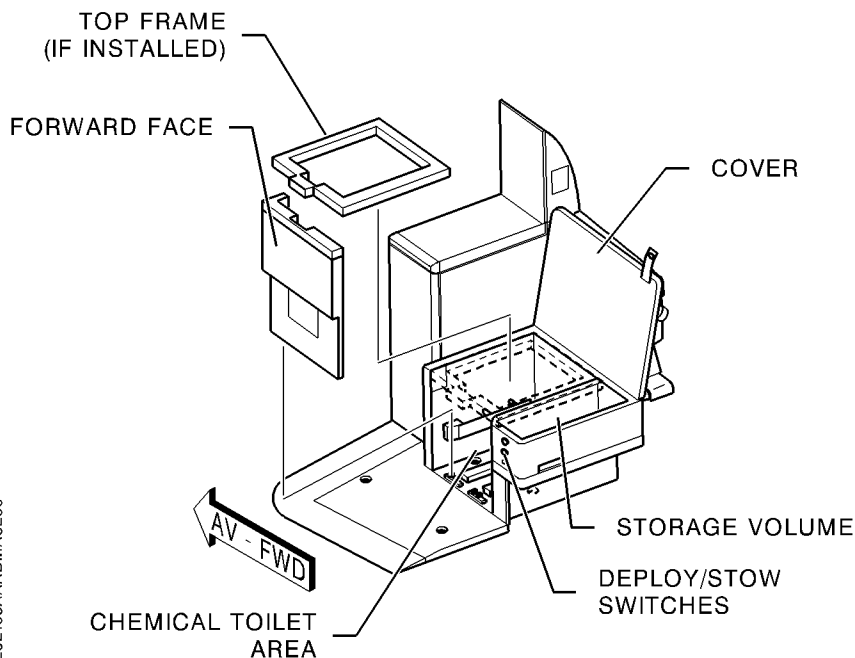
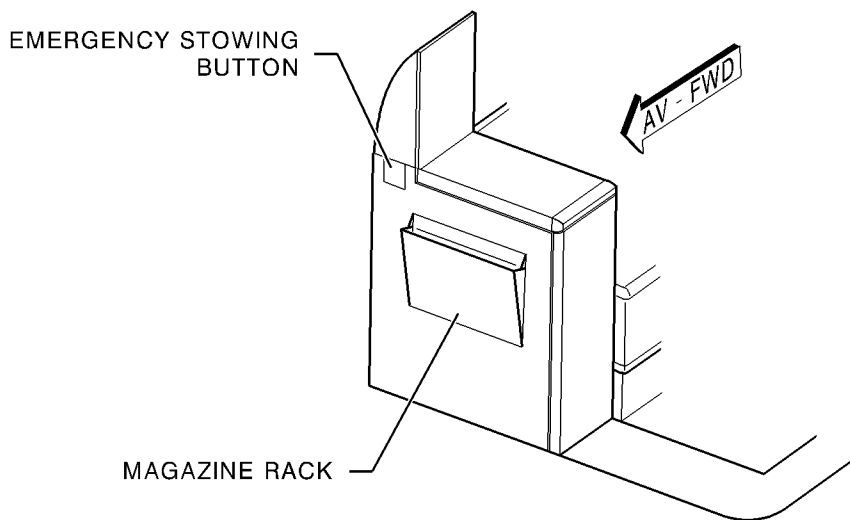


Figure 9.63.3 (2/2) - Lavatory compartment

## **SECTION 8**

### **Handling, servicing and maintenance**

Installation and operation of Lavatory compartment do not change the basic handling, servicing and maintenance procedures of the airplane described in section 8 Handling, servicing and maintenance of the basic POH.

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# **SUPPLEMENT**

## **DATA COLLECTION AND TRANSMISSION SYSTEM (FASTBOX)**

### **TABLE OF CONTENTS**

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| 5 - PERFORMANCE .....                         | 9.67.3 |
| 6 - WEIGHT AND BALANCE .....                  | 9.67.3 |
| 7 - DESCRIPTION .....                         | 9.67.4 |
| 8 - HANDLING, SERVICING AND MAINTENANCE ..... | 9.67.4 |

## **SECTION 1**

### **GENERAL**

This supplement is intended to inform the pilot about the equipment limitations, description and operations necessary for operation when the airplane is equipped with the option DATA COLLECTION AND TRANSMISSION SYSTEM (FASTBOX).

## **SECTION 2**

### **LIMITATIONS**

Installation and operation of DATA COLLECTION AND TRANSMISSION SYSTEM (FASTBOX) do not change the basic performance of the airplane described in Section 2 Limitations of the basic Pilot's Operating Handbook.

## **SECTION 3**

### **EMERGENCY PROCEDURES**

Installation and operation of DATA COLLECTION AND TRANSMISSION SYSTEM (FASTBOX) do not change the basic performance of the airplane described in Section 3 Emergency Procedures of the basic Pilot's Operating Handbook.

## **SECTION 4**

### **NORMAL PROCEDURES**

Installation and operation of DATA COLLECTION AND TRANSMISSION SYSTEM (FASTBOX) do not change the basic performance of the airplane described in Section 4 Normal Procedures of the basic Pilot's Operating Handbook.

## SECTION 5

### PERFORMANCE

Installation and operation of DATA COLLECTION AND TRANSMISSION SYSTEM (FASTBOX) do not change the basic performance of the airplane described in Section 5 Performance of the basic Pilot's Operating Handbook.

## SECTION 6

### WEIGHT AND BALANCE

The weight and balance hereafter supplement or replace those of the standard airplane described in Section 6 Weight and balance of the basic Pilot's Operating Handbook when the airplane is equipped with the option DATA COLLECTION AND TRANSMISSION SYSTEM (FASTBOX).

| S/<br>R/<br>A/<br>O | ITEM<br>OPT70<br>or<br>MOD70 | REQUIRED (R) OR STANDARD (S) OR OPTIONAL<br>(A or O) EQUIPMENT | WEIGHT<br>per unit<br>lb<br>(kg) | ARM<br>in.<br>(m) |
|---------------------|------------------------------|--|----------------------------------|-------------------|
|                     |                              | <b>31 - INDICATING/RECORDING SYSTEMS</b>                       |                                  |                   |
| A                   | 0578-31A                     | Data collection and transmission system PWC                    | 3.62<br>(1.64)                   | 125.2<br>(3.18)   |

## **SECTION 7**

### **DESCRIPTION**

Information hereafter supplement or replace those of the standard airplane described in Section 7 Description of the basic Pilot's Operating Handbook when the airplane is equipped with the DATA COLLECTION AND TRANSMISSION SYSTEM (FASTBOX).

#### **7.14 - MISCELLANEOUS EQUIPMENT**

##### **DATA COLLECTION AND TRANSMISSION SYSTEM (FASTBOX)**

The data collection and transmission system collects data from aircraft data buses and discrete inputs and store it in resident non-volatile memory.

When the aircraft is on the ground and a few minutes after the engine shut-down, recorded data are automatically transmitted to a ground station via the cellular or WiFi network. These data are intended to be used for maintenance and trend monitoring.

The data collection and transmission system records data from the GASC and data from the GIA integrated avionics unit #1 (GARMIN integrated flight deck system).

The data collection and transmission system starts recording data as soon as the crash lever is set upwards, and stops recording data when the crash lever is set downwards.

The data collection and transmission system is installed in the front cargo compartment, and does not require a pilot input to operate.

The data collection and transmission system is powered from the BATT BUS and protected by the "REC" circuit breaker.

## **SECTION 8**

### **HANDLING, SERVICING AND MAINTENANCE**

Installation and operation of DATA COLLECTION AND TRANSMISSION SYSTEM (FASTBOX) do not change the basic handling, servicing and maintenance procedures of the airplane described in Section 8 Handling, Servicing and Maintenance of the basic Pilot's Operating Handbook.

# SUPPLEMENT

## TBM930 (2018)

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## **SECTION 1**

### **GENERAL**

This supplement is intended to inform the pilot about the equipment limitations, description and operations necessary for operation of the airplane TBM930 (2018).

The list of systems equipping the airplane delivered from January 2018 is :

- TBM930 fitted with SW GDU V6.6 and with MOD70-0549-00 capability ,
- Heated seats MOD70-0588-25 ,
- O<sub>2</sub> smart mike mask MOD70-0597-35 ,
- Landing and taxi light rotary switch MOD70-0595-92.

This supplement is valid until the TBM930 Pilot's Operating Handbook at edition 1 is issued and airplanes concerned by this supplement have applied MOD70-0549-00.

## SECTION 2

### LIMITATIONS

Operation of the airplane TBM930 (2018) does not change the limitations of the airplane as described in Section 2 Limitations of the basic Pilot's Operating Handbook.

**SECTION 3**  
**EMERGENCY PROCEDURES**

The emergency procedures hereafter supplement or replace those of the standard airplane described in Section 3 Emergency Procedures of the basic Pilot's Operating Handbook when the airplane uses the equipment of the TBM930 (2018).

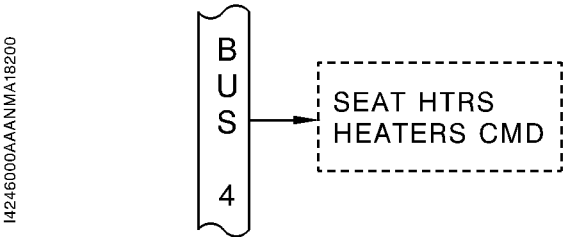
**3.9 - ELECTRICAL SYSTEM**

MAIN GEN

AND

LOW VOLTAGE

1 - SEATS HTRS MASTER ..... OFF



NOTE: CIRCUIT BREAKERS ON C13 BIS FRAME

Figure 3.9.1 - ELECTRICAL DISTRIBUTION OF BUS BARS

## SECTION 4

### NORMAL PROCEDURES

The normal procedures hereafter supplement or replace those of the standard airplane described in Section 4 Normal Procedures of the basic Pilot's Operating Handbook when the airplane uses the equipment of the TBM930 (2018).

#### 4.3 - CHECK-LIST PROCEDURES

##### INSIDE INSPECTION

2 - SEATS HTRS MASTER ..... **OFF**

*Other procedures in the basic Pilot's Operating Handbook are unchanged.*

##### AFTER ENGINE START

1 - SEATS HTRS MASTER ..... **As required**

*Other procedures in the basic Pilot's Operating Handbook are unchanged.*

##### SHUT-DOWN

1 - SEATS HTRS MASTER ..... **OFF**

*Other procedures in the basic Pilot's Operating Handbook are unchanged.*

**4.3 - AMPLIFIED PROCEDURES**  
**PREFLIGHT INSPECTION**

1 - SEATS HTRS MASTER ..... **OFF**

*Other procedures in the basic Pilot's Operating Handbook are unchanged.*

**INSIDE INSPECTION**

1 - SEATS HTRS MASTER ..... **OFF**

*Other procedures in the basic Pilot's Operating Handbook are unchanged.*

**AFTER ENGINE START**

1 - SEATS HTRS MASTER ..... **As required**

*Other procedures in the basic Pilot's Operating Handbook are unchanged.*

**SHUT-DOWN**

1 - SEATS HTRS MASTER ..... **OFF**

*Other procedures in the basic Pilot's Operating Handbook are unchanged.*

## PILOT'S OPERATING HANDBOOK

**SECTION 5**  
**PERFORMANCE**

Operation of the airplane TBM930 (2018) does not change the performance of the airplane described in Section 5 Performance of the basic Pilot's Operating Handbook.

**SECTION 6**  
**WEIGHT AND BALANCE**

Operation of the airplane TBM930 (2018) does not change the weight and balance of the airplane described in Section 6 Weight and Balance of the basic Pilot's Operating Handbook.

## **SECTION 7**

### **DESCRIPTION**

Information hereafter supplement the one given for the standard airplane in Section 7 Description of the basic Pilot's Operating Handbook when the airplane uses the equipment of the TBM930 (2018).

### **7.3 - ACCOMODATIONS**

#### **SEAT, BELTS AND HARNESSSES**

##### **Heated seats (Figure 7.3.1)**

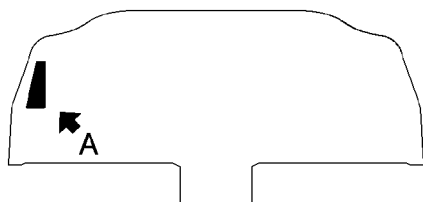
Cockpit and cabin seats are equipped with a heating system providing comfort to pilot and passengers.

The system consists of :

- One heating element in the cushion and one heating element in the backrest of each seat,
- The SEATS HTRS MASTER switch located on the instrument panel,
- HI/OFF/LOW three positions switch located on each cockpit and cabin seat,
- The seats heaters control box and relays located under the floor panel.

Each seat is equipped with a power supply wire with a connector. A clip provides a holding mean for the connector to prevent its damage during seat operation or seat storage.

PILOT'S OPERATING HANDBOOK



- 1) SEATS HTRS MASTER switch
- 2) Backrest surface heating
- 3) Seat surface heating
- 4) HI/OFF/LOW three position switch
- 5) Tactile marks

14251204AAA MA18000

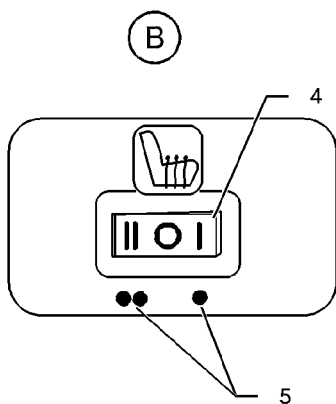
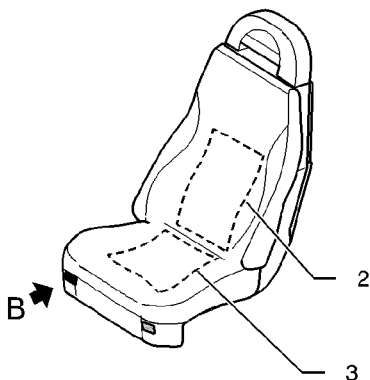
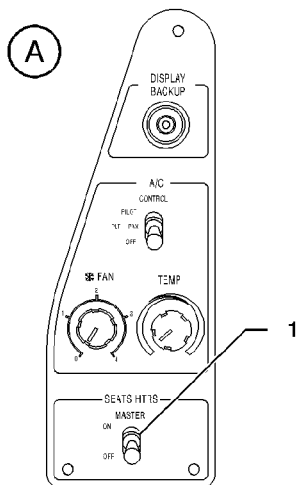


Figure 7.3 1 - Heated seat



## PILOT'S OPERATING HANDBOOK

The seats heating is only available when the airplane is connected to a GPU or when the main generator is supplying power.

The system does not operate if GENERATOR selector is set to ST-BY.

The SEATS HTRS MASTER switch allows the pilot to enable or not the electrical supply of all seats heaters.

Each seat is then individually controlled by the HI/OFF/LOW switch :

**NOTE :**     ***Two tactile marks located under the HI / OFF / LOW switch enables to determinate which position is selected.***

- OFF position is obtained when the switch is in the central position. In this position the seat does not heat.
- HI position is obtained by positioning the switch to the right. In this position the seat heats at its maximum capacity.
- LOW position is obtained by positioning the switch to the left. In this position the seat heats less than HI position.

**NOTE :**     ***In HI position, the heating sensation comes up after approximately 3 minutes.***

To avoid overheating, each seat is equipped with thermal sensors which remove power supply in case of overtemperature detection.

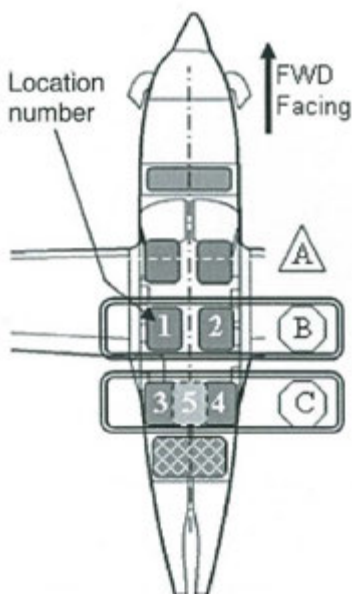
Precaution of use of the seats heaters system :

- Do not place any sharp or heavy objects on the seat, as the seat heater could otherwise be damaged.
- Persons with an impaired sensitivity to heat should only operate the seat heater at low level.
- Do not place any heat insulating objects, such as blankets or coats, on the seat when the seat heater is switched on.
- The seat heater can be damaged by fluids spilt on the seat.
- Never switch the seat heater on when it is wet.

## PILOT'S OPERATING HANDBOOK

Possible configurations of seats equipped with the seat heaters system :

| Configuration name | Location number |   |          |   |   |
|--------------------|-----------------|---|----------|---|---|
|                    | 1               | 2 | 3        | 4 | 5 |
| C1                 | X               | X | X        | X |   |
| C2 (2)             | X               | X |          |   | X |
| C3                 | X               | X |          | X |   |
| C4 (1)             | X               | X |          |   |   |
| C5                 | X               | X | X        |   |   |
| C6                 | X               |   | X        | X |   |
| C7                 | X               |   | X        |   |   |
| C8                 | X               |   |          | X |   |
| C9 (2)             | X               |   |          |   | X |
| C10 (1)            | X               |   |          |   |   |
| C11                |                 | X | X        | X |   |
| C12 (2)            |                 | X |          |   | X |
| C13                |                 | X | X        |   |   |
| C14                |                 | X |          | X |   |
| C15 (1)            |                 | X |          |   |   |
| C16                |                 |   | X        | X |   |
| C17                |                 |   | X        |   |   |
| C18                |                 |   |          | X |   |
| C19 (2)            |                 |   |          |   | X |
| C20 (1)            |                 |   |          |   |   |
|                    | Zone (B)        |   | Zone (C) |   |   |



(1) This configuration accepts small net or large net

(2) Rear seat, with seats heaters system, cannot be connected to electrical supply in this configuration

Each cross indicates that you have a seat at the correspondent location number.

- 1) DISPLAY BACKUP, A/C CONTROL and SEATS HTRS panels
- 2) General alarm red and amber indicators
- 3) PFD1
- 4) MD302
- 5) Landing gear configuration and control panel
- 6) Parking brake control
- 7) Left station control wheel tube
- 8) Deicing control and check panel
- 9) L.H. station rudder pedals adjusting handle
- 10) Left station reception-micro jacks
- 11) Pitch & Yaw trim setting management
- 12) Push To Talk button (PTT)
- 13) AP / TRIM DISC red push-button
- 14) CWS
- 15) Paper clip
- 16) Chronometer management
- 17) Transponder Ident sequence
- 18) Stormscope clear
- 19) COM 2 (Stand-by / active)
- 20) Flight conditions and instruction placard
- 21) Adjustable air outlet
- 22) Circuit breaker panel lighting switch
- 23) Oxygen mask microphone switch
- 24) Hourmeter
- 25) USB servicing plug

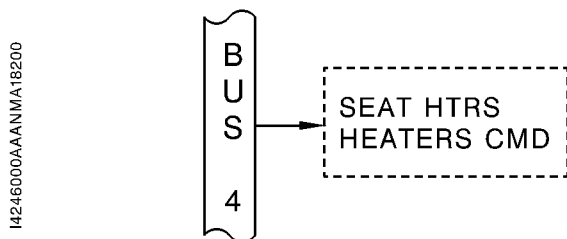
Figure 7.3 2 (1/2) - LEFT INSTRUMENT PANEL

PILOT'S OPERATING HANDBOOK



Figure 7.3 2 (2/2) - LEFT INSTRUMENT PANEL  
(Typical arrangement)

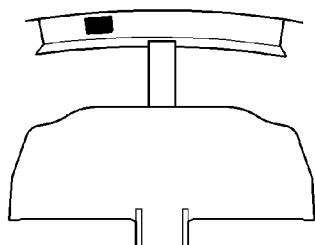
## 7.8 - ELECTRICAL SYSTEM



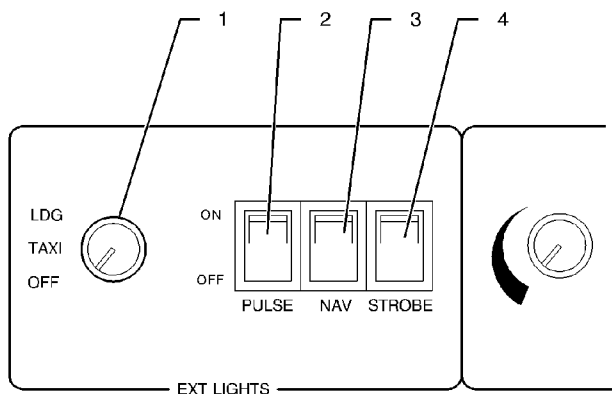
NOTE: CIRCUIT BREAKERS ON C13 BIS FRAME

Figure 7.8.3 - ELECTRICAL DISTRIBUTION OF BUS BARS

PILOT'S OPERATING HANDBOOK



- 1) Taxi and landing light switch
- 2) Pulse system switch
- 3) Navigation lights switch
- 4) Strobe lights switch



14240000AAVVA18000

Figure 7.8.6 - EXTERNAL LIGHTING CONTROLS

*Other information in the Paragraph 7.8 of the Pilot's Operating Handbook are unchanged.*

## 7.10 - EMERGENCY OXYGEN SYSTEM

### O<sub>2</sub> SMART MIKE MASK

The two cockpit masks, O<sub>2</sub> smart mike masks, are equipped with a system that detects when a communication is made operating one of the push-to-talk buttons. It activates a noise reduction system to attenuate the oxygen flow noise in the headsets.

Noise reduction function operates when the switch located on O<sub>2</sub> connecting line is set to ON.

14351100AAAEIA8000

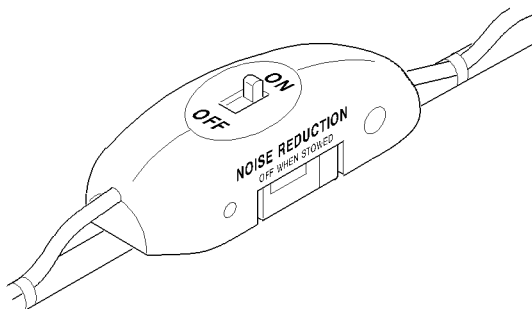


Figure 7.10.2 - NOISE REDUCTION SWITCH

*Other information in the Paragraph 7.10 of the Pilot's Operating Handbook are unchanged.*

## **SECTION 8**

### **HANDLING, SERVICING AND MAINTENANCE**

Information hereafter supplement the given for the standard airplane in Section 8 Handling, Servicing and Maintenance of the basic Pilot's Operating Handbook when the airplane uses the equipment of the TBM930 (2018).

#### **8.10 - PREPARING OF THE AIRPLANE (EQUIPMENT AND FURNISHING)**

##### **CONVERSIONS OF SEATS ACCOMODATION**

###### **CAUTION**

**MAKE SURE TO DISCONNECT THE SEAT HEATING SYSTEM BEFORE THE REMOVAL OF THE SEAT TO PREVENT DAMAGE TO THE HARNESS.**

Before a seat removal, disconnect heating system harness, and clip the loose connectors to the holders located on the seat backrest and under the cabin floor, close floor access door.

After seat installation or re-configuration, reconnect heating system harness, and clip connector to the dedicated holder on the seat.



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